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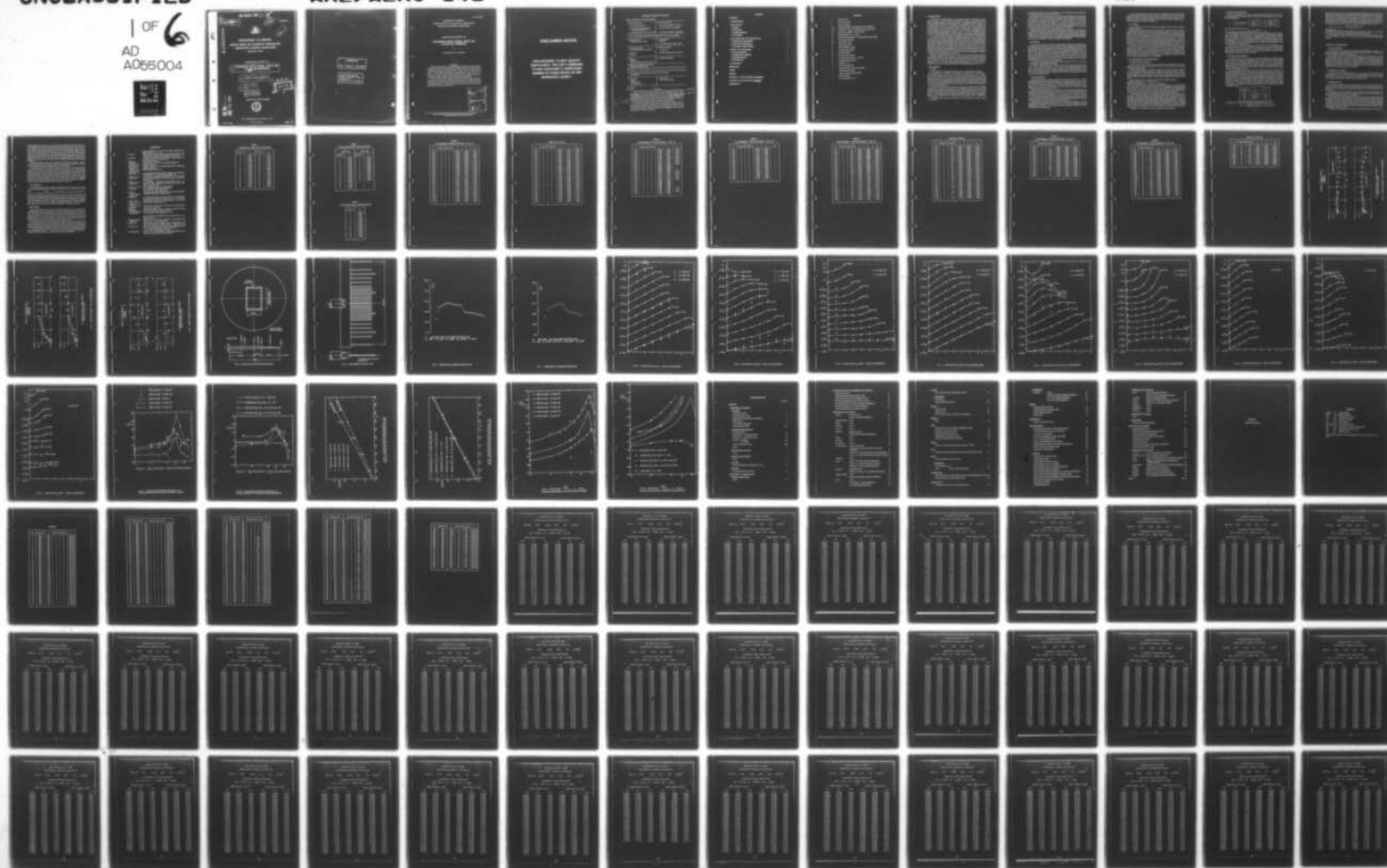
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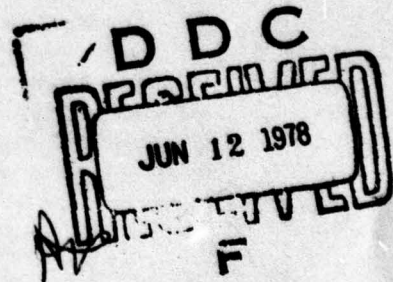
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10 N. POLLOCK & B. D. FAIRLIE

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AERODYNAMICS REPORT 148

**TRANSONIC WIND TUNNEL TESTS ON
A NACA 0012 AEROFOIL**

by

N. POLLOCK & B. D. FAIRLIE

SUMMARY

Transonic wind tunnel tests on three geometrically similar NACA 0012 aerofoil models are reported. The tests which comprised surface pressure distribution and wake traverse measurements covered a Mach number range of 0.50 to 0.82 and an incidence range of 0° to 8°. The test Reynolds numbers for the three models were 1.68×10^6 , 0.84×10^6 and 0.42×10^6 for the large, intermediate and small models respectively. Brief tests on the two larger models at reduced Reynolds number were also included. All tests were conducted with artificial boundary layer transition fixing on both surfaces.

The main purpose of this investigation was to provide data to evaluate wall interference effects in the transonic wind tunnel. The data should also be useful for checking the accuracy of interference theories at high subsonic speed. This work forms part of a Commonwealth Advisory Aeronautical Research Council cooperative research program of aerofoil tests in transonic wind tunnels.

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Transonic wind tunnel tests on three geometrically similar NACA 0012 aerofoil models are reported. The tests which comprised surface pressure distribution and wake traverse measurements covered a Mach number range of 0.50 to 0.82 and an incidence range of 0° to 8°. The test Reynolds numbers for the three models were 1.68×10^6 , 0.84×10^6 and 0.42×10^6 for the large, intermediate and small models respectively. Brief tests on the two larger models at reduced Reynolds number were also included. All tests were conducted with artificial boundary layer transition fixing on both surfaces.

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NOTATION

A	aspect ratio (b/c)
b	width of wind tunnel
c	model chord length
C_{DW}	wake drag coefficient = drag force from wake traverse/ qS
C'_{DW}	contribution to wake drag coefficient from any point in wake
C_N	normal force coefficient = normal force/ qS
C_P	pressure coefficient = $(P - P_0)/q$
C_m	pitching moment coefficient = (pitching moment about $c/4$)/ qSc
H	free stream total pressure
H_W	total pressure in wake
h	height of wind tunnel
K	torsional stiffness of model
M	free stream Mach number
P	local static pressure
P_0	free stream static pressure
q	free stream kinetic pressure = $\gamma P_0 M^2/2$
R	Reynolds number based on c
S	wing area ($b \times c$)
T	free stream temperature
x	chordwise ordinate; origin at leading edge
y	aerofoil thickness ordinate
z	spanwise ordinate; origin at midspan.
α	angle of incidence
γ	ratio of specific heats (1.4)

1. INTRODUCTION

The problems in obtaining accurate two dimensional transonic aerofoil data from wind tunnel tests, and in trying to eliminate uncertainties in transonic two dimensional aerofoil theory were recognised by the Commonwealth Advisory Aeronautical Research Council*. That Council endorsed¹ a proposal by the Coordinators in Aerodynamics² to "test a small number of suitable aerofoils in as many as possible of the transonic wind tunnels, throughout the Commonwealth, in which two dimensional research is normally done". The proposal for such a research program, put forward by the C.A.A.R.C. U.K. coordinator in aerodynamics³, stated that the purpose of such a program would be "to add to our mutual knowledge and understanding of experimental techniques in a subject which is at once of great practical importance and at the same time afflicted with serious uncertainties, both theoretical and experimental".

Two aerofoil shapes were selected for the cooperative program. The first the NACA 0012 section was chosen to provide a "standard" symmetrical aerofoil which, since it had been tested in a number of wind tunnels throughout the world, would provide a reliable calibration of wind tunnels under both subcritical and supercritical conditions. The second, an aerofoil designed by the method of Bauer, Garabedian and Korn⁴ (referred to as BGK 1), is an example of a modern "supercritical" section, with a reasonable extent of shock free supersonic flow on the upper surface and some degree of rear loading. This aerofoil had the added advantage of having already been extensively tested at high Reynolds number in the NAE wind tunnel in Canada^{5, 6}. Since both of these sections already formed part of a theoretical and experimental investigation of transonic aerofoils underway at ARL it was decided to extend the experimental investigation to include the test program proposed in Reference 3.

This report contains results from tests on three NACA 0012 aerofoil models of differing chord lengths. These tests were conducted in the ARL transonic wind tunnel during October 1975 and April 1976. The results include measurements of both surface pressure distributions and wake pressure surveys. The range of Mach number ($0.50 \leq M \leq 0.82$) and angle of incidence ($0^\circ \leq \alpha \leq 8^\circ$) covered by these tests are those suggested in the C.A.A.R.C. proposal³. The results of the tests on supercritical aerofoil BGK 1 have been published previously^{7, 8, 9}.

2. TEST DETAILS

2.1 Models

The NACA 0012 is a symmetrical aerofoil, 12% thick, whose ordinates are given by:
 $y/c = 0.6 \{0.29690 \sqrt{x/c} - 0.12600 x/c - 0.35160 (x/c)^2 + 0.28430 (x/c)^3 - 0.10150 (x/c)^4\}$
with leading edge radius $0.0158 c$. The basic aerofoil section has a blunt base of $0.00252 c$ (i.e. 0.51 mm , 0.26 mm and 0.13 mm for the 203.2 mm , 101.6 mm and 50.8 mm models respectively). However, to avoid confusing comparisons between the models arising from base pressure variations with Reynolds number, the upper and lower surfaces were linearly extended to a sharp (0.1 mm thick) trailing edge. Due to this extension the actual physical chord length of the models was slightly greater than the nominal chord. The calculated force coefficients are based on the nominal chord (c).

The testing of two-dimensional aerofoils in transonic wind tunnels presents conflicting requirements on model size; a large model increases Reynolds number, but a small model reduces tunnel interference effects. Studies of wind tunnel interference suggest that c/h ratios of 0.063 or less lead to very nearly interference free conditions¹⁰ and that c/h ratios of up to about 0.3 can be used before conventional wall interference concepts cease to be applicable³. Based on these considerations, and considering the model support systems available, three

* Australia, Canada, India, New Zealand and the United Kingdom are active members of C.A.A.R.C.

models of the NACA 0012 section were manufactured, with c/h ratios of 0.25, 0.125 and 0.063 (i.e. chords of 203.2 mm, 101.6 mm and 50.8 mm).

The models were cast in epoxy resin in a wax mould using a development of the technique described in Ref. 11. The models were pressure tapped to enable surface pressure distributions to be measured. The pressure hole diameter was 0.4 mm for all models. For the 203.2 mm chord model, the pressure holes were distributed over a spanwise region $0.32c$ wide, centred at midspan, with 38 pressure holes located in one surface only. The 101.6 mm model had 18 holes in the upper surface and 14 holes in the lower surface, distributed over $0.29c$. The 50.8 mm model had 15 pressure holes in one surface only, distributed over $0.33c$. The chord-wise locations of the pressure holes for all models are given in Table 1.

Measured profile errors for the completed models are plotted in Figs. 1 to 3. The absolute accuracy achieved on all three models was similar. This led to a deterioration of non-dimensional accuracy as the models became smaller. The inaccuracies of the 50.8 mm chord model are sufficiently large to require caution in the use of the experimental pressure distributions. However it is believed that the integrated normal force coefficients obtained from this model should be accurate, since this coefficient is relatively insensitive to section shape.

2.2 Wind Tunnel

All tests were carried out in the ARL variable pressure transonic wind tunnel. The test section fitted for these tests (Fig. 4) was 533 mm wide and 813 mm high at the model location with solid sidewalls and longitudinally slotted top and bottom walls (open area ratio 16.5% at model location). These walls are normally used for two dimensional testing in this tunnel. The open area ratio, which was selected to provide supersonic operation with diffuser suction, is known to be larger than that giving interference free subsonic test conditions. Lack of time prevented the manufacture of a new set of walls of smaller open area ratio for the CAARC program. Mach number was derived from measurements of the pressure in the plenum chamber surrounding the test section, and in the entry to the contraction, assuming these to be the static and total pressures of the test section flow respectively.

The models were supported by integral end tongues clamped in slots in the steel discs which replaced the usual sidewall glass windows.

The models completely spanned the width of the test section giving aspect ratios (A) of 2.62, 5.25 and 10.50 for the 203.2 mm, 101.6 mm and 50.8 mm aerofoils respectively. These aspect ratios should be large enough to ensure reasonably two-dimensional conditions, at least over the central measuring sections. Small gaps, approximately 0.6 mm wide existed between the ends of the models and the steel "windows". For the present tests these gaps were sealed with a silicone rubber compound.

2.3 Measuring Equipment

Model surface pressures were measured using two 48 port Scanivalves fitted with ± 33.5 kPa Statham differential pressure transducers referenced to P_0 . Stagger scanning was used to ensure the maximum possible settling time between readings. Scanning speed was normally 14 ports per second, with free stream conditions (P_0 , H , T) being recorded after each group of 12 ports. Values of P_0 and H used to calculate P/H and C_p were the average of the P_0 and H readings bounding the particular pressure measurement.

Wake traverse measurements of $(H - H_w)$ and $(P_0 - P)$ were made using a fixed drag rake mounted on the tunnel sting support strut, located on the tunnel centre line just downstream of the diffuser inlet. The rake was located at $x/c \approx 2.0$, $x/c \approx 2.0$ and $x/c \approx 3.5$ for the 203.2 mm, 101.6 mm and 50.8 mm models respectively. The arrangement of the drag rake is shown in Fig. 5. The spacing of the tubes on the rake was limited by the number of pressure tubes which could be led out of the tunnel strut, and by the need to include the whole width of the wake at high Mach numbers (up to one chord) within the span of the rake.

Wake pressures were measured with two 48 port Scanivalves, one fitted with a ± 33.5 kPa Statham differential transducer, and the other with a ± 9.2 kPa Scanco differential transducer, measuring $(P_0 - P)$ and $(H - H_w)$ respectively. Scanivalve operation followed that used for the surface pressure measurements.

All pressure measurements were processed by the tunnel PDP 8/I computer data processing system, the data being reduced to pressure coefficient and pressure ratio forms for on-line printing and display. The real time display allowed the operator to detect system faults immediately and to repeat the affected readings. The data were also recorded on magnetic tape for further processing by the central site computer.

2.4 Test Program

The test program, which in general follows that suggested in the C.A.A.R.C. proposal³, is presented in Table 2.

The Reynolds numbers for the majority of these tests were chosen such that the maximum available Reynolds number was utilized at the highest Mach number ($M = 0.82$), and constant Reynolds number maintained for each model for all other Mach numbers by adjusting tunnel pressure. Due to tunnel temperature variations and inaccuracies in setting tunnel pressure, the Reynolds number varied slightly, the values being in the range $1.68 \pm 0.07 \times 10^6$ for the 203.2 mm model, $0.84 \pm 0.04 \times 10^6$ for the 101.6 mm model and $0.42 \pm 0.02 \times 10^6$ for the 50.8 mm model. To help separate the Reynolds number effects from those of tunnel interference, brief tests were conducted on the 203.2 mm model at Reynolds numbers of $0.84 \pm 0.04 \times 10^6$ and $0.42 \pm 0.02 \times 10^6$, and on the 101.6 mm model at a Reynolds number of $0.42 \pm 0.02 \times 10^6$.

For all tests boundary layer trips consisting of spanwise bands of carborundum particles attached with a thin (0.03 mm) layer of lacquer were used on both surfaces of the models. The bands were located at $x/c = 0.05$ and were 0.01 c wide, with a particle coverage of 10–20%; particle size was 0.15 mm for the 203.2 mm model and 0.10 mm for the 101.6 mm and 50.8 mm models. Surface oil flow observations indicated that at the maximum test Reynolds number for each model the roughness bands effectively produced boundary layer transition. Oil flow tests at reduced Reynolds number were not undertaken.

3. DATA REDUCTION AND PRESENTATION

3.1 Surface Pressure Measurements

The 203.2 mm and 50.8 mm models were pressure tapped on one surface only. To obtain "upper" and "lower" surface pressures these two models were tested twice at equal positive and negative incidence angles.

All surface pressure data were reduced to pressure ratio (P/H) and pressure coefficient (C_P) form and are tabulated in Appendix A*. Appendix B* contains plots of P/H versus x/c for all test conditions. The data were plotted in this form rather than C_P versus x/c to allow direct comparisons with other tunnel tests or with theory to be made: adjustments to the values of Mach number and angle of incidence to take account of interference do not affect the P/H values.

Surface pressure data were integrated to obtain force coefficients as follows:

normal force coefficient $C_N = 1/c \oint C_P \cos \theta \, ds$

pitching moment coefficient $C_m = 1/c^2 \oint C_P (x - 0.25) \cos \theta \, ds$

where s is arc length, measured clockwise from the leading edge, and θ is the angle between the local tangent to the surface, and the chord line. The full expression for pitching moment also contains a term $1/c^2 \oint C_P y \sin \theta \, ds$. However due to the absence of leading edge pressure holes on the three models, this integral was poorly defined and since rough check calculations showed the term to be small, it was not included in the present calculations of C_m .

The integrations were carried out numerically using the method of Reference 18. Due to the scarcity of pressure tappings near the leading and trailing edges of the models some extra information was required to define these integrals properly. After investigating a number of approaches, the following method was adopted. A fictitious pressure point was generated at the trailing edge by linearly extrapolating the upper surface pressure distribution. At the leading edge the upper and lower surfaces were extrapolated using parabolae which were tangential to the C_P axis. It was found that this procedure gave values of C_m and C_N which were in good agreement with those obtained from curves manually fitted to the experimental data points.

* Appendix A and Appendix B have been reproduced in microfiche form.

3.2 Wake Traverse Measurements

Based on the assumption that entropy is constant along stream lines in the wake, Lock, Hilton and Goldstein¹² have shown that the contribution to drag coefficient from any point in the wake is given by:

$$C'_{DW} = 2 \left(\frac{P}{P_o} \right)^{\frac{1}{\gamma}} \left(\frac{H_w}{H} \right)^{\frac{\gamma-1}{\gamma}} \left\{ \frac{1 - \left(\frac{P}{H_w} \right)^{\frac{\gamma-1}{\gamma}}}{1 - \left(\frac{P_o}{H} \right)^{\frac{\gamma-1}{\gamma}}} \right\}^{\frac{1}{2}} \left\{ 1 - \left[\frac{1 - \left(\frac{P_o}{H_w} \right)^{\frac{\gamma-1}{\gamma}}}{1 - \left(\frac{P_o}{H} \right)^{\frac{\gamma-1}{\gamma}}} \right]^{\frac{1}{2}} \right\}$$

The wake drag coefficient is given by:

$$C_{DW} = \int C'_{DW} d(y/c)$$

where the integral is evaluated between points at which C'_{DW} is zero.

When evaluating this integral, allowance should be made for the displacement of the effective centre of the mouth of a total head tube in a velocity gradient towards the region of higher velocity. The magnitude of this effect has been investigated by Young and Maas,¹³ who found that, at low speeds for tubes of the dimensions employed in the wake rake used for these tests, it is sufficiently accurate to take the displacement as 0.18 times the external diameter of the tube. In the absence of more detailed information relevant to higher speeds, this simple correction has been applied to all the present tests.

Values of static pressure were obtained at positions in the wake corresponding to total pressure tube locations by linear interpolation and extrapolation of the four measured wake static pressures. Wake drag coefficients were then found by integrating C'_{DW} numerically; parabolas were fitted by least squares to each group of four points, and were then used to find the contribution arising from the centre two points.

3.3 Correction to Angle of Incidence

The model incidence was measured with a transducer attached to the starboard tunnel window. Since the port window was mechanically slaved to the starboard one it was assumed that the model was not twisted by differential window rotations. The zero incidence position for each model was determined from tests at one Mach number ($M = 0.6$) with the model upright and inverted. The measured incidence angles are given in the tables and graphs reproduced in Appendices A and B; they are also listed as "Nominal α " in Table 2 and in the tables of contents of Appendices A and B.

Conventional metal aerofoil models tested in the ARL transonic tunnel normally have sufficient torsional stiffness to ensure that the central pressure tapped section of the model can be regarded as having the same angle of incidence as that measured at the ends. However composite plastic models such as those used for these tests, particularly in small sizes, show significant spanwise twist due to aerodynamic loading. The angles of incidence from these tests were therefore corrected for twisting under aerodynamic loading as follows. By appropriately loading the models with weights the torsional stiffness per unit length (K) and the location of the shear centre ($(x/c)_{sc}$), i.e. the chordwise position at which an applied normal force produces no rotation of the section, were determined. The torsional characteristics of the three models are summarised in the following table:

Model chord (mm)	K (N. rad ⁻¹)	$(x/c)_{sc}$
203.2	0.956×10^7	0.30
101.6	0.291×10^6	0.35
50.8	0.427×10^4	0.32

The angle of twist of the model with respect to the midspan section is given by:

$$\delta\alpha|_{z=z^1} = \frac{qc^2}{K} \int_0^{z^1} \int_0^{z^1} \left\{ C_N \left[\left(\frac{x}{c} \right)_{sc} - 0.25 \right] + C_m \right\} dz \cdot dz$$

NB. By working with the z origin at midspan the problem is simplified since from considerations of symmetry it is evident that the torsional moment acting on the midspan section is zero. The angle of twist of the midspan section with respect to the sidewalls $\delta\alpha|_{z=b/2}$ is calculated from the above equation by successive approximation. For the first iteration C_N and C_m are assumed constant at their midspan values (obtained by integrating the pressure distribution). The expression for $\delta\alpha$ is evaluated numerically for 20 stations equally spaced over the semispan. For all successive iterations C_N and C_m are assumed to vary linearly with α so that:

$$C_N|_{z=z^1} = C_{N \text{ midspan}} \times \frac{\alpha_{\text{sidewall}} + \alpha|_{z=b/2} - \alpha|_{z=z^1}}{\alpha_{\text{sidewall}} + \alpha|_{z=b/2}}$$

and similarly for C_m .

The calculation was continued until the angle of twist converged to within $\pm 0.01^\circ$ of its final value. For the present results, a maximum of three iterations was required.

Incidence angles corrected for model twist are used for the plotted results (Figs. 8-16), they are also listed as "Corrected α " in Table 2 and in the tables of contents of Appendices A and B.

4. RESULTS AND DISCUSSION

4.1 Pressure Distributions

The general form of the experimental pressure distributions tabulated in Appendix A and plotted in Appendix B are in good agreement with other published results^{10,14,16}. However one feature of the pressure distributions should be noted.

The test results for the 101.6 mm and 203.2 mm models at a Mach number of 0.80 and a Reynolds number of 0.42×10^6 (Fig. 6) show a small pressure rise, followed by a pressure plateau, and then a rapid recompression; these are characteristic of laminar boundary layer separation and turbulent reattachment.¹⁷ It is believed that the low unit Reynolds number of these tests led to the failure of the standard roughness band to "trip" the boundary layer. This is supported by the fact that tests on the 50.8 mm model at the same chord Reynolds number (Fig. 7) did not show any signs of laminar separation.

4.2 Integrated Force Coefficients

The integrated force and moment coefficients C_N , C_m and C_{DW} are listed in Table 2 and plotted in Figures 8-16. All the coefficients are in reasonable agreement with other published test results^{10,14,16}. The scatter in the values of C_{DW} evident in Figures 10, 13 and 16 is caused by the relatively small number of tubes in the wake rake (Section 2.3).

4.3 Blockage Interference

The evaluation of wind tunnel wall interference will be the subject of a separate publication. However a brief comparison between the present results and other published data will be presented to give an indication of the nature and magnitude of the tunnel interference in the current tests.

Two parameters have been suggested as sensitive indicators of blockage interference in tests on symmetrical aerofoil sections: the minimum surface P/H value at zero incidence, and the chordwise shock wave location at zero incidence.

The minimum surface P/H approach⁵ will be considered first. In Fig. 17 $\delta(P/H)_{\min}$ is plotted against M for the three models. $\delta(P/H)_{\min}$ is equal to the experimental minimum P/H minus the minimum P/H value computed using the inviscid transonic program of Ref. 4. The data were plotted in this form rather than simply $(P/H)_{\min}$ versus M so that the small differences in $(P/H)_{\min}$ could be more clearly shown. Irrespective of the nature of the blockage interference (solid tunnel or open jet) the curve for the 101.6 mm model should lie between the curves for the 50.8 mm and 203.2 mm models.

From Fig. 17 it can be seen that this does not occur. Since $(P/H)_{\min}$ is very sensitive to

section shape and the accuracy of the 50.8 mm model is significantly worse than the other two it is considered that the 50.8 mm model data are at fault. Comparing the results for the two larger models, it can be seen that the 203.2 mm model experiences a higher value of $(P/H)_{\min}$ at a given free stream Mach number than the 101.6 mm model. This is consistent with the known open jet nature of the tunnel walls (Section 2.2). In Fig. 18 $\delta(P/H)_{\min}$ is plotted against M for the 101.6 mm model and for some other published NACA 0012 tests. All of the comparative data presented in Fig. 18 are claimed to have "small" blockage interference. The main conclusion that can be drawn from Fig. 18 is that considering the scatter in results between different tests, the present 101.6 mm model data can be regarded as being nearly free of blockage interference.

The second parameter which has been suggested as an indicator of blockage interference is the zero lift shock position.¹⁴ In Fig. 19 the shock position (arbitrarily defined as the midpoint of the surface pressure rise) is plotted against M for the three models.

The shock position on the 101.6 mm model shows a clear trend to be further aft than that on the 203.2 mm model. This confirms the previous observation that the tunnel blockage interference in these tests is of the open jet type. The shock position on the 50.8 mm model is suspect, again due to model inaccuracies. In Fig. 20 shock position is plotted against M for the 101.6 mm model and for some other published NACA 0012 tests. Except for the NPL results (Ref. 15), there is good agreement in shock position between all the tests. This supplies further evidence that the present tests on the 101.6 mm model are nearly free of blockage interference. From Figs 18 and 20 it appears that the NPL results are subject to a significant closed unnel type blockage interference at Mach numbers in excess of 0.7.

4.4 Lift Interference

The standard parameter used to measure lift interference is the zero lift—lift curve slope which, for a symmetrical section, is equal to the zero normal force—normal force curve slope,

$\left. \frac{\partial C_N}{\partial \alpha} \right|_{C_N=0}$. In Fig. 21 $\left. \frac{\partial C_N}{\partial \alpha} \right|_{C_N=0}$ is plotted against M for the three models. The figure shows a marked increase in normal force curve slope as the model chord is reduced. This indicates that the tunnel lift interference is also of the open jet type. In Fig. 22 the normal force curve slope for the 50.8 mm model (which should be closest to interference free) is compared with other published NACA 0012 results and with viscous theory. This comparison suggests that even the 50.8 mm model may be subject to measurable interference. It also appears that NAE results (Ref. 16) are subject to serious lift interference at high Mach numbers.

5. CONCLUSION

Three geometrically similar models of a NACA 0012 aerofoil section have been subject to detailed measurements of surface pressure distribution and wake traverses in the ARL transonic wind tunnel. The three models which had chord to tunnel height ratios of 0.25, 0.125 and 0.06 were tested over a Mach number range from 0.50 to 0.82 and a range of angles of incidence from 0° to 8°. The main purpose of the tests was to provide data to evaluate wall interference effects in the transonic tunnel. The data should also be useful for checking the accuracy of interference theories at high subsonic and transonic speeds.

A preliminary analysis of the data shows that with respect to both blockage and lift, the tunnel walls are more open than those giving interference free conditions. This is not surprising since the top and bottom wall open area ratio (16.5% at the model location) was chosen for the generation of supersonic Mach numbers using diffuser suction and not for minimum interference.

In the present tests, lift interference is a more serious problem than blockage interference since the intermediate model ($c/h = 0.125$) appears to be close to being free of blockage interference while the small model ($c/h = 0.06$) still shows signs of lift interference.

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TABLE 1

(a) Pressure Hole Locations—203.2 mm Chord Model

Hole No.	x/c	Hole No.	x/c
1	0.0119	20	0.4871
2	0.0371	21	0.5120
3	0.0623	22	0.5371
4	0.0872	23	0.5621
5	0.1123	24	0.5870
6	0.1372	25	0.6122
7	0.1623	26	0.6371
8	0.1872	27	0.6661
9	0.2122	28	0.6869
10	0.2372	29	0.7121
11	0.2621	30	0.7369
12	0.2872	31	0.7620
13	0.3122	32	0.7870
14	0.3376	33	0.8117
15	0.3618	34	0.8369
16	0.3873	35	0.8619
17	0.4124	36	0.8869
18	0.4371	37	0.9120
19	0.4621	38	0.9416

TABLE 1**(b) Pressure Hole Locations—101.6 mm Chord Model**

Upper Surface		Lower Surface	
Hole No.	x/c	Hole No.	x/c
1	0.0207	1	0.0757
2	0.0787	2	0.1294
3	0.1299	3	0.1830
4	0.1813	4	0.2368
5	0.2307	5	0.2864
6	0.2836	6	0.3383
7	0.3337	7	0.3904
8	0.3858	8	0.4376
9	0.4358	9	0.4870
10	0.4860	10	0.5371
11	0.5380	11	0.5879
12	0.5882	12	0.6371
13	0.6395	13	0.6874
14	0.6894	14	0.7359
15	0.7443		
16	0.7954		
17	0.8414		
18	0.8891		

TABLE 1**(c) Pressure Hole Locations—50.8 mm Chord Model**

Hole No.	x/c
1	0.0253
2	0.0794
3	0.1374
4	0.1911
5	0.2368
6	0.2892
7	0.3371
8	0.3867
9	0.4396
10	0.4913
11	0.5369
12	0.5802
13	0.6275
14	0.6769
15	0.7346

TABLE 2

(a) Test Programme—203.2 mm Chord Model $R = 1.68 \times 10^4$

M	Nominal α	Corrected α	C_N	C_m	C_{DW}
0.50	0	0	-0.0034	0.0000	0.0105
0.50	1	1	0.0841	0.0016	0.0131
0.50	2	2	0.1704	0.0030	0.0091
0.50	3	3	0.2583	0.0044	0.0104
0.50	4	4	0.3447	0.0062	0.0107
0.50	5	5	0.4316	0.0078	0.0112
0.50	6	6	0.5183	0.0100	0.0110
0.50	7	7	0.6052	0.0124	0.0118
0.50	8	8	0.6868	0.0149	0.0112
0.60	0	0	-0.0014	0.0002	0.0100
0.60	1	1	0.0880	0.0025	0.0136
0.60	2	2	0.1786	0.0042	0.0109
0.60	3	3	0.2706	0.0064	0.0100
0.60	4	4	0.3598	0.0089	0.0109
0.60	5	5	0.4560	0.0130	0.0123
0.60	6	6	0.5391	0.0157	0.0127
0.60	7	7	0.6311	0.0211	0.0137
0.65	0	0	-0.0015	-0.0002	0.0091
0.65	1	1	0.0927	0.0026	0.0125
0.65	2	2	0.1824	0.0053	0.0108
0.65	3	3	0.2784	0.0086	0.0105
0.65	4	4	0.3728	0.0119	0.0112
0.65	5	5	0.4714	0.0166	0.0128
0.65	6	6	0.5726	0.0213	0.0159
0.675	0	0	-0.0060	0.0002	0.0093
0.675	1	1	0.0941	0.0027	0.0138
0.675	2	2	0.1867	0.0060	0.0106
0.675	3	3	0.2835	0.0101	0.0115
0.675	4	4	0.3831	0.0145	0.0123
0.675	5	5	0.4816	0.0187	0.0148

TABLE 2 (a) (Continued)

M	Nominal α	Corrected α	C_N	C_m	C_{DW}
0.70	0	0	-0.0050	-0.0002	0.0098
0.70	1	1	0.0970	0.0031	0.0124
0.70	2	2	0.1941	0.0075	0.0122
0.70	3	3	0.2930	0.0117	0.0115
0.70	4	4	0.3936	0.0168	0.0127
0.70	5	5	0.4998	0.0193	0.0171
0.72	0	0	-0.0065	0.0001	0.0107
0.72	1	1	0.0983	0.0040	0.0125
0.72	2	2	0.1981	0.0085	0.0112
0.72	3	3	0.3028	0.0133	0.0115
0.72	4	4	0.4100	0.0168	0.0140
0.74	0	0	-0.0057	0.0000	0.0106
0.74	1	1	0.0992	0.0043	0.0141
0.74	2	2	0.2052	0.0095	0.0119
0.74	3	3	0.3149	0.0135	0.0134
0.74	4	4	0.4219	0.0150	0.0187
0.76	0	0	-0.0065	-0.0002	0.0109
0.76	1	1	0.1069	0.0052	0.0138
0.76	2	2	0.2144	0.0090	0.0132
0.76	3	3	0.3227	0.0118	0.0170
0.76	4	4	0.4239	0.0112	0.0236
0.78	0	0	-0.0072	0.0000	0.0106
0.78	1	1	0.1138	0.0045	0.0146
0.78	2	2	0.2243	0.0063	0.0152
0.78	3	3	0.3319	0.0060	0.0214
0.80	0	0	-0.0049	0.0000	0.0146
0.80	1	1	0.1199	0.0014	0.0171
0.80	2	2	0.2299	0.0006	0.0216
0.82	0	0	-0.0137	0.0017	0.0198
0.82	1	1	0.1182	-0.0039	0.0233
0.82	2	2	0.2141	-0.0037	0.0283

TABLE 2

(b) Test Programme—203.2 mm Chord Model $R = 0.84 \times 10^4$

M	Nominal α	Corrected α	C_N	C_m	C_{DW}
0.50	0	0	0.0000	0.0000	0.0126
0.50	1	1	0.0808	0.0018	0.0140
0.50	2	2	0.1708	0.0031	0.0123
0.50	3	3	0.2561	0.0051	0.0124
0.50	4	4	0.3438	0.0066	0.0124
0.50	5	5	0.4286	0.0088	0.0127
0.50	6	6	0.5147	0.0111	0.0129
0.50	7	7	0.5942	0.0142	0.0123
0.50	8	8	0.6801	0.0182	0.0132
0.60	0	0	0.0000	0.0000	—
0.65	0	0	0.0000	0.0000	—
0.675	0	0	0.0000	0.0000	—
0.70	0	0	0.0000	0.0000	0.0117
0.70	1	1	0.0969	0.0028	0.0144
0.70	2	2	0.1930	0.0074	0.0123
0.70	3	3	0.2954	0.0115	0.0127
0.70	4	4	0.3932	0.0167	0.0127
0.70	5	5	0.4987	0.0204	0.0177
0.72	0	0	0.0000	0.0000	—
0.74	0	0	0.0000	0.0000	—
0.76	0	0	0.0000	0.0000	—
0.78	0	0	0.0000	0.0000	—
0.80	0	0	0.0000	0.0000	0.0146
0.80	1	1	0.1227	0.0002	0.0171
0.80	2	2	0.2384	-0.0012	0.0216
0.82	0	0	0.0000	0.0000	—

TABLE 2

(c) Test Programme — 203.2 mm Cord Model $R = 0.42 \times 10^4$

M	Nominal α	Corrected α	C_N	C_m	C_{DW}
0.50	0	0	0.0000	0.0000	—
0.50	1	1	0.0768	0.0025	—
0.50	2	2	0.1573	0.0055	—
0.50	3	3	0.2336	0.0106	—
0.50	4	4	0.3093	0.0152	—
0.50	5	5	0.3833	0.0175	—
0.50	6	6	0.4966	0.0136	—
0.50	7	7	0.6062	0.0084	—
0.50	8	8	0.6711	0.0143	—
0.70	0	0	0.0000	0.0000	—
0.70	1	1	0.0913	0.0051	—
0.70	2	2	0.1770	0.0108	—
0.70	3	3	0.2689	0.0182	—
0.70	4	4	0.3603	0.0253	—
0.70	5	5	0.4446	0.0311	—
0.80	0	0	0.0000	0.0000	—
0.80	1	1	0.0983	0.0024	—
0.80	2	2	0.1875	0.0012	—

TABLE 2

(d) Test Programme — 101.6 mm Chord Model $R = 0.84 \times 10^6$

M	Nominal α	Corrected α	C_N	C_m	C_{DW}
0.50	0	0.01	0.0214	0.0036	0.0109
0.50	1	1.02	0.0943	0.0012	0.0134
0.50	2	2.03	0.1932	0.0026	0.0144
0.50	3	3.05	0.2909	0.0043	0.0150
0.50	4	4.06	0.3855	0.0064	0.0140
0.50	5	5.07	0.4765	0.0094	0.0155
0.50	6	6.09	0.5771	0.0126	0.0150
0.50	7	7.10	0.6700	0.0153	0.0101
0.50	8	8.11	0.7212	0.0177	0.0125
0.60	0	0.01	0.0197	0.0040	0.0115
0.60	1	1.03	0.0927	0.0016	0.0116
0.60	2	2.04	0.2024	0.0045	0.0142
0.60	3	3.06	0.3032	0.0074	0.0153
0.60	4	4.08	0.4016	0.0109	0.0132
0.60	5	5.09	0.4981	0.0155	0.0169
0.60	6	6.12	0.6052	0.0215	0.0181
0.60	7	7.15	0.6950	0.0274	0.0176
0.65	0	0.01	0.0230	0.0045	0.0139
0.65	1	1.03	0.1070	0.0028	0.0109
0.65	2	2.05	0.2112	0.0059	0.0146
0.65	3	3.07	0.3176	0.0111	0.0161
0.65	4	4.10	0.4334	0.0178	0.0157
0.65	5	5.12	0.5291	0.0203	0.0191
0.65	6	6.14	0.6315	0.0248	0.0246
0.675	0	0.01	0.0258	0.0047	0.0136
0.675	1	1.03	0.1112	0.0038	0.0122
0.675	2	2.06	0.2206	0.0083	0.0135
0.675	3	3.09	0.3564	0.0177	0.0147
0.675	4	4.11	0.4468	0.0182	0.0176
0.675	5	5.13	0.5504	0.0215	0.0240

TABLE 2(d) (Continued)

M	Nominal α	Corrected α	C_N	C_m	C_{DW}
0.70	0	0.01	0.0275	0.0048	0.0124
0.70	1	1.04	0.1189	0.0054	0.0126
0.70	2	2.06	0.2550	0.0149	0.0143
0.70	3	3.10	0.3477	0.0158	0.0169
0.70	4	4.12	0.4641	0.0194	0.0199
0.70	5	5.14	0.5594	0.0194	0.0301
0.72	0	0.01	0.0237	0.0055	0.0130
0.72	1	1.04	0.1265	0.0080	0.0131
0.72	2	2.07	0.2733	0.0177	0.0145
0.72	3	3.10	0.3648	0.0163	0.0172
0.72	4	4.12	0.4683	0.0181	0.0241
0.74	0	0.01	0.0254	0.0063	0.0127
0.74	1	1.04	0.1446	0.0108	0.0127
0.74	2	2.08	0.2708	0.0168	0.0144
0.74	3	3.10	0.3734	0.0148	0.0202
0.74	4	4.11	0.4646	0.0098	0.0303
0.76	0	0.01	0.0166	0.0045	0.0124
0.76	1	1.05	0.1530	0.0118	0.0133
0.76	2	2.08	0.2819	0.0143	0.0166
0.76	3	3.09	0.3753	0.0101	0.0246
0.76	4	4.09	0.4051	0.0016	0.0444
0.78	0	0	0.0033	0.0028	0.0137
0.78	1	1.04	0.1497	0.0068	0.0149
0.78	2	2.07	0.2867	0.0080	0.0200
0.78	3	3.07	0.3387	0.0013	0.0395
0.80	0	0	0.0023	0.0019	0.0168
0.80	1	1.03	0.1390	-0.0012	0.0193
0.80	2	2.05	0.2480	0.0025	0.0282
0.82	0	0	0.0006	0.0020	0.0233
0.82	1	1.02	0.1034	-0.0046	0.0243
0.82	2	2.05	0.1429	0.0077	0.0343

TABLE 2

(c) Test Programme — 101.6 mm Chord Model $R = 0.42 \times 10^6$

M	Nominal α	Corrected α	C_N	C_m	C_{DW}
0.50	0	0	0.0159	0.0033	0.0127
0.50	1	1.01	0.0908	0.0008	0.0120
0.50	2	2.02	0.1844	0.0018	0.0158
0.50	3	3.03	0.2815	0.0035	0.0172
0.50	4	4.03	0.3783	0.0056	0.0157
0.50	5	5.04	0.4720	0.0084	0.0177
0.50	6	6.04	0.5740	0.0105	0.0137
0.50	7	7.05	0.6783	0.0105	0.0099
0.50	8	8.06	0.7589	0.0176	0.0205
0.70	0	0	0.0100	0.0029	0.0126
0.70	1	1.02	0.1098	0.0029	0.0126
0.70	2	2.03	0.2482	0.0111	0.0143
0.70	3	3.05	0.3516	0.0145	0.0168
0.70	4	4.06	0.4464	0.0177	0.0197
0.70	5	5.07	0.5140	0.0170	0.0315
0.80	0	0	0.0012	0.0015	0.0169
0.80	1	1.01	0.1539	0.0024	0.0195
0.80	2	2.02	0.2076	-0.0023	0.0291

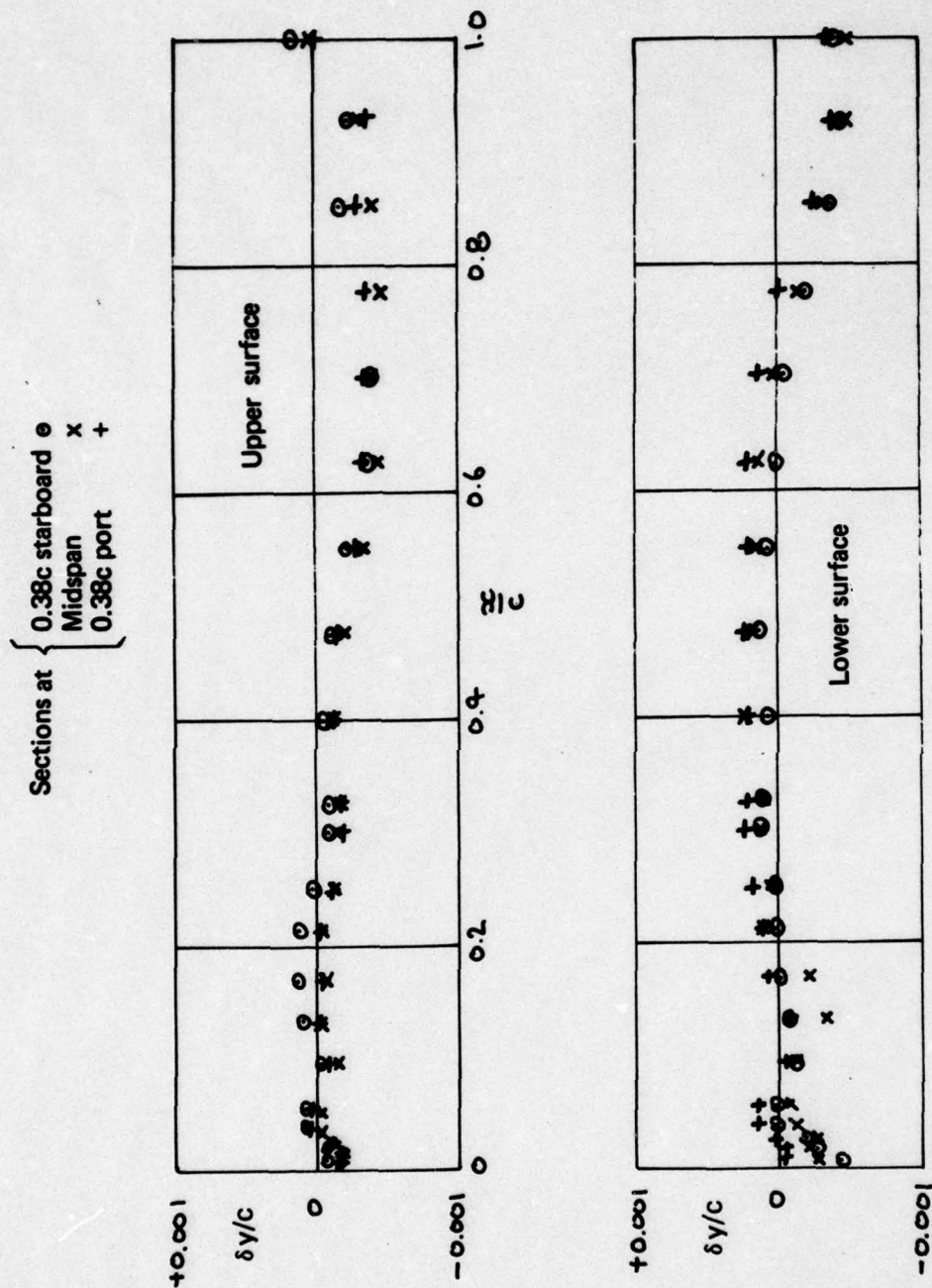
TABLE 2

(f) Test Programme — 50.8 mm Chord Model $R = 0.42 \times 10^6$

M	Nominal α	Corrected α	C_N	C_m	C_{DW}
0.50	0	-0.02	-0.0034	-0.0006	0.0145
0.50	0.5	0.61	0.0683	0.0001	0.0152
0.50	1.0	1.25	0.1364	0.0012	0.0152
0.50	1.5	1.89	0.2044	0.0026	0.0161
0.50	2.0	2.52	0.2739	0.0036	0.0182
0.60	0	-0.02	-0.0041	-0.0006	0.0127
0.60	0.5	0.72	0.0790	0.0003	0.0140
0.60	1.0	1.36	0.1604	0.0026	0.0108
0.60	1.5	2.05	0.2390	0.0043	0.0127
0.60	2.0	2.75	0.3138	0.0067	0.0128
0.65	0	-0.03	-0.0027	-0.0009	0.0160
0.65	0.5	0.72	0.0895	0.0015	0.0155
0.65	1.0	1.45	0.1781	0.0039	0.0134
0.65	1.5	2.24	0.2688	0.0080	0.0168
0.675	0	-0.02	-0.0046	-0.0006	0.0178
0.675	0.5	0.75	0.0951	0.0021	0.0157
0.675	1.0	1.53	0.1909	0.0052	0.0180
0.675	1.5	2.40	0.2907	0.0111	0.0143
0.70	0	-0.05	-0.0099	-0.0012	0.0162
0.70	0.5	0.80	0.1045	0.0029	0.0169
0.70	1.0	1.62	0.2100	0.0067	0.0159
0.72	0	-0.03	-0.0030	-0.0011	0.0146
0.72	0.5	0.85	0.1150	0.0039	0.0147
0.72	1.0	1.75	0.2298	0.0092	0.0168
0.74	0	-0.04	-0.0104	-0.0008	0.0140
0.74	0.5	0.93	0.1285	0.0053	0.0157
0.74	1.0	1.93	0.2658	0.0129	0.0176

TABLE 2 (f) (Continued)

M	Nominal α	Corrected α	C_N	C_m	C_{DW}
0.76	0	-0.07	-0.0126	-0.0015	0.0171
0.76	0.5	1.01	0.1463	0.0067	0.0174
0.76	1.0	1.95	0.2887	0.0110	0.0248
0.78	0	-0.10	-0.0188	-0.0024	0.0146
0.78	0.5	1.11	0.1738	0.0080	0.0186
0.78	1.0	1.80	0.2745	0.0066	0.0252
0.80	0	-0.05	-0.0173	-0.0006	0.0171
0.80	0.5	0.93	0.1523	0.0029	0.0242
0.80	1.0	1.54	0.2511	-0.0009	0.0300
0.82	0	-0.01	-0.0158	0.0008	0.0273
0.82	0.5	0.45	0.1014	-0.0086	0.0280
0.82	1.0	1.21	0.1873	-0.0073	0.0336



δy = Measured ordinates — design ordinates
(positive for excess material)

FIG. 1. PROFILE ERRORS — 203.2 mm CHORD MODEL

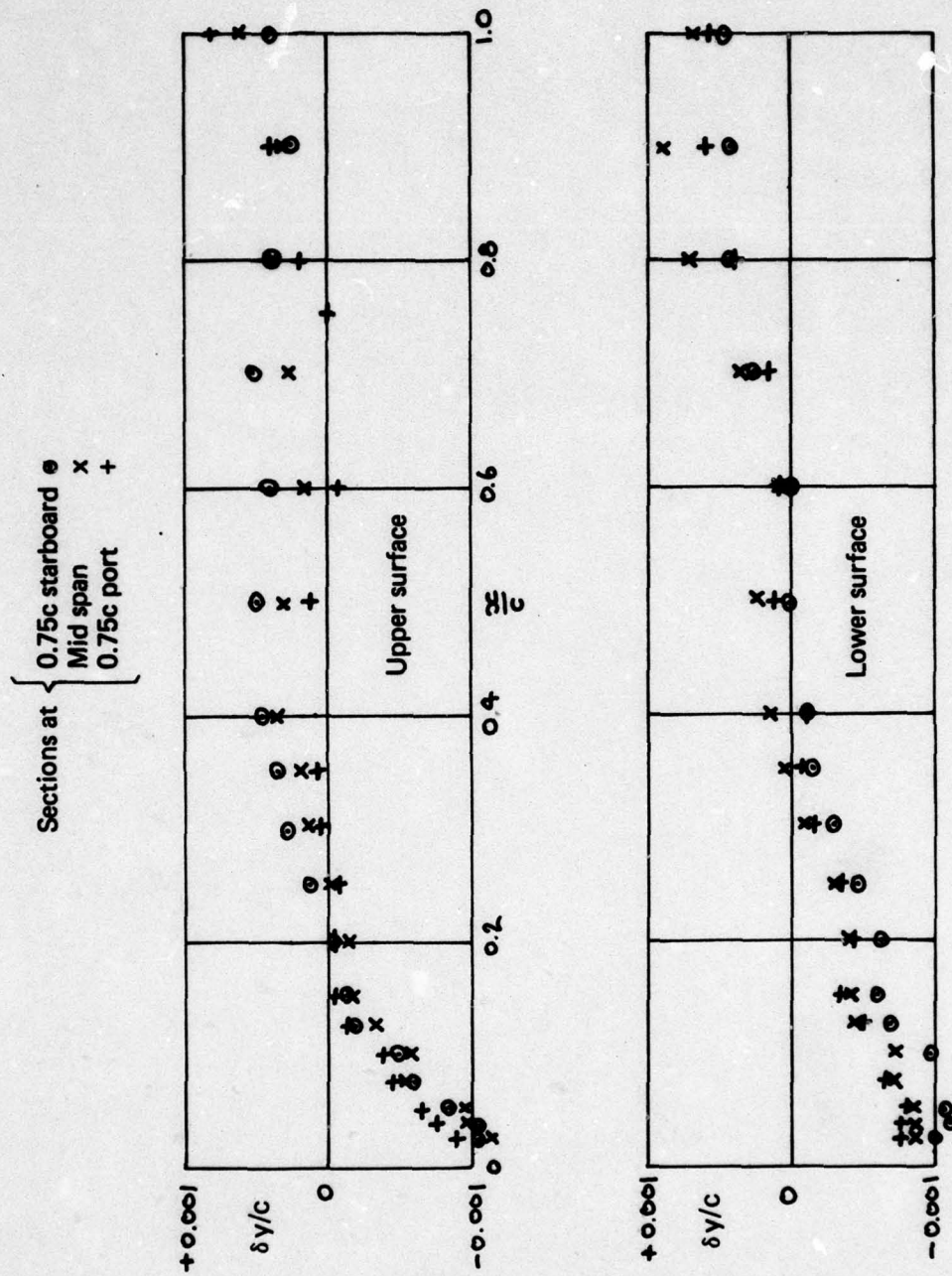
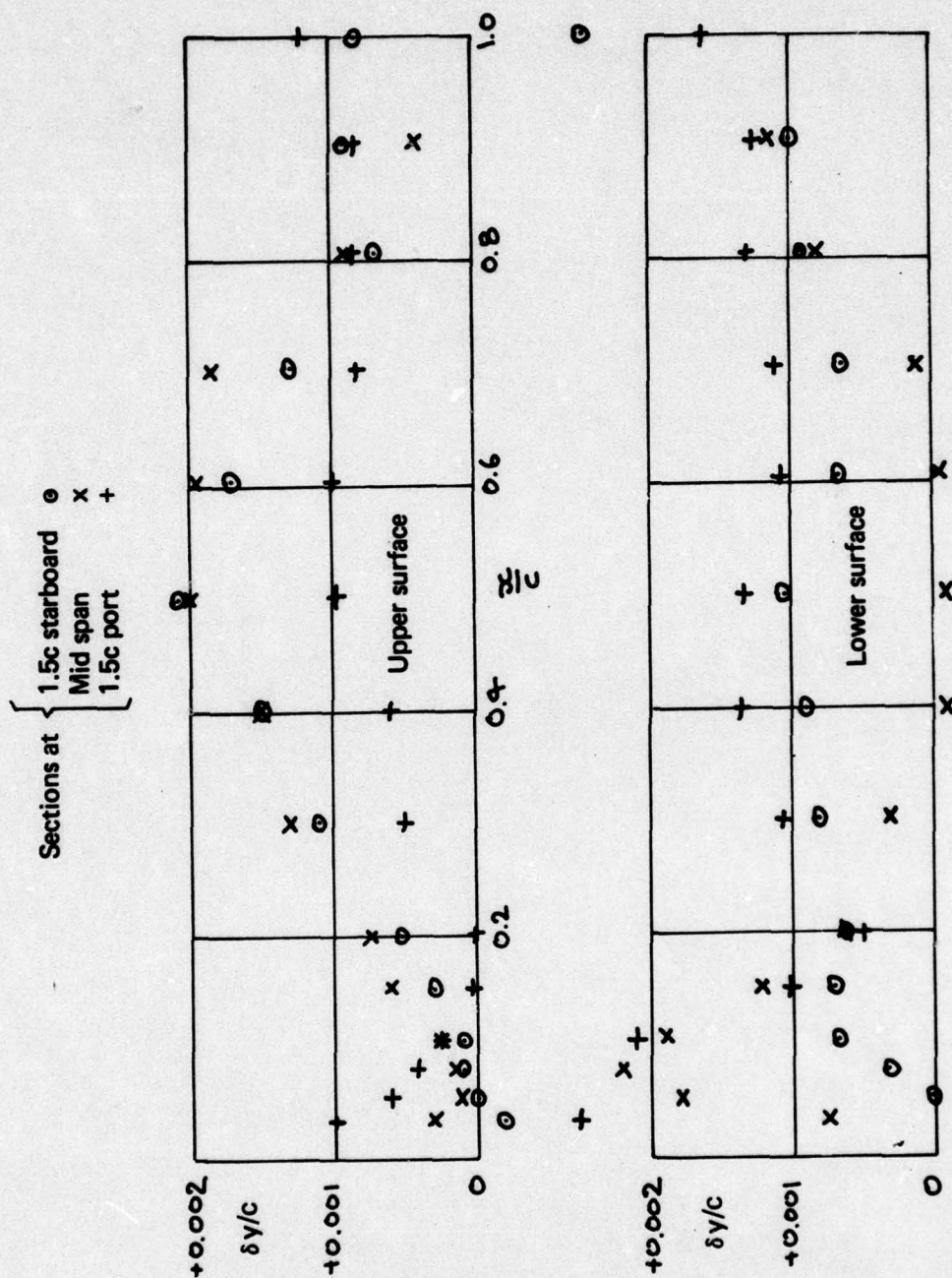


FIG. 2. PROFILE ERRORS - 101.6 mm CHORD MODEL



δy = Measured ordinates - design ordinates
(positive for excess material)

FIG. 3. PROFILE ERRORS - 50.8 mm CHORD MODEL

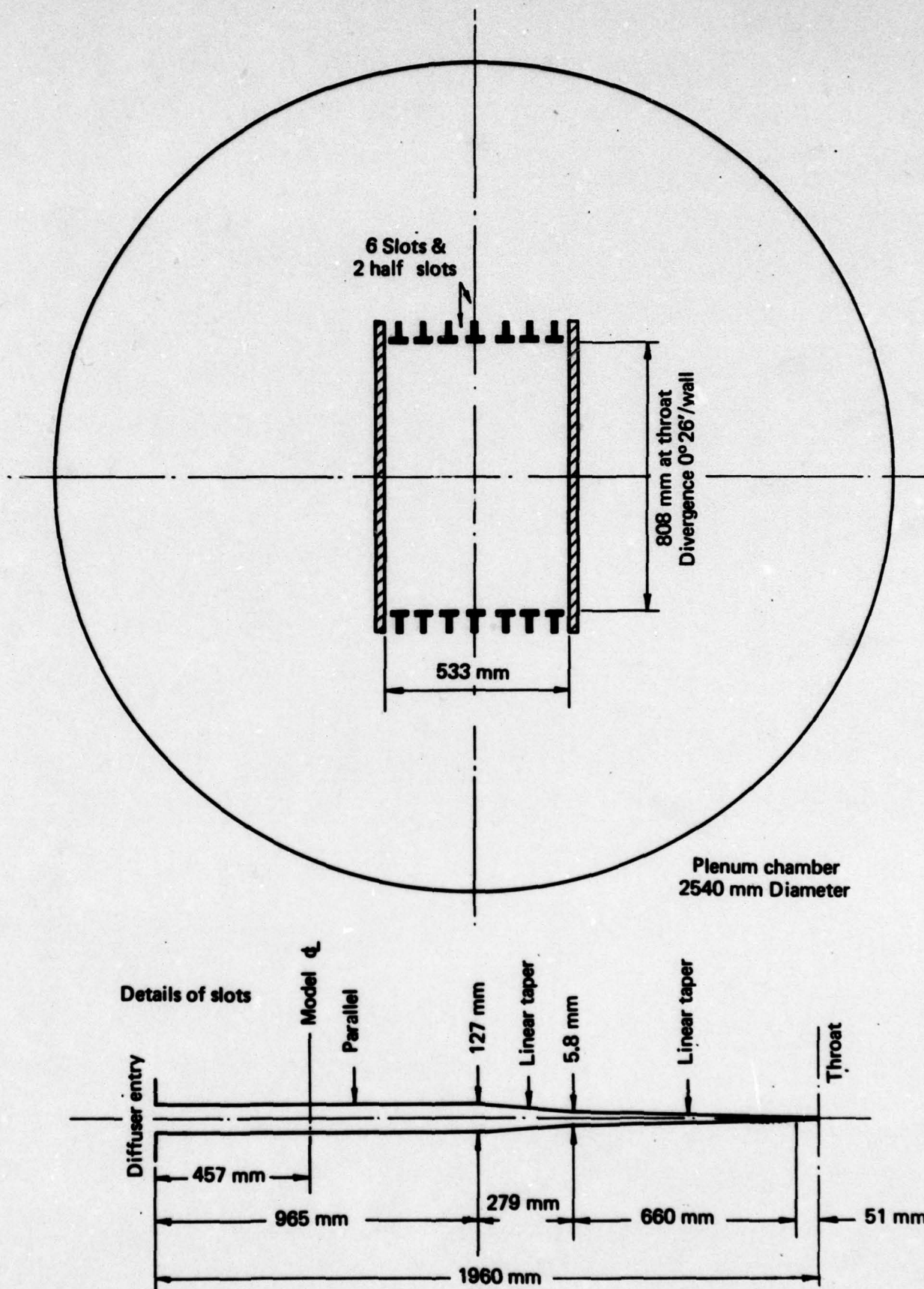


FIG. 4. DETAILS OF SLOTTED WORKING SECTION

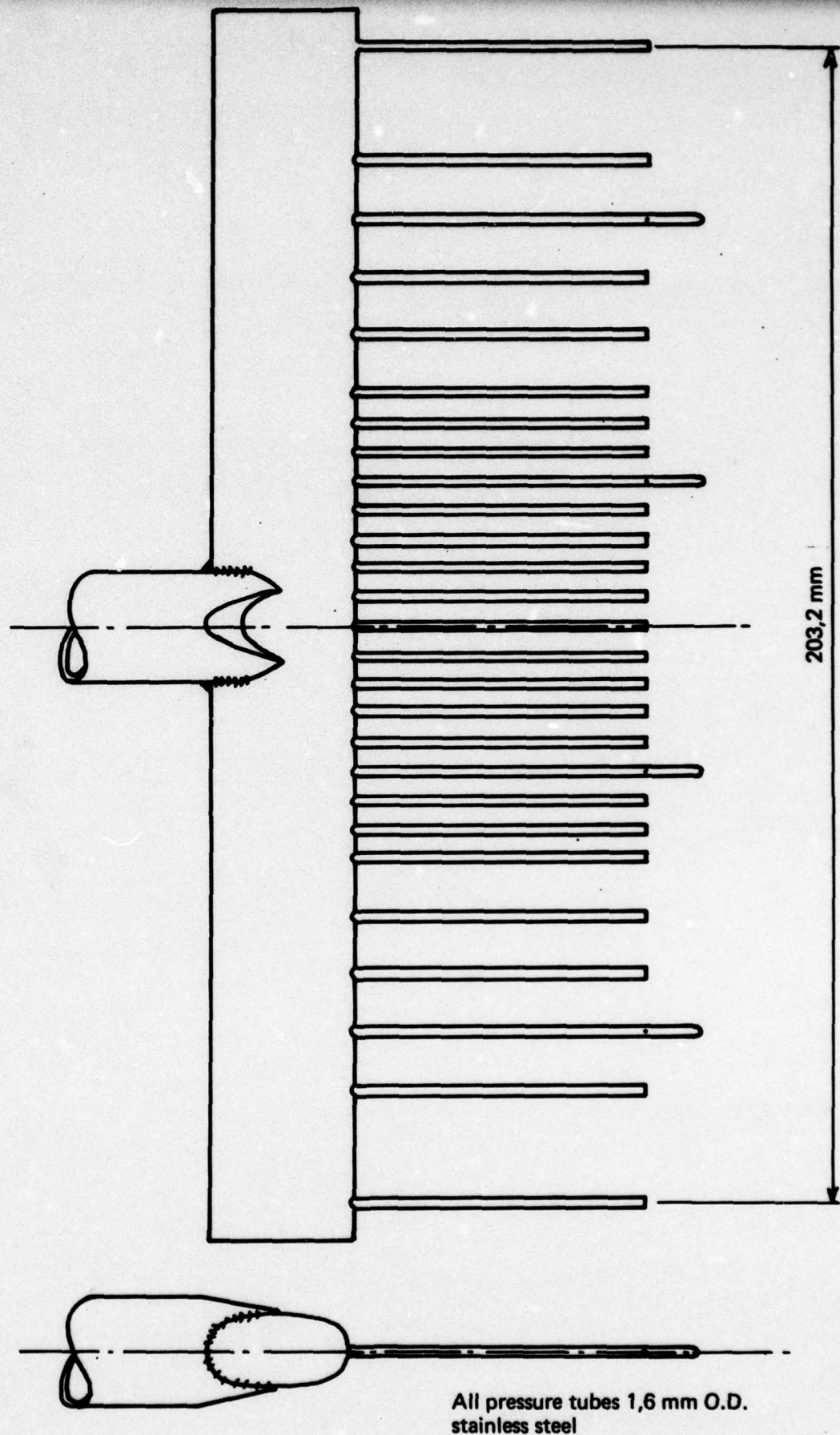
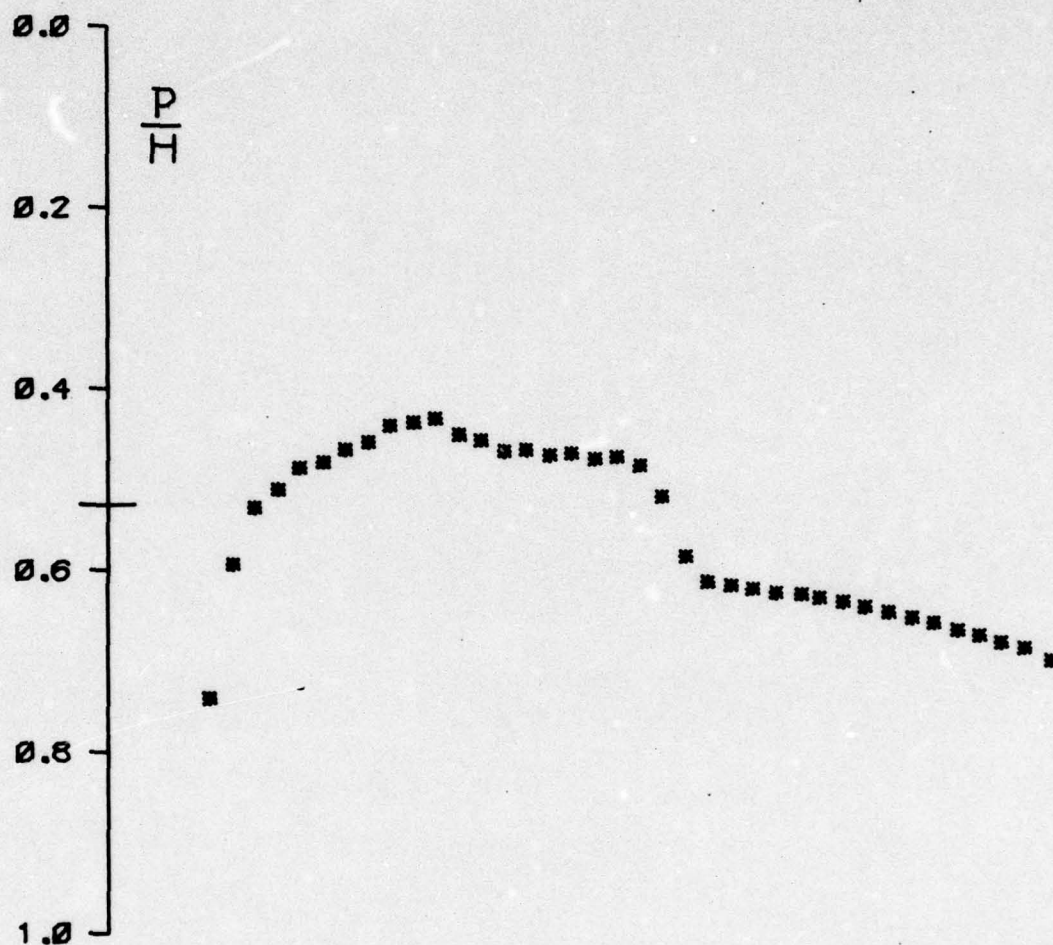
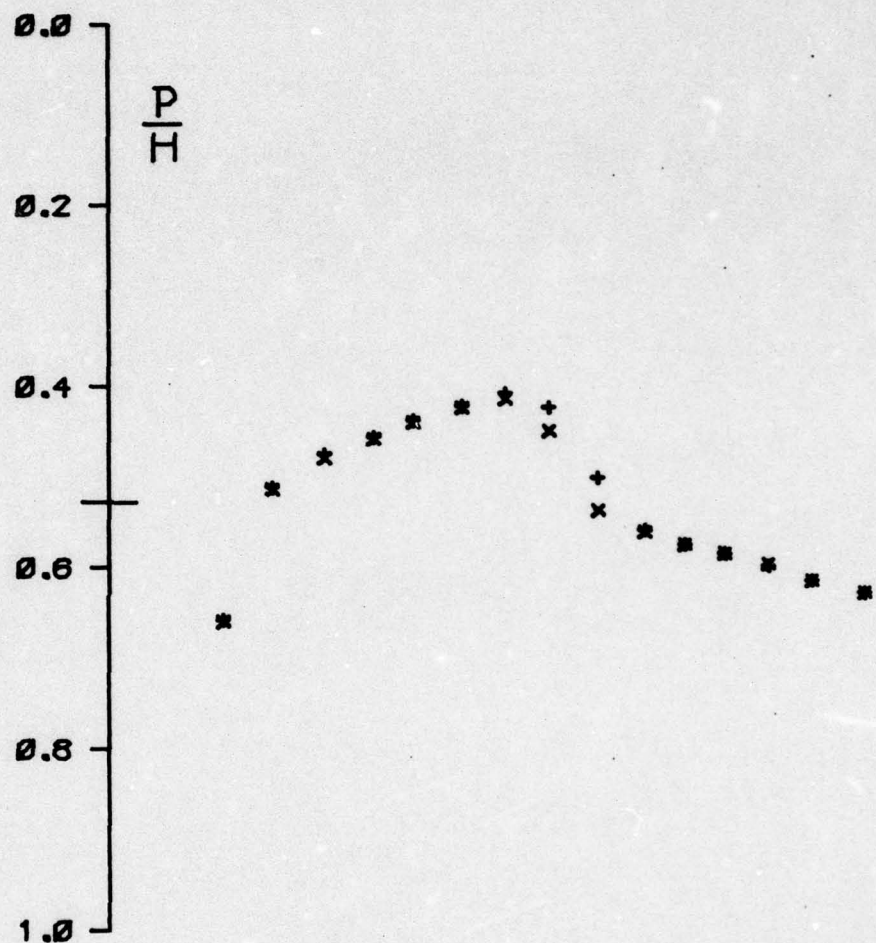


FIG. 5. ARRANGEMENT OF DRAG RAKE



+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .800$ $AL = 0.00$ $CN = 0.000$ $CM = 0.000$ $R = 0.439$

FIG. 6. EXPERIMENTAL PRESSURE DISTRIBUTION



+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .799$ $AL = 0.00$ $CN = -0.017$ $CM = -0.001$ $R = 0.399$

FIG. 7. EXPERIMENTAL PRESSURE DISTRIBUTION

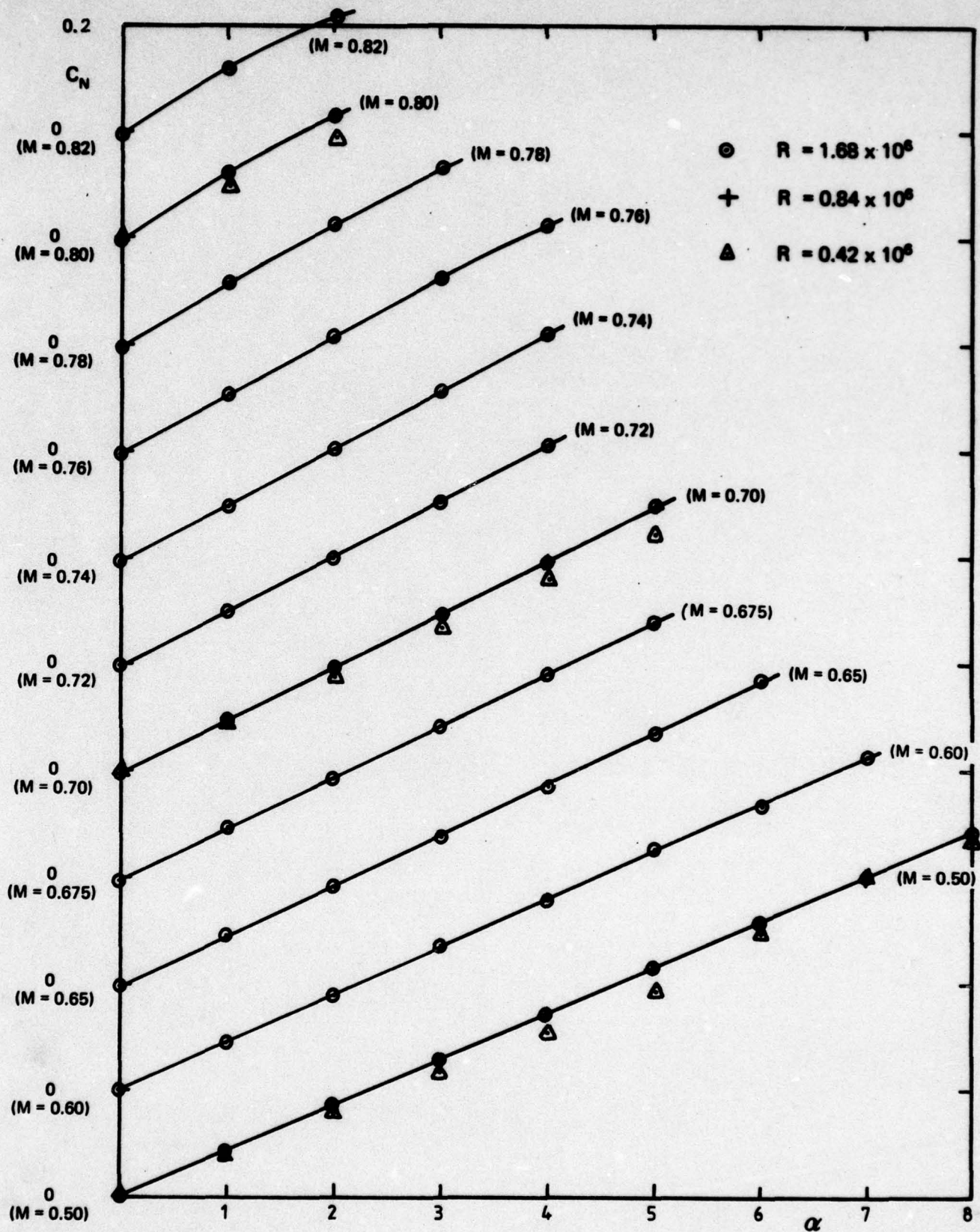


FIG. 8. VARIATION OF C_N WITH α - 203.2 mm CHORD MODEL

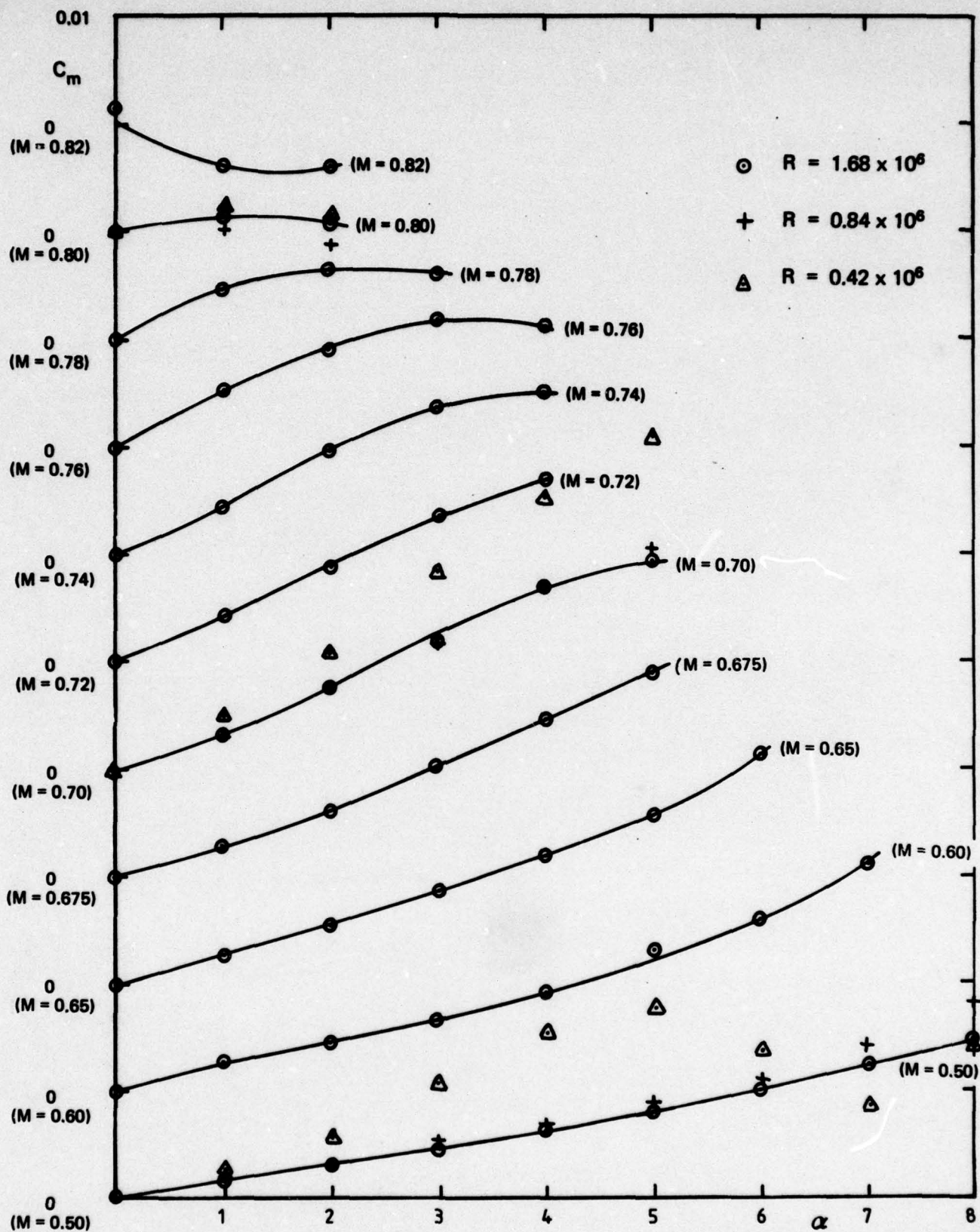


FIG. 9. VARIATION OF C_m WITH α - 203.2 mm CHORD MODEL

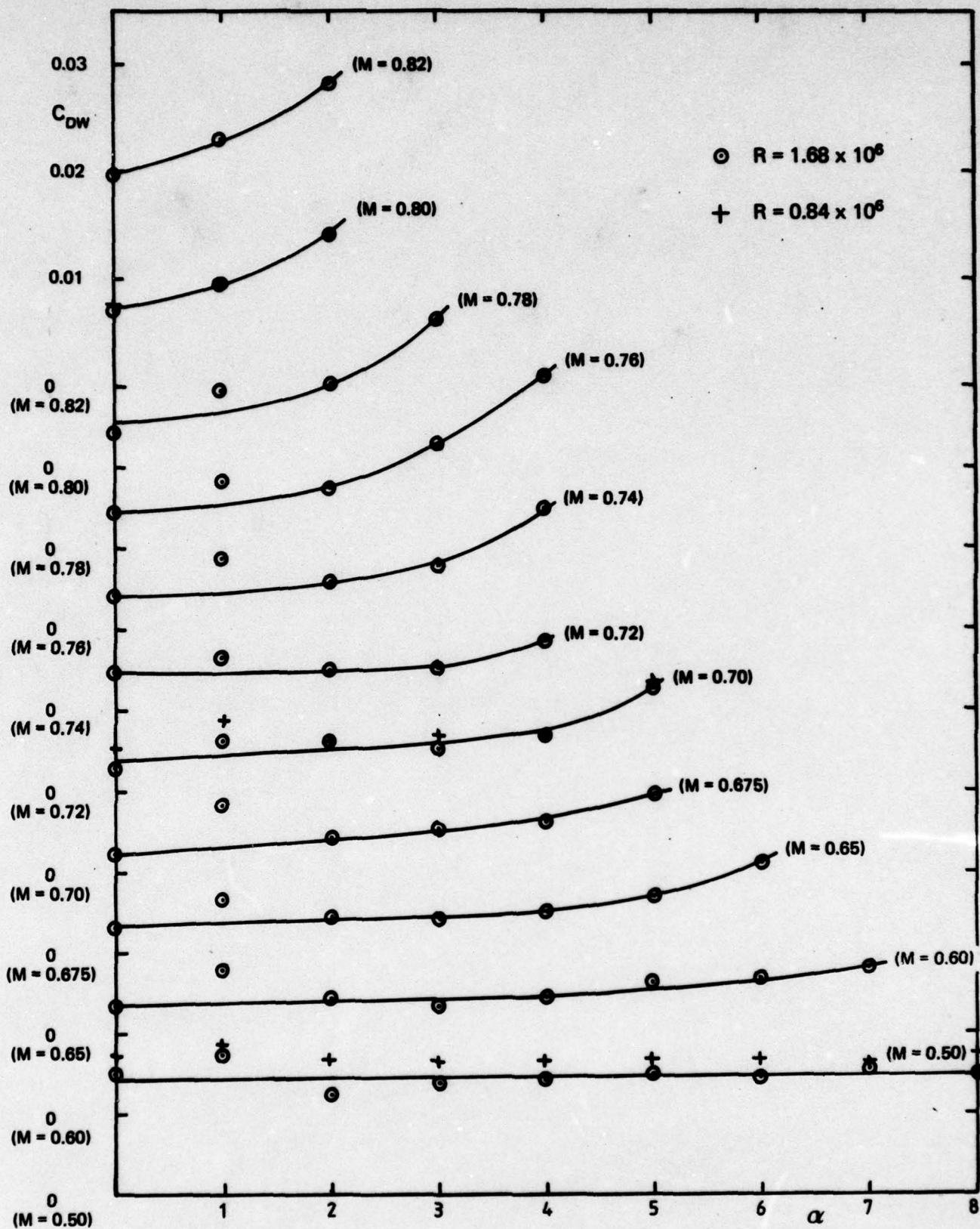


FIG. 10. VARIATION OF C_{DW} WITH α - 203.2 mm CHORD MODEL

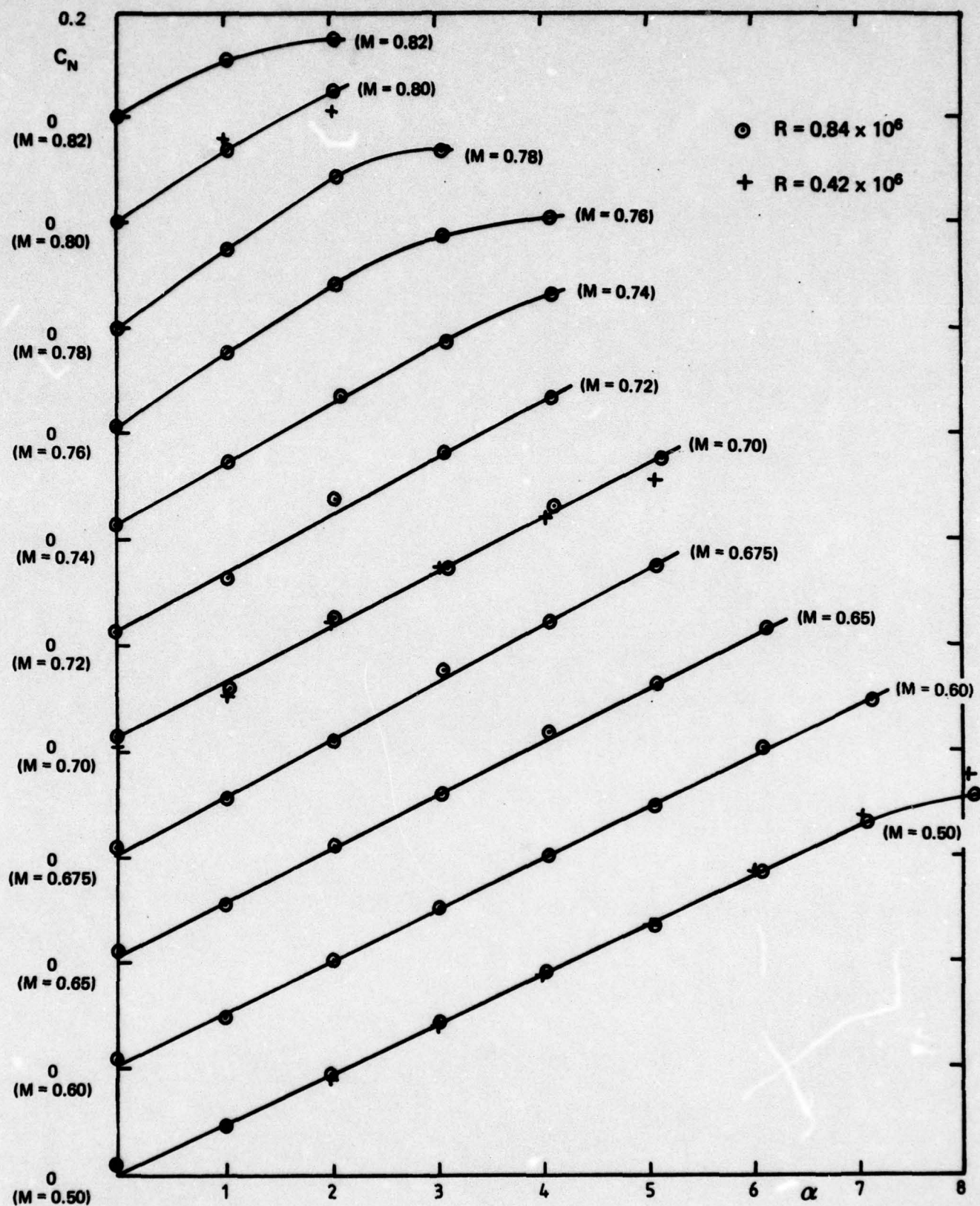


FIG. 11. VARIATION OF C_N WITH α 101.6 mm CHORD MODEL

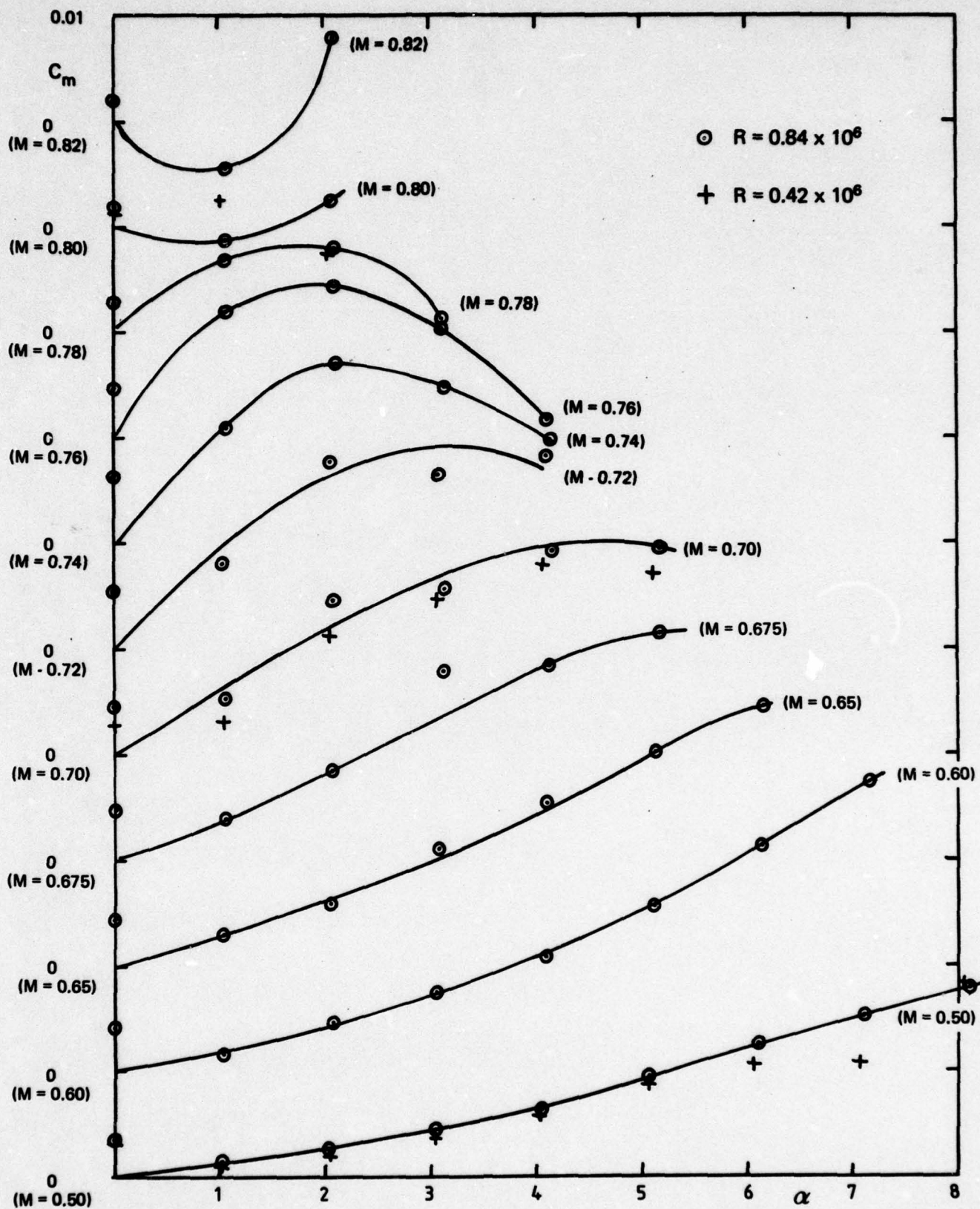


FIG. 12. VARIATION OF C_m WITH α 101.6 mm CHORD MODEL

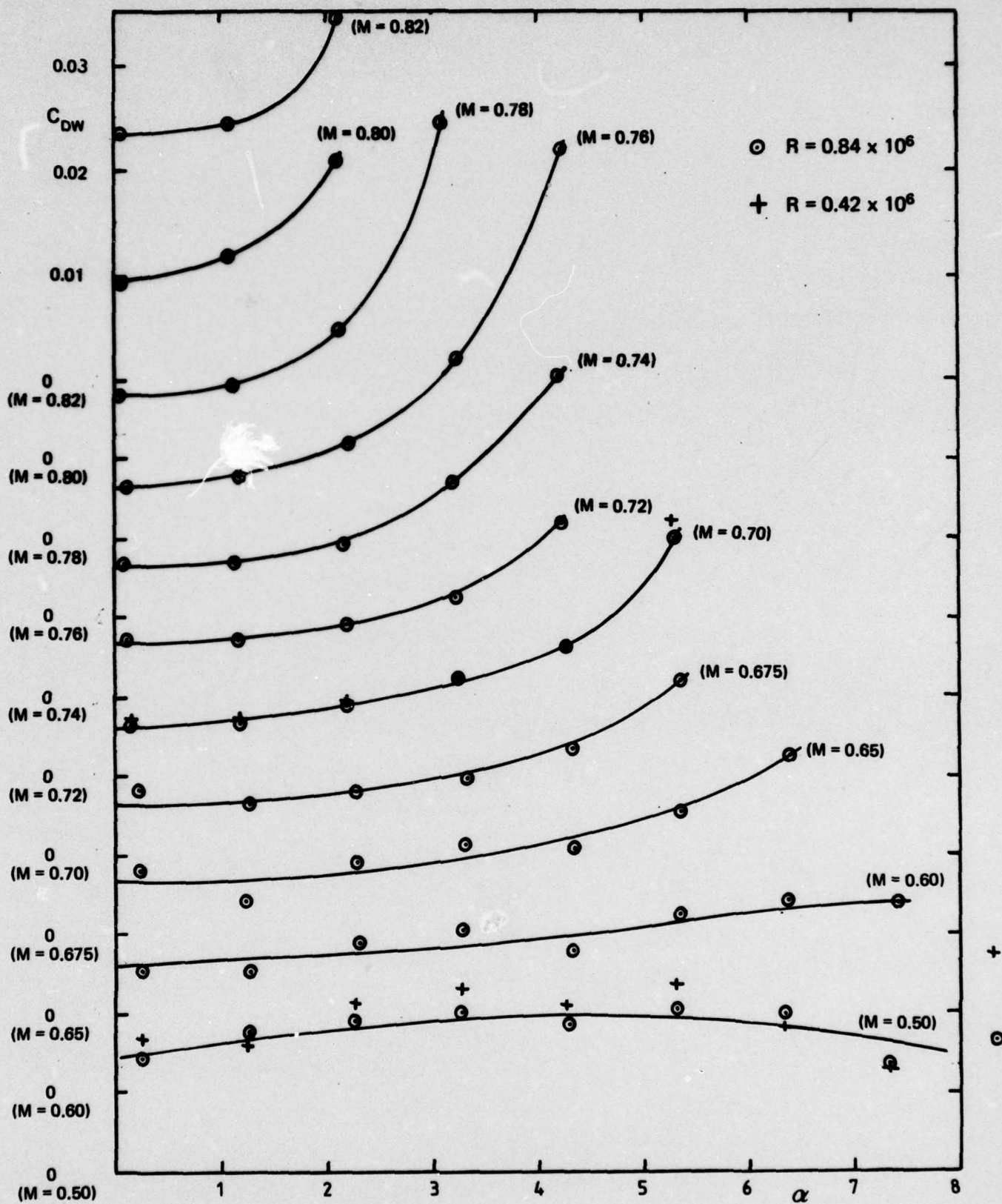


FIG. 13. VARIATION OF C_{DW} WITH α 101.6 mm CHORD MODEL

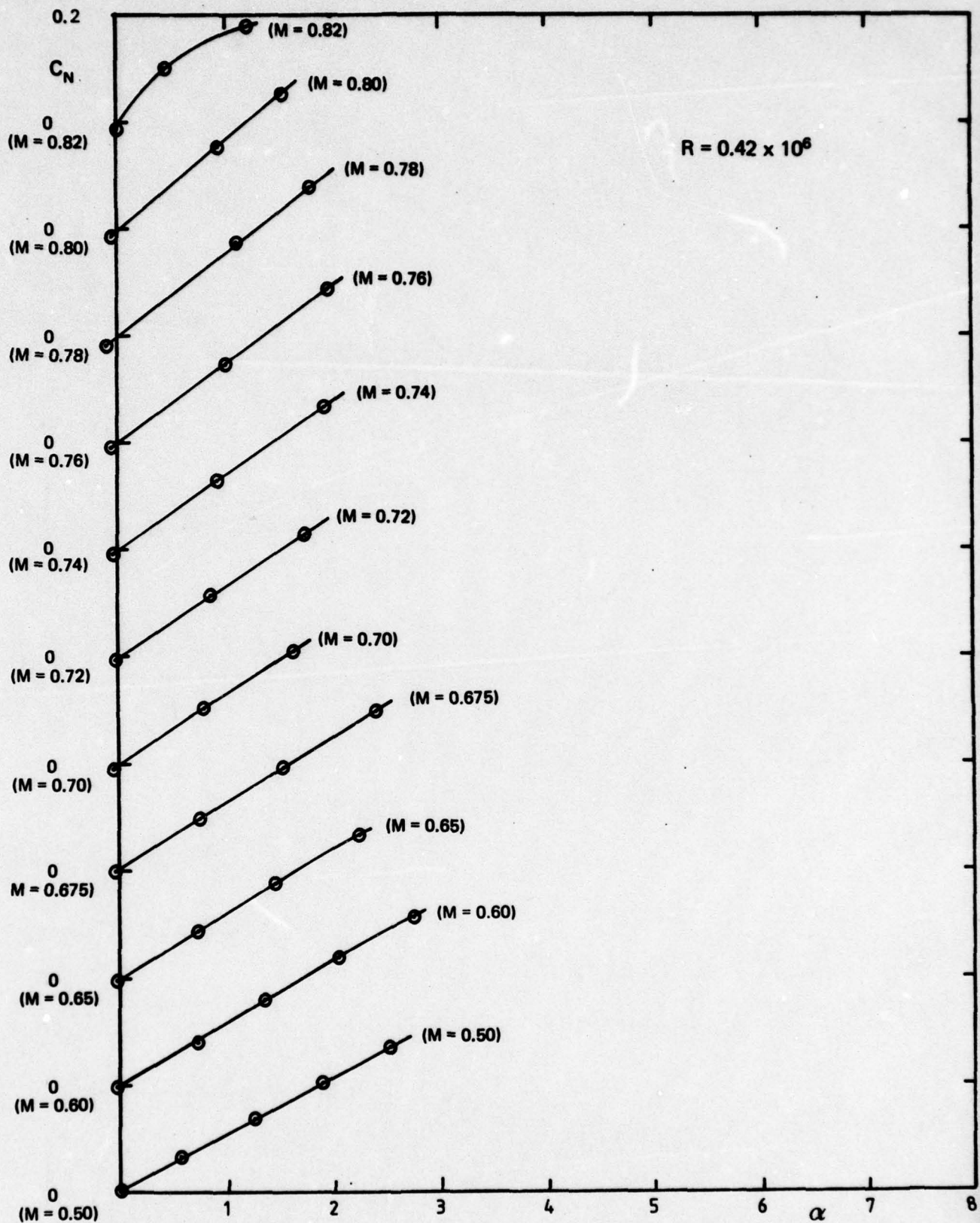


FIG. 14. VARIATION OF C_N WITH α - 50.8 mm CHORD MODEL

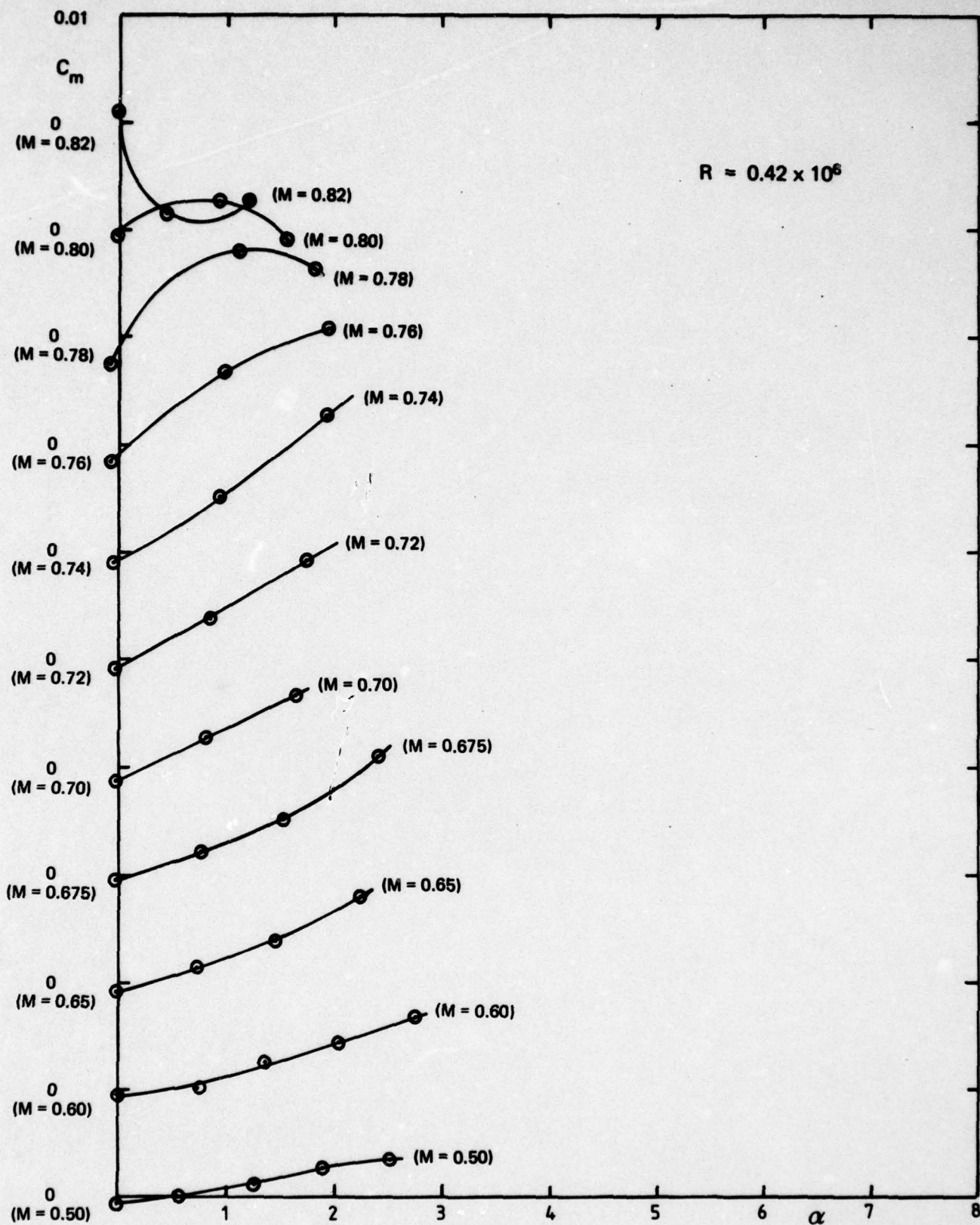


FIG. 15. VARIATION OF C_m WITH α - 50.8 mm CHORD MODEL

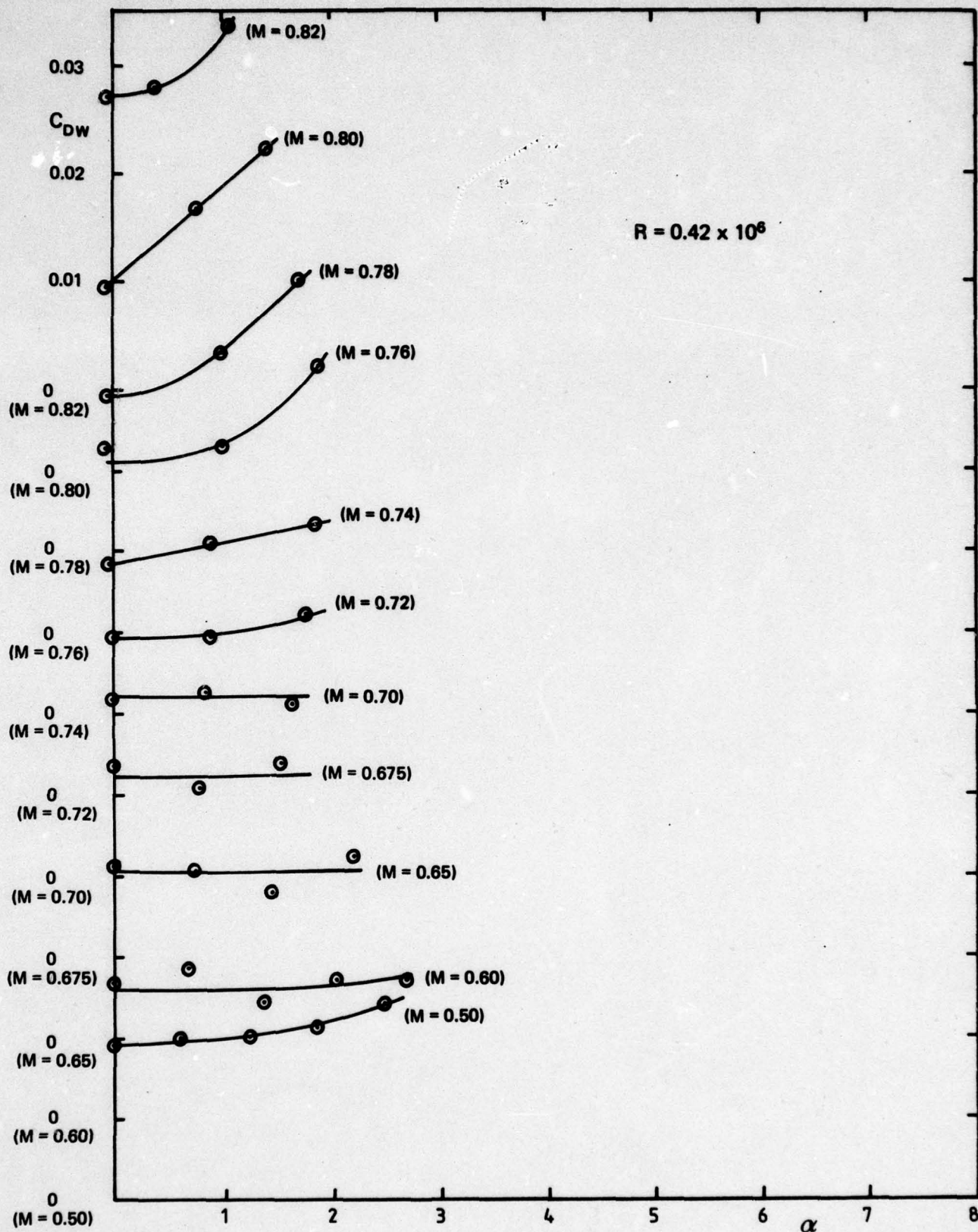
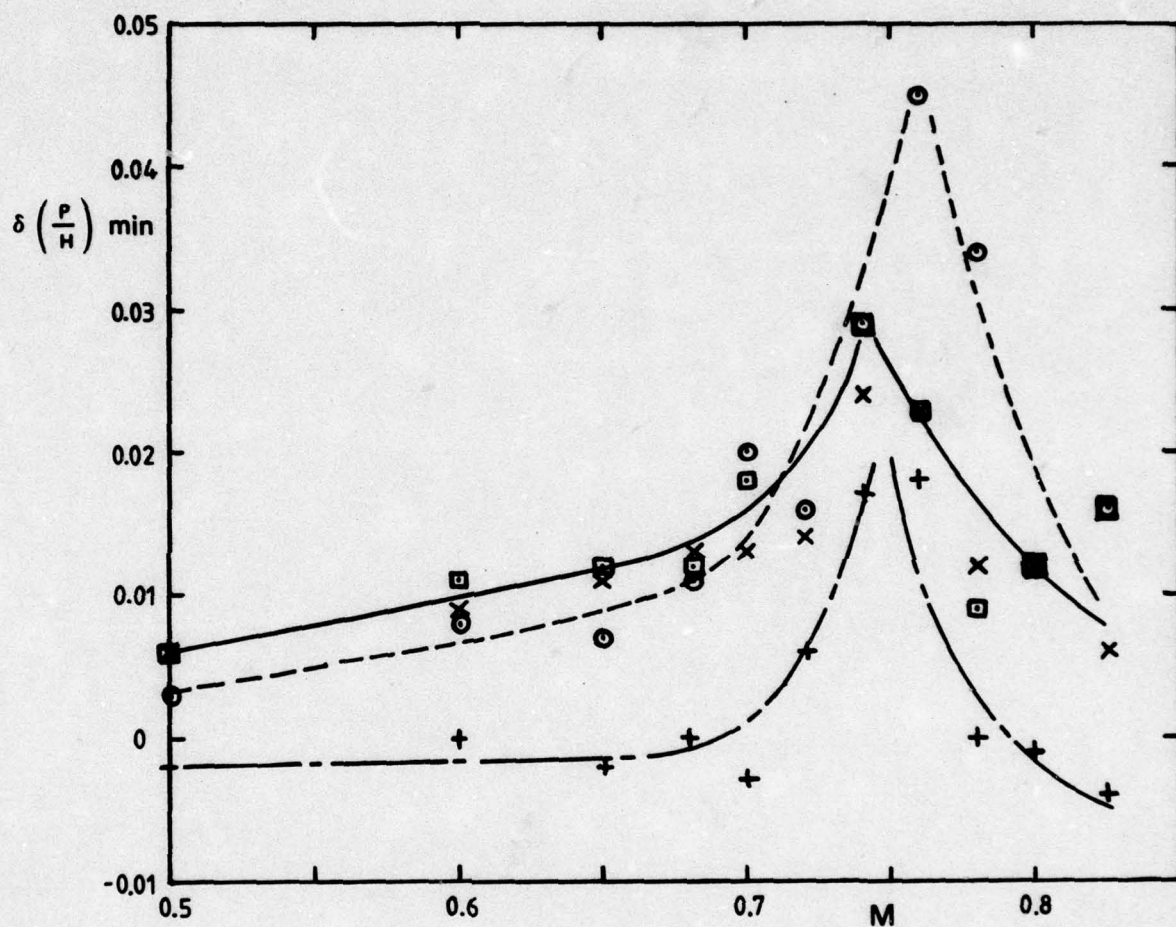


FIG. 16. VARIATION OF C_{DW} WITH α - 50.8 mm CHORD MODEL

- x 203.2 mm model $R = 1.68 \times 10^6$
- 203.2 mm model $R = 0.84 \times 10^6$
- 203.2 mm model mean curve
- + - - - 101.6 mm model $R = 0.84 \times 10^6$
- - - - 50.8 mm model $R = 0.42 \times 10^6$



$$\delta \left(\frac{P}{H} \right) \text{ min.} = \left[\left(\frac{P}{H} \right) \text{ min} \right] \text{ experiment} - \left[\left(\frac{P}{H} \right) \text{ min} \right] \text{ inviscid computation}$$

FIG. 17. VARIATION OF MINIMUM P/H WITH M ($\alpha = 0^\circ$)
(Comparison between 203.2 mm, 101.6 mm and 50.8 mm models)

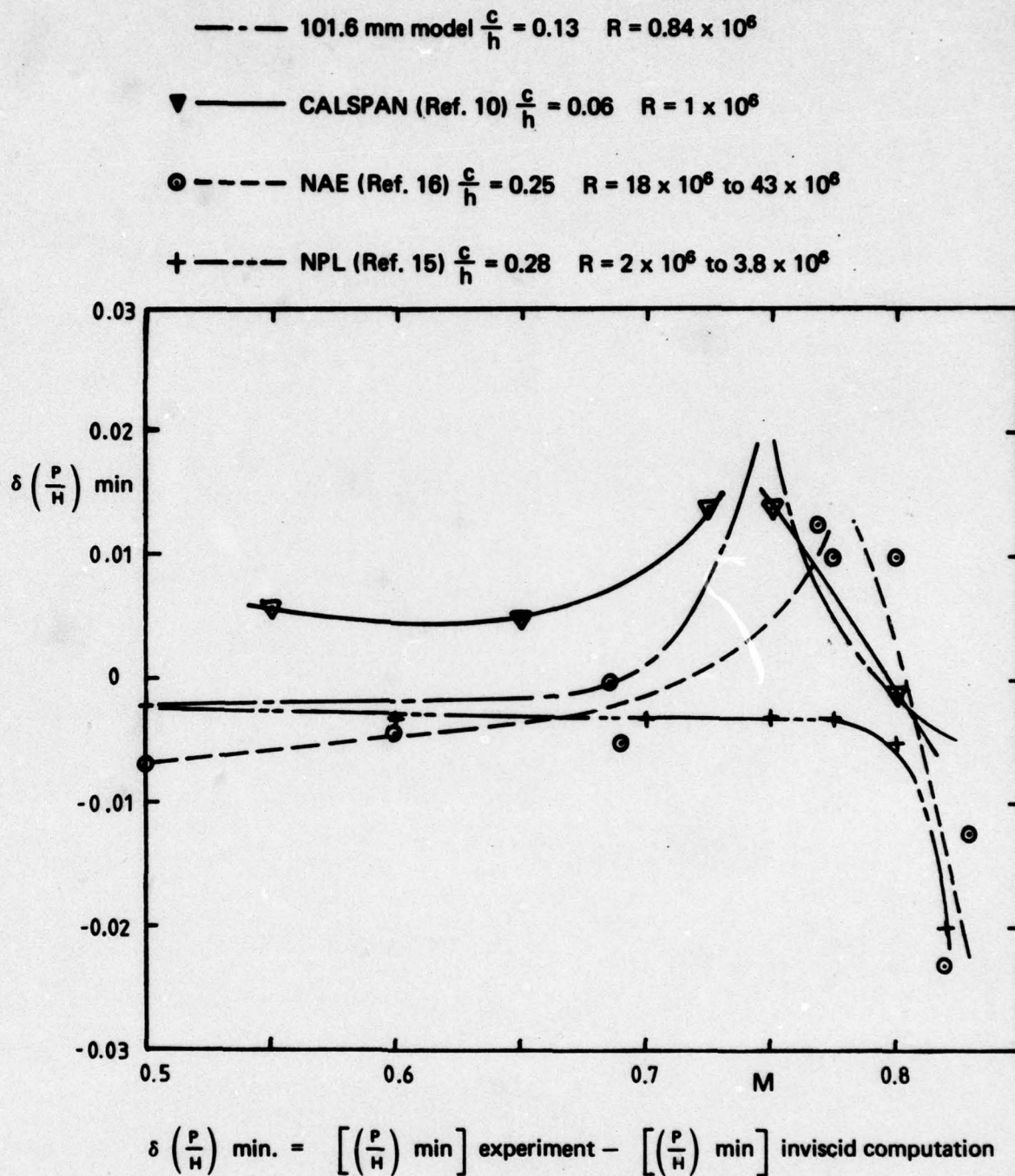


FIG. 18. VARIATION OF MINIMUM P/H WITH M ($\alpha = 0^\circ$)
 (Comparison between 101.6 mm model and other published results)

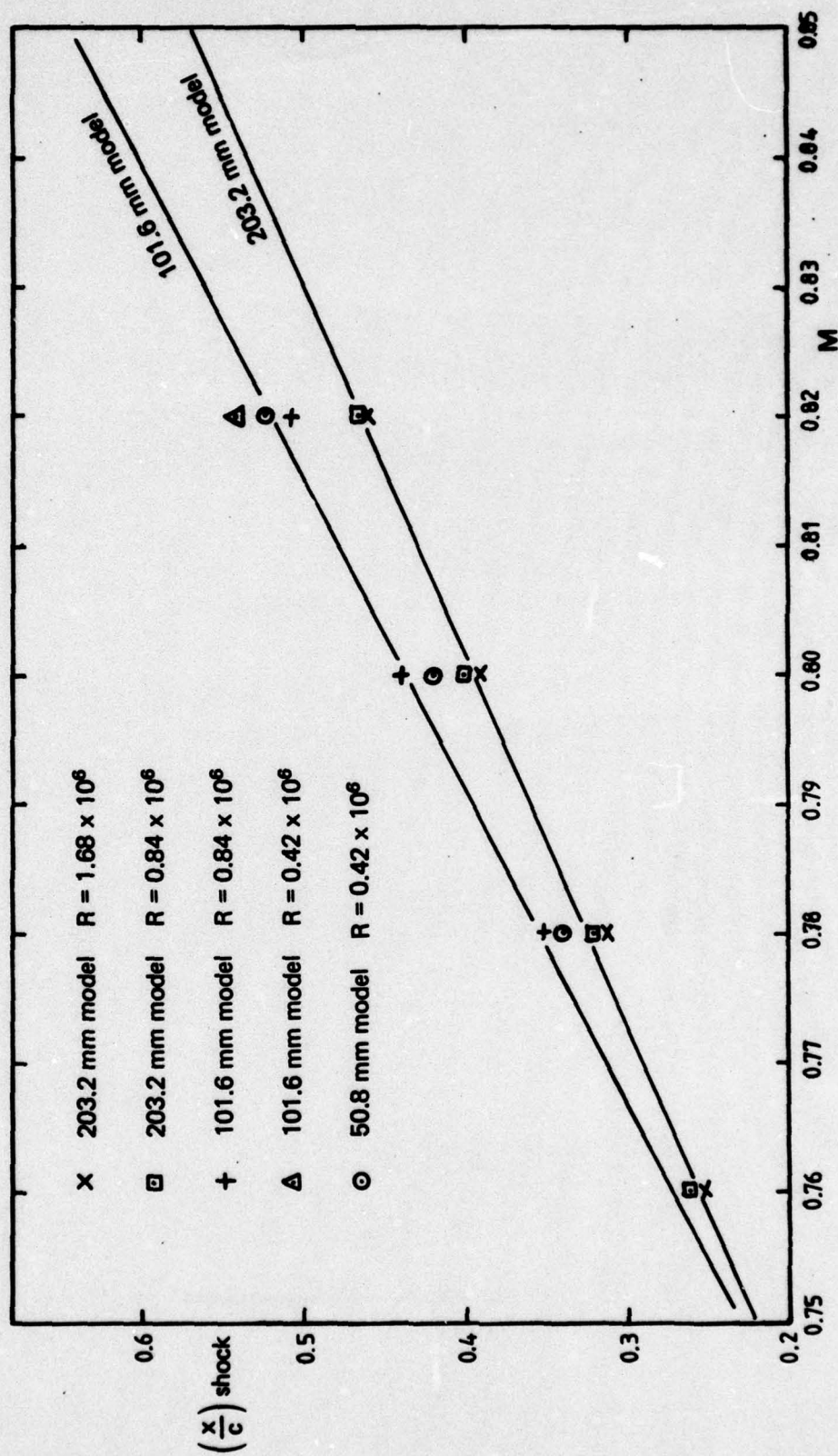


FIG. 19. VARIATION OF SHOCK POSITION WITH M ($\alpha = 0^\circ$)
(Comparison between 203.2 mm, 101.6 mm and 50.8 mm models)

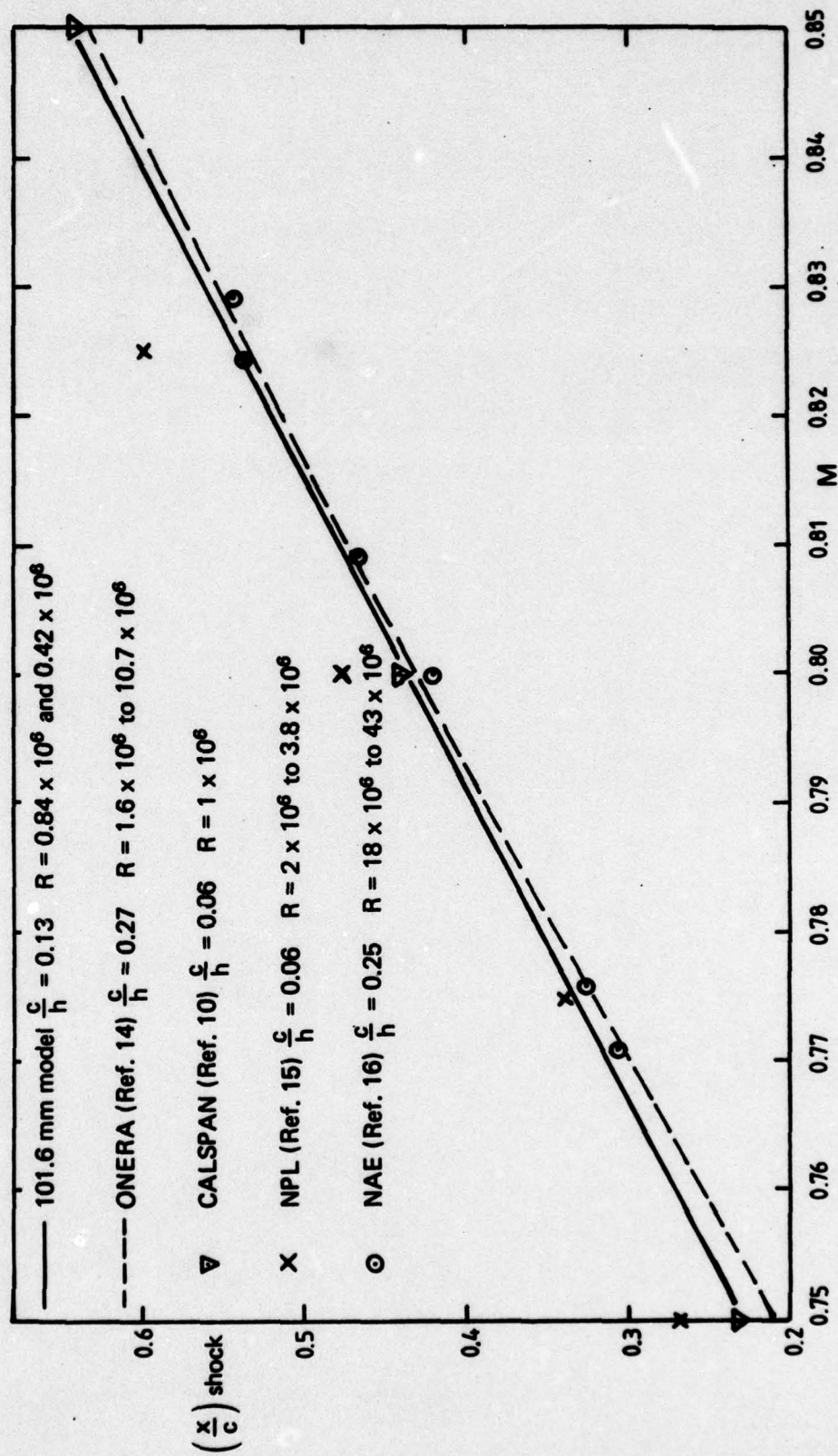


FIG. 20. VARIATION OF SHOCK POSITION WITH $M (\alpha = 0^\circ)$
(Comparison between 101.6 mm model and other published results)

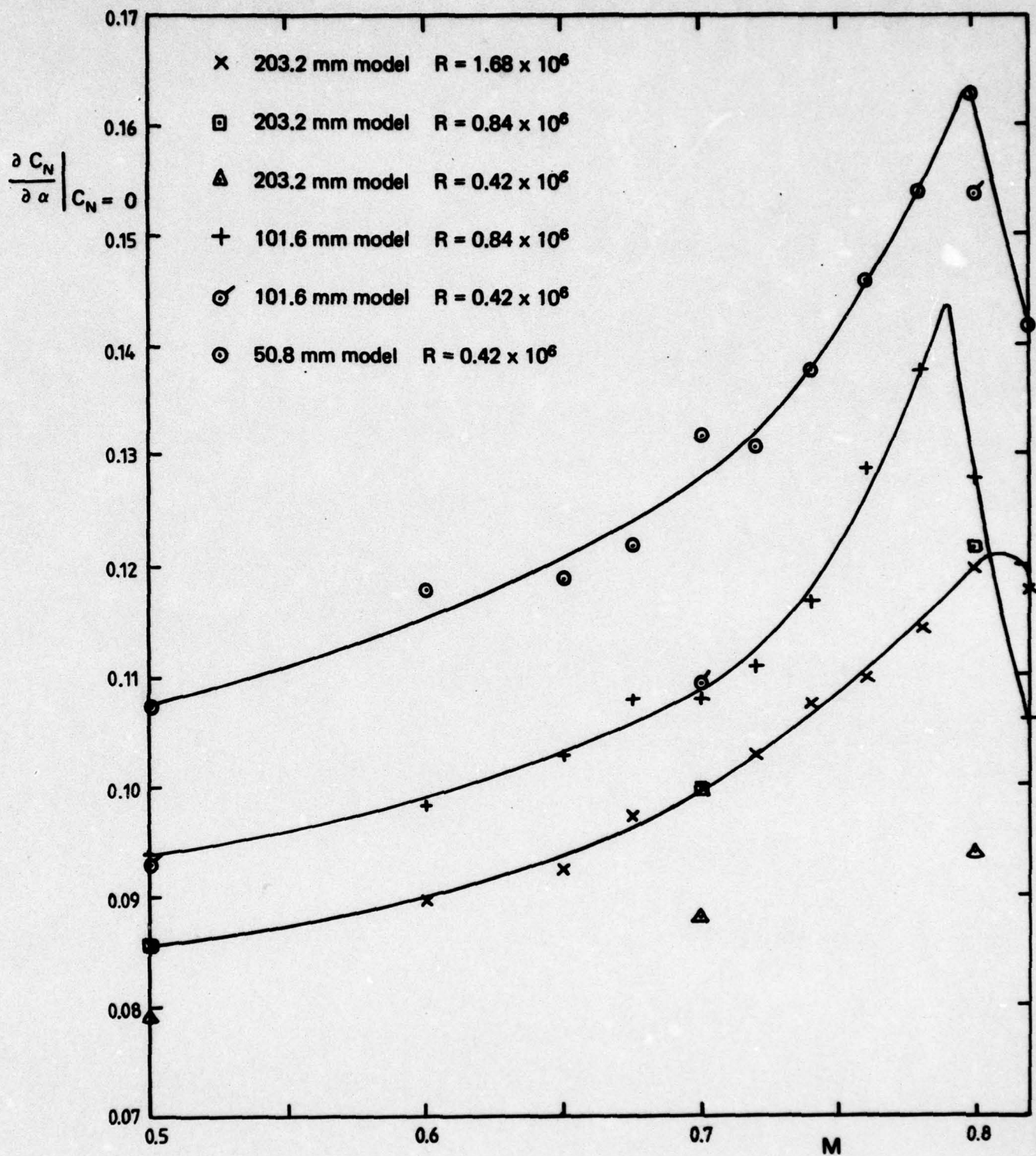


FIG. 21. VARIATION OF $\frac{\partial C_N}{\partial \alpha} \Big|_{C_N = 0}$ WITH M
 (Comparison between 203.2 mm, 101.6 mm and 50.8 mm models)

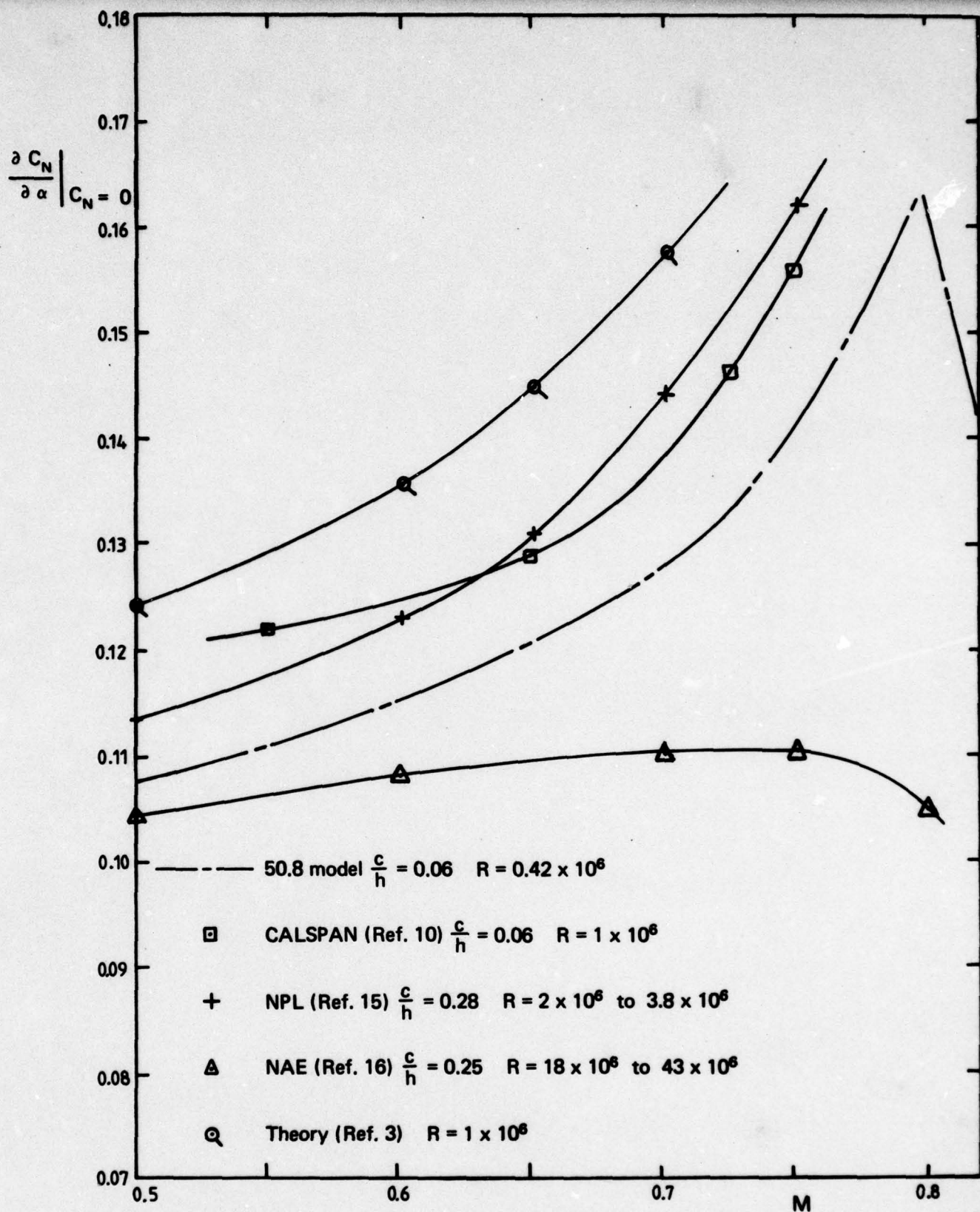


FIG. 22. VARIATION OF $\left. \frac{\partial C_N}{\partial \alpha} \right|_{C_N = 0}$ WITH M
 (Comparison between 50.8 mm model and other published results)

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SPARES

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APPENDIX A
TABULATED RESULTS

NOTATION

ALPHA	(α)	Angle of incidence.
CDW	(C_{DW})	Wake drag coefficient.
CM	(C_m)	Pitching moment coefficient.
CN	(C_N)	Normal force coefficient
CP	(C_P)	Pressure coefficient
MACH NO	(M)	Free stream Mach number
P/H	(P/H)	Local static to free stream total pressure ratio.
REY	(R)	Reynolds number based on chord.
X/C	(x/c)	Chordwise position

Note: Symbols in brackets are the corresponding symbols used in the main part of the report.

CONTENTS

No.	Model chord	M	Nominal α	Corrected α	R
1	203.2	0.50	0	0	1.68×10^6
2	203.2	0.50	1	1	1.68×10^6
3	203.2	0.50	2	2	1.68×10^6
4	203.2	0.50	3	3	1.68×10^6
5	203.2	0.50	4	4	1.68×10^6
6	203.2	0.50	5	5	1.68×10^6
7	203.2	0.50	6	6	1.68×10^6
8	203.2	0.50	7	7	1.68×10^6
9	203.2	0.50	8	8	1.68×10^6
10	203.2	0.60	0	0	1.68×10^6
11	203.2	0.60	1	1	1.68×10^6
12	203.2	0.60	2	2	1.68×10^6
13	203.2	0.60	3	3	1.68×10^6
14	203.2	0.60	4	4	1.68×10^6
15	203.2	0.60	5	5	1.68×10^6
16	203.2	0.60	6	6	1.68×10^6
17	203.2	0.60	7	7	1.68×10^6
18	203.2	0.65	0	0	1.68×10^6
19	203.2	0.65	1	1	1.68×10^6
20	203.2	0.65	2	2	1.68×10^6
21	203.2	0.65	3	3	1.68×10^6
22	203.2	0.65	4	4	1.68×10^6
23	203.2	0.65	5	5	1.68×10^6
24	203.2	0.65	6	6	1.68×10^6
25	203.2	0.675	0	0	1.68×10^6
26	203.2	0.675	1	1	1.68×10^6
27	203.2	0.675	2	2	1.68×10^6
28	203.2	0.675	3	3	1.68×10^6
29	203.2	0.675	4	4	1.68×10^6
30	203.2	0.675	5	5	1.68×10^6
31	203.2	0.70	0	0	1.68×10^6
32	203.2	0.70	1	1	1.68×10^6
33	203.2	0.70	2	2	1.68×10^6
34	203.2	0.70	3	3	1.68×10^6
35	203.2	0.70	4	4	1.68×10^6
36	203.2	0.70	5	5	1.68×10^6
37	203.2	0.72	0	0	1.68×10^6
38	203.2	0.72	1	1	1.68×10^6
39	203.2	0.72	2	2	1.68×10^6
40	203.2	0.72	3	3	1.68×10^6
41	203.2	0.72	4	4	1.68×10^6
42	203.2	0.74	0	0	1.68×10^6
43	203.2	0.74	1	1	1.68×10^6
44	203.2	0.74	2	2	1.68×10^6

No.	Model chord	M	Nominal α	Corrected α	R
45	203.2	0.74	3	3	1.68×10^6
46	203.2	0.74	4	4	1.68×10^6
47	203.2	0.76	0	0	1.68×10^6
48	203.2	0.76	1	1	1.68×10^6
49	203.2	0.76	2	2	1.68×10^6
50	203.2	0.76	3	3	1.68×10^6
51	203.2	0.76	4	4	1.68×10^6
52	203.2	0.78	0	0	1.68×10^6
53	203.2	0.78	1	1	1.68×10^6
54	203.2	0.78	2	2	1.68×10^6
55	203.2	0.78	3	3	1.68×10^6
56	203.2	0.80	0	0	1.68×10^6
57	203.2	0.80	1	1	1.52×10^6
58	203.2	0.80	2	2	1.68×10^6
59	203.2	0.82	0	0	1.68×10^6
60	203.2	0.82	1	1	1.68×10^6
61	203.2	0.82	2	2	1.68×10^6
62	203.2	0.50	0	0	0.84×10^6
63	203.2	0.50	1	1	0.84×10^6
64	203.2	0.50	2	2	0.84×10^6
65	203.2	0.50	3	3	0.84×10^6
66	203.2	0.50	4	4	0.84×10^6
67	203.2	0.50	5	5	0.84×10^6
68	203.2	0.50	6	6	0.84×10^6
69	203.2	0.50	7	7	0.84×10^6
70	203.2	0.50	8	8	0.84×10^6
71	203.2	0.60	0	0	0.84×10^6
72	203.2	0.65	0	0	0.84×10^6
73	203.2	0.675	0	0	0.84×10^6
74	203.2	0.70	0	0	0.84×10^6
75	203.2	0.70	1	1	0.84×10^6
76	203.2	0.70	2	2	0.84×10^6
77	203.2	0.70	3	3	0.84×10^6
78	203.2	0.70	4	4	0.84×10^6
79	203.2	0.70	5	5	0.84×10^6
80	203.2	0.72	0	0	0.84×10^6
81	203.2	0.74	0	0	0.84×10^6
82	203.2	0.76	0	0	0.84×10^6
83	203.2	0.78	0	0	0.84×10^6
84	203.2	0.80	0	0	0.84×10^6
85	203.2	0.80	1	1	0.84×10^6
86	203.2	0.80	2	2	0.84×10^6
87	203.2	0.50	0	0	0.42×10^6
88	203.2	0.50	1	1	0.42×10^6
89	203.2	0.50	2	2	0.42×10^6
90	203.2	0.50	3	3	0.42×10^6
91	203.2	0.50	4	4	0.42×10^6
92	203.2	0.50	5	5	0.42×10^6
93	203.2	0.50	6	6	0.42×10^6
94	203.2	0.50	7	7	0.42×10^6
95	203.2	0.50	8	8	0.42×10^6

No.	Model chord	M	Nominal α	Corrected α	R
96	203.2	0.70	0	0	0.42×10^6
97	203.2	0.70	1	1	0.42×10^6
98	203.2	0.70	2	2	0.42×10^6
99	203.2	0.70	3	3	0.42×10^6
100	203.2	0.70	4	4	0.42×10^6
101	203.2	0.70	5	5	0.42×10^6
102	203.2	0.80	0	0	0.42×10^6
103	203.2	0.80	1	1	0.42×10^6
104	203.2	0.80	2	2	0.42×10^6
105	101.6	0.50	0	0.01	0.84×10^6
106	101.6	0.50	1	1.02	0.84×10^6
107	101.6	0.50	2	2.03	0.84×10^6
108	101.6	0.50	3	3.05	0.84×10^6
109	101.6	0.50	4	4.06	0.84×10^6
110	101.6	0.50	5	5.07	0.84×10^6
111	101.6	0.50	6	6.09	0.84×10^6
112	101.6	0.50	7	7.10	0.84×10^6
113	101.6	0.50	8	8.11	0.84×10^6
114	101.6	0.60	0	0.01	0.84×10^6
115	101.6	0.60	1	1.03	0.84×10^6
116	101.6	0.60	2	2.04	0.84×10^6
117	101.6	0.60	3	3.06	0.84×10^6
118	101.6	0.60	4	4.08	0.84×10^6
119	101.6	0.60	5	5.09	0.84×10^6
120	101.6	0.60	6	6.12	0.84×10^6
121	101.6	0.60	7	7.15	0.84×10^6
122	101.6	0.65	0	0.01	0.84×10^6
123	101.6	0.65	1	1.03	0.84×10^6
124	101.6	0.65	2	2.05	0.84×10^6
125	101.6	0.65	3	3.07	0.84×10^6
126	101.6	0.65	4	4.10	0.84×10^6
127	101.6	0.65	5	5.12	0.84×10^6
128	101.6	0.65	6	6.14	0.84×10^6
129	101.6	0.675	0	0.01	0.84×10^6
130	101.6	0.675	1	1.03	0.84×10^6
131	101.6	0.675	2	2.06	0.84×10^6
132	101.6	0.675	3	3.09	0.84×10^6
133	101.6	0.675	4	4.11	0.84×10^6
134	101.6	0.675	5	5.13	0.84×10^6
135	101.6	0.70	0	0.01	0.84×10^6
136	101.6	0.70	1	1.04	0.84×10^6
137	101.6	0.70	2	2.06	0.84×10^6
138	101.6	0.70	3	3.10	0.84×10^6
139	101.6	0.70	4	4.12	0.84×10^6
140	101.6	0.70	5	5.14	0.84×10^6
141	101.6	0.72	0	0.01	0.84×10^6
142	101.6	0.72	1	1.04	0.84×10^6
143	101.6	0.72	2	2.07	0.84×10^6
144	101.6	0.72	3	3.10	0.84×10^6
145	101.6	0.72	4	4.12	0.84×10^6
146	101.6	0.74	0	0.01	0.84×10^6

No.	Model chord	M	Nominal α	Corrected α	R
147	101-6	0-74	1	1-04	$0-84 \times 10^6$
148	101-6	0-74	2	2-08	$0-84 \times 10^6$
149	101-6	0-74	3	3-10	$0-84 \times 10^6$
150	101-6	0-74	4	4-11	$0-84 \times 10^6$
151	101-6	0-76	0	0-01	$0-84 \times 10^6$
152	101-6	0-76	1	1-05	$0-84 \times 10^6$
153	101-6	0-76	2	2-08	$0-84 \times 10^6$
154	101-6	0-76	3	3-09	$0-84 \times 10^6$
155	101-6	0-76	4	4-09	$0-84 \times 10^6$
156	101-6	0-78	0	0	$0-84 \times 10^6$
157	101-6	0-78	1	1-04	$0-84 \times 10^6$
158	101-6	0-78	2	2-07	$0-84 \times 10^6$
159	101-6	0-78	3	3-07	$0-84 \times 10^6$
160	101-6	0-80	0	0	$0-84 \times 10^6$
161	101-6	0-80	1	1-03	$0-84 \times 10^6$
162	101-6	0-80	2	2-05	$0-84 \times 10^6$
163	101-6	0-82	0	0	$0-84 \times 10^6$
164	101-6	0-82	1	1-02	$0-84 \times 10^6$
165	101-6	0-82	2	2-05	$0-84 \times 10^6$
166	101-6	0-50	0	0	$0-42 \times 10^6$
167	101-6	0-50	1	1-01	$0-42 \times 10^6$
168	101-6	0-50	2	2-02	$0-42 \times 10^6$
169	101-6	0-50	3	3-03	$0-42 \times 10^6$
170	101-6	0-50	4	4-03	$0-42 \times 10^6$
171	101-6	0-50	5	5-04	$0-42 \times 10^6$
172	101-6	0-50	6	6-04	$0-42 \times 10^6$
173	101-6	0-50	7	7-05	$0-42 \times 10^6$
174	101-6	0-50	8	8-06	$0-42 \times 10^6$
175	101-6	0-70	0	0	$0-42 \times 10^6$
176	101-6	0-70	1	1-02	$0-42 \times 10^6$
177	101-6	0-70	2	2-03	$0-42 \times 10^6$
178	101-6	0-70	3	3-05	$0-42 \times 10^6$
179	101-6	0-70	4	4-06	$0-42 \times 10^6$
180	101-6	0-70	5	5-07	$0-42 \times 10^6$
181	101-6	0-80	0	0	$0-42 \times 10^6$
182	101-6	0-80	1	1-01	$0-42 \times 10^6$
183	101-6	0-80	2	2-02	$0-42 \times 10^6$
184	50-8	0-50	0	-0-02	$0-42 \times 10^6$
185	50-8	0-50	0-5	0-61	$0-42 \times 10^6$
186	50-8	0-50	1-0	1-25	$0-42 \times 10^6$
187	50-8	0-50	1-5	1-89	$0-42 \times 10^6$
188	50-8	0-50	2-0	2-52	$0-42 \times 10^6$
189	50-8	0-60	0	-0-02	$0-42 \times 10^6$
190	50-8	0-60	0-5	0-72	$0-42 \times 10^6$
191	50-8	0-60	1-0	1-36	$0-42 \times 10^6$
192	50-8	0-60	1-5	2-05	$0-42 \times 10^6$
193	50-8	0-60	2-0	2-75	$0-42 \times 10^6$
194	50-8	0-65	0	-0-03	$0-42 \times 10^6$
195	50-8	0-65	0-5	0-72	$0-42 \times 10^6$
196	50-8	0-65	1-0	1-45	$0-42 \times 10^6$
197	50-8	0-65	1-5	2-24	$0-42 \times 10^6$

No.	Model chord	M	Nominal α	Corrected α	R
198	50.8	0.675	0	-0.02	0.42×10^6
199	50.8	0.675	0.5	0.75	0.42×10^6
200	50.8	0.675	1.0	1.53	0.42×10^6
201	50.8	0.675	1.5	2.40	0.42×10^6
202	50.8	0.70	0	-0.05	0.42×10^6
203	50.8	0.70	0.5	0.80	0.42×10^6
204	50.8	0.70	1.0	1.62	0.42×10^6
205	50.8	0.72	0	-0.03	0.42×10^6
206	50.8	0.72	0.5	0.85	0.42×10^6
207	50.8	0.72	1.0	1.75	0.42×10^6
208	50.8	0.74	0	-0.04	0.42×10^6
209	50.8	0.74	0.5	0.93	0.42×10^6
210	50.8	0.74	1.0	1.93	0.42×10^6
211	50.8	0.76	0	-0.07	0.42×10^6
212	50.8	0.76	0.5	1.01	0.42×10^6
213	50.8	0.76	1.0	1.95	0.42×10^6
214	50.8	0.78	0	-0.10	0.42×10^6
215	50.8	0.78	0.5	1.11	0.42×10^6
216	50.8	0.78	1.0	1.80	0.42×10^6
217	50.8	0.80	0	-0.05	0.42×10^6
218	50.8	0.80	0.5	0.93	0.42×10^6
219	50.8	0.80	1.0	1.54	0.42×10^6
220	50.8	0.82	0	-0.01	0.42×10^6
221	50.8	0.82	0.5	0.45	0.42×10^6
222	50.8	0.82	1.0	1.21	0.42×10^6

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.531 ALPHA 0.00 REYNOLDS 1.73*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.0034 CM = 0.0000 CDW = 0.0105

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.100	0.857	0.0119	0.095	0.856
0.0371	-0.294	0.799	0.0371	-0.287	0.800
0.0623	-0.383	0.766	0.0623	-0.388	0.785
0.0872	-0.424	0.780	0.0872	-0.476	0.772
0.1122	-0.431	0.778	0.1122	-0.433	0.778
0.1372	-0.425	0.779	0.1372	-0.428	0.779
0.1623	-0.432	0.778	0.1623	-0.437	0.777
0.1872	-0.426	0.779	0.1872	-0.435	0.778
0.2122	-0.424	0.780	0.2122	-0.427	0.779
0.2372	-0.405	0.783	0.2372	-0.410	0.781
0.2623	-0.387	0.785	0.2623	-0.389	0.784
0.2872	-0.361	0.789	0.2872	-0.367	0.788
0.3122	-0.348	0.791	0.3122	-0.353	0.790
0.3375	-0.326	0.794	0.3375	-0.334	0.792
0.3618	-0.309	0.797	0.3618	-0.315	0.795
0.3873	-0.291	0.800	0.3873	-0.296	0.798
0.4124	-0.282	0.801	0.4124	-0.285	0.800
0.4371	-0.257	0.805	0.4371	-0.261	0.803
0.4621	-0.251	0.806	0.4621	-0.251	0.805
0.4871	-0.235	0.808	0.4871	-0.240	0.806
0.5120	-0.222	0.810	0.5120	-0.221	0.809
0.5371	-0.197	0.814	0.5371	-0.199	0.812
0.5621	-0.176	0.817	0.5621	-0.180	0.815
0.5870	-0.164	0.819	0.5870	-0.166	0.817
0.6122	-0.136	0.823	0.6122	-0.137	0.821
0.6371	-0.126	0.824	0.6371	-0.128	0.823
0.6661	-0.112	0.826	0.6661	-0.114	0.825
0.6869	-0.100	0.828	0.6869	-0.102	0.827
0.7121	-0.085	0.830	0.7121	-0.082	0.830
0.7369	-0.066	0.833	0.7369	-0.064	0.833
0.7620	-0.053	0.835	0.7620	-0.052	0.834
0.7870	-0.032	0.838	0.7870	-0.033	0.837
0.8117	-0.013	0.841	0.8117	-0.013	0.840
0.8370	0.006	0.844	0.8370	0.005	0.843
0.8619	0.028	0.847	0.8619	0.029	0.847
0.8869	0.052	0.850	0.8869	0.050	0.850
0.9120	0.078	0.854	0.9120	0.077	0.854
0.9416	0.114	0.860	0.9416	0.113	0.859

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 1.00 REYNOLDS 1.72*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.3841 CM = 0.0016 CDW = 0.0131

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.109	0.827	0.0119	0.296	0.867
0.0371	-0.464	0.774	0.0371	-0.130	0.824
0.0623	-0.523	0.765	0.0623	-0.248	0.807
0.0872	-0.543	0.762	0.0872	-0.309	0.793
0.1122	-0.534	0.764	0.1122	-0.329	0.795
0.1372	-0.517	0.766	0.1372	-0.336	0.794
0.1623	-0.514	0.767	0.1623	-0.349	0.792
0.1872	-0.502	0.768	0.1872	-0.354	0.791
0.2122	-0.491	0.770	0.2122	-0.353	0.791
0.2372	-0.465	0.774	0.2372	-0.343	0.792
0.2620	-0.441	0.777	0.2620	-0.329	0.794
0.2872	-0.412	0.781	0.2872	-0.310	0.797
0.3122	-0.397	0.784	0.3122	-0.301	0.799
0.3375	-0.369	0.788	0.3375	-0.283	0.801
0.3618	-0.350	0.791	0.3618	-0.267	0.804
0.3873	-0.329	0.794	0.3873	-0.254	0.805
0.4124	-0.317	0.796	0.4124	-0.246	0.807
0.4371	-0.290	0.800	0.4371	-0.225	0.810
0.4621	-0.260	0.801	0.4621	-0.219	0.811
0.4871	-0.264	0.803	0.4871	-0.207	0.812
0.5120	-0.245	0.806	0.5120	-0.194	0.814
0.5371	-0.220	0.810	0.5371	-0.172	0.817
0.5621	-0.199	0.813	0.5621	-0.154	0.820
0.5870	-0.184	0.815	0.5870	-0.144	0.822
0.6122	-0.153	0.820	0.6122	-0.116	0.826
0.6371	-0.143	0.821	0.6371	-0.110	0.827
0.6601	-0.127	0.824	0.6601	-0.098	0.828
0.6869	-0.113	0.826	0.6869	-0.087	0.830
0.7121	-0.096	0.828	0.7121	-0.070	0.832
0.7369	-0.075	0.831	0.7369	-0.055	0.835
0.7620	-0.062	0.833	0.7620	-0.041	0.836
0.7870	-0.038	0.837	0.7870	-0.023	0.839
0.8117	-0.019	0.840	0.8117	-0.005	0.842
0.8370	0.002	0.843	0.8370	0.012	0.844
0.8619	0.026	0.846	0.8619	0.032	0.847
0.8869	0.050	0.850	0.8869	0.055	0.851
0.9120	0.079	0.854	0.9120	0.081	0.855
0.9416	0.116	0.860	0.9416	0.115	0.862

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.502 ALPHA 2.00 REYNOLDS 1.72*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1744 CM = 0.0030 CDW = 0.0091

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.354	0.709	0.0119	0.464	0.911
0.0371	-0.657	0.744	0.0371	0.018	0.845
0.0623	-0.677	0.741	0.0623	-0.119	0.825
0.0872	-0.671	0.742	0.0872	-0.197	0.813
0.1122	-0.645	0.746	0.1122	-0.230	0.808
0.1372	-0.602	0.752	0.1372	-0.249	0.805
0.1623	-0.601	0.752	0.1623	-0.268	0.803
0.1872	-0.581	0.755	0.1872	-0.283	0.801
0.2122	-0.562	0.758	0.2122	-0.287	0.800
0.2372	-0.530	0.762	0.2372	-0.281	0.801
0.2620	-0.501	0.767	0.2620	-0.270	0.802
0.2872	-0.464	0.772	0.2872	-0.255	0.804
0.3122	-0.444	0.775	0.3122	-0.250	0.805
0.3375	-0.413	0.780	0.3375	-0.236	0.807
0.3618	-0.368	0.784	0.3618	-0.213	0.811
0.3873	-0.366	0.787	0.3873	-0.213	0.811
0.4124	-0.351	0.789	0.4124	-0.208	0.811
0.4371	-0.320	0.794	0.4371	-0.190	0.814
0.4621	-0.308	0.796	0.4621	-0.186	0.814
0.4871	-0.289	0.799	0.4871	-0.178	0.816
0.5120	-0.272	0.801	0.5120	-0.165	0.817
0.5371	-0.242	0.806	0.5371	-0.145	0.820
0.5621	-0.219	0.809	0.5621	-0.128	0.823
0.5870	-0.203	0.811	0.5870	-0.121	0.824
0.6122	-0.173	0.816	0.6122	-0.094	0.828
0.6371	-0.158	0.818	0.6371	-0.089	0.829
0.6661	-0.140	0.821	0.6661	-0.078	0.830
0.6869	-0.125	0.823	0.6869	-0.070	0.832
0.7121	-0.106	0.826	0.7121	-0.056	0.834
0.7369	-0.084	0.829	0.7369	-0.041	0.836
0.7620	-0.068	0.832	0.7620	-0.030	0.838
0.7870	-0.043	0.835	0.7870	-0.012	0.840
0.8117	-0.023	0.836	0.8117	0.004	0.845
0.8370	-0.002	0.842	0.8370	0.020	0.845
0.8619	0.023	0.845	0.8619	0.037	0.847
0.8869	0.049	0.849	0.8869	0.059	0.851
0.9120	0.077	0.853	0.9120	0.083	0.854
0.9416	0.117	0.859	0.9416	0.116	0.859

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 3.00 REY 1.72*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2583 CM = 0.0044 CDW = 0.0104

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.631	0.750	0.0119	0.621	0.934
0.0371	-0.856	0.717	0.0371	0.162	0.866
0.0623	-0.832	0.721	0.0623	0.005	0.843
0.0872	-0.796	0.726	0.0872	-0.088	0.829
0.1122	-0.753	0.757	0.1122	-0.132	0.823
0.1372	-0.707	0.739	0.1372	-0.167	0.819
0.1623	-0.686	0.742	0.1623	-0.186	0.815
0.1872	-0.656	0.747	0.1872	-0.205	0.812
0.2122	-0.632	0.750	0.2122	-0.216	0.810
0.2372	-0.592	0.756	0.2372	-0.215	0.811
0.2623	-0.559	0.761	0.2623	-0.209	0.812
0.2872	-0.516	0.767	0.2872	-0.200	0.813
0.3122	0.494	0.771	0.3122	-0.197	0.813
0.3375	-0.458	0.776	0.3375	-0.188	0.815
0.3618	-0.432	0.780	0.3618	-0.177	0.816
0.3873	-0.404	0.784	0.3873	-0.171	0.817
0.4124	-0.386	0.786	0.4124	-0.168	0.818
0.4371	-0.352	0.791	0.4371	-0.152	0.820
0.4621	-0.349	0.793	0.4621	-0.151	0.822
0.4871	-0.316	0.797	0.4871	-0.145	0.821
0.5120	-0.295	0.800	0.5120	-0.135	0.822
0.5371	-0.264	0.804	0.5371	-0.117	0.825
0.5621	-0.240	0.808	0.5621	-0.100	0.827
0.5870	-0.221	0.811	0.5870	-0.096	0.828
0.6122	-0.189	0.816	0.6122	-0.070	0.832
0.6371	-0.172	0.818	0.6371	-0.069	0.832
0.6651	-0.153	0.821	0.6651	-0.058	0.834
0.6869	-0.136	0.823	0.6869	-0.051	0.835
0.7121	-0.117	0.826	0.7121	-0.040	0.836
0.7369	-0.093	0.830	0.7369	-0.029	0.838
0.7620	-0.076	0.832	0.7620	-0.016	0.840
0.7870	-0.048	0.836	0.7870	0.000	0.842
0.8117	-0.028	0.839	0.8117	0.015	0.844
0.8370	-0.003	0.843	0.8370	0.029	0.846
0.8619	0.022	0.847	0.8619	0.043	0.848
0.8869	0.050	0.851	0.8869	0.064	0.852
0.9120	0.078	0.855	0.9120	0.088	0.855
0.9416	0.119	0.861	0.9416	0.117	0.859

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 4.00 REYNOLDS 1.73*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.3447 CM = 0.0062 CDW = 0.0107

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.930	0.705	0.0119	0.743	0.952
0.0371	-1.073	0.684	0.0371	0.294	0.886
0.0623	-0.998	0.695	0.0623	0.118	0.860
0.0872	-0.930	0.705	0.0872	0.014	0.844
0.1122	-0.869	0.714	0.1122	-0.037	0.837
0.1372	-0.805	0.724	0.1372	-0.073	0.831
0.1623	-0.771	0.729	0.1623	-0.106	0.826
0.1872	-0.732	0.734	0.1872	-0.129	0.823
0.2122	-0.698	0.739	0.2122	-0.146	0.820
0.2372	-0.653	0.746	0.2372	-0.150	0.820
0.2622	-0.614	0.752	0.2622	-0.150	0.820
0.2872	-0.568	0.759	0.2872	-0.143	0.821
0.3122	-0.540	0.763	0.3122	-0.144	0.821
0.3375	-0.500	0.769	0.3375	-0.137	0.822
0.3618	-0.468	0.773	0.3618	-0.128	0.823
0.3873	-0.438	0.778	0.3873	-0.126	0.823
0.4124	-0.416	0.781	0.4124	-0.127	0.823
0.4371	-0.381	0.786	0.4371	-0.112	0.825
0.4621	-0.364	0.789	0.4621	-0.114	0.825
0.4871	-0.339	0.793	0.4871	-0.111	0.826
0.5120	-0.316	0.796	0.5120	-0.103	0.827
0.5371	-0.283	0.801	0.5371	-0.087	0.829
0.5621	-0.257	0.805	0.5621	-0.075	0.831
0.5870	-0.236	0.808	0.5870	-0.070	0.832
0.6122	-0.202	0.813	0.6122	-0.047	0.835
0.6371	-0.185	0.815	0.6371	-0.045	0.835
0.6661	-0.163	0.818	0.6661	-0.039	0.836
0.6869	-0.146	0.821	0.6869	-0.034	0.837
0.7121	-0.125	0.824	0.7121	-0.023	0.839
0.7369	-0.099	0.828	0.7369	-0.013	0.840
0.7620	-0.079	0.831	0.7620	-0.002	0.842
0.7870	-0.051	0.835	0.7870	0.013	0.844
0.8117	-0.030	0.838	0.8117	0.025	0.846
0.8370	-0.005	0.842	0.8370	0.038	0.848
0.8619	0.023	0.846	0.8619	0.050	0.849
0.8869	0.051	0.850	0.8869	0.071	0.852
0.9120	0.081	0.855	0.9120	0.091	0.855
0.9416	0.121	0.861	0.9416	0.120	0.860

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 5.00 REYNOLDS 1.72*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4316 CM = 0.0078 COW = 0.0112

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-1.272	0.656	0.0119	0.851	0.969
0.0371	-1.313	0.649	0.0371	0.413	0.904
0.0623	-1.173	0.670	0.0623	0.227	0.876
0.0872	-1.066	0.686	0.0872	0.114	0.860
0.1122	-0.982	0.698	0.1122	0.051	0.850
0.1372	-0.901	0.710	0.1372	0.007	0.844
0.1623	-0.858	0.717	0.1623	-0.027	0.839
0.1872	-0.807	0.724	0.1872	-0.056	0.835
0.2122	-0.767	0.730	0.2122	-0.077	0.832
0.2372	-0.712	0.738	0.2372	-0.087	0.830
0.2620	-0.667	0.745	0.2620	-0.089	0.830
0.2872	-0.615	0.752	0.2872	-0.087	0.832
0.3122	-0.564	0.757	0.3122	-0.092	0.830
0.3375	-0.540	0.763	0.3375	-0.088	0.830
0.3618	-0.507	0.768	0.3618	-0.084	0.831
0.3873	-0.473	0.773	0.3873	-0.083	0.831
0.4124	-0.449	0.777	0.4124	-0.085	0.831
0.4371	-0.408	0.783	0.4371	-0.074	0.832
0.4621	-0.389	0.786	0.4621	-0.078	0.832
0.4871	-0.362	0.790	0.4871	-0.076	0.832
0.5123	-0.336	0.794	0.5123	-0.070	0.833
0.5371	-0.302	0.799	0.5371	-0.057	0.835
0.5621	-0.275	0.802	0.5621	-0.045	0.837
0.5873	-0.252	0.806	0.5873	-0.044	0.837
0.6122	-0.217	0.811	0.6122	-0.020	0.840
0.6371	-0.193	0.814	0.6371	-0.025	0.839
0.6661	-0.174	0.817	0.6661	-0.016	0.841
0.6869	-0.155	0.820	0.6869	-0.015	0.841
0.7121	-0.132	0.824	0.7121	-0.002	0.843
0.7369	-0.105	0.828	0.7369	0.005	0.844
0.7620	-0.084	0.831	0.7620	0.014	0.845
0.7870	-0.055	0.835	0.7870	0.026	0.847
0.8117	-0.032	0.838	0.8117	0.029	0.847
0.8370	-0.006	0.842	0.8370	0.048	0.850
0.8619	0.022	0.846	0.8619	0.060	0.852
0.8869	0.052	0.851	0.8869	0.078	0.855
0.9120	0.082	0.855	0.9120	0.097	0.857
0.9416	0.122	0.861	0.9416	0.121	0.861

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 6.00 REY 1.72*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.5185 CM = 0.0100 CDW = 0.0110

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-1.652	0.600	0.0119	0.933	0.951
0.0371	-1.573	0.611	0.0371	0.919	0.919
0.0623	-1.360	0.643	0.0623	0.926	0.891
0.0872	-1.211	0.665	0.0872	0.227	0.873
0.1122	-1.105	0.680	0.1122	0.136	0.863
0.1372	-1.006	0.695	0.1372	0.097	0.856
0.1623	-0.953	0.703	0.1623	0.043	0.849
0.1872	-0.890	0.712	0.1872	0.013	0.844
0.2122	-0.841	0.719	0.2122	-0.012	0.841
0.2372	-0.779	0.728	0.2372	-0.029	0.838
0.2620	-0.729	0.735	0.2620	-0.034	0.838
0.2872	-0.668	0.744	0.2872	-0.037	0.837
0.3122	-0.631	0.750	0.3122	-0.043	0.836
0.3375	-0.581	0.757	0.3375	-0.041	0.837
0.3618	-0.544	0.763	0.3618	-0.041	0.837
0.3873	-0.503	0.769	0.3873	-0.041	0.837
0.4124	-0.478	0.772	0.4124	-0.047	0.836
0.4371	-0.435	0.779	0.4371	-0.038	0.837
0.4621	-0.414	0.782	0.4621	-0.013	0.836
0.4871	-0.383	0.786	0.4871	-0.042	0.837
0.5120	-0.356	0.790	0.5120	-0.039	0.837
0.5371	-0.318	0.796	0.5371	-0.029	0.838
0.5621	-0.292	0.800	0.5621	-0.017	0.840
0.5870	-0.265	0.804	0.5870	-0.020	0.840
0.6122	-0.230	0.809	0.6122	0.004	0.843
0.6371	-0.206	0.812	0.6371	-0.002	0.842
0.6661	-0.164	0.816	0.6661	0.003	0.843
0.6869	-0.161	0.819	0.6869	0.006	0.844
0.7121	-0.138	0.823	0.7121	0.016	0.845
0.7369	-0.109	0.827	0.7369	0.021	0.846
0.7620	-0.089	0.830	0.7620	0.029	0.847
0.7870	-0.056	0.835	0.7870	0.041	0.849
0.8117	-0.032	0.838	0.8117	0.046	0.850
0.8370	-0.004	0.842	0.8370	0.060	0.852
0.8619	0.024	0.847	0.8619	0.068	0.853
0.8869	0.055	0.851	0.8869	0.085	0.855
0.9120	0.083	0.855	0.9120	0.101	0.858
0.9416	0.125	0.861	0.9416	0.125	0.861

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.520 ALPHA 7.00 REYNOLDS 1.72×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.6052 CM = 0.0124 CDW = 0.0118

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-2.088	0.534	0.0119	0.992	0.939
0.0371	-1.858	0.568	0.0371	0.617	0.933
0.0623	-1.554	0.613	0.0623	0.424	0.905
0.0872	-1.351	0.643	0.0872	0.299	0.886
0.1122	-1.219	0.662	0.1122	0.222	0.875
0.1372	-1.105	0.679	0.1372	0.167	0.867
0.1623	-1.033	0.690	0.1623	0.118	0.859
0.1872	-0.965	0.700	0.1872	0.082	0.854
0.2122	-0.905	0.709	0.2122	0.051	0.852
0.2372	-0.837	0.719	0.2372	0.035	0.847
0.2623	-0.777	0.727	0.2623	0.026	0.846
0.2872	-0.714	0.737	0.2872	0.020	0.845
0.3122	-0.671	0.743	0.3122	0.010	0.843
0.3375	-0.620	0.751	0.3375	0.007	0.843
0.3618	-0.577	0.757	0.3618	0.009	0.843
0.3873	-0.535	0.763	0.3873	0.002	0.842
0.4124	-0.505	0.768	0.4124	-0.005	0.841
0.4371	-0.461	0.774	0.4371	-0.002	0.842
0.4621	-0.437	0.778	0.4621	-0.004	0.841
0.4871	-0.405	0.783	0.4871	-0.008	0.841
0.5120	-0.375	0.787	0.5120	-0.005	0.841
0.5371	-0.335	0.793	0.5371	0.004	0.843
0.5621	-0.304	0.797	0.5621	0.013	0.844
0.5870	-0.277	0.801	0.5870	0.010	0.843
0.6122	-0.240	0.807	0.6122	0.030	0.846
0.6371	-0.215	0.811	0.6371	0.025	0.846
0.6601	-0.189	0.815	0.6601	0.027	0.846
0.6869	-0.165	0.818	0.6869	0.026	0.846
0.7121	-0.141	0.822	0.7121	0.033	0.847
0.7369	-0.112	0.826	0.7369	0.042	0.848
0.7620	-0.087	0.830	0.7620	0.042	0.848
0.7870	-0.056	0.834	0.7870	0.057	0.850
0.8117	-0.031	0.838	0.8117	0.061	0.851
0.8370	-0.002	0.842	0.8370	0.073	0.853
0.8619	0.027	0.847	0.8619	0.079	0.854
0.8869	0.058	0.851	0.8869	0.094	0.856
0.9120	0.088	0.855	0.9120	0.107	0.858
0.9416	0.127	0.861	0.9416	0.128	0.861

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 8.00 REY 1.72*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.5868 CM = 0.0149 CDW = 0.0112

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-2.537	0.470	0.0119	1.032	0.995
0.0371	-2.167	0.524	0.0371	0.701	0.947
0.0623	-1.767	0.583	0.0623	0.510	0.918
0.0872	-1.469	0.627	0.0872	0.381	0.899
0.1122	-1.324	0.648	0.1122	0.300	0.887
0.1372	-1.186	0.669	0.1372	0.242	0.879
0.1623	-1.111	0.680	0.1623	0.193	0.871
0.1872	-1.024	0.693	0.1872	0.150	0.865
0.2122	-0.962	0.702	0.2122	0.116	0.860
0.2372	-0.879	0.714	0.2372	0.095	0.857
0.2620	-0.823	0.722	0.2620	0.080	0.855
0.2872	-0.749	0.733	0.2872	0.074	0.854
0.3122	-0.712	0.739	0.3122	0.058	0.852
0.3375	-0.651	0.748	0.3375	0.055	0.851
0.3618	-0.613	0.753	0.3618	0.049	0.850
0.3873	-0.560	0.761	0.3873	0.044	0.850
0.4124	-0.535	0.765	0.4124	0.033	0.848
0.4371	-0.481	0.773	0.4371	0.041	0.849
0.4621	-0.463	0.776	0.4621	0.031	0.848
0.4871	-0.417	0.782	0.4871	0.027	0.847
0.5120	-0.392	0.786	0.5120	0.027	0.847
0.5371	-0.345	0.793	0.5371	0.035	0.848
0.5621	-0.319	0.796	0.5621	0.039	0.849
0.5870	-0.283	0.802	0.5870	0.039	0.849
0.6122	-0.254	0.806	0.6122	0.053	0.851
0.6371	-0.219	0.811	0.6371	0.051	0.851
0.6661	-0.197	0.814	0.6661	0.047	0.850
0.6869	-0.167	0.819	0.6869	0.051	0.851
0.7121	-0.147	0.822	0.7121	0.052	0.851
0.7369	-0.109	0.827	0.7369	0.061	0.852
0.7620	-0.091	0.830	0.7620	0.059	0.852
0.7870	-0.051	0.836	0.7870	0.073	0.854
0.8117	-0.032	0.839	0.8117	0.074	0.854
0.8370	0.004	0.844	0.8370	0.085	0.856
0.8619	0.027	0.847	0.8619	0.090	0.856
0.8869	0.065	0.853	0.8869	0.102	0.858
0.9120	0.086	0.856	0.9120	0.113	0.858
0.9416	0.132	0.863	0.9416	0.131	0.863

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.599 ALPHA 0.00 REYNOLDS 1.70*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.0014 CM = 0.0002 CDW = 0.0100

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.137	0.811	0.0119	0.132	0.809
0.0371	-0.295	0.725	0.0371	-0.285	0.726
0.0623	-0.414	0.702	0.0623	-0.416	0.700
0.0872	-0.461	0.693	0.0872	-0.503	0.683
0.1122	-0.471	0.691	0.1122	-0.466	0.690
0.1372	-0.468	0.691	0.1372	-0.462	0.691
0.1623	-0.477	0.690	0.1623	-0.476	0.688
0.1872	-0.476	0.690	0.1872	-0.478	0.688
0.2122	-0.470	0.691	0.2122	-0.468	0.690
0.2372	-0.449	0.695	0.2372	-0.451	0.694
0.2620	-0.427	0.699	0.2620	-0.429	0.693
0.2872	-0.401	0.704	0.2872	-0.402	0.703
0.3122	-0.387	0.707	0.3122	-0.386	0.706
0.3375	-0.363	0.712	0.3375	-0.366	0.710
0.3618	-0.342	0.716	0.3618	-0.346	0.714
0.3873	-0.322	0.720	0.3873	-0.325	0.718
0.4124	-0.310	0.722	0.4124	-0.311	0.721
0.4371	-0.285	0.727	0.4371	-0.285	0.726
0.4621	-0.275	0.729	0.4621	-0.274	0.729
0.4871	-0.258	0.732	0.4871	-0.260	0.732
0.5120	-0.241	0.736	0.5120	-0.241	0.736
0.5371	-0.216	0.741	0.5371	-0.216	0.741
0.5621	-0.193	0.745	0.5621	-0.195	0.745
0.5870	-0.181	0.747	0.5870	-0.180	0.748
0.6122	-0.149	0.754	0.6122	-0.151	0.754
0.6371	-0.140	0.756	0.6371	-0.140	0.756
0.6661	-0.121	0.759	0.6661	-0.122	0.759
0.6869	-0.110	0.762	0.6869	-0.109	0.762
0.7121	-0.091	0.765	0.7121	-0.091	0.766
0.7369	-0.071	0.769	0.7369	-0.069	0.770
0.7620	-0.055	0.772	0.7620	-0.056	0.773
0.7870	-0.025	0.778	0.7870	-0.032	0.778
0.8117	-0.012	0.781	0.8117	-0.012	0.781
0.8370	0.007	0.785	0.8370	0.007	0.785
0.8619	0.029	0.789	0.8619	0.030	0.790
0.8869	0.056	0.794	0.8869	0.056	0.795
0.9120	0.086	0.800	0.9120	0.085	0.801
0.9416	0.126	0.808	0.9416	0.122	0.808

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.599 ALPHA 1.00 REY 1.70*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0880 CM = 0.0025 CDW = 0.0136

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.356	0.712	0.0119	0.523	0.847
0.0371	-0.472	0.691	0.0371	-0.127	0.756
0.0623	-0.563	0.674	0.0623	-0.266	0.731
0.0872	-0.588	0.669	0.0872	-0.336	0.717
0.1122	-0.582	0.670	0.1122	-0.361	0.712
0.1372	-0.566	0.673	0.1372	-0.369	0.710
0.1623	-0.565	0.673	0.1623	-0.387	0.707
0.1872	-0.557	0.675	0.1872	-0.394	0.705
0.2122	-0.543	0.677	0.2122	-0.395	0.705
0.2372	-0.515	0.683	0.2372	-0.382	0.708
0.2622	-0.487	0.688	0.2622	-0.366	0.711
0.2872	-0.455	0.695	0.2872	-0.344	0.715
0.3122	-0.435	0.699	0.3122	-0.335	0.717
0.3375	-0.408	0.704	0.3375	-0.315	0.721
0.3618	-0.383	0.709	0.3618	-0.297	0.725
0.3873	-0.363	0.713	0.3873	-0.284	0.727
0.4124	-0.345	0.716	0.4124	-0.275	0.729
0.4371	-0.316	0.722	0.4371	-0.251	0.734
0.4621	-0.304	0.725	0.4621	-0.243	0.735
0.4871	-0.285	0.728	0.4871	-0.230	0.737
0.5120	-0.265	0.732	0.5120	-0.215	0.740
0.5371	-0.237	0.738	0.5371	-0.192	0.745
0.5621	-0.212	0.743	0.5621	-0.171	0.749
0.5870	-0.199	0.745	0.5870	-0.159	0.752
0.6122	-0.166	0.752	0.6122	-0.129	0.757
0.6371	-0.156	0.754	0.6371	-0.123	0.759
0.6661	-0.134	0.758	0.6661	-0.106	0.762
0.6869	-0.122	0.760	0.6869	-0.095	0.764
0.7121	-0.101	0.765	0.7121	-0.078	0.767
0.7369	-0.081	0.768	0.7369	-0.060	0.771
0.7620	-0.062	0.772	0.7620	-0.045	0.773
0.7870	-0.038	0.777	0.7870	-0.025	0.778
0.8117	-0.017	0.781	0.8117	-0.005	0.781
0.8370	0.005	0.785	0.8370	0.013	0.785
0.8619	0.027	0.790	0.8619	0.033	0.789
0.8869	0.056	0.796	0.8869	0.059	0.794
0.9120	0.087	0.802	0.9120	0.085	0.799
0.9416	0.124	0.809	0.9416	0.121	0.827

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.601 ALPHA 2.00 REY 1.71×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1756 CM = 0.0042 CDW = 0.0109

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0117	-0.301	0.723	0.0117	0.488	0.880
0.0371	-0.669	0.650	0.0371	0.022	0.798
0.0623	-0.738	0.636	0.0623	-0.129	0.759
0.0872	-0.729	0.638	0.0872	-0.216	0.741
0.1122	-0.704	0.643	0.1122	-0.251	0.734
0.1372	-0.669	0.650	0.1372	-0.272	0.730
0.1623	-0.660	0.652	0.1623	-0.295	0.726
0.1872	-0.642	0.655	0.1872	-0.311	0.723
0.2122	-0.619	0.660	0.2122	-0.317	0.722
0.2372	-0.582	0.668	0.2372	-0.311	0.723
0.2623	-0.549	0.674	0.2623	-0.300	0.725
0.2872	-0.509	0.682	0.2872	-0.285	0.728
0.3122	-0.466	0.687	0.3122	-0.279	0.729
0.3375	-0.451	0.694	0.3375	-0.264	0.732
0.3618	-0.425	0.699	0.3618	-0.249	0.735
0.3873	-0.400	0.704	0.3873	-0.239	0.737
0.4124	-0.380	0.708	0.4124	-0.232	0.739
0.4371	-0.347	0.714	0.4371	-0.211	0.743
0.4621	-0.334	0.717	0.4621	-0.206	0.744
0.4871	-0.311	0.722	0.4871	-0.196	0.746
0.5120	-0.288	0.727	0.5120	-0.182	0.748
0.5371	-0.259	0.732	0.5371	-0.162	0.752
0.5621	-0.234	0.737	0.5621	-0.143	0.756
0.5870	-0.216	0.741	0.5870	-0.136	0.758
0.6122	-0.183	0.747	0.6122	-0.106	0.763
0.6371	-0.169	0.750	0.6371	-0.102	0.764
0.6601	-0.147	0.754	0.6601	-0.088	0.767
0.6869	-0.132	0.757	0.6869	-0.079	0.769
0.7121	-0.111	0.762	0.7121	-0.062	0.772
0.7369	-0.088	0.766	0.7369	-0.048	0.775
0.7620	-0.069	0.770	0.7620	-0.034	0.778
0.7870	-0.043	0.775	0.7870	-0.016	0.781
0.8117	-0.021	0.780	0.8117	0.003	0.785
0.8370	0.002	0.784	0.8370	0.018	0.788
0.8619	0.025	0.789	0.8619	0.038	0.792
0.8869	0.055	0.795	0.8869	0.060	0.796
0.9120	0.085	0.801	0.9120	0.088	0.802
0.9416	0.124	0.808	0.9416	0.120	0.808

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.620 ALPHA 3.00 REY 1.70×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2706 CM = 0.0064 CDW = 0.0100

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.560	0.673	0.0119	0.632	0.909
0.0371	-0.890	0.607	0.0371	0.165	0.816
0.0623	-0.929	0.599	0.0623	-0.004	0.783
0.0872	-0.878	0.610	0.0872	-0.103	0.763
0.1122	-0.832	0.619	0.1122	-0.150	0.754
0.1372	-0.781	0.629	0.1372	-0.181	0.748
0.1623	-0.758	0.633	0.1623	-0.210	0.742
0.1872	-0.728	0.639	0.1872	-0.232	0.737
0.2122	-0.697	0.645	0.2122	-0.244	0.735
0.2372	-0.652	0.654	0.2372	-0.244	0.735
0.2620	-0.611	0.662	0.2620	-0.237	0.736
0.2872	-0.564	0.672	0.2872	-0.226	0.739
0.3122	-0.537	0.677	0.3122	-0.222	0.739
0.3375	-0.499	0.685	0.3375	-0.212	0.741
0.3613	-0.467	0.691	0.3613	-0.200	0.744
0.3873	-0.439	0.696	0.3873	-0.193	0.745
0.4124	-0.416	0.701	0.4124	-0.190	0.746
0.4371	-0.379	0.708	0.4371	-0.172	0.749
0.4621	-0.361	0.712	0.4621	-0.172	0.749
0.4871	-0.337	0.717	0.4871	-0.162	0.751
0.5120	-0.312	0.722	0.5120	-0.152	0.753
0.5371	-0.281	0.728	0.5371	-0.133	0.757
0.5621	-0.253	0.733	0.5621	-0.108	0.762
0.5873	-0.234	0.737	0.5873	-0.110	0.762
0.6122	-0.197	0.744	0.6122	-0.082	0.767
0.6371	-0.162	0.747	0.6371	-0.080	0.768
0.6621	-0.158	0.752	0.6621	-0.067	0.770
0.6869	-0.141	0.755	0.6869	-0.060	0.771
0.7121	-0.119	0.760	0.7121	-0.047	0.774
0.7369	-0.095	0.765	0.7369	-0.032	0.777
0.7623	-0.073	0.769	0.7623	-0.021	0.779
0.7870	-0.046	0.774	0.7870	-0.004	0.783
0.8117	-0.023	0.779	0.8117	0.012	0.786
0.8370	0.001	0.784	0.8370	0.027	0.789
0.8619	0.027	0.789	0.8619	0.043	0.792
0.8869	0.057	0.795	0.8869	0.067	0.797
0.9120	0.088	0.801	0.9120	0.090	0.801
0.9416	0.127	0.809	0.9416	0.122	0.808

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.599 ALPHA 4.00 REYNOLDS 1.70×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3598 CM = 0.0089 CDW = 0.0109

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.845	0.618	0.0119	0.754	0.933
0.0371	-1.123	0.563	0.0371	0.292	0.842
0.0623	-1.144	0.559	0.0623	0.111	0.826
0.0872	-1.027	0.582	0.0872	0.002	0.785
0.1122	-0.954	0.596	0.1122	-0.052	0.774
0.1372	-0.883	0.610	0.1372	-0.092	0.766
0.1623	-0.853	0.616	0.1623	-0.125	0.762
0.1872	-0.812	0.624	0.1872	-0.153	0.754
0.2122	-0.772	0.632	0.2122	-0.170	0.751
0.2372	-0.718	0.643	0.2372	-0.175	0.750
0.2620	-0.671	0.652	0.2620	-0.174	0.750
0.2872	-0.617	0.663	0.2872	-0.167	0.751
0.3122	-0.564	0.669	0.3122	-0.168	0.751
0.3375	-0.541	0.678	0.3375	-0.161	0.753
0.3618	-0.506	0.685	0.3618	-0.152	0.754
0.3873	-0.473	0.691	0.3873	-0.149	0.755
0.4124	-0.447	0.696	0.4124	-0.147	0.755
0.4371	-0.408	0.704	0.4371	-0.133	0.758
0.4621	-0.387	0.708	0.4621	-0.133	0.758
0.4871	-0.360	0.713	0.4871	-0.128	0.759
0.5120	-0.333	0.719	0.5120	-0.119	0.761
0.5371	-0.297	0.726	0.5371	-0.103	0.764
0.5621	-0.270	0.731	0.5621	-0.088	0.767
0.5870	-0.247	0.736	0.5870	-0.084	0.768
0.6122	-0.213	0.742	0.6122	-0.058	0.773
0.6371	-0.193	0.746	0.6371	-0.057	0.773
0.6661	-0.167	0.751	0.6661	-0.047	0.775
0.6869	-0.149	0.755	0.6869	-0.041	0.776
0.7121	-0.126	0.760	0.7121	-0.029	0.778
0.7369	-0.099	0.765	0.7369	-0.019	0.780
0.7620	-0.077	0.769	0.7620	-0.007	0.783
0.7870	-0.047	0.775	0.7870	0.008	0.786
0.8117	-0.024	0.780	0.8117	0.023	0.789
0.8370	0.002	0.785	0.8370	0.036	0.791
0.8619	0.028	0.790	0.8619	0.050	0.794
0.8869	0.060	0.796	0.8869	0.070	0.798
0.9120	0.091	0.802	0.9120	0.093	0.803
0.9416	0.131	0.810	0.9416	0.121	0.808

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.598 ALPHA 5.00 REYNOLDS 1.70⁶ × 10⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.4560 C_D = 0.0130 C_{DH} = 0.0123

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	C _P	P/H	X/C	C _P	P/H
0.0119	-1.157	0.557	0.0119	0.856	0.953
0.0371	-1.389	0.511	0.0371	0.412	0.865
0.0623	-1.645	0.460	0.0623	0.221	0.828
0.0872	-1.187	0.551	0.0872	0.104	0.804
0.1122	-1.093	0.569	0.1122	0.038	0.791
0.1372	-0.996	0.568	0.1372	-0.026	0.782
0.1623	-0.950	0.597	0.1623	-0.044	0.775
0.1872	-0.898	0.608	0.1872	-0.076	0.769
0.2122	-0.843	0.618	0.2122	-0.099	0.764
0.2372	-0.778	0.631	0.2372	-0.138	0.762
0.2620	-0.726	0.641	0.2620	-0.112	0.762
0.2872	-0.667	0.653	0.2872	-0.108	0.762
0.3122	-0.629	0.660	0.3122	-0.114	0.761
0.3375	-0.582	0.669	0.3375	-0.109	0.762
0.3618	-0.543	0.677	0.3618	-0.105	0.763
0.3873	-0.506	0.684	0.3873	-0.104	0.763
0.4124	-0.475	0.690	0.4124	-0.106	0.763
0.4371	-0.433	0.699	0.4371	-0.093	0.765
0.4621	-0.408	0.703	0.4621	-0.096	0.765
0.4871	-0.378	0.709	0.4871	-0.093	0.765
0.5120	-0.349	0.714	0.5120	-0.086	0.767
0.5371	-0.314	0.721	0.5371	-0.072	0.770
0.5621	-0.284	0.727	0.5621	-0.060	0.772
0.5870	-0.259	0.732	0.5870	-0.056	0.773
0.6122	-0.222	0.740	0.6122	-0.033	0.777
0.6371	-0.201	0.744	0.6371	-0.034	0.777
0.6661	-0.174	0.749	0.6661	-0.026	0.779
0.6869	-0.154	0.753	0.6869	-0.023	0.779
0.7121	-0.130	0.758	0.7121	-0.011	0.781
0.7369	-0.102	0.763	0.7369	-0.002	0.783
0.7620	-0.078	0.768	0.7620	0.007	0.785
0.7870	-0.048	0.774	0.7870	0.022	0.788
0.8117	-0.022	0.779	0.8117	0.031	0.790
0.8370	0.005	0.784	0.8370	0.047	0.793
0.8619	0.031	0.789	0.8619	0.059	0.795
0.8869	0.064	0.796	0.8869	0.078	0.799
0.9120	0.094	0.802	0.9120	0.098	0.803
0.9416	0.132	0.810	0.9416	0.124	0.808

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.598 ALPHA 6.00 REYNOLDS 1.70*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.5391 CM = 0.0157 CDW = 0.0127

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-1.446	0.501	0.0119	0.938	0.969
0.0371	-1.675	0.456	0.0371	0.518	0.885
0.0623	-1.944	0.403	0.0623	0.323	0.847
0.0872	-1.295	0.531	0.0872	0.198	0.822
0.1122	-1.170	0.555	0.1122	0.128	0.808
0.1372	-1.373	0.574	0.1372	0.076	0.798
0.1623	-1.023	0.584	0.1623	0.031	0.789
0.1872	-0.960	0.597	0.1872	-0.005	0.782
0.2122	-0.904	0.608	0.2122	-0.028	0.777
0.2372	-0.831	0.622	0.2372	-0.043	0.774
0.2620	-0.773	0.633	0.2620	-0.049	0.773
0.2872	-0.705	0.647	0.2872	-0.050	0.772
0.3122	-0.605	0.655	0.3122	-0.059	0.773
0.3375	-0.611	0.665	0.3375	-0.058	0.771
0.3618	-0.570	0.673	0.3618	-0.055	0.771
0.3873	-0.527	0.682	0.3873	-0.058	0.771
0.4124	-0.496	0.698	0.4124	-0.061	0.770
0.4371	-0.452	0.708	0.4371	-0.051	0.772
0.4621	-0.427	0.714	0.4621	-0.056	0.771
0.4871	-0.392	0.721	0.4871	-0.056	0.771
0.5120	-0.363	0.727	0.5120	-0.052	0.772
0.5371	-0.324	0.733	0.5371	-0.038	0.774
0.5621	-0.294	0.740	0.5621	-0.026	0.777
0.5870	-0.265	0.744	0.5870	-0.029	0.776
0.6122	-0.231	0.750	0.6122	-0.006	0.781
0.6371	-0.207	0.755	0.6371	-0.013	0.780
0.6661	-0.179	0.759	0.6661	-0.002	0.782
0.6869	-0.155	0.765	0.6869	-0.002	0.782
0.7121	-0.131	0.770	0.7121	0.009	0.784
0.7369	-0.101	0.777	0.7369	0.016	0.786
0.7620	-0.078	0.781	0.7620	0.024	0.787
0.7870	-0.044	0.787	0.7870	0.037	0.793
0.8117	-0.020	0.792	0.8117	0.045	0.791
0.8370	0.008	0.799	0.8370	0.057	0.794
0.8619	0.033	0.804	0.8619	0.068	0.796
0.8869	0.068	0.812	0.8869	0.084	0.799
0.9120	0.096		0.9120	0.102	0.803
0.9416	0.135		0.9416	0.125	0.807

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.601 ALPHA 7.00 REY 1.70*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.6311 CM = 0.0211 CDW = 0.0137

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-1.692	0.448	0.0119	1.000	0.982
0.0371	-2.011	0.384	0.0371	0.611	0.925
0.0623	-2.165	0.354	0.0623	0.418	0.867
0.0872	-2.054	0.376	0.0872	0.290	0.842
0.1122	-1.362	0.519	0.1122	0.213	0.826
0.1372	-1.093	0.567	0.1372	0.157	0.815
0.1623	-1.045	0.576	0.1623	0.109	0.806
0.1872	-0.992	0.586	0.1872	0.069	0.798
0.2122	-0.937	0.598	0.2122	0.038	0.792
0.2372	-0.865	0.612	0.2372	0.021	0.789
0.2620	-0.804	0.624	0.2620	0.013	0.787
0.2872	-0.737	0.637	0.2872	0.006	0.786
0.3122	-0.692	0.646	0.3122	-0.003	0.784
0.3375	-0.637	0.657	0.3375	-0.007	0.783
0.3618	-0.591	0.666	0.3618	-0.005	0.783
0.3873	-0.547	0.675	0.3873	-0.012	0.782
0.4124	-0.513	0.682	0.4124	-0.017	0.781
0.4371	-0.466	0.691	0.4371	-0.011	0.782
0.4621	-0.439	0.697	0.4621	-0.016	0.782
0.4871	-0.403	0.704	0.4871	-0.019	0.781
0.5123	-0.371	0.710	0.5123	-0.016	0.782
0.5371	-0.332	0.718	0.5371	-0.007	0.783
0.5621	-0.301	0.724	0.5621	0.003	0.785
0.5870	-0.271	0.730	0.5870	0.000	0.785
0.6122	-0.234	0.737	0.6122	0.022	0.789
0.6371	-0.209	0.742	0.6371	0.017	0.788
0.6651	-0.178	0.748	0.6651	0.020	0.789
0.6869	-0.155	0.753	0.6869	0.021	0.789
0.7121	-0.129	0.753	0.7121	0.025	0.790
0.7359	-0.099	0.764	0.7359	0.036	0.792
0.7620	-0.073	0.769	0.7620	0.037	0.792
0.7873	-0.042	0.775	0.7873	0.052	0.795
0.8117	-0.015	0.780	0.8117	0.057	0.796
0.8370	0.013	0.786	0.8370	0.069	0.799
0.8619	0.039	0.791	0.8619	0.076	0.800
0.8869	0.071	0.798	0.8869	0.091	0.803
0.9123	0.098	0.803	0.9123	0.106	0.806
0.9416	0.134	0.810	0.9416	0.127	0.810

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.519 ALPHA 0.00 REYNOLDS 1.72*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.0015 CM = -0.0002 CDW = 0.0091

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.159	0.789	0.0119	0.155	0.787
0.0371	-0.289	0.689	0.0371	-0.284	0.689
0.0623	-0.430	0.657	0.0623	-0.438	0.654
0.0872	-0.483	0.646	0.0872	-0.537	0.632
0.1122	-0.498	0.642	0.1122	-0.497	0.641
0.1372	-0.495	0.643	0.1372	-0.494	0.642
0.1623	-0.510	0.640	0.1623	-0.529	0.638
0.1872	-0.509	0.640	0.1872	-0.514	0.637
0.2122	-0.506	0.641	0.2122	-0.503	0.640
0.2372	-0.480	0.647	0.2372	-0.481	0.645
0.2620	-0.459	0.651	0.2620	-0.458	0.650
0.2872	-0.426	0.659	0.2872	-0.427	0.657
0.3122	-0.413	0.662	0.3122	-0.410	0.660
0.3375	-0.385	0.668	0.3375	-0.387	0.666
0.3618	-0.362	0.673	0.3618	-0.364	0.671
0.3873	-0.343	0.677	0.3873	-0.344	0.675
0.4124	-0.329	0.680	0.4124	-0.327	0.679
0.4371	-0.299	0.687	0.4371	-0.299	0.685
0.4621	-0.291	0.689	0.4621	-0.290	0.688
0.4871	-0.271	0.693	0.4871	-0.273	0.692
0.5120	-0.253	0.697	0.5120	-0.252	0.697
0.5371	-0.225	0.703	0.5371	-0.226	0.702
0.5621	-0.203	0.708	0.5621	-0.204	0.707
0.5870	-0.188	0.712	0.5870	-0.189	0.711
0.6122	-0.156	0.719	0.6122	-0.157	0.718
0.6371	-0.147	0.721	0.6371	-0.146	0.720
0.6661	-0.126	0.725	0.6661	-0.126	0.725
0.6869	-0.113	0.728	0.6869	-0.113	0.728
0.7121	-0.095	0.732	0.7121	-0.095	0.733
0.7369	-0.072	0.737	0.7369	-0.070	0.738
0.7620	-0.057	0.741	0.7620	-0.055	0.741
0.7870	-0.032	0.746	0.7870	-0.032	0.747
0.8117	-0.011	0.751	0.8117	-0.010	0.751
0.8370	0.011	0.756	0.8370	0.010	0.756
0.8619	0.033	0.761	0.8619	0.033	0.761
0.8869	0.059	0.767	0.8869	0.060	0.767
0.9120	0.090	0.773	0.9120	0.090	0.774
0.9416	0.128	0.782	0.9416	0.126	0.782

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.651 ALPHA 1.00 REYNOLDS 1.72*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0927 CM = 0.0026 CDW = 0.0125

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.043	0.742	0.0119	0.342	0.823
0.0371	-0.469	0.647	0.0371	-0.123	0.724
0.0623	-0.599	0.618	0.0623	-0.278	0.692
0.0872	-0.625	0.612	0.0872	-0.351	0.673
0.1122	-0.625	0.612	0.1122	-0.383	0.656
0.1372	-0.614	0.617	0.1372	-0.393	0.655
0.1623	-0.612	0.615	0.1623	-0.413	0.659
0.1872	-0.603	0.617	0.1872	-0.422	0.657
0.2122	-0.572	0.623	0.2122	-0.426	0.657
0.2372	-0.555	0.627	0.2372	-0.437	0.651
0.2623	-0.527	0.633	0.2623	-0.393	0.654
0.2872	-0.486	0.642	0.2872	-0.368	0.673
0.3122	-0.468	0.647	0.3122	-0.357	0.672
0.3375	-0.434	0.654	0.3375	-0.335	0.677
0.3618	-0.409	0.660	0.3618	-0.318	0.681
0.3873	-0.386	0.665	0.3873	-0.303	0.684
0.4124	-0.367	0.669	0.4124	-0.291	0.687
0.4371	-0.334	0.676	0.4371	-0.266	0.692
0.4621	-0.322	0.679	0.4621	-0.259	0.693
0.4871	-0.300	0.684	0.4871	-0.242	0.697
0.5123	-0.279	0.688	0.5123	-0.225	0.701
0.5371	-0.245	0.696	0.5371	-0.202	0.706
0.5621	-0.225	0.703	0.5621	-0.180	0.711
0.5873	-0.208	0.704	0.5873	-0.169	0.714
0.6122	-0.175	0.712	0.6122	-0.137	0.721
0.6371	-0.162	0.715	0.6371	-0.130	0.722
0.6661	-0.139	0.720	0.6661	-0.111	0.727
0.6869	-0.125	0.723	0.6869	-0.100	0.729
0.7121	-0.106	0.727	0.7121	-0.082	0.732
0.7369	-0.082	0.732	0.7369	-0.063	0.736
0.7623	-0.063	0.737	0.7623	-0.046	0.742
0.7873	-0.038	0.742	0.7873	-0.026	0.745
0.8117	-0.016	0.747	0.8117	-0.008	0.749
0.8370	0.006	0.752	0.8370	0.022	0.755
0.8619	0.034	0.758	0.8619	0.036	0.759
0.8869	0.060	0.764	0.8869	0.062	0.765
0.9123	0.091	0.771	0.9123	0.100	0.773
0.9416	0.130	0.783	0.9416	0.126	0.779

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.651 ALPHA 2.00 REY 1.72*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1824 CM = 0.0053 CDW = 0.0108

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.251	0.696	0.0119	0.499	0.863
0.0371	-0.600	0.604	0.0371	0.025	0.757
0.0623	-0.706	0.576	0.0623	-0.142	0.720
0.0872	-0.772	0.579	0.0872	-0.229	0.700
0.1122	-0.753	0.584	0.1122	-0.269	0.691
0.1372	-0.714	0.592	0.1372	-0.290	0.687
0.1623	-0.709	0.593	0.1623	-0.318	0.680
0.1872	-0.669	0.598	0.1872	-0.334	0.677
0.2122	-0.667	0.594	0.2122	-0.342	0.675
0.2372	-0.621	0.614	0.2372	-0.335	0.677
0.2620	-0.587	0.621	0.2620	-0.325	0.679
0.2872	-0.539	0.632	0.2872	-0.305	0.683
0.3122	-0.515	0.637	0.3122	-0.300	0.684
0.3375	-0.477	0.646	0.3375	-0.282	0.689
0.3618	-0.447	0.652	0.3618	-0.268	0.692
0.3873	-0.420	0.659	0.3873	-0.256	0.694
0.4124	-0.398	0.663	0.4124	-0.249	0.696
0.4371	-0.362	0.671	0.4371	-0.226	0.701
0.4621	-0.349	0.675	0.4621	-0.222	0.702
0.4871	-0.323	0.681	0.4871	-0.208	0.705
0.5120	-0.300	0.686	0.5120	-0.196	0.708
0.5371	-0.268	0.693	0.5371	-0.173	0.713
0.5621	-0.242	0.699	0.5621	-0.153	0.717
0.5870	-0.222	0.703	0.5870	-0.144	0.719
0.6122	-0.189	0.711	0.6122	-0.115	0.726
0.6371	-0.174	0.714	0.6371	-0.108	0.727
0.6651	-0.149	0.720	0.6651	-0.094	0.731
0.6869	-0.133	0.723	0.6869	-0.083	0.733
0.7121	-0.113	0.726	0.7121	-0.068	0.737
0.7369	-0.087	0.734	0.7369	-0.050	0.741
0.7620	-0.067	0.738	0.7620	-0.037	0.744
0.7870	-0.040	0.744	0.7870	-0.017	0.748
0.8117	-0.017	0.749	0.8117	0.002	0.752
0.8370	0.007	0.755	0.8370	0.019	0.756
0.8619	0.032	0.760	0.8619	0.039	0.760
0.8869	0.062	0.767	0.8869	0.064	0.766
0.9120	0.093	0.774	0.9120	0.091	0.772
0.9416	0.131	0.782	0.9416	0.124	0.780

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.649 ALPHA 3.00 REY 1.72×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2784 CM = 0.0086 CDW = 0.0105

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.493	0.643	0.0119	0.042	0.895
0.0371	-0.871	0.559	0.0371	0.169	0.789
0.0623	-1.073	0.515	0.0623	-0.010	0.749
0.0872	-0.941	0.544	0.0872	-0.111	0.726
0.1122	-0.897	0.554	0.1122	-0.162	0.715
0.1372	-0.833	0.508	0.1372	-0.193	0.708
0.1623	-0.816	0.572	0.1623	-0.227	0.700
0.1872	-0.784	0.579	0.1872	-0.250	0.695
0.2122	-0.750	0.586	0.2122	-0.263	0.692
0.2372	-0.694	0.599	0.2372	-0.261	0.692
0.2620	-0.652	0.608	0.2620	-0.256	0.693
0.2872	-0.596	0.621	0.2872	-0.242	0.697
0.3122	-0.567	0.627	0.3122	-0.242	0.697
0.3375	-0.523	0.637	0.3375	-0.228	0.700
0.3618	-0.489	0.644	0.3618	-0.216	0.702
0.3873	-0.458	0.651	0.3873	-0.209	0.704
0.4124	-0.432	0.657	0.4124	-0.204	0.705
0.4371	-0.393	0.666	0.4371	-0.184	0.710
0.4621	-0.375	0.669	0.4621	-0.184	0.710
0.4871	-0.346	0.676	0.4871	-0.173	0.713
0.5122	-0.319	0.682	0.5122	-0.163	0.715
0.5371	-0.287	0.689	0.5371	-0.142	0.720
0.5621	-0.259	0.695	0.5621	-0.125	0.723
0.5870	-0.238	0.700	0.5870	-0.117	0.725
0.6122	-0.203	0.708	0.6122	-0.091	0.731
0.6371	-0.185	0.711	0.6371	-0.086	0.732
0.6651	-0.158	0.718	0.6651	-0.073	0.735
0.6859	-0.140	0.722	0.6859	-0.065	0.737
0.7121	-0.117	0.726	0.7121	-0.052	0.740
0.7369	-0.091	0.732	0.7369	-0.035	0.744
0.7620	-0.069	0.737	0.7620	-0.023	0.747
0.7870	-0.041	0.743	0.7870	-0.005	0.751
0.8117	-0.015	0.749	0.8117	0.012	0.754
0.8370	0.008	0.754	0.8370	0.028	0.758
0.8619	0.035	0.760	0.8619	0.045	0.762
0.8869	0.066	0.767	0.8869	0.067	0.767
0.9120	0.098	0.774	0.9120	0.094	0.773
0.9416	0.136	0.783	0.9416	0.124	0.779

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.649 ALPHA 4.00 REYNOLDS 1.72×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.3728$ $C_M = 0.0119$ $C_{DW} = 0.0112$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	C _p	P/H	X/C	C _p	P/H
0.0119	-0.754	0.590	0.0119	0.761	0.922
0.0371	-1.070	0.516	0.0371	0.294	0.818
0.0623	-1.534	0.412	0.0623	0.107	0.777
0.0872	-1.390	0.511	0.0872	-0.003	0.752
0.1122	-1.017	0.527	0.1122	-0.061	0.740
0.1372	-0.941	0.544	0.1372	-0.099	0.731
0.1623	-0.916	0.550	0.1623	-0.138	0.722
0.1872	-0.871	0.560	0.1872	-0.167	0.716
0.2122	-0.827	0.570	0.2122	-0.187	0.711
0.2372	-0.761	0.584	0.2372	-0.189	0.710
0.2620	-0.708	0.596	0.2620	-0.190	0.710
0.2872	-0.649	0.609	0.2872	-0.181	0.712
0.3122	-0.511	0.618	0.3122	-0.164	0.712
0.3375	-0.563	0.626	0.3375	-0.173	0.714
0.3618	-0.525	0.637	0.3618	-0.165	0.716
0.3873	-0.490	0.644	0.3873	-0.162	0.717
0.4124	-0.460	0.651	0.4124	-0.161	0.717
0.4371	-0.420	0.660	0.4371	-0.143	0.721
0.4621	-0.398	0.665	0.4621	-0.145	0.720
0.4871	-0.367	0.672	0.4871	-0.137	0.722
0.5120	-0.339	0.678	0.5120	-0.129	0.723
0.5371	-0.304	0.686	0.5371	-0.111	0.726
0.5621	-0.274	0.692	0.5621	-0.097	0.731
0.5873	-0.250	0.698	0.5873	-0.090	0.732
0.6122	-0.212	0.706	0.6122	-0.065	0.733
0.6371	-0.194	0.710	0.6371	-0.062	0.738
0.6661	-0.165	0.716	0.6661	-0.052	0.741
0.6869	-0.147	0.720	0.6869	-0.045	0.742
0.7121	-0.122	0.726	0.7121	-0.034	0.745
0.7369	-0.095	0.732	0.7369	-0.020	0.748
0.7620	-0.070	0.737	0.7620	-0.001	0.750
0.7873	-0.042	0.744	0.7873	0.009	0.754
0.8117	-0.014	0.750	0.8117	0.020	0.757
0.8370	0.010	0.755	0.8370	0.037	0.761
0.8619	0.038	0.761	0.8619	0.051	0.764
0.8869	0.068	0.768	0.8869	0.073	0.768
0.9120	0.100	0.775	0.9120	0.096	0.774
0.9415	0.137	0.784	0.9415	0.125	0.780

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.650 ALPHA 5.00 REY 1.72⁶ × 10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4714 CM = 0.0166 CDW = 0.0128

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.960	0.538	0.0119	0.860	0.944
0.0371	-1.289	0.465	0.0371	0.409	0.844
0.0623	-1.613	0.392	0.0623	0.217	0.801
0.0872	-1.610	0.343	0.0872	0.098	0.774
0.1122	-1.626	0.390	0.1122	0.031	0.763
0.1372	-1.813	0.526	0.1372	-0.014	0.749
0.1623	-0.930	0.545	0.1623	-0.053	0.741
0.1872	-0.901	0.551	0.1872	-0.088	0.733
0.2122	-0.867	0.559	0.2122	-0.111	0.727
0.2372	-0.798	0.574	0.2372	-0.120	0.725
0.2620	-0.746	0.586	0.2620	-0.122	0.725
0.2872	-0.680	0.601	0.2872	-0.120	0.725
0.3122	-0.643	0.609	0.3122	-0.125	0.724
0.3375	-0.589	0.621	0.3375	-0.120	0.725
0.3618	-0.551	0.630	0.3618	-0.114	0.727
0.3873	-0.511	0.639	0.3873	-0.115	0.727
0.4124	-0.480	0.646	0.4124	-0.115	0.726
0.4371	-0.436	0.655	0.4371	-0.101	0.729
0.4621	-0.414	0.661	0.4621	-0.105	0.728
0.4871	-0.379	0.668	0.4871	-0.100	0.729
0.5120	-0.351	0.675	0.5120	-0.094	0.731
0.5371	-0.312	0.683	0.5371	-0.078	0.734
0.5621	-0.253	0.690	0.5621	-0.064	0.737
0.5873	-0.256	0.696	0.5873	-0.062	0.738
0.6122	-0.220	0.704	0.6122	-0.038	0.743
0.6371	-0.198	0.709	0.6371	-0.039	0.743
0.6661	-0.169	0.715	0.6661	-0.029	0.745
0.6869	-0.147	0.720	0.6869	-0.022	0.747
0.7121	-0.124	0.726	0.7121	-0.013	0.748
0.7369	-0.093	0.732	0.7369	-0.004	0.750
0.7620	-0.070	0.738	0.7620	0.008	0.753
0.7870	-0.039	0.744	0.7870	0.022	0.756
0.8117	-0.014	0.750	0.8117	0.034	0.759
0.8370	0.015	0.756	0.8370	0.048	0.762
0.8619	0.039	0.762	0.8619	0.070	0.767
0.8869	0.074	0.769	0.8869	0.080	0.769
0.9120	0.101	0.776	0.9120	0.103	0.774
0.9416	0.141	0.784	0.9416	0.128	0.780

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.652 ALPHA 6.00 REYNOLDS 1.73⁶ × 10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.5726 CM = 0.0213 CDW = 0.0159

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-1.160	0.491	0.0119	0.437	0.961
0.0371	-1.533	0.408	0.0371	0.511	0.867
0.0623	-1.746	0.360	0.0623	0.519	0.824
0.0872	-1.703	0.370	0.0872	0.194	0.796
0.1122	-1.822	0.343	0.1122	0.124	0.781
0.1372	-1.762	0.357	0.1372	0.072	0.769
0.1623	-1.600	0.393	0.1623	0.026	0.759
0.1872	-0.995	0.528	0.1872	-0.011	0.751
0.2122	-0.864	0.557	0.2122	-0.037	0.744
0.2372	-0.766	0.574	0.2372	-0.049	0.742
0.2623	-0.738	0.585	0.2623	-0.056	0.740
0.2872	-0.683	0.598	0.2872	-0.058	0.740
0.3122	-0.649	0.605	0.3122	-0.066	0.738
0.3375	-0.593	0.617	0.3375	-0.063	0.738
0.3618	-0.560	0.625	0.3618	-0.062	0.739
0.3873	-0.521	0.634	0.3873	-0.064	0.738
0.4124	-0.491	0.641	0.4124	-0.067	0.738
0.4371	-0.447	0.651	0.4371	-0.056	0.742
0.4621	-0.422	0.656	0.4621	-0.063	0.738
0.4871	-0.387	0.664	0.4871	-0.059	0.739
0.5120	-0.358	0.670	0.5120	-0.056	0.740
0.5371	-0.319	0.679	0.5371	-0.042	0.743
0.5621	-0.288	0.686	0.5621	-0.031	0.745
0.5870	-0.261	0.692	0.5870	-0.032	0.745
0.6122	-0.224	0.700	0.6122	-0.009	0.752
0.6371	-0.200	0.706	0.6371	-0.012	0.749
0.6661	-0.171	0.712	0.6661	-0.004	0.751
0.6869	-0.149	0.717	0.6869	-0.003	0.752
0.7121	-0.124	0.723	0.7121	0.008	0.754
0.7369	-0.093	0.730	0.7369	0.017	0.756
0.7623	-0.069	0.735	0.7623	0.022	0.757
0.7873	-0.037	0.742	0.7873	0.038	0.761
0.8117	-0.013	0.748	0.8117	0.046	0.763
0.8370	0.016	0.754	0.8370	0.061	0.766
0.8619	0.042	0.760	0.8619	0.073	0.768
0.8869	0.075	0.767	0.8869	0.090	0.772
0.9120	0.104	0.774	0.9120	0.109	0.776
0.9416	0.141	0.782	0.9416	0.133	0.782

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.675 ALPHA 0.00 REY 1.61×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0060 CH = 0.0002 CDW = 0.0093

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.185	0.780	0.0119	0.168	0.777
0.0371	-0.279	0.671	0.0371	-0.281	0.671
0.0623	-0.430	0.636	0.0623	-0.450	0.632
0.0872	-0.490	0.621	0.0872	-0.554	0.607
0.1122	-0.507	0.617	0.1122	-0.515	0.616
0.1372	-0.509	0.617	0.1372	-0.511	0.617
0.1623	-0.523	0.614	0.1623	-0.533	0.613
0.1872	-0.531	0.612	0.1872	-0.538	0.611
0.2122	-0.520	0.614	0.2122	-0.528	0.613
0.2372	-0.497	0.620	0.2372	-0.502	0.619
0.2620	-0.471	0.626	0.2620	-0.477	0.625
0.2872	-0.441	0.633	0.2872	-0.445	0.633
0.3122	-0.423	0.637	0.3122	-0.428	0.637
0.3375	-0.396	0.644	0.3375	-0.404	0.642
0.3618	-0.374	0.649	0.3618	-0.381	0.648
0.3873	-0.354	0.654	0.3873	-0.359	0.653
0.4124	-0.334	0.658	0.4124	-0.341	0.657
0.4371	-0.308	0.664	0.4371	-0.313	0.664
0.4621	-0.294	0.668	0.4621	-0.301	0.667
0.4871	-0.277	0.672	0.4871	-0.283	0.671
0.5120	-0.256	0.676	0.5120	-0.261	0.676
0.5371	-0.233	0.683	0.5371	-0.234	0.682
0.5621	-0.206	0.688	0.5621	-0.212	0.687
0.5873	-0.191	0.692	0.5870	-0.195	0.691
0.6122	-0.159	0.699	0.6122	-0.162	0.699
0.6371	-0.148	0.702	0.6371	-0.151	0.702
0.6661	-0.126	0.707	0.6661	-0.130	0.707
0.6869	-0.113	0.710	0.6869	-0.117	0.710
0.7121	-0.094	0.715	0.7121	-0.095	0.715
0.7369	-0.072	0.720	0.7369	-0.073	0.720
0.7620	-0.054	0.724	0.7620	-0.056	0.724
0.7873	-0.028	0.730	0.7870	-0.039	0.728
0.8117	-0.009	0.735	0.8117	-0.009	0.735
0.8370	0.012	0.739	0.8370	0.011	0.740
0.8619	0.034	0.745	0.8619	0.040	0.747
0.8869	0.064	0.752	0.8869	0.062	0.752
0.9123	0.093	0.759	0.9120	0.093	0.759
0.9416	0.131	0.768	0.9410	0.130	0.768

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.674 ALPHA 1.00 REY 1.61*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0941 CM = 0.0027 CDW = 0.0138

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.001	0.737	0.0119	0.356	0.820
0.0371	-0.457	0.630	0.0371	-0.117	0.739
0.0623	-0.606	0.595	0.0623	-0.278	0.671
0.0872	-0.638	0.587	0.0872	-0.357	0.633
0.1122	-0.636	0.588	0.1122	-0.388	0.645
0.1372	-0.623	0.591	0.1372	-0.402	0.642
0.1623	-0.628	0.590	0.1623	-0.422	0.637
0.1872	-0.627	0.590	0.1872	-0.437	0.634
0.2122	-0.606	0.594	0.2122	-0.437	0.634
0.2372	-0.573	0.602	0.2372	-0.423	0.637
0.2620	-0.541	0.610	0.2620	-0.403	0.642
0.2872	-0.503	0.619	0.2872	-0.380	0.648
0.3122	-0.479	0.624	0.3122	-0.365	0.651
0.3375	-0.448	0.632	0.3375	-0.346	0.656
0.3618	-0.419	0.638	0.3618	-0.325	0.660
0.3873	-0.396	0.644	0.3873	-0.310	0.664
0.4124	-0.373	0.649	0.4124	-0.296	0.667
0.4371	-0.342	0.656	0.4371	-0.271	0.673
0.4621	-0.328	0.660	0.4621	-0.263	0.675
0.4871	-0.305	0.665	0.4871	-0.247	0.679
0.5120	-0.281	0.671	0.5120	-0.228	0.684
0.5371	-0.254	0.677	0.5371	-0.204	0.689
0.5621	-0.227	0.684	0.5621	-0.183	0.694
0.5870	-0.211	0.687	0.5870	-0.170	0.697
0.6122	-0.177	0.695	0.6122	-0.139	0.704
0.6371	-0.164	0.698	0.6371	-0.130	0.707
0.6601	-0.140	0.704	0.6601	-0.111	0.711
0.6869	-0.126	0.707	0.6869	-0.100	0.714
0.7121	-0.104	0.712	0.7121	-0.082	0.716
0.7369	-0.082	0.718	0.7369	-0.061	0.723
0.7620	-0.061	0.723	0.7620	-0.045	0.727
0.7870	-0.035	0.729	0.7870	-0.025	0.731
0.8117	-0.013	0.734	0.8117	-0.003	0.736
0.8370	0.010	0.739	0.8370	0.016	0.741
0.8619	0.034	0.745	0.8619	0.037	0.746
0.8869	0.064	0.752	0.8869	0.064	0.752
0.9120	0.094	0.759	0.9120	0.093	0.759
0.9416	0.133	0.768	0.9416	0.129	0.767

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.676 ALPHA 2.00 REY 1.62*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1867 CM = 0.0060 CDW = 0.0106

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.205	0.688	0.0119	0.513	0.857
0.0371	-0.642	0.585	0.0371	0.033	0.743
0.0623	-0.821	0.543	0.0623	-0.138	0.743
0.0872	-0.800	0.548	0.0872	-0.232	0.681
0.1122	-0.778	0.553	0.1122	-0.272	0.671
0.1372	-0.741	0.562	0.1372	-0.297	0.666
0.1623	-0.736	0.563	0.1623	-0.325	0.659
0.1872	-0.725	0.565	0.1872	-0.347	0.654
0.2122	-0.692	0.573	0.2122	-0.354	0.652
0.2372	-0.648	0.584	0.2372	-0.348	0.654
0.2620	-0.609	0.593	0.2620	-0.334	0.657
0.2872	-0.559	0.605	0.2872	-0.316	0.661
0.3122	-0.502	0.611	0.3122	-0.308	0.663
0.3375	-0.493	0.620	0.3375	-0.293	0.667
0.3618	-0.461	0.628	0.3618	-0.276	0.671
0.3873	-0.433	0.634	0.3873	-0.264	0.673
0.4124	-0.407	0.640	0.4124	-0.255	0.676
0.4371	-0.372	0.649	0.4371	-0.234	0.681
0.4621	-0.354	0.653	0.4621	-0.223	0.683
0.4871	-0.329	0.659	0.4871	-0.214	0.686
0.5120	-0.302	0.665	0.5120	-0.198	0.690
0.5371	-0.272	0.672	0.5371	-0.176	0.695
0.5621	-0.245	0.679	0.5621	-0.157	0.700
0.5870	-0.224	0.684	0.5870	-0.147	0.702
0.6122	-0.190	0.692	0.6122	-0.119	0.709
0.6371	-0.174	0.695	0.6371	-0.111	0.710
0.6661	-0.149	0.701	0.6661	-0.094	0.715
0.6869	-0.132	0.705	0.6869	-0.084	0.717
0.7121	-0.109	0.711	0.7121	-0.068	0.721
0.7369	-0.086	0.716	0.7369	-0.050	0.725
0.7620	-0.064	0.721	0.7620	-0.036	0.729
0.7870	-0.033	0.727	0.7870	-0.016	0.734
0.8117	-0.013	0.733	0.8117	0.004	0.738
0.8370	0.010	0.739	0.8370	0.021	0.742
0.8619	0.035	0.745	0.8619	0.040	0.747
0.8869	0.065	0.752	0.8869	0.066	0.753
0.9120	0.098	0.759	0.9120	0.093	0.759
0.9416	0.136	0.768	0.9416	0.127	0.767

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.674 ALPHA 3.00 REY 1.62×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2835 CM = 0.0101 CDW = 0.0115

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.423	0.638	0.0119	0.642	0.888
0.0371	-0.829	2.543	0.0371	0.170	2.777
0.0623	-1.256	3.443	0.0623	-0.011	0.735
0.0872	-0.946	2.515	0.0872	-0.115	0.710
0.1122	-0.938	0.517	0.1122	-0.167	0.698
0.1372	-0.871	2.533	0.1372	-0.199	0.690
0.1623	-0.857	2.536	0.1623	-0.234	0.682
0.1872	-0.831	2.542	0.1872	-0.261	0.676
0.2122	-0.780	2.553	0.2122	-0.274	0.673
0.2372	-0.721	2.567	0.2372	-0.272	0.673
0.2620	-0.669	2.579	0.2620	-0.266	0.675
0.2872	-0.615	2.592	0.2872	-0.252	0.678
0.3122	-0.578	2.601	0.3122	-0.251	0.678
0.3375	-0.538	2.610	0.3375	-0.238	0.681
0.3618	-0.500	2.619	0.3618	-0.227	0.684
0.3873	-0.467	2.627	0.3873	-0.218	0.686
0.4124	-0.436	2.634	0.4124	-0.211	0.687
0.4371	-0.399	2.643	0.4371	-0.191	0.692
0.4621	-0.377	2.648	0.4621	-0.190	0.693
0.4871	-0.349	2.654	0.4871	-0.180	0.695
0.5123	-0.321	2.661	0.5120	-0.168	0.696
0.5371	-0.289	2.668	0.5371	-0.147	0.703
0.5621	-0.260	2.675	0.5621	-0.130	0.707
0.5870	-0.238	2.680	0.5870	-0.122	0.709
0.6122	-0.202	2.689	0.6122	-0.095	0.715
0.6371	-0.164	2.693	0.6371	-0.090	0.716
0.6661	-0.157	2.700	0.6661	-0.076	0.719
0.6869	-0.138	2.704	0.6869	-0.067	0.721
0.7121	-0.114	2.709	0.7121	-0.053	0.725
0.7369	-0.089	2.715	0.7369	-0.036	0.729
0.7620	-0.066	2.721	0.7620	-0.025	0.732
0.7870	-0.038	2.727	0.7870	-0.006	0.736
0.8117	-0.013	2.733	0.8117	0.011	0.740
0.8370	0.012	2.739	0.8370	0.027	0.744
0.8619	0.037	2.745	0.8619	0.045	0.748
0.8869	0.070	2.753	0.8869	0.069	0.754
0.9120	0.100	2.760	0.9120	0.093	0.759
0.9416	0.139	2.769	0.9416	0.127	0.767

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.675 ALPHA 4.00 REY 1.62×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3831 CM = 0.0145 CDW = 0.0123

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.640	2.586	0.0119	0.759	0.915
0.0371	-1.003	2.501	0.0371	0.297	0.807
0.0623	-1.422	2.402	0.0623	0.107	0.762
0.0872	-1.401	2.407	0.0872	-0.006	0.736
0.1122	-1.412	2.405	0.1122	-0.063	0.722
0.1372	-0.933	2.517	0.1372	-0.105	0.712
0.1623	-0.893	2.527	0.1623	-0.143	0.703
0.1872	-0.882	2.529	0.1872	-0.177	0.695
0.2122	-0.842	2.539	0.2122	-0.193	0.692
0.2372	-0.775	2.554	0.2372	-0.199	0.690
0.2620	-0.719	2.568	0.2620	-0.198	0.691
0.2872	-0.656	2.582	0.2872	-0.190	0.692
0.3122	-0.619	2.591	0.3122	-0.191	0.692
0.3375	-0.571	2.602	0.3375	-0.182	0.694
0.3618	-0.532	2.612	0.3618	-0.175	0.696
0.3873	-0.494	2.621	0.3873	-0.170	0.697
0.4124	-0.463	2.628	0.4124	-0.167	0.698
0.4371	-0.422	2.637	0.4371	-0.153	0.702
0.4621	-0.400	2.643	0.4621	-0.150	0.701
0.4871	-0.367	2.650	0.4871	-0.144	0.703
0.5120	-0.339	2.657	0.5120	-0.134	0.705
0.5371	-0.303	2.666	0.5371	-0.116	0.709
0.5621	-0.275	2.672	0.5621	-0.100	0.713
0.5870	-0.247	2.678	0.5870	-0.085	0.714
0.6122	-0.213	2.687	0.6122	-0.069	0.720
0.6371	-0.190	2.692	0.6371	-0.066	0.721
0.6661	-0.164	2.698	0.6661	-0.054	0.724
0.6869	-0.142	2.703	0.6869	-0.047	0.725
0.7121	-0.119	2.708	0.7121	-0.034	0.728
0.7369	-0.089	2.715	0.7369	-0.021	0.731
0.7620	-0.068	2.720	0.7620	-0.009	0.734
0.7870	-0.036	2.728	0.7870	0.008	0.738
0.8117	-0.012	2.734	0.8117	0.023	0.741
0.8370	0.016	2.740	0.8370	0.037	0.745
0.8619	0.040	2.746	0.8619	0.053	0.748
0.8869	0.075	2.754	0.8869	0.075	0.754
0.9120	0.103	2.761	0.9120	0.098	0.759
0.9416	0.144	2.770	0.9416	0.129	0.766

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.674 ALPHA 5.00 REYNOLDS 1.62*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4816 CM = 0.0187 CDW = 0.0148

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.847	2.538	0.0119	0.850	0.936
0.0371	-1.207	2.454	0.0371	0.405	0.832
0.0623	-1.466	2.393	0.0623	0.213	0.786
0.0872	-1.534	2.377	0.0872	0.094	0.758
0.1122	-1.585	2.365	0.1122	0.029	0.743
0.1372	-1.563	2.370	0.1372	-0.019	0.732
0.1623	-1.531	2.378	0.1623	-0.059	0.722
0.1872	-1.459	2.491	0.1872	-0.096	0.714
0.2122	-0.827	2.543	0.2122	-0.117	0.708
0.2372	-0.755	2.560	0.2372	-0.127	0.706
0.2620	-0.710	2.571	0.2620	-0.130	0.705
0.2872	-0.661	2.582	0.2872	-0.126	0.706
0.3122	-0.626	2.591	0.3122	-0.131	0.705
0.3375	-0.581	2.601	0.3375	-0.127	0.706
0.3618	-0.541	2.611	0.3618	-0.122	0.707
0.3873	-0.504	2.619	0.3873	-0.120	0.708
0.4124	-0.471	2.627	0.4124	-0.120	0.708
0.4371	-0.432	2.636	0.4371	-0.106	0.711
0.4621	-0.407	2.642	0.4621	-0.108	0.710
0.4871	-0.375	2.649	0.4871	-0.103	0.712
0.5120	-0.344	2.657	0.5120	-0.096	0.713
0.5371	-0.318	2.665	0.5371	-0.080	0.717
0.5621	-0.278	2.672	0.5621	-0.067	0.720
0.5870	-0.252	2.678	0.5870	-0.063	0.721
0.6122	-0.215	2.687	0.6122	-0.040	0.726
0.6371	-0.193	2.692	0.6371	-0.039	0.727
0.6661	-0.164	2.699	0.6661	-0.031	0.729
0.6869	-0.143	2.704	0.6869	-0.026	0.732
0.7121	-0.118	2.710	0.7121	-0.013	0.733
0.7369	-0.088	2.717	0.7369	-0.002	0.735
0.7620	-0.065	2.722	0.7620	0.007	0.738

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.700 ALPHA 0.00 REYNOLDS 1.66*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.0050 CM = -0.0002 CDW = 0.0098

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.200	0.770	0.0119	0.187	0.768
0.0371	-0.273	0.653	0.0371	-0.273	0.654
0.0623	-0.445	0.611	0.0623	-0.461	0.608
0.0872	-0.509	0.595	0.0872	-0.569	0.581
0.1122	-0.532	0.589	0.1122	-0.536	0.589
0.1372	-0.535	0.588	0.1372	-0.538	0.589
0.1623	-0.553	0.584	0.1623	-0.561	0.583
0.1872	-0.563	0.581	0.1872	-0.570	0.581
0.2122	-0.553	0.584	0.2122	-0.561	0.582
0.2372	-0.526	0.591	0.2372	-0.532	0.590
0.2623	-0.499	0.597	0.2623	-0.506	0.596
0.2872	-0.463	0.606	0.2872	-0.470	0.605
0.3122	-0.445	0.611	0.3122	-0.452	0.609
0.3375	-0.416	0.618	0.3375	-0.424	0.616
0.3619	-0.391	0.624	0.3619	-0.398	0.623
0.3873	-0.371	0.629	0.3873	-0.376	0.628
0.4124	-0.350	0.634	0.4124	-0.354	0.634
0.4371	-0.321	0.641	0.4371	-0.325	0.641
0.4621	-0.309	0.644	0.4621	-0.312	0.643
0.4871	-0.286	0.650	0.4871	-0.292	0.648
0.5120	-0.265	0.655	0.5120	-0.268	0.654
0.5371	-0.236	0.662	0.5371	-0.241	0.661
0.5621	-0.213	0.668	0.5621	-0.217	0.667
0.5870	-0.197	0.672	0.5870	-0.200	0.671
0.6122	-0.164	0.680	0.6122	-0.167	0.679
0.6371	-0.152	0.683	0.6371	-0.155	0.682
0.6661	-0.131	0.688	0.6661	-0.132	0.688
0.6869	-0.116	0.692	0.6869	-0.119	0.691
0.7121	-0.097	0.697	0.7121	-0.097	0.696
0.7369	-0.074	0.702	0.7369	-0.073	0.702
0.7620	-0.055	0.707	0.7620	-0.055	0.706
0.7870	-0.031	0.713	0.7870	-0.030	0.712
0.8117	-0.008	0.718	0.8117	-0.007	0.718
0.8370	0.014	0.724	0.8370	0.013	0.723
0.8619	0.037	0.730	0.8619	0.038	0.729
0.8869	0.066	0.737	0.8869	0.066	0.736
0.9120	0.095	0.744	0.9120	0.098	0.744
0.9416	0.135	0.754	0.9416	0.135	0.753

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.701 ALPHA 1.00 REY 1.66×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3970 CH = 0.0031 CDW = 0.0124

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.025	0.726	0.0119	0.366	0.812
0.0371	-0.442	0.610	0.0371	-0.109	0.694
0.0623	-0.627	0.564	0.0623	-0.283	0.651
0.0872	-0.663	0.555	0.0872	-0.365	0.631
0.1122	-0.673	0.553	0.1122	-0.401	0.622
0.1372	-0.657	0.557	0.1372	-0.417	0.619
0.1623	-0.658	0.554	0.1623	-0.442	0.612
0.1872	-0.672	0.553	0.1872	-0.459	0.608
0.2122	-0.651	0.557	0.2122	-0.460	0.608
0.2372	-0.612	0.567	0.2372	-0.442	0.612
0.2622	-0.573	0.577	0.2622	-0.424	0.617
0.2872	-0.528	0.588	0.2872	-0.396	0.624
0.3122	-0.501	0.595	0.3122	-0.385	0.626
0.3372	-0.468	0.603	0.3372	-0.361	0.632
0.3618	-0.434	0.611	0.3618	-0.339	0.636
0.3873	-0.411	0.617	0.3873	-0.323	0.642
0.4124	-0.384	0.624	0.4124	-0.308	0.645
0.4371	-0.353	0.632	0.4371	-0.281	0.652
0.4621	-0.336	0.635	0.4621	-0.274	0.653
0.4871	-0.312	0.642	0.4871	-0.255	0.658
0.5120	-0.288	0.647	0.5120	-0.237	0.662
0.5371	-0.259	0.655	0.5371	-0.211	0.669
0.5621	-0.232	0.661	0.5621	-0.190	0.674
0.5870	-0.215	0.666	0.5870	-0.175	0.678
0.6122	-0.179	0.674	0.6122	-0.145	0.685
0.6371	-0.166	0.678	0.6371	-0.135	0.688
0.6661	-0.140	0.684	0.6661	-0.115	0.692
0.6869	-0.126	0.688	0.6869	-0.102	0.696
0.7121	-0.103	0.693	0.7121	-0.084	0.699
0.7369	-0.081	0.699	0.7369	-0.062	0.705
0.7620	-0.060	0.704	0.7620	-0.046	0.709
0.7870	-0.034	0.711	0.7870	-0.024	0.714
0.8117	-0.010	0.717	0.8117	-0.004	0.719
0.8370	0.012	0.722	0.8370	0.019	0.725
0.8619	0.038	0.729	0.8619	0.040	0.730
0.8869	0.067	0.736	0.8869	0.068	0.737
0.9120	0.099	0.741	0.9120	0.097	0.744
0.9416	0.136	0.753	0.9416	0.134	0.753

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.700 ALPHA 2.00 REY 1.66*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1941 CM = 0.0075 CDW = 0.0122

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.169	0.679	0.0119	0.515	0.847
0.0371	-0.617	0.568	0.0371	0.034	0.728
0.0623	-0.946	0.487	0.0623	-0.143	0.684
0.0872	-0.831	0.515	0.0872	-0.240	0.662
0.1122	-0.832	0.515	0.1122	-0.286	0.649
0.1372	-0.790	0.525	0.1372	-0.312	0.642
0.1623	-0.788	0.526	0.1623	-0.342	0.635
0.1872	-0.785	0.526	0.1872	-0.366	0.629
0.2122	-0.748	0.536	0.2122	-0.374	0.627
0.2372	-0.689	0.520	0.2372	-0.364	0.632
0.2620	-0.640	0.582	0.2620	-0.351	0.633
0.2872	-0.586	0.576	0.2872	-0.332	0.638
0.3122	-0.553	0.584	0.3122	-0.323	0.640
0.3375	-0.513	0.594	0.3375	-0.305	0.644
0.3618	-0.475	0.643	0.3618	-0.290	0.648
0.3873	-0.447	0.610	0.3873	-0.277	0.651
0.4124	-0.417	0.617	0.4124	-0.265	0.654
0.4371	-0.381	0.626	0.4371	-0.243	0.660
0.4621	-0.363	0.630	0.4621	-0.237	0.662
0.4871	-0.334	0.638	0.4871	-0.222	0.665
0.5120	-0.308	0.644	0.5120	-0.207	0.669
0.5371	-0.277	0.652	0.5371	-0.184	0.675
0.5621	-0.249	0.659	0.5621	-0.164	0.680
0.5870	-0.226	0.664	0.5870	-0.153	0.682
0.6122	-0.192	0.673	0.6122	-0.123	0.690
0.6371	-0.178	0.676	0.6371	-0.115	0.692
0.6651	-0.150	0.683	0.6651	-0.098	0.696
0.6869	-0.134	0.687	0.6869	-0.087	0.699
0.7121	-0.109	0.693	0.7121	-0.071	0.703
0.7369	-0.085	0.699	0.7369	-0.052	0.708
0.7620	-0.061	0.704	0.7620	-0.037	0.712
0.7870	-0.035	0.711	0.7870	-0.016	0.717
0.8117	-0.009	0.717	0.8117	0.003	0.722
0.8370	0.013	0.723	0.8370	0.022	0.726
0.8619	0.040	0.729	0.8619	0.041	0.731
0.8869	0.070	0.737	0.8869	0.067	0.738
0.9120	0.102	0.745	0.9120	0.094	0.744
0.9416	0.140	0.754	0.9416	0.129	0.753

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.702 ALPHA 3.00 REY 1.66*10⁶

INTEGRATED FORCE COEFFICIENTS

CM = 0.2930 CM = 0.0117 CDW = 0.0115

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.359	0.630	0.0119	0.642	0.880
0.0371	-0.766	0.529	0.0371	0.171	0.763
0.0623	-1.229	0.414	0.0623	-0.013	0.716
0.0872	-1.142	0.436	0.0872	-0.119	0.571
0.1122	-1.093	0.448	0.1122	-0.174	0.678
0.1372	-0.914	0.492	0.1372	-0.208	0.670
0.1623	-0.872	0.503	0.1623	-0.245	0.660
0.1872	-0.884	0.500	0.1872	-0.273	0.654
0.2122	-0.837	0.512	0.2122	-0.287	0.650
0.2372	-0.760	0.531	0.2372	-0.284	0.651
0.2620	-0.699	0.546	0.2620	-0.279	0.652
0.2872	-0.635	0.562	0.2872	-0.265	0.655
0.3122	-0.595	0.572	0.3122	-0.263	0.656
0.3375	-0.550	0.583	0.3375	-0.250	0.659
0.3618	-0.510	0.593	0.3618	-0.239	0.661
0.3873	-0.477	0.601	0.3873	-0.228	0.661
0.4124	-0.445	0.609	0.4124	-0.221	0.666
0.4371	-0.406	0.619	0.4371	-0.200	0.671
0.4621	-0.384	0.624	0.4621	-0.199	0.671
0.4871	-0.353	0.632	0.4871	-0.187	0.674
0.5120	-0.326	0.639	0.5120	-0.174	0.675
0.5371	-0.291	0.647	0.5371	-0.153	0.683
0.5621	-0.263	0.655	0.5621	-0.136	0.687
0.5870	-0.239	0.660	0.5870	-0.127	0.689
0.6122	-0.204	0.669	0.6122	-0.099	0.696
0.6371	-0.185	0.674	0.6371	-0.093	0.698
0.6661	-0.156	0.681	0.6661	-0.079	0.701
0.6869	-0.138	0.685	0.6869	-0.069	0.704
0.7121	-0.113	0.692	0.7121	-0.056	0.707
0.7369	-0.087	0.698	0.7369	-0.038	0.711
0.7620	-0.064	0.704	0.7620	-0.025	0.714
0.7870	-0.035	0.711	0.7870	-0.005	0.719
0.8117	-0.010	0.717	0.8117	0.012	0.723
0.8370	0.016	0.724	0.8370	0.029	0.728
0.8619	0.042	0.730	0.8619	0.047	0.732
0.8869	0.074	0.738	0.8869	0.072	0.738
0.9120	0.104	0.746	0.9120	0.097	0.745
0.9416	0.143	0.755	0.9416	0.131	0.753

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.098 ALPHA 4.00 REY 1.65*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.3936 CM = 0.0168 CDW = 0.0127

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.552	0.586	0.0119	0.155	0.937
0.0371	-0.930	0.493	0.0371	0.320	0.795
0.0623	-1.317	0.397	0.0623	0.180	0.745
0.0872	-1.352	0.394	0.0872	-0.014	0.717
0.1122	-1.368	0.385	0.1122	-0.074	0.702
0.1372	-1.356	0.387	0.1372	-0.115	0.692
0.1623	-1.330	0.394	0.1623	-0.156	0.682
0.1872	-1.256	0.412	0.1872	-0.189	0.673
0.2122	-0.864	0.509	0.2122	-0.208	0.668
0.2372	-0.730	0.542	0.2372	-0.212	0.667
0.2623	-0.684	0.553	0.2623	-0.210	0.667
0.2872	-0.636	0.565	0.2872	-0.202	0.669
0.3122	-0.603	0.573	0.3122	-0.203	0.669
0.3375	-0.561	0.583	0.3375	-0.194	0.671
0.3618	-0.523	0.593	0.3618	-0.186	0.673
0.3873	-0.489	0.601	0.3873	-0.180	0.675
0.4124	-0.456	0.609	0.4124	-0.175	0.676
0.4371	-0.419	0.618	0.4371	-0.157	0.681
0.4621	-0.394	0.624	0.4621	-0.158	0.680
0.4871	-0.364	0.632	0.4871	-0.149	0.682
0.5120	-0.334	0.639	0.5120	-0.139	0.685
0.5371	-0.298	0.648	0.5371	-0.121	0.689
0.5621	-0.270	0.655	0.5621	-0.105	0.693
0.5870	-0.245	0.661	0.5870	-0.099	0.695
0.6122	-0.207	0.670	0.6122	-0.072	0.701
0.6371	-0.187	0.675	0.6371	-0.069	0.702
0.6661	-0.158	0.682	0.6661	-0.056	0.705
0.6869	-0.139	0.687	0.6869	-0.049	0.707
0.7121	-0.113	0.693	0.7121	-0.036	0.710
0.7369	-0.086	0.699	0.7369	-0.023	0.714
0.7620	-0.061	0.705	0.7620	-0.009	0.717
0.7870	-0.032	0.713	0.7870	0.008	0.721
0.8117	-0.006	0.719	0.8117	0.024	0.725
0.8370	0.020	0.725	0.8370	0.039	0.729
0.8619	0.047	0.732	0.8619	0.056	0.733
0.8869	0.079	0.740	0.8869	0.078	0.739
0.9120	0.111	0.748	0.9120	0.102	0.745
0.9416	0.149	0.757	0.9416	0.133	0.752

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.698 ALPHA 5.00 REY 1.66×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.4998 CM = 0.0193 CDW = 0.0171

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.754	0.541	0.0119	0.846	0.930
0.0371	-1.106	0.449	0.0371	0.401	0.820
0.0623	-1.359	0.387	0.0623	0.206	0.773
0.0872	-1.454	0.368	0.0872	0.088	0.743
0.1122	-1.511	0.349	0.1122	0.024	0.727
0.1372	-1.490	0.355	0.1372	-0.025	0.715
0.1623	-1.499	0.352	0.1623	-0.065	0.706
0.1872	-1.480	0.357	0.1872	-0.103	0.696
0.2122	-1.513	0.349	0.2122	-0.125	0.691
0.2372	-1.264	0.410	0.2372	-0.135	0.683
0.2620	-0.844	0.514	0.2620	-0.137	0.688
0.2872	-0.703	0.549	0.2872	-0.134	0.688
0.3122	-0.618	0.570	0.3122	-0.137	0.688
0.3375	-0.554	0.585	0.3375	-0.135	0.688
0.3610	-0.511	0.590	0.3610	-0.129	0.693
0.3873	-0.472	0.606	0.3873	-0.128	0.692
0.4124	-0.442	0.613	0.4124	-0.126	0.691
0.4371	-0.406	0.622	0.4371	-0.112	0.694
0.4621	-0.384	0.627	0.4621	-0.114	0.693
0.4871	-0.354	0.634	0.4871	-0.108	0.695
0.5120	-0.327	0.641	0.5120	-0.101	0.696
0.5371	-0.292	0.649	0.5371	-0.084	0.701
0.5621	-0.264	0.656	0.5621	-0.071	0.704
0.5870	-0.239	0.662	0.5870	-0.067	0.705
0.6122	-0.205	0.671	0.6122	-0.042	0.711
0.6371	-0.183	0.676	0.6371	-0.040	0.711
0.6661	-0.156	0.683	0.6661	-0.030	0.714
0.6869	-0.135	0.688	0.6869	-0.026	0.715
0.7121	-0.111	0.694	0.7121	-0.013	0.718
0.7369	-0.082	0.701	0.7369	-0.002	0.720
0.7620	-0.059	0.706	0.7620	0.010	0.723
0.7870	-0.029	0.714	0.7870	0.025	0.727
0.8117	-0.004	0.720	0.8117	0.037	0.730
0.8370	0.023	0.727	0.8370	0.053	0.734
0.8619	0.048	0.733	0.8619	0.068	0.738
0.8869	0.081	0.741	0.8869	0.087	0.743
0.9120	0.110	0.748	0.9120	0.111	0.748
0.9416	0.149	0.756	0.9416	0.140	0.756

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.721 ALPHA 0.00 REY 1.66*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.2065 CM = 0.0001 CDW = 0.0107

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.216	0.763	0.0119	0.201	0.760
0.0371	-0.261	0.640	0.0371	-0.262	0.641
0.0623	-0.449	0.591	0.0623	-0.468	0.585
0.0872	-0.519	0.573	0.0872	-0.532	0.558
0.1122	-0.548	0.566	0.1122	-0.554	0.566
0.1372	-0.553	0.564	0.1372	-0.556	0.565
0.1623	-0.575	0.559	0.1623	-0.584	0.558
0.1872	-0.592	0.554	0.1872	-0.601	0.554
0.2122	-0.582	0.557	0.2122	-0.591	0.556
0.2372	-0.552	0.565	0.2372	-0.559	0.564
0.2620	-0.521	0.573	0.2620	-0.529	0.572
0.2872	-0.482	0.582	0.2872	-0.490	0.582
0.3122	-0.460	0.588	0.3122	-0.468	0.588
0.3375	-0.431	0.596	0.3375	-0.441	0.595
0.3618	-0.401	0.604	0.3618	-0.412	0.602
0.3873	-0.380	0.609	0.3873	-0.389	0.608
0.4124	-0.359	0.614	0.4124	-0.366	0.614
0.4371	-0.329	0.622	0.4371	-0.336	0.622
0.4621	-0.315	0.626	0.4621	-0.321	0.625
0.4871	-0.293	0.632	0.4871	-0.303	0.631
0.5120	-0.270	0.638	0.5120	-0.275	0.637
0.5371	-0.242	0.645	0.5371	-0.247	0.644
0.5621	-0.216	0.651	0.5621	-0.223	0.653
0.5870	-0.201	0.655	0.5870	-0.204	0.655
0.6122	-0.166	0.664	0.6122	-0.170	0.664
0.6371	-0.155	0.667	0.6371	-0.159	0.667
0.6661	-0.130	0.674	0.6661	-0.134	0.673
0.6859	-0.118	0.677	0.6859	-0.123	0.677
0.7121	-0.095	0.683	0.7121	-0.097	0.682
0.7369	-0.074	0.688	0.7369	-0.074	0.688
0.7620	-0.054	0.694	0.7620	-0.054	0.693
0.7870	-0.030	0.700	0.7870	-0.031	0.699
0.8117	-0.005	0.706	0.8117	-0.005	0.706
0.8370	0.016	0.711	0.8370	0.014	0.711
0.8619	0.041	0.718	0.8619	0.041	0.718
0.8859	0.068	0.725	0.8859	0.068	0.725
0.9120	0.100	0.733	0.9120	0.101	0.734
0.9416	0.137	0.743	0.9416	0.136	0.743

AD-A055 004

AERONAUTICAL RESEARCH LABS MELBOURNE (AUSTRALIA)
TRANSONIC WIND TUNNEL TESTS ON A NACA 0012 AEROFOIL, (U)
APR 77 N POLLOCK, B D FAIRLIE

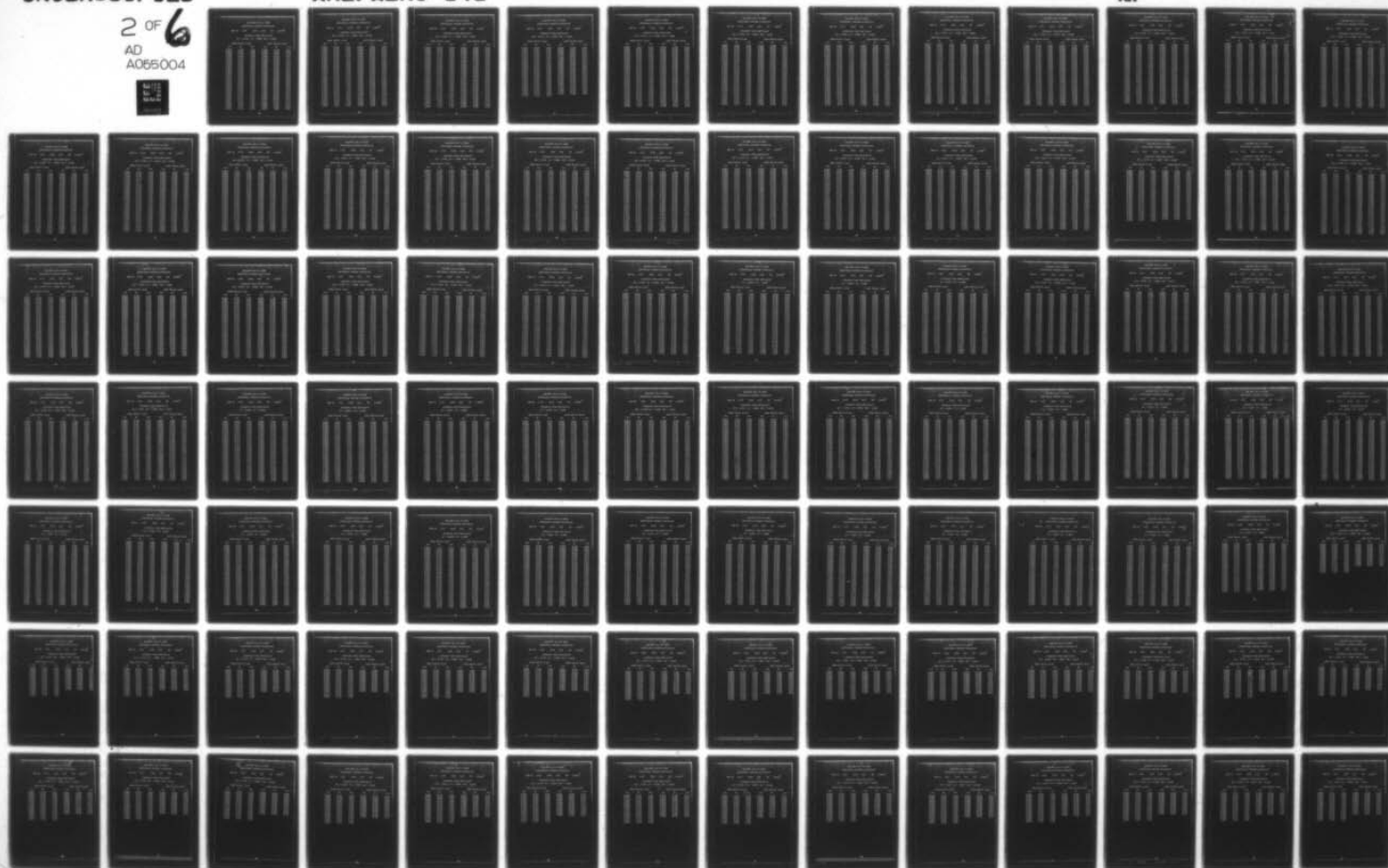
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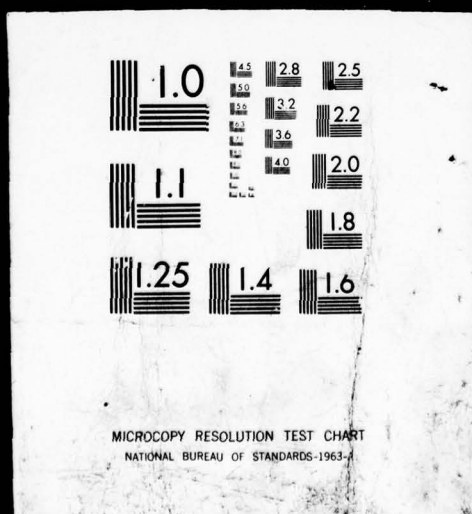


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NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.720 ALPHA 1.00 REYNOLDS 1.65*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2985 CM = 0.0040 CDW = 0.0125

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.044	0.719	0.0119	0.377	0.825
0.0371	-0.421	0.599	0.0371	-0.104	0.661
0.0623	-0.644	0.542	0.0623	-0.290	0.634
0.0872	-0.677	0.533	0.0872	-0.375	0.612
0.1122	-0.699	0.528	0.1122	-0.414	0.602
0.1372	-0.685	0.531	0.1372	-0.432	0.597
0.1623	-0.702	0.527	0.1623	-0.461	0.590
0.1872	-0.712	0.524	0.1872	-0.482	0.584
0.2122	-0.690	0.529	0.2122	-0.483	0.584
0.2372	-0.642	0.541	0.2372	-0.463	0.589
0.2623	-0.602	0.552	0.2623	-0.443	0.594
0.2872	-0.551	0.565	0.2872	-0.413	0.602
0.3122	-0.523	0.572	0.3122	-0.401	0.605
0.3375	-0.484	0.582	0.3375	-0.375	0.611
0.3618	-0.451	0.591	0.3618	-0.355	0.617
0.3873	-0.423	0.598	0.3873	-0.338	0.621
0.4124	-0.397	0.605	0.4124	-0.321	0.625
0.4371	-0.362	0.614	0.4371	-0.292	0.633
0.4621	-0.345	0.617	0.4621	-0.284	0.634
0.4871	-0.318	0.624	0.4871	-0.264	0.639
0.5120	-0.296	0.630	0.5120	-0.245	0.644
0.5371	-0.263	0.638	0.5371	-0.218	0.651
0.5621	-0.237	0.645	0.5621	-0.196	0.657
0.5870	-0.218	0.650	0.5870	-0.181	0.661
0.6122	-0.183	0.659	0.6122	-0.150	0.669
0.6371	-0.166	0.663	0.6371	-0.139	0.672
0.6661	-0.142	0.670	0.6661	-0.118	0.677
0.6869	-0.126	0.674	0.6869	-0.105	0.680
0.7121	-0.104	0.679	0.7121	-0.087	0.685
0.7369	-0.080	0.686	0.7369	-0.065	0.691
0.7620	-0.059	0.691	0.7620	-0.047	0.695
0.7870	-0.032	0.698	0.7870	-0.033	0.700
0.8117	-0.008	0.704	0.8117	-0.003	0.706
0.8370	0.016	0.710	0.8370	0.018	0.712
0.8619	0.041	0.717	0.8619	0.041	0.718
0.8869	0.072	0.725	0.8869	0.069	0.725
0.9120	0.103	0.733	0.9120	0.098	0.732
0.9416	0.142	0.743	0.9416	0.135	0.742

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.721 ALPHA 2.00 REY 1.66*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1981 CM = 0.0085 CDW = 0.0112

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.131	0.674	0.0119	0.516	0.840
0.0371	-0.575	0.559	0.0371	0.036	0.717
0.0623	-1.024	0.444	0.0623	-0.148	0.670
0.0872	-0.810	0.499	0.0872	-0.245	0.645
0.1122	-0.900	0.476	0.1122	-0.292	0.633
0.1372	-0.856	0.467	0.1372	-0.319	0.626
0.1623	-0.824	0.495	0.1623	-0.354	0.617
0.1872	-0.844	0.490	0.1872	-0.380	0.610
0.2122	-0.804	0.501	0.2122	-0.390	0.608
0.2372	-0.722	0.522	0.2372	-0.381	0.611
0.2623	-0.666	0.536	0.2623	-0.368	0.614
0.2872	-0.606	0.552	0.2872	-0.346	0.620
0.3122	-0.569	0.561	0.3122	-0.338	0.622
0.3375	-0.524	0.573	0.3375	-0.319	0.627
0.3618	-0.486	0.583	0.3618	-0.302	0.631
0.3873	-0.454	0.591	0.3873	-0.289	0.634
0.4124	-0.423	0.599	0.4124	-0.277	0.637
0.4371	-0.387	0.608	0.4371	-0.253	0.644
0.4621	-0.369	0.613	0.4621	-0.246	0.645
0.4871	-0.339	0.621	0.4871	-0.230	0.650
0.5120	-0.313	0.628	0.5120	-0.214	0.654
0.5371	-0.279	0.636	0.5371	-0.190	0.660
0.5621	-0.251	0.644	0.5621	-0.170	0.665
0.5870	-0.229	0.649	0.5870	-0.158	0.668
0.6122	-0.194	0.653	0.6122	-0.128	0.676
0.6371	-0.176	0.663	0.6371	-0.120	0.678
0.6661	-0.150	0.670	0.6661	-0.101	0.683
0.6869	-0.132	0.674	0.6869	-0.090	0.685
0.7121	-0.109	0.680	0.7121	-0.074	0.690
0.7369	-0.083	0.687	0.7369	-0.054	0.695
0.7620	-0.061	0.692	0.7620	-0.039	0.698
0.7870	-0.033	0.700	0.7870	-0.018	0.704
0.8117	-0.009	0.706	0.8117	0.002	0.709
0.8370	0.016	0.712	0.8370	0.022	0.714
0.8619	0.043	0.719	0.8619	0.042	0.719
0.8869	0.073	0.727	0.8869	0.069	0.726
0.9120	0.105	0.735	0.9120	0.098	0.734
0.9416	0.143	0.745	0.9416	0.133	0.743

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.718 ALPHA 3.00 REYNOLDS 1.66*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.3028 CM = 0.0133 CDW = 0.0115

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.308	0.630	0.0119	0.643	0.873
0.0371	-0.712	0.527	0.0371	-0.167	0.751
0.0623	-1.170	0.409	0.0623	-0.021	0.733
0.0872	-1.116	0.423	0.0872	-0.128	0.675
0.1122	-1.173	0.408	0.1122	-0.184	0.661
0.1372	-1.077	0.433	0.1372	-0.220	0.652
0.1623	-1.107	0.425	0.1623	-0.259	0.642
0.1872	-1.114	0.424	0.1872	-0.289	0.634
0.2122	-1.032	0.444	0.2122	-0.303	0.629
0.2372	-0.741	0.518	0.2372	-0.300	0.633
0.2623	-0.682	0.534	0.2623	-0.294	0.632
0.2872	-0.631	0.547	0.2872	-0.278	0.636
0.3122	-0.597	0.555	0.3122	-0.276	0.636
0.3375	-0.552	0.567	0.3375	-0.262	0.640
0.3618	-0.513	0.577	0.3618	-0.249	0.643
0.3873	-0.479	0.586	0.3873	-0.240	0.646
0.4124	-0.447	0.594	0.4124	-0.231	0.648
0.4371	-0.407	0.604	0.4371	-0.210	0.653
0.4621	-0.364	0.609	0.4621	-0.208	0.654
0.4871	-0.325	0.617	0.4871	-0.195	0.657
0.5120	-0.325	0.625	0.5120	-0.182	0.661
0.5371	-0.291	0.633	0.5371	-0.159	0.666
0.5621	-0.261	0.641	0.5621	-0.141	0.671
0.5870	-0.239	0.647	0.5870	-0.132	0.673
0.6122	-0.202	0.656	0.6122	-0.104	0.681
0.6371	-0.183	0.661	0.6371	-0.097	0.682
0.6661	-0.154	0.669	0.6661	-0.082	0.686
0.6869	-0.135	0.673	0.6869	-0.073	0.689
0.7121	-0.111	0.680	0.7121	-0.057	0.693
0.7369	-0.085	0.686	0.7369	-0.039	0.697
0.7620	-0.060	0.693	0.7620	-0.025	0.701
0.7870	-0.032	0.700	0.7870	-0.005	0.706
0.8117	-0.005	0.707	0.8117	0.013	0.711
0.8370	0.020	0.713	0.8370	0.031	0.715
0.8619	0.047	0.720	0.8619	0.050	0.720
0.8869	0.078	0.728	0.8869	0.075	0.727
0.9120	0.109	0.736	0.9120	0.101	0.733
0.9416	0.147	0.746	0.9416	0.135	0.742

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.719 ALPHA 4.00 REY 1.66×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.4100 CM = 0.0168 CDW = 0.0140

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.408	0.588	0.0119	0.748	0.920
0.0371	-0.847	0.490	0.0371	0.286	0.751
0.0623	-1.232	0.392	0.0623	0.091	0.730
0.0872	-1.254	0.386	0.0872	-0.022	0.702
0.1122	-1.296	0.375	0.1122	-0.081	0.636
0.1372	-1.299	0.375	0.1372	-0.122	0.676
0.1623	-1.296	0.375	0.1623	-0.164	0.655
0.1872	-1.302	0.374	0.1872	-0.199	0.656
0.2122	-1.308	0.371	0.2122	-0.217	0.651
0.2372	-1.311	0.371	0.2372	-0.220	0.650
0.2623	-1.131	0.417	0.2623	-0.220	0.650
0.2872	-0.733	0.519	0.2872	-0.209	0.652
0.3122	-0.605	0.552	0.3122	-0.211	0.652
0.3375	-0.535	0.570	0.3375	-0.201	0.654
0.3618	-0.491	0.582	0.3618	-0.185	0.659
0.3873	-0.456	0.591	0.3873	-0.186	0.658
0.4124	-0.428	0.598	0.4124	-0.182	0.660
0.4371	-0.392	0.607	0.4371	-0.164	0.664
0.4621	-0.371	0.612	0.4621	-0.163	0.664
0.4871	-0.343	0.619	0.4871	-0.154	0.667
0.5120	-0.318	0.625	0.5120	-0.144	0.669
0.5371	-0.283	0.634	0.5371	-0.124	0.674
0.5621	-0.257	0.641	0.5621	-0.109	0.678
0.5870	-0.232	0.648	0.5870	-0.101	0.680
0.6122	-0.198	0.656	0.6122	-0.075	0.687
0.6371	-0.178	0.661	0.6371	-0.070	0.688
0.6661	-0.150	0.669	0.6661	-0.058	0.692
0.6869	-0.130	0.674	0.6869	-0.049	0.694
0.7121	-0.107	0.679	0.7121	-0.036	0.698
0.7369	-0.079	0.687	0.7369	-0.021	0.722
0.7620	-0.056	0.693			

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.740 ALPHA 0.00 REY 1.65×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0057 CM = -0.0000 CDW = 0.0106

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.253	0.757	0.0119	0.218	0.753
0.0371	-0.244	0.629	0.0371	-0.245	0.629
0.0623	-0.452	0.574	0.0623	-0.472	0.568
0.0872	-0.526	0.554	0.0872	-0.565	0.538
0.1122	-0.565	0.543	0.1122	-0.574	0.541
0.1372	-0.571	0.542	0.1372	-0.578	0.540
0.1623	-0.602	0.534	0.1623	-0.611	0.531
0.1872	-0.627	0.527	0.1872	-0.634	0.525
0.2122	-0.622	0.528	0.2122	-0.631	0.527
0.2372	-0.584	0.538	0.2372	-0.591	0.537
0.2620	-0.552	0.547	0.2620	-0.559	0.546
0.2872	-0.506	0.559	0.2872	-0.513	0.558
0.3122	-0.466	0.565	0.3122	-0.492	0.563
0.3375	-0.451	0.574	0.3375	-0.458	0.573
0.3618	-0.423	0.582	0.3618	-0.428	0.580
0.3873	-0.395	0.589	0.3873	-0.401	0.588
0.4124	-0.373	0.595	0.4124	-0.378	0.594
0.4371	-0.358	0.604	0.4371	-0.341	0.604
0.4621	-0.325	0.608	0.4621	-0.330	0.607
0.4871	-0.299	0.615	0.4871	-0.304	0.614
0.5120	-0.278	0.620	0.5120	-0.281	0.620
0.5371	-0.247	0.629	0.5371	-0.251	0.628
0.5621	-0.224	0.635	0.5621	-0.228	0.634
0.5870	-0.204	0.640	0.5870	-0.208	0.640
0.6122	-0.172	0.649	0.6122	-0.176	0.646
0.6371	-0.156	0.653	0.6371	-0.159	0.652
0.6661	-0.133	0.659	0.6661	-0.137	0.658
0.6869	-0.117	0.663	0.6869	-0.119	0.663
0.7121	-0.097	0.668	0.7121	-0.099	0.669
0.7369	-0.072	0.675	0.7369	-0.073	0.676
0.7620	-0.053	0.680	0.7620	-0.055	0.680
0.7870	-0.027	0.687	0.7870	-0.029	0.687
0.8117	-0.005	0.693	0.8117	-0.005	0.694
0.8370	0.019	0.700	0.8370	0.018	0.700
0.8619	0.042	0.706	0.8619	0.042	0.706
0.8869	0.074	0.714	0.8869	0.074	0.715
0.9120	0.102	0.722	0.9120	0.104	0.723
0.9416	0.142	0.732	0.9416	0.142	0.733

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.742 ALPHA 1.00 REY 1.65×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0992 CM = 0.0043 CDW = 0.0141

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.075	0.714	0.0119	0.383	0.797
0.0371	-0.392	0.589	0.0371	-0.099	0.668
0.0623	-0.665	0.516	0.0623	-0.292	0.617
0.0872	-0.679	0.512	0.0872	-0.385	0.592
0.1122	-0.729	0.499	0.1122	-0.429	0.580
0.1372	-0.713	0.503	0.1372	-0.447	0.575
0.1623	-0.732	0.498	0.1623	-0.483	0.566
0.1872	-0.771	0.488	0.1872	-0.510	0.558
0.2122	-0.758	0.492	0.2122	-0.513	0.557
0.2372	-0.684	0.512	0.2372	-0.492	0.563
0.2620	-0.620	0.526	0.2620	-0.470	0.569
0.2872	-0.572	0.541	0.2872	-0.438	0.578
0.3122	-0.540	0.550	0.3122	-0.423	0.581
0.3375	-0.501	0.560	0.3375	-0.398	0.588
0.3618	-0.464	0.570	0.3618	-0.372	0.595
0.3873	-0.435	0.578	0.3873	-0.354	0.600
0.4124	-0.405	0.586	0.4124	-0.335	0.605
0.4371	-0.369	0.596	0.4371	-0.306	0.613
0.4621	-0.351	0.601	0.4621	-0.295	0.615
0.4871	-0.325	0.608	0.4871	-0.272	0.621
0.5120	-0.299	0.615	0.5120	-0.252	0.627
0.5371	-0.268	0.623	0.5371	-0.225	0.634
0.5621	-0.239	0.631	0.5621	-0.202	0.640
0.5870	-0.220	0.636	0.5870	-0.187	0.644
0.6122	-0.184	0.646	0.6122	-0.155	0.653
0.6371	-0.168	0.650	0.6371	-0.143	0.656
0.6661	-0.142	0.657	0.6661	-0.120	0.662
0.6869	-0.126	0.661	0.6869	-0.107	0.666
0.7121	-0.103	0.668	0.7121	-0.087	0.670
0.7369	-0.078	0.674	0.7369	-0.066	0.676
0.7620	-0.056	0.680	0.7620	-0.046	0.681
0.7870	-0.031	0.687	0.7870	-0.024	0.687
0.8117	-0.006	0.694	0.8117	-0.001	0.693
0.8370	0.018	0.700	0.8370	0.020	0.699
0.8619	0.043	0.707	0.8619	0.044	0.705
0.8869	0.074	0.715	0.8869	0.072	0.713
0.9120	0.106	0.723	0.9120	0.102	0.721
0.9416	0.144	0.733	0.9416	0.139	0.731

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.741 ALPHA 2.00 REY 1.65×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2052 CM = 0.0095 CDW = 0.0119

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.078	0.673	0.0119	0.518	0.832
0.0371	-0.523	0.555	0.0371	0.038	0.734
0.0623	-0.979	0.433	0.0623	-0.153	0.653
0.0872	-0.808	0.457	0.0872	-0.255	0.626
0.1122	-0.644	0.469	0.1122	-0.306	0.612
0.1372	-0.900	0.454	0.1372	-0.334	0.605
0.1623	-0.898	0.455	0.1623	-0.374	0.594
0.1872	-0.904	0.453	0.1872	-0.405	0.586
0.2122	-0.951	0.440	0.2122	-0.414	0.583
0.2372	-0.953	0.440	0.2372	-0.402	0.586
0.2620	-0.808	0.479	0.2620	-0.389	0.590
0.2872	-0.603	0.533	0.2872	-0.365	0.596
0.3122	-0.561	0.545	0.3122	-0.356	0.599
0.3375	-0.523	0.555	0.3375	-0.336	0.604
0.3618	-0.468	0.564	0.3618	-0.318	0.609
0.3873	-0.458	0.572	0.3873	-0.303	0.610
0.4124	-0.427	0.580	0.4124	-0.289	0.617
0.4371	-0.392	0.590	0.4371	-0.262	0.624
0.4621	-0.370	0.596	0.4621	-0.257	0.625
0.4871	-0.341	0.603	0.4871	-0.239	0.630
0.5120	-0.312	0.611	0.5120	-0.221	0.635
0.5371	-0.281	0.619	0.5371	-0.197	0.641
0.5621	-0.251	0.627	0.5621	-0.178	0.646
0.5870	-0.230	0.633	0.5870	-0.163	0.650
0.6122	-0.193	0.643	0.6122	-0.132	0.659
0.6371	-0.176	0.647	0.6371	-0.122	0.661
0.6661	-0.147	0.655	0.6661	-0.103	0.666
0.6869	-0.131	0.660	0.6869	-0.091	0.670
0.7121	-0.105	0.665	0.7121	-0.074	0.675
0.7369	-0.080	0.672	0.7369	-0.054	0.680
0.7620	-0.056	0.679	0.7620	-0.038	0.684
0.7870	-0.029	0.686	0.7870	-0.016	0.690
0.8117	-0.003	0.693	0.8117	0.005	0.696
0.8370	0.021	0.699	0.8370	0.025	0.701
0.8619	0.048	0.707	0.8619	0.046	0.707
0.8869	0.079	0.715	0.8869	0.073	0.714
0.9120	0.111	0.723	0.9120	0.100	0.721
0.9416	0.148	0.733	0.9416	0.137	0.731

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.740 ALPHA 3.00 REY 1.65×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3149 CM = 0.0135 CDW = 0.0134

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.239	0.631	0.0119	0.637	0.865
0.0371	-0.641	0.524	0.0371	0.163	0.739
0.0623	-1.080	0.407	0.0623	-0.025	0.659
0.0872	-1.057	0.413	0.0872	-0.135	0.660
0.1122	-1.114	0.398	0.1122	-0.192	0.645
0.1372	-1.092	0.404	0.1372	-0.229	0.635
0.1623	-1.083	0.406	0.1623	-0.268	0.624
0.1872	-1.111	0.399	0.1872	-0.303	0.615
0.2122	-1.145	0.390	0.2122	-0.317	0.611
0.2372	-1.145	0.390	0.2372	-0.314	0.612
0.2620	-1.157	0.386	0.2620	-0.306	0.614
0.2872	-0.937	0.445	0.2872	-0.291	0.618
0.3122	-0.626	0.528	0.3122	-0.286	0.619
0.3375	-0.524	0.555	0.3375	-0.274	0.622
0.3618	-0.475	0.568	0.3618	-0.259	0.626
0.3873	-0.444	0.577	0.3873	-0.250	0.629
0.4124	-0.416	0.584	0.4124	-0.239	0.632
0.4371	-0.385	0.592	0.4371	-0.219	0.637
0.4621	-0.365	0.598	0.4621	-0.215	0.638
0.4871	-0.340	0.605	0.4871	-0.232	0.642
0.5120	-0.311	0.612	0.5120	-0.186	0.646
0.5371	-0.280	0.621	0.5371	-0.164	0.652
0.5621	-0.251	0.628	0.5621	-0.145	0.657
0.5870	-0.228	0.634	0.5870	-0.137	0.659
0.6122	-0.194	0.643	0.6122	-0.107	0.667
0.6371	-0.174	0.649	0.6371	-0.100	0.669
0.6661	-0.146	0.656	0.6661	-0.083	0.673
0.6869	-0.129	0.661	0.6869	-0.073	0.676
0.7121	-0.104	0.668	0.7121	-0.057	0.680
0.7369	-0.078	0.675	0.7369	-0.040	0.685
0.7620	-0.054	0.681	0.7620	-0.024	0.689
0.7870	-0.027	0.688	0.7870	-0.004	0.694
0.8117	0.000	0.695	0.8117	0.015	0.699
0.8370	0.025	0.702	0.8370	0.033	0.704
0.8619	0.051	0.709	0.8619	0.053	0.710
0.8869	0.082	0.717	0.8869	0.078	0.716
0.9120	0.114	0.726	0.9120	0.105	0.723
0.9416	0.151	0.736	0.9416	0.139	0.732

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.739 ALPHA 4.00 REY 1.65×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.4219 CM = 0.0150 CDW = 0.0187

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.391	0.591	0.0119	0.738	0.892
0.0371	-0.775	0.489	0.0371	0.277	0.770
0.0623	-1.140	0.392	0.0623	0.082	0.718
0.0872	-1.174	0.383	0.0872	-0.023	0.690
0.1122	-1.218	0.371	0.1122	-0.090	0.672
0.1372	-1.232	0.367	0.1372	-0.131	0.661
0.1623	-1.235	0.367	0.1623	-0.173	0.650
0.1872	-1.237	0.361	0.1872	-0.211	0.643
0.2122	-1.207	0.353	0.2122	-0.230	0.635
0.2372	-1.270	0.357	0.2372	-0.234	0.634
0.2620	-1.312	0.346	0.2620	-0.232	0.635
0.2872	-1.290	0.332	0.2872	-0.221	0.637
0.3122	-1.124	0.396	0.3122	-0.223	0.637
0.3375	-0.716	0.505	0.3375	-0.214	0.639
0.3618	-0.590	0.538	0.3618	-0.204	0.642
0.3873	-0.510	0.560	0.3873	-0.197	0.644
0.4124	-0.445	0.577	0.4124	-0.190	0.646
0.4371	-0.392	0.591	0.4371	-0.172	0.650
0.4621	-0.352	0.601	0.4621	-0.171	0.651
0.4871	-0.321	0.610	0.4871	-0.161	0.653
0.5120	-0.292	0.618	0.5120	-0.149	0.656
0.5371	-0.262	0.626	0.5371	-0.130	0.662
0.5621	-0.235	0.633	0.5621	-0.113	0.666
0.5870	-0.212	0.639	0.5870	-0.106	0.668
0.6122	-0.180	0.647	0.6122	-0.079	0.675
0.6371	-0.161	0.653	0.6371	-0.075	0.676
0.6661	-0.134	0.660	0.6661	-0.060	0.680
0.6869	-0.117	0.664	0.6869	-0.052	0.682
0.7121	-0.093	0.671	0.7121	-0.038	0.686
0.7369	-0.068	0.677	0.7369	-0.023	0.690
0.7620	-0.045	0.683	0.7620	-0.010	0.693
0.7870	-0.020	0.690	0.7870	0.010	0.699
0.8117	0.006	0.697	0.8117	0.026	0.703
0.8370	0.031	0.704	0.8370	0.043	0.707
0.8619	0.057	0.710	0.8619	0.061	0.712
0.8869	0.087	0.718	0.8869	0.085	0.719
0.9120	0.116	0.726	0.9120	0.111	0.725
0.9416	0.153	0.736	0.9416	0.143	0.734

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.759 ALPHA 0.00 REYNOLDS 1.64*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.0065 CM = -0.0002 CDW = 0.0109

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.252	0.752	0.0119	0.238	0.748
0.0371	-0.250	0.619	0.0371	-0.229	0.619
0.0623	-0.453	0.558	0.0623	-0.475	0.551
0.0872	-0.533	0.536	0.0872	-0.594	0.518
0.1122	-0.586	0.521	0.1122	-0.595	0.516
0.1372	-0.598	0.518	0.1372	-0.586	0.515
0.1623	-0.635	0.507	0.1623	-0.643	0.505
0.1872	-0.683	0.494	0.1872	-0.694	0.491
0.2122	-0.704	0.488	0.2122	-0.720	0.483
0.2372	-0.651	0.503	0.2372	-0.673	0.496
0.2620	-0.598	0.517	0.2620	-0.611	0.513
0.2872	-0.540	0.533	0.2872	-0.549	0.530
0.3122	-0.513	0.541	0.3122	-0.521	0.538
0.3375	-0.476	0.551	0.3375	-0.486	0.548
0.3618	-0.442	0.560	0.3618	-0.452	0.557
0.3873	-0.416	0.567	0.3873	-0.422	0.565
0.4124	-0.389	0.575	0.4124	-0.394	0.573
0.4371	-0.353	0.585	0.4371	-0.359	0.583
0.4621	-0.338	0.588	0.4621	-0.341	0.587
0.4871	-0.313	0.596	0.4871	-0.316	0.594
0.5120	-0.287	0.602	0.5120	-0.289	0.601
0.5371	-0.256	0.611	0.5371	-0.261	0.609
0.5621	-0.228	0.619	0.5621	-0.233	0.617
0.5870	-0.210	0.624	0.5870	-0.213	0.622
0.6122	-0.175	0.633	0.6122	-0.179	0.632
0.6371	-0.160	0.637	0.6371	-0.164	0.636
0.6661	-0.134	0.644	0.6661	-0.137	0.643
0.6869	-0.119	0.649	0.6869	-0.121	0.648
0.7121	-0.097	0.655	0.7121	-0.098	0.654
0.7369	-0.073	0.661	0.7369	-0.073	0.661
0.7620	-0.052	0.667	0.7620	-0.053	0.666
0.7870	-0.028	0.674	0.7870	-0.022	0.675
0.8117	-0.003	0.681	0.8117	-0.002	0.680
0.8370	0.021	0.687	0.8370	0.020	0.686
0.8619	0.046	0.694	0.8619	0.047	0.694
0.8869	0.075	0.702	0.8869	0.076	0.702
0.9120	0.106	0.711	0.9120	0.108	0.711
0.9416	0.145	0.722	0.9416	0.145	0.721

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.761 ALPHA 1.00 REY 1.65*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1069 CM = 0.0052 CDW = 0.0130

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.116	0.713	0.0119	0.590	3.790
0.0371	-0.350	0.584	0.0371	-0.093	0.655
0.0623	-0.704	0.486	0.0623	-0.295	0.601
0.0872	-0.653	0.500	0.0872	-0.393	0.571
0.1122	-0.743	0.475	0.1122	-0.446	0.550
0.1372	-0.760	0.470	0.1372	-0.468	0.553
0.1623	-0.765	0.469	0.1623	-0.507	0.542
0.1872	-0.764	0.464	0.1872	-0.544	0.532
0.2122	-0.840	0.447	0.2122	-0.550	0.530
0.2372	-0.868	0.440	0.2372	-0.529	0.536
0.2623	-0.885	0.435	0.2623	-0.505	0.543
0.2872	-0.751	0.472	0.2872	-0.466	0.554
0.3122	-0.537	0.531	0.3122	-0.448	0.558
0.3375	-0.489	0.545	0.3375	-0.423	0.566
0.3618	-0.460	0.553	0.3618	-0.393	0.574
0.3873	-0.433	0.560	0.3873	-0.370	0.580
0.4124	-0.406	0.568	0.4124	-0.349	0.586
0.4371	-0.371	0.576	0.4371	-0.318	0.594
0.4621	-0.352	0.583	0.4621	-0.307	0.597
0.4871	-0.325	0.590	0.4871	-0.283	0.604
0.5120	-0.297	0.598	0.5120	-0.261	0.610
0.5371	-0.266	0.607	0.5371	-0.232	0.618
0.5621	-0.238	0.614	0.5621	-0.208	0.625
0.5870	-0.219	0.620	0.5870	-0.191	0.629
0.6122	-0.183	0.630	0.6122	-0.158	0.639
0.6371	-0.166	0.634	0.6371	-0.145	0.642
0.6661	-0.143	0.641	0.6661	-0.123	0.646
0.6869	-0.124	0.646	0.6869	-0.108	0.652
0.7121	-0.100	0.653	0.7121	-0.087	0.659
0.7369	-0.075	0.660	0.7369	-0.065	0.664
0.7620	-0.052	0.666	0.7620	-0.047	0.669
0.7870	-0.028	0.673	0.7870	-0.022	0.676
0.8117	-0.001	0.680	0.8117	0.001	0.682
0.8370	0.023	0.687	0.8370	0.022	0.688
0.8619	0.049	0.694	0.8619	0.047	0.695
0.8869	0.078	0.702	0.8869	0.076	0.703
0.9120	0.110	0.711	0.9120	0.106	0.711
0.9416	0.148	0.721	0.9416	0.143	0.721

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.701 ALPHA 2.00 REY 1.65×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2144 CM = 0.0096 CDW = 0.0132

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.335	0.672	0.0119	0.519	0.825
0.0371	-0.472	0.552	0.0371	0.040	0.692
0.0623	-0.921	0.428	0.0623	-0.155	0.639
0.0872	-0.850	0.447	0.0872	-0.260	0.629
0.1122	-0.841	0.450	0.1122	-0.315	0.594
0.1372	-0.862	0.438	0.1372	-0.346	0.586
0.1623	-0.913	0.430	0.1623	-0.387	0.574
0.1872	-0.924	0.427	0.1872	-0.423	0.564
0.2122	-0.958	0.419	0.2122	-0.437	0.563
0.2372	-0.971	0.415	0.2372	-0.427	0.563
0.2620	-1.000	0.407	0.2620	-0.412	0.567
0.2872	-0.995	0.408	0.2872	-0.386	0.575
0.3122	-0.909	0.432	0.3122	-0.376	0.577
0.3375	-0.573	0.525	0.3375	-0.355	0.583
0.3618	-0.472	0.552	0.3618	-0.336	0.588
0.3873	-0.427	0.565	0.3873	-0.319	0.593
0.4124	-0.400	0.572	0.4124	-0.302	0.598
0.4371	-0.371	0.580	0.4371	-0.276	0.605
0.4621	-0.352	0.586	0.4621	-0.268	0.607
0.4871	-0.327	0.593	0.4871	-0.248	0.613
0.5120	-0.300	0.600	0.5120	-0.233	0.618
0.5371	-0.271	0.608	0.5371	-0.204	0.625
0.5621	-0.243	0.616	0.5621	-0.181	0.631
0.5870	-0.222	0.622	0.5870	-0.166	0.635
0.6122	-0.187	0.631	0.6122	-0.136	0.644
0.6371	-0.169	0.636	0.6371	-0.125	0.647
0.6661	-0.142	0.644	0.6661	-0.105	0.652
0.6869	-0.125	0.648	0.6869	-0.093	0.656
0.7121	-0.100	0.655	0.7121	-0.074	0.661
0.7369	-0.076	0.662	0.7369	-0.053	0.667
0.7620	-0.052	0.669	0.7620	-0.037	0.671
0.7870	-0.027	0.676	0.7870	-0.014	0.677
0.8117	0.000	0.683	0.8117	0.007	0.683
0.8370	0.024	0.690	0.8370	0.028	0.689
0.8619	0.052	0.697	0.8619	0.050	0.695
0.8869	0.082	0.706	0.8869	0.078	0.703
0.9120	0.113	0.714	0.9120	0.107	0.711
0.9416	0.151	0.725	0.9416	0.144	0.721

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.752 ALPHA 3.00 REY 1.64×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3227 CM = 0.0118 CDW = 0.0170

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.169	0.636	0.0119	0.632	0.856
0.0371	-0.574	0.525	0.0371	0.159	0.725
0.0623	-1.102	0.407	0.0623	-0.034	0.672
0.0872	-0.993	0.409	0.0872	-0.145	0.641
0.1122	-1.049	0.394	0.1122	-0.205	0.625
0.1372	-1.045	0.395	0.1372	-0.243	0.614
0.1623	-1.047	0.394	0.1623	-0.286	0.602
0.1872	-1.063	0.390	0.1872	-0.322	0.592
0.2122	-1.107	0.378	0.2122	-0.339	0.587
0.2372	-1.128	0.372	0.2372	-0.337	0.588
0.2623	-1.144	0.368	0.2623	-0.330	0.590
0.2872	-1.153	0.370	0.2872	-0.313	0.595
0.3122	-1.144	0.367	0.3122	-0.307	0.596
0.3375	-1.034	0.398	0.3375	-0.292	0.600
0.3618	-0.660	0.501	0.3618	-0.276	0.605
0.3873	-0.503	0.544	0.3873	-0.266	0.608
0.4124	-0.428	0.565	0.4124	-0.254	0.611
0.4371	-0.380	0.578	0.4371	-0.232	0.617
0.4621	-0.343	0.568	0.4621	-0.226	0.619
0.4871	-0.311	0.597	0.4871	-0.211	0.623
0.5120	-0.282	0.605	0.5120	-0.195	0.627
0.5371	-0.254	0.613	0.5371	-0.173	0.633
0.5621	-0.226	0.620	0.5621	-0.154	0.639
0.5873	-0.206	0.626	0.5873	-0.142	0.642
0.6122	-0.174	0.634	0.6122	-0.113	0.650
0.6371	-0.156	0.639	0.6371	-0.104	0.652
0.6661	-0.130	0.647	0.6661	-0.086	0.657
0.6869	-0.113	0.651	0.6869	-0.076	0.660
0.7121	-0.090	0.657	0.7121	-0.059	0.665
0.7369	-0.067	0.664	0.7369	-0.040	0.670
0.7623	-0.043	0.670	0.7623	-0.024	0.675
0.7873	-0.018	0.677	0.7873	-0.004	0.680
0.8117	0.008	0.685	0.8117	0.016	0.686
0.8370	0.032	0.691	0.8370	0.035	0.691
0.8619	0.058	0.698	0.8619	0.056	0.697
0.8869	0.087	0.706	0.8869	0.082	0.704
0.9120	0.118	0.715	0.9120	0.110	0.712
0.9416	0.155	0.725	0.9416	0.145	0.721

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.758 ALPHA 4.00 REY 1.64×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.4239 CII = 0.0112 CDW = 0.0236

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.307	0.598	0.0119	0.126	0.883
0.0371	-0.694	0.492	0.0371	0.267	0.756
0.0623	-1.057	0.392	0.0623	0.074	0.703
0.0872	-1.096	0.361	0.0872	-0.039	0.672
0.1122	-1.141	0.309	0.1122	-0.103	0.654
0.1372	-1.161	0.363	0.1372	-0.145	0.642
0.1623	-1.167	0.362	0.1623	-0.189	0.630
0.1872	-1.191	0.355	0.1872	-0.228	0.620
0.2122	-1.226	0.345	0.2122	-0.249	0.614
0.2372	-1.229	0.345	0.2372	-0.251	0.613
0.2623	-1.255	0.338	0.2623	-0.250	0.614
0.2872	-1.248	0.340	0.2872	-0.239	0.617
0.3122	-1.266	0.335	0.3122	-0.239	0.617
0.3375	-1.289	0.330	0.3375	-0.229	0.620
0.3613	-0.813	0.459	0.3613	-0.218	0.623
0.3873	-0.628	0.510	0.3873	-0.212	0.624
0.4124	-0.552	0.531	0.4124	-0.204	0.626
0.4371	-0.490	0.548	0.4371	-0.187	0.631
0.4621	-0.436	0.563	0.4621	-0.186	0.632
0.4871	-0.379	0.579	0.4871	-0.173	0.635
0.5123	-0.331	0.592	0.5123	-0.160	0.639
0.5371	-0.288	0.604	0.5371	-0.141	0.644
0.5621	-0.245	0.615	0.5621	-0.123	0.649
0.5873	-0.212	0.625	0.5873	-0.115	0.651
0.6122	-0.175	0.635	0.6122	-0.089	0.658
0.6371	-0.149	0.642	0.6371	-0.083	0.660
0.6661	-0.119	0.650	0.6661	-0.068	0.664
0.6869	-0.101	0.655	0.6869	-0.057	0.667
0.7121	-0.076	0.662	0.7121	-0.043	0.671
0.7369	-0.054	0.668	0.7369	-0.028	0.675
0.7623	-0.031	0.674	0.7623	-0.014	0.679
0.7873	-0.007	0.681	0.7873	0.005	0.684
0.8117	0.018	0.688	0.8117	0.024	0.689
0.8373	0.040	0.694	0.8373	0.041	0.694
0.8619	0.066	0.701	0.8619	0.058	0.699
0.8869	0.093	0.708	0.8869	0.082	0.706
0.9120	0.122	0.716	0.9120	0.108	0.713
0.9416	0.155	0.726	0.9416	0.140	0.722

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.790 ALPHA 0.00 REY 1.61*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.0372 CM = -0.0000 CDW = 0.0106

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.277	0.747	0.0119	0.264	0.744
0.0371	-0.202	0.611	0.0371	-0.201	0.611
0.0623	-0.446	0.541	0.0623	-0.467	0.535
0.0872	-0.519	0.520	0.0872	-0.577	0.533
0.1122	-0.591	0.500	0.1122	-0.595	0.498
0.1372	-0.603	0.497	0.1372	-0.608	0.494
0.1623	-0.631	0.489	0.1623	-0.634	0.487
0.1872	-0.689	0.472	0.1872	-0.693	0.476
0.2122	-0.747	0.455	0.2122	-0.752	0.453
0.2372	-0.776	0.447	0.2372	-0.781	0.445
0.2623	-0.799	0.441	0.2623	-0.806	0.438
0.2872	-0.774	0.446	0.2872	-0.800	0.439
0.3122	-0.573	0.525	0.3122	-0.712	0.465
0.3375	-0.463	0.536	0.3375	-0.485	0.529
0.3618	-0.433	0.545	0.3618	-0.431	0.545
0.3873	-0.413	0.551	0.3873	-0.409	0.551
0.4124	-0.388	0.558	0.4124	-0.386	0.558
0.4371	-0.356	0.567	0.4371	-0.357	0.566
0.4621	-0.339	0.572	0.4621	-0.339	0.571
0.4871	-0.313	0.579	0.4871	-0.314	0.576
0.5120	-0.286	0.587	0.5120	-0.287	0.586
0.5371	-0.256	0.596	0.5371	-0.259	0.594
0.5621	-0.228	0.603	0.5621	-0.231	0.602
0.5873	-0.210	0.609	0.5873	-0.211	0.608
0.6122	-0.174	0.619	0.6122	-0.176	0.618
0.6371	-0.159	0.623	0.6371	-0.161	0.622
0.6661	-0.133	0.631	0.6661	-0.135	0.629
0.6859	-0.118	0.635	0.6859	-0.119	0.634
0.7121	-0.094	0.642	0.7121	-0.094	0.641
0.7369	-0.072	0.648	0.7369	-0.070	0.648
0.7623	-0.049	0.655	0.7623	-0.048	0.654
0.7873	-0.024	0.662	0.7873	-0.023	0.661
0.8117	0.002	0.669	0.8117	0.003	0.669
0.8373	0.025	0.676	0.8373	0.024	0.675
0.8619	0.051	0.683	0.8619	0.052	0.683
0.8869	0.080	0.691	0.8869	0.080	0.691
0.9120	0.112	0.701	0.9120	0.113	0.700
0.9416	0.150	0.711	0.9416	0.149	0.710

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.780 ALPHA 1.00 REY 1.61×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1138 CM = 0.0045 CDW = 0.0146

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.147	0.711	0.0119	0.400	0.703
0.0371	-0.314	0.579	0.0371	-0.081	0.646
0.0623	-0.710	0.466	0.0623	-0.295	0.585
0.0872	-0.611	0.494	0.0872	-0.394	0.557
0.1122	-0.714	0.465	0.1122	-0.458	0.539
0.1372	-0.742	0.457	0.1372	-0.484	0.531
0.1623	-0.768	0.449	0.1623	-0.531	0.518
0.1872	-0.767	0.444	0.1872	-0.585	0.502
0.2122	-0.834	0.431	0.2122	-0.618	0.493
0.2372	-0.864	0.422	0.2372	-0.594	0.500
0.2620	-0.897	0.413	0.2620	-0.556	0.511
0.2872	-0.898	0.412	0.2872	-0.503	0.526
0.3122	-0.920	0.406	0.3122	-0.480	0.532
0.3375	-0.883	0.417	0.3375	-0.446	0.542
0.3618	-0.684	0.496	0.3618	-0.417	0.550
0.3873	-0.422	0.548	0.3873	-0.390	0.558
0.4124	-0.374	0.562	0.4124	-0.366	0.565
0.4371	-0.342	0.571	0.4371	-0.333	0.574
0.4621	-0.326	0.575	0.4621	-0.318	0.578
0.4871	-0.304	0.582	0.4871	-0.291	0.585
0.5120	-0.282	0.588	0.5120	-0.269	0.592
0.5371	-0.255	0.596	0.5371	-0.240	0.600
0.5621	-0.230	0.603	0.5621	-0.214	0.607
0.5870	-0.210	0.609	0.5870	-0.196	0.612
0.6122	-0.176	0.618	0.6122	-0.162	0.622
0.6371	-0.160	0.623	0.6371	-0.147	0.626
0.6661	-0.134	0.630	0.6661	-0.123	0.633
0.6869	-0.117	0.635	0.6869	-0.109	0.637
0.7121	-0.094	0.642	0.7121	-0.087	0.643
0.7369	-0.071	0.648	0.7369	-0.064	0.649
0.7620	-0.048	0.655	0.7620	-0.044	0.655
0.7870	-0.022	0.662	0.7870	-0.020	0.662
0.8117	0.003	0.669	0.8117	0.004	0.669
0.8370	0.028	0.676	0.8370	0.027	0.675
0.8619	0.053	0.684	0.8619	0.051	0.682
0.8869	0.084	0.692	0.8869	0.081	0.691
0.9120	0.114	0.701	0.9120	0.112	0.700
0.9416	0.153	0.712	0.9416	0.150	0.711

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.780 ALPHA 2.00 REYNOLDS 1.61*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2245 CM = 0.0063 CDW = 0.0152

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.025	0.676	0.0119	0.015	0.815
0.0371	-0.409	2.552	0.0371	0.036	0.679
0.0623	-0.852	2.426	0.0623	-0.168	0.620
0.0872	-0.788	2.444	0.0872	-0.270	0.591
0.1122	-0.819	2.435	0.1122	-0.331	0.574
0.1372	-0.838	2.430	0.1372	-0.364	0.564
0.1623	-0.860	2.418	0.1623	-0.409	0.552
0.1872	-0.902	2.411	0.1872	-0.450	0.540
0.2122	-0.936	2.402	0.2122	-0.473	0.535
0.2372	-0.952	2.397	0.2372	-0.459	0.538
0.2620	-0.979	2.389	0.2620	-0.444	0.542
0.2872	-0.964	2.388	0.2872	-0.414	0.551
0.3122	-1.011	2.380	0.3122	-0.402	0.554
0.3375	-1.004	2.382	0.3375	-0.378	0.561
0.3618	-1.009	2.381	0.3618	-0.355	0.568
0.3873	-0.788	2.444	0.3873	-0.337	0.573
0.4124	-0.498	2.527	0.4124	-0.319	0.578
0.4371	-0.397	2.555	0.4371	-0.289	0.587
0.4621	-0.359	2.572	0.4621	-0.280	0.590
0.4871	-0.301	2.563	0.4871	-0.257	0.596
0.5120	-0.269	2.592	0.5120	-0.238	0.602
0.5371	-0.242	2.600	0.5371	-0.212	0.609
0.5621	-0.217	2.607	0.5621	-0.188	0.616
0.5870	-0.197	2.613	0.5870	-0.173	0.620
0.6122	-0.167	2.621	0.6122	-0.141	0.629
0.6371	-0.150	2.626	0.6371	-0.129	0.633
0.6661	-0.125	2.633	0.6661	-0.108	0.639
0.6869	-0.109	2.638	0.6869	-0.095	0.643
0.7121	-0.085	2.645	0.7121	-0.075	0.648
0.7369	-0.063	2.651	0.7369	-0.054	0.654
0.7620	-0.039	2.658	0.7620	-0.036	0.659
0.7870	-0.016	2.665	0.7870	-0.014	0.666
0.8117	0.011	2.672	0.8117	0.010	0.672
0.8370	0.033	2.679	0.8370	0.030	0.678
0.8619	0.059	2.686	0.8619	0.053	0.685
0.8869	0.088	2.694	0.8869	0.081	0.692
0.9120	0.119	2.703	0.9120	0.111	0.701
0.9416	0.155	2.713	0.9416	0.148	0.711

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.781 ALPHA 3.00 REYNOLDS 1.61*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.3319 CM = 0.0060 CDW = 0.0214

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.093	0.642	0.0119	0.625	0.848
0.0371	-0.499	0.526	0.0371	0.151	0.713
0.0623	-0.915	0.407	0.0623	-0.047	0.657
0.0872	-0.911	0.408	0.0872	-0.155	0.626
0.1122	-0.972	0.391	0.1122	-0.220	0.628
0.1372	-0.976	0.389	0.1372	-0.258	0.597
0.1623	-0.994	0.384	0.1623	-0.303	0.584
0.1872	-0.998	0.383	0.1872	-0.343	0.573
0.2122	-1.051	0.368	0.2122	-0.365	0.566
0.2372	-1.076	0.361	0.2372	-0.361	0.567
0.2620	-1.092	0.357	0.2620	-0.356	0.559
0.2872	-1.093	0.356	0.2872	-0.337	0.574
0.3122	-1.112	0.351	0.3122	-0.331	0.576
0.3375	-1.106	0.353	0.3375	-0.314	0.581
0.3618	-1.120	0.349	0.3618	-0.297	0.586
0.3873	-1.058	0.366	0.3873	-0.283	0.589
0.4124	-0.660	0.480	0.4124	-0.271	0.593
0.4371	-0.514	0.522	0.4371	-0.245	0.600
0.4621	-0.445	0.541	0.4621	-0.240	0.601
0.4871	-0.390	0.557	0.4871	-0.223	0.606
0.5120	-0.363	0.573	0.5120	-0.207	0.611
0.5371	-0.291	0.585	0.5371	-0.182	0.618
0.5621	-0.248	0.597	0.5621	-0.161	0.624
0.5870	-0.214	0.607	0.5870	-0.150	0.627
0.6122	-0.176	0.618	0.6122	-0.120	0.636
0.6371	-0.147	0.626	0.6371	-0.110	0.635
0.6661	-0.113	0.636	0.6661	-0.092	0.643
0.6869	-0.097	0.641	0.6869	-0.080	0.647
0.7121	-0.070	0.648	0.7121	-0.062	0.652
0.7369	-0.049	0.654	0.7369	-0.043	0.657
0.7620	-0.025	0.661	0.7620	-0.028	0.662
0.7870	-0.002	0.668	0.7870	-0.007	0.668
0.8117	0.025	0.675	0.8117	0.014	0.674
0.8370	0.045	0.681	0.8370	0.034	0.679
0.8619	0.071	0.689	0.8619	0.055	0.685
0.8869	0.097	0.696	0.8869	0.081	0.693
0.9120	0.126	0.704	0.9120	0.109	0.701
0.9416	0.158	0.713	0.9416	0.144	0.711

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.801 ALPHA 0.00 REYNOLDS 1.63*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.0049 CM = 0.0000 CDW = 0.0146

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.299	0.743	0.0119	0.282	0.738
0.0371	-0.171	0.604	0.0371	-0.170	0.605
0.0623	-0.438	0.526	0.0623	-0.456	0.521
0.0872	-0.489	0.511	0.0872	-0.547	0.494
0.1122	-0.574	0.486	0.1122	-0.581	0.484
0.1372	-0.601	0.478	0.1372	-0.605	0.477
0.1623	-0.615	0.474	0.1623	-0.626	0.471
0.1872	-0.600	0.460	0.1872	-0.662	0.460
0.2122	-0.726	0.441	0.2122	-0.733	0.440
0.2372	-0.759	0.432	0.2372	-0.764	0.431
0.2620	-0.794	0.421	0.2620	-0.802	0.420
0.2872	-0.801	0.419	0.2872	-0.804	0.419
0.3122	-0.824	0.412	0.3122	-0.832	0.411
0.3375	-0.822	0.413	0.3375	-0.829	0.412
0.3618	-0.816	0.415	0.3618	-0.827	0.412
0.3873	-0.576	0.406	0.3873	-0.585	0.484
0.4124	-0.380	0.543	0.4124	-0.384	0.543
0.4371	-0.322	0.560	0.4371	-0.324	0.560
0.4621	-0.302	0.557	0.4621	-0.307	0.566
0.4871	-0.281	0.573	0.4871	-0.283	0.574
0.5120	-0.261	0.579	0.5120	-0.266	0.578
0.5371	-0.237	0.586	0.5371	-0.237	0.587
0.5621	-0.214	0.593	0.5621	-0.219	0.592
0.5870	-0.196	0.598	0.5870	-0.196	0.599
0.6122	-0.165	0.607	0.6122	-0.169	0.607
0.6371	-0.148	0.612	0.6371	-0.148	0.613
0.6661	-0.124	0.619	0.6661	-0.128	0.619
0.6869	-0.108	0.624	0.6869	-0.109	0.625
0.7121	-0.087	0.631	0.7121	-0.090	0.630
0.7369	-0.063	0.638	0.7369	-0.061	0.639
0.7623	-0.041	0.644	0.7623	-0.043	0.644
0.7870	-0.016	0.651	0.7870	-0.016	0.652
0.8117	0.009	0.659	0.8117	0.007	0.659
0.8370	0.032	0.665	0.8370	0.033	0.667
0.8619	0.058	0.673	0.8619	0.056	0.673
0.8869	0.088	0.682	0.8869	0.088	0.683
0.9120	0.118	0.691	0.9120	0.117	0.691
0.9416	0.157	0.702	0.9416	0.157	0.703

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.800 ALPHA 1.00 REYNOLDS 1.63*10⁶

INTEGRATED FORCE COEFFICIENTS

C_N = 0.1199 C_M = 0.0014 C_{DM} = 0.0171

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	C _P	P/H	X/C	C _P	P/H
0.0119	0.186	0.711	0.0119	0.410	0.776
0.0371	-0.269	0.577	0.0371	-0.067	0.635
0.0623	-0.668	0.454	0.0623	-0.296	0.568
0.0872	-0.562	0.491	0.0872	-0.386	0.541
0.1122	-0.668	0.460	0.1122	-0.456	0.521
0.1372	-0.705	0.448	0.1372	-0.481	0.513
0.1623	-0.738	0.439	0.1623	-0.525	0.500
0.1872	-0.764	0.431	0.1872	-0.587	0.482
0.2122	-0.816	0.416	0.2122	-0.648	0.465
0.2372	-0.841	0.409	0.2372	-0.675	0.457
0.2623	-0.873	0.399	0.2623	-0.701	0.450
0.2872	-0.882	0.397	0.2872	-0.690	0.453
0.3122	-0.908	0.389	0.3122	-0.607	0.477
0.3375	-0.908	0.389	0.3375	-0.448	0.524
0.3618	-0.923	0.385	0.3618	-0.410	0.535
0.3873	-0.926	0.384	0.3873	-0.390	0.541
0.4124	-0.813	0.417	0.4124	-0.369	0.547
0.4371	-0.469	0.518	0.4371	-0.336	0.557
0.4621	-0.353	0.553	0.4621	-0.322	0.562
0.4871	-0.296	0.569	0.4871	-0.295	0.572
0.5120	-0.258	0.580	0.5120	-0.271	0.577
0.5371	-0.229	0.589	0.5371	-0.242	0.585
0.5621	-0.202	0.597	0.5621	-0.216	0.593
0.5870	-0.183	0.603	0.5870	-0.197	0.598
0.6122	-0.155	0.611	0.6122	-0.163	0.609
0.6371	-0.138	0.616	0.6371	-0.148	0.613
0.6661	-0.115	0.623	0.6661	-0.123	0.620
0.6869	-0.100	0.627	0.6869	-0.106	0.625
0.7121	-0.078	0.634	0.7121	-0.086	0.631
0.7369	-0.056	0.640	0.7369	-0.062	0.638
0.7620	-0.034	0.647	0.7620	-0.042	0.644
0.7870	-0.011	0.653	0.7870	-0.017	0.651
0.8117	0.014	0.661	0.8117	0.008	0.659
0.8370	0.036	0.667	0.8370	0.031	0.665
0.8619	0.053	0.675	0.8619	0.056	0.673
0.8869	0.091	0.683	0.8869	0.086	0.682
0.9120	0.123	0.692	0.9120	0.117	0.691
0.9416	0.158	0.703	0.9416	0.155	0.702

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.801 ALPHA 2.00 REY 1.63*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2299 CM = 0.0006 CDW = 0.0216

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.081	0.679	0.0119	0.518	0.808
0.0371	-0.344	0.554	0.0371	0.043	0.668
0.0623	-0.786	0.424	0.0623	-0.172	0.605
0.0872	-0.721	0.443	0.0872	-0.274	0.575
0.1122	-0.783	0.425	0.1122	-0.344	0.554
0.1372	-0.780	0.426	0.1372	-0.378	0.544
0.1623	-0.830	0.411	0.1623	-0.432	0.528
0.1872	-0.857	0.443	0.1872	-0.485	0.512
0.2122	-0.900	0.390	0.2122	-0.526	0.500
0.2372	-0.916	0.385	0.2372	-0.520	0.502
0.2623	-0.944	0.377	0.2623	-0.505	0.507
0.2872	-0.945	0.377	0.2872	-0.461	0.520
0.3122	-0.973	0.368	0.3122	-0.444	0.524
0.3375	-0.975	0.368	0.3375	-0.415	0.533
0.3618	-0.993	0.363	0.3618	-0.388	0.541
0.3871	-0.993	0.363	0.3871	-0.364	0.548
0.4124	-1.023	0.354	0.4124	-0.342	0.554
0.4371	-0.857	0.403	0.4371	-0.309	0.564
0.4621	-0.532	0.498	0.4621	-0.298	0.567
0.4871	-0.413	0.533	0.4871	-0.271	0.575
0.5120	-0.347	0.553	0.5120	-0.251	0.581
0.5371	-0.296	0.568	0.5371	-0.222	0.590
0.5621	-0.251	0.581	0.5621	-0.190	0.597
0.5870	-0.212	0.592	0.5870	-0.180	0.602
0.6122	-0.174	0.603	0.6122	-0.148	0.611
0.6371	-0.142	0.613	0.6371	-0.134	0.615
0.6661	-0.109	0.623	0.6661	-0.113	0.622
0.6869	-0.090	0.628	0.6869	-0.097	0.626
0.7121	-0.065	0.636	0.7121	-0.079	0.632
0.7369	-0.040	0.643	0.7369	-0.055	0.639
0.7620	-0.019	0.649	0.7620	-0.037	0.645
0.7870	0.005	0.656	0.7870	-0.013	0.652
0.8117	0.026	0.662	0.8117	0.009	0.658
0.8370	0.050	0.669	0.8370	0.032	0.665
0.8619	0.075	0.677	0.8619	0.055	0.672
0.8869	0.101	0.685	0.8869	0.084	0.680
0.9120	0.129	0.693	0.9120	0.113	0.689
0.9416	0.163	0.703	0.9416	0.150	0.700

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.820 ALPHA 0.00 REY 1.68*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.0137 CM = 0.0017 CDW = 0.0198

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.323	0.741	0.0119	0.512	0.737
0.0371	-0.139	0.601	0.0371	-0.137	0.621
0.0623	-0.425	0.514	0.0623	-0.444	0.538
0.0872	-0.456	0.505	0.0872	-0.511	0.488
0.1122	-0.549	0.477	0.1122	-0.550	0.476
0.1372	-0.580	0.458	0.1372	-0.583	0.466
0.1623	-0.609	0.459	0.1623	-0.615	0.457
0.1872	-0.633	0.451	0.1872	-0.639	0.449
0.2122	-0.698	0.453	0.2122	-0.699	0.431
0.2372	-0.732	0.422	0.2372	-0.735	0.420
0.2620	-0.770	0.411	0.2620	-0.773	0.409
0.2872	-0.779	0.408	0.2872	-0.780	0.406
0.3122	-0.809	0.399	0.3122	-0.812	0.397
0.3375	-0.809	0.399	0.3375	-0.816	0.396
0.3618	-0.828	0.393	0.3618	-0.821	0.394
0.3873	-0.804	0.401	0.3873	-0.839	0.389
0.4124	-0.858	0.385	0.4124	-0.868	0.380
0.4371	-0.752	0.417	0.4371	-0.847	0.386
0.4621	-0.436	0.512	0.4621	-0.660	0.442
0.4871	-0.312	0.549	0.4871	-0.385	0.526
0.5120	-0.253	0.567	0.5120	-0.291	0.554
0.5371	-0.216	0.578	0.5371	-0.233	0.572
0.5621	-0.187	0.587	0.5621	-0.199	0.582
0.5870	-0.169	0.593	0.5870	-0.170	0.591
0.6122	-0.140	0.601	0.6122	-0.141	0.600
0.6371	-0.126	0.606	0.6371	-0.121	0.606
0.6661	-0.102	0.613	0.6661	-0.098	0.613
0.6869	-0.090	0.617	0.6869	-0.082	0.618
0.7121	-0.068	0.623	0.7121	-0.063	0.623
0.7369	-0.047	0.629	0.7369	-0.040	0.630
0.7620	-0.025	0.636	0.7620	-0.022	0.636
0.7870	-0.005	0.642	0.7870	0.001	0.643
0.8117	0.022	0.650	0.8117	0.022	0.649
0.8370	0.043	0.657	0.8370	0.047	0.657
0.8619	0.071	0.665	0.8619	0.072	0.664
0.8869	0.096	0.673	0.8869	0.100	0.673
0.9120	0.128	0.682	0.9120	0.128	0.681
0.9416	0.163	0.693	0.9416	0.165	0.692

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.819 ALPHA 1.00 REYNOLDS 1.68*10⁶

INTEGRATED FORCE COEFFICIENTS

C_N = 0.1102 C_M = -0.0039 C_{DM} = 0.0233

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	C _P	P/H	X/C	C _P	P/H
0.0119	0.230	0.713	0.0119	0.422	0.771
0.0371	-0.219	0.577	0.0371	-0.049	0.628
0.0623	-0.637	0.450	0.0623	-0.285	0.557
0.0872	-0.506	0.490	0.0872	-0.370	0.531
0.1122	-0.612	0.458	0.1122	-0.447	0.508
0.1372	-0.658	0.444	0.1372	-0.471	0.501
0.1623	-0.693	0.453	0.1623	-0.510	0.489
0.1872	-0.722	0.425	0.1872	-0.566	0.472
0.2122	-0.771	0.409	0.2122	-0.633	0.452
0.2372	-0.801	0.420	0.2372	-0.666	0.442
0.2623	-0.833	0.391	0.2623	-0.702	0.431
0.2872	-0.843	0.388	0.2872	-0.708	0.429
0.3122	-0.868	0.380	0.3122	-0.734	0.421
0.3375	-0.875	0.378	0.3375	-0.732	0.422
0.3618	-0.891	0.373	0.3618	-0.743	0.419
0.3873	-0.900	0.370	0.3873	-0.732	0.422
0.4124	-0.925	0.363	0.4124	-0.759	0.474
0.4371	-0.917	0.365	0.4371	-0.739	0.541
0.4621	-0.918	0.364	0.4621	-0.787	0.557
0.4871	-0.900	0.400	0.4871	-0.761	0.565
0.5120	-0.398	0.521	0.5120	-0.741	0.571
0.5371	-0.317	0.546	0.5371	-0.719	0.577
0.5621	-0.261	0.563	0.5621	-0.198	0.584
0.5870	-0.224	0.574	0.5870	-0.183	0.586
0.6122	-0.174	0.589	0.6122	-0.152	0.597
0.6371	-0.145	0.598	0.6371	-0.138	0.602
0.6661	-0.104	0.611	0.6661	-0.114	0.609
0.6869	-0.086	0.616	0.6869	-0.100	0.613
0.7121	-0.753	0.626	0.7121	-0.079	0.620
0.7369	-0.735	0.631	0.7369	-0.057	0.626
0.7623	-0.706	0.640	0.7623	-0.036	0.632
0.7870	0.712	0.645	0.7870	-0.012	0.640
0.8117	0.839	0.653	0.8117	0.013	0.647
0.8370	0.858	0.659	0.8370	0.035	0.654
0.8619	0.768	0.668	0.8619	0.061	0.662
0.8869	0.109	0.675	0.8869	0.089	0.673
0.9120	0.143	0.685	0.9120	0.121	0.680
0.9416	0.162	0.697	0.9416	0.156	0.691

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.319 ALPHA 2.00 REYNOLDS 1.68*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2141 CM = -0.0037 CDW = 0.0283

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.129	0.682	0.0119	0.518	0.800
0.0371	-0.293	0.594	0.0371	0.045	0.657
0.0623	-0.728	0.423	0.0623	-0.173	0.592
0.0872	-0.666	0.442	0.0872	-0.274	0.561
0.1122	-0.736	0.420	0.1122	-0.347	0.539
0.1372	-0.729	0.423	0.1372	-0.383	0.528
0.1623	-0.780	0.407	0.1623	-0.437	0.512
0.1872	-0.811	0.398	0.1872	-0.497	0.494
0.2122	-0.856	0.384	0.2122	-0.558	0.475
0.2372	-0.873	0.379	0.2372	-0.586	0.467
0.2620	-0.904	0.377	0.2620	-0.613	0.459
0.2872	-0.906	0.369	0.2872	-0.609	0.460
0.3122	-0.936	0.360	0.3122	-0.608	0.460
0.3375	-0.935	0.360	0.3375	-0.520	0.457
0.3618	-0.956	0.354	0.3618	-0.401	0.523
0.3873	-0.957	0.354	0.3873	-0.374	0.531
0.4124	-0.992	0.343	0.4124	-0.355	0.537
0.4371	-0.901	0.346	0.4371	-0.324	0.546
0.4621	-0.870	0.360	0.4621	-0.313	0.550
0.4871	-0.530	0.483	0.4871	-0.287	0.557
0.5120	-0.415	0.518	0.5120	-0.265	0.564
0.5371	-0.363	0.533	0.5371	-0.236	0.573
0.5621	-0.324	0.545	0.5621	-0.211	0.580
0.5870	-0.260	0.558	0.5870	-0.195	0.585
0.6122	-0.246	0.569	0.6122	-0.160	0.595
0.6371	-0.209	0.580	0.6371	-0.148	0.599
0.6661	-0.163	0.594	0.6661	-0.126	0.606
0.6869	-0.135	0.602	0.6869	-0.113	0.610
0.7121	-0.099	0.613	0.7121	-0.092	0.616
0.7369	-0.063	0.624	0.7369	-0.071	0.622
0.7620	-0.033	0.633	0.7620	-0.051	0.628
0.7870	-0.003	0.642			

NACA-0012 243.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 3.20 REYNOLDS 0.85*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0000 CM = 0.0000 CDW = 0.0126

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.097	0.857	0.0119	0.097	0.857
0.0371	-0.289	0.801	0.0371	-0.289	0.801
0.0623	-0.404	0.784	0.0623	-0.404	0.784
0.0872	-0.418	0.781	0.0872	-0.418	0.781
0.1122	-0.432	0.779	0.1122	-0.432	0.779
0.1372	-0.416	0.782	0.1372	-0.416	0.782
0.1623	-0.431	0.780	0.1623	-0.431	0.780
0.1872	-0.417	0.782	0.1872	-0.417	0.782
0.2122	-0.424	0.781	0.2122	-0.424	0.781
0.2372	-0.396	0.785	0.2372	-0.396	0.785
0.2623	-0.388	0.786	0.2623	-0.388	0.786
0.2872	-0.354	0.791	0.2872	-0.354	0.791
0.3122	-0.353	0.791	0.3122	-0.353	0.791
0.3375	-0.321	0.796	0.3375	-0.321	0.796
0.3618	-0.313	0.797	0.3618	-0.313	0.797
0.3873	-0.286	0.801	0.3873	-0.286	0.801
0.4124	-0.283	0.801	0.4124	-0.283	0.801
0.4371	-0.249	0.806	0.4371	-0.249	0.806
0.4621	-0.251	0.806	0.4621	-0.251	0.806
0.4871	-0.228	0.810	0.4871	-0.228	0.810
0.5120	-0.221	0.811	0.5120	-0.221	0.811
0.5371	-0.190	0.815	0.5371	-0.190	0.815
0.5621	-0.181	0.816	0.5621	-0.181	0.816
0.5870	-0.159	0.820	0.5870	-0.159	0.820
0.6122	-0.141	0.822	0.6122	-0.141	0.822
0.6371	-0.123	0.825	0.6371	-0.123	0.825
0.6661	-0.114	0.826	0.6661	-0.114	0.826
0.6869	-0.094	0.829	0.6869	-0.094	0.829
0.7121	-0.087	0.830	0.7121	-0.087	0.830
0.7369	-0.066	0.833	0.7369	-0.066	0.833
0.7623	-0.056	0.835	0.7623	-0.056	0.835
0.7873	-0.028	0.839	0.7873	-0.028	0.839
0.8117	-0.014	0.841	0.8117	-0.014	0.841
0.8370	0.010	0.845	0.8370	0.010	0.845
0.8619	0.025	0.847	0.8619	0.025	0.847
0.8869	0.056	0.851	0.8869	0.056	0.851
0.9120	0.072	0.854	0.9120	0.072	0.854
0.9416	0.117	0.860	0.9416	0.117	0.860

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 1.00 REYNOLDS 0.90×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3808 CM = 0.0018 CDW = 0.0140

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.096	0.830	0.0119	0.294	0.837
0.0371	-0.455	0.777	0.0371	-0.133	0.824
0.0623	-0.537	0.765	0.0623	-0.267	0.815
0.0872	-0.528	0.766	0.0872	-0.310	0.793
0.1122	-0.534	0.766	0.1122	-0.334	0.795
0.1372	-0.526	0.770	0.1372	-0.333	0.795
0.1623	-0.511	0.769	0.1623	-0.353	0.792
0.1872	-0.493	0.772	0.1872	-0.349	0.793
0.2122	-0.489	0.773	0.2122	-0.356	0.791
0.2372	-0.460	0.777	0.2372	-0.339	0.794
0.2623	-0.443	0.779	0.2623	-0.332	0.795
0.2872	-0.408	0.784	0.2872	-0.328	0.799
0.3122	-0.397	0.786	0.3122	-0.303	0.799
0.3375	-0.367	0.790	0.3375	-0.281	0.802
0.3618	-0.349	0.793	0.3618	-0.271	0.804
0.3873	-0.325	0.797	0.3873	-0.253	0.807
0.4124	-0.316	0.798	0.4124	-0.251	0.807
0.4371	-0.296	0.802	0.4371	-0.223	0.811
0.4621	-0.290	0.803	0.4621	-0.224	0.811
0.4871	-0.259	0.806	0.4871	-0.204	0.814
0.5120	-0.244	0.809	0.5120	-0.195	0.815
0.5371	-0.217	0.812	0.5371	-0.173	0.819
0.5621	-0.201	0.815	0.5621	-0.163	0.820
0.5870	-0.183	0.818	0.5870	-0.143	0.823
0.6122	-0.158	0.821	0.6122	-0.125	0.825
0.6371	-0.142	0.823	0.6371	-0.109	0.828
0.6661	-0.129	0.825	0.6661	-0.101	0.829
0.6859	-0.112	0.828	0.6859	-0.086	0.831
0.7121	-0.098	0.830	0.7121	-0.075	0.833
0.7369	-0.077	0.833	0.7369	-0.059	0.835
0.7620	-0.061	0.835	0.7620	-0.045	0.837
0.7870	-0.041	0.838	0.7870	-0.026	0.840
0.8117	-0.020	0.841	0.8117	-0.010	0.842
0.8370	0.000	0.844	0.8370	0.013	0.845
0.8619	0.025	0.848	0.8619	0.030	0.848
0.8869	0.049	0.851	0.8869	0.053	0.852
0.9120	0.074	0.855	0.9120	0.072	0.854
0.9416	0.113	0.861	0.9416	0.112	0.860

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 2.00 REYNOLDS 0.85×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1708 CM = 0.0031 CDW = 0.0123

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.346	0.792	0.0119	0.464	0.911
0.0371	-0.653	0.747	0.0371	0.016	0.845
0.0623	-0.705	0.739	0.0623	-0.133	0.823
0.0872	-0.664	0.745	0.0872	-0.196	0.814
0.1122	-0.647	0.746	0.1122	-0.235	0.808
0.1372	-0.609	0.753	0.1372	-0.245	0.807
0.1623	-0.601	0.755	0.1623	-0.269	0.803
0.1872	-0.575	0.758	0.1872	-0.275	0.802
0.2122	-0.564	0.760	0.2122	-0.288	0.800
0.2372	-0.527	0.766	0.2372	-0.276	0.802
0.2620	-0.504	0.769	0.2620	-0.273	0.802
0.2872	-0.466	0.775	0.2872	-0.253	0.806
0.3122	-0.448	0.777	0.3122	-0.253	0.806
0.3375	-0.417	0.782	0.3375	-0.234	0.808
0.3618	-0.396	0.785	0.3618	-0.227	0.809
0.3873	-0.368	0.789	0.3873	-0.212	0.911
0.4124	-0.354	0.791	0.4124	-0.211	0.812
0.4371	-0.322	0.796	0.4371	-0.188	0.815
0.4621	-0.311	0.797	0.4621	-0.189	0.815
0.4871	-0.289	0.801	0.4871	-0.175	0.817
0.5120	-0.270	0.803	0.5120	-0.167	0.818
0.5371	-0.243	0.807	0.5371	-0.144	0.822
0.5621	-0.224	0.810	0.5621	-0.135	0.823
0.5870	-0.205	0.813	0.5870	-0.120	0.825
0.6122	-0.180	0.817	0.6122	-0.102	0.828
0.6371	-0.160	0.820	0.6371	-0.090	0.830
0.6661	-0.145	0.822	0.6661	-0.083	0.831
0.6869	-0.126	0.825	0.6869	-0.074	0.832
0.7121	-0.110	0.827	0.7121	-0.060	0.834
0.7369	-0.086	0.830	0.7369	-0.046	0.836
0.7620	-0.070	0.833	0.7620	-0.033	0.838
0.7870	-0.049	0.836	0.7870	-0.015	0.841
0.8117	-0.027	0.839	0.8117	0.000	0.843
0.8370	-0.004	0.843	0.8370	0.017	0.845
0.8619	0.021	0.846	0.8619	0.035	0.848
0.8869	0.047	0.850	0.8869	0.058	0.851
0.9120	0.072	0.854	0.9120	0.073	0.854
0.9416	0.113	0.860	0.9416	0.113	0.859

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 3.00 REY 0.85*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2561 CH = 0.0051 CDW = 0.0124

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.623	0.751	0.0119	0.617	0.934
0.0371	-0.860	0.717	0.0371	0.156	0.865
0.0623	-0.875	0.714	0.0623	-0.008	0.841
0.0872	-0.792	0.727	0.0872	-0.093	0.828
0.1122	-0.758	0.732	0.1122	-0.139	0.821
0.1372	-0.706	0.739	0.1372	-0.162	0.818
0.1623	-0.690	0.742	0.1623	-0.191	0.814
0.1872	-0.653	0.747	0.1872	-0.206	0.811
0.2122	-0.636	0.749	0.2122	-0.218	0.810
0.2372	-0.590	0.756	0.2372	-0.215	0.810
0.2620	-0.563	0.760	0.2620	-0.215	0.813
0.2872	-0.516	0.767	0.2872	-0.201	0.812
0.3122	-0.498	0.769	0.3122	-0.204	0.812
0.3375	-0.458	0.775	0.3375	-0.191	0.814
0.3618	-0.437	0.778	0.3618	-0.183	0.815
0.3873	-0.404	0.783	0.3873	-0.173	0.816
0.4124	-0.390	0.785	0.4124	-0.172	0.816
0.4371	-0.351	0.791	0.4371	-0.152	0.819
0.4621	-0.340	0.792	0.4621	-0.155	0.819
0.4871	-0.312	0.796	0.4871	-0.144	0.821
0.5120	-0.294	0.799	0.5120	-0.136	0.822
0.5371	-0.262	0.804	0.5371	-0.119	0.824
0.5621	-0.244	0.807	0.5621	-0.107	0.826
0.5870	-0.219	0.810	0.5870	-0.099	0.827
0.6122	-0.194	0.814	0.6122	-0.079	0.830
0.6371	-0.171	0.817	0.6371	-0.076	0.830
0.6661	-0.155	0.820	0.6661	-0.063	0.832
0.6869	-0.134	0.823	0.6869	-0.060	0.833
0.7121	-0.119	0.825	0.7121	-0.042	0.835
0.7369	-0.091	0.829	0.7369	-0.034	0.836
0.7620	-0.077	0.831	0.7620	-0.019	0.839
0.7870	-0.051	0.835	0.7870	-0.005	0.841
0.8117	-0.030	0.838	0.8117	0.011	0.843
0.8370	-0.004	0.842	0.8370	0.024	0.845
0.8619	0.021	0.846	0.8619	0.044	0.848
0.8869	0.050	0.850	0.8869	0.062	0.851
0.9120	0.071	0.853	0.9120	0.080	0.853
0.9416	0.118	0.860	0.9416	0.114	0.858

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 4.00 REY 0.85*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.3438 CM = 0.0066 CDW = 0.0124

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.925	0.706	0.0119	0.744	0.953
0.0371	-1.074	0.684	0.0371	0.287	0.885
0.0623	-1.055	0.667	0.0623	0.103	0.858
0.0872	-0.927	0.706	0.0872	0.009	0.844
0.1122	-0.871	0.714	0.1122	-0.045	0.836
0.1372	-0.824	0.724	0.1372	-0.081	0.831
0.1623	-0.778	0.728	0.1623	-0.110	0.827
0.1872	-0.733	0.734	0.1872	-0.128	0.824
0.2122	-0.705	0.738	0.2122	-0.149	0.821
0.2372	-0.653	0.746	0.2372	-0.149	0.821
0.2620	-0.620	0.751	0.2620	-0.156	0.820
0.2872	-0.569	0.758	0.2872	-0.144	0.822
0.3122	-0.544	0.762	0.3122	-0.151	0.821
0.3375	-0.502	0.768	0.3375	-0.141	0.822
0.3618	-0.476	0.772	0.3618	-0.137	0.823
0.3873	-0.439	0.778	0.3873	-0.129	0.824
0.4124	-0.422	0.780	0.4124	-0.132	0.823
0.4371	-0.382	0.786	0.4371	-0.114	0.826
0.4621	-0.359	0.788	0.4621	-0.119	0.825
0.4871	-0.339	0.792	0.4871	-0.111	0.827
0.5120	-0.319	0.795	0.5120	-0.106	0.827
0.5371	-0.283	0.800	0.5371	-0.090	0.830
0.5621	-0.263	0.803	0.5621	-0.081	0.831
0.5870	-0.237	0.807	0.5870	-0.078	0.831
0.6122	-0.210	0.811	0.6122	-0.057	0.834
0.6371	-0.186	0.815	0.6371	-0.054	0.835
0.6661	-0.167	0.817	0.6661	-0.045	0.836
0.6869	-0.144	0.821	0.6869	-0.039	0.837
0.7121	-0.127	0.823	0.7121	-0.029	0.839
0.7369	-0.100	0.827	0.7369	-0.017	0.840
0.7620	-0.053	0.829	0.7620	-0.006	0.842
0.7870	-0.006	0.833	0.7870	0.008	0.844
0.8117	0.033	0.837	0.8117	0.021	0.846
0.8370	0.068	0.841	0.8370	0.033	0.848
0.8619	0.019	0.845	0.8619	0.051	0.850
0.8869	0.049	0.849	0.8869	0.068	0.853
0.9120	0.072	0.853	0.9120	0.083	0.855
0.9416	0.117	0.859	0.9416	0.115	0.862

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 5.00 REYNOLDS 0.85*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4286 CM = 0.0088 CDW = 0.0127

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-1.267	2.655	0.0119	0.854	0.969
0.0371	-1.318	2.647	0.0371	0.429	0.924
0.0623	-1.257	2.656	0.0623	0.216	0.875
0.0872	-1.059	2.685	0.0872	0.109	0.860
0.1122	-0.976	2.698	0.1122	0.048	0.851
0.1372	-0.896	2.710	0.1372	0.004	0.844
0.1623	-0.856	2.715	0.1623	-0.032	0.839
0.1872	-0.804	2.723	0.1872	-0.063	0.834
0.2122	-0.766	2.728	0.2122	-0.080	0.832
0.2372	-0.710	2.737	0.2372	-0.067	0.831
0.2620	-0.669	2.743	0.2620	-0.094	0.830
0.2872	-0.614	2.751	0.2872	-0.091	0.830
0.3122	-0.582	2.756	0.3122	-0.096	0.829
0.3375	-0.557	2.762	0.3375	-0.092	0.830
0.3618	-0.507	2.767	0.3618	-0.089	0.830
0.3873	-0.468	2.773	0.3873	-0.086	0.831
0.4124	-0.447	2.776	0.4124	-0.089	0.830
0.4371	-0.405	2.782	0.4371	-0.081	0.832
0.4621	-0.369	2.784	0.4621	-0.081	0.831
0.4871	-0.357	2.789	0.4871	-0.082	0.831
0.5120	-0.335	2.792	0.5120	-0.073	0.832
0.5371	-0.300	2.797	0.5371	-0.064	0.834
0.5621	-0.279	2.801	0.5621	-0.051	0.836
0.5870	-0.250	2.805	0.5870	-0.050	0.836
0.6122	-0.220	2.809	0.6122	-0.030	0.839
0.6371	-0.196	2.813	0.6371	-0.030	0.839
0.6661	-0.176	2.816	0.6661	-0.021	0.840
0.6869	-0.152	2.819	0.6869	-0.020	0.840
0.7121	-0.132	2.822	0.7121	-0.006	0.842
0.7369	-0.103	2.827	0.7369	-0.001	0.843
0.7620	-0.087	2.829	0.7620	0.011	0.844
0.7870	-0.058	2.833	0.7870	0.021	0.846
0.8117	-0.031	2.837	0.8117	0.033	0.848
0.8370	-0.006	2.841	0.8370	0.044	0.849
0.8619	0.022	2.845	0.8619	0.059	0.852
0.8869	0.052	2.850	0.8869	0.076	0.854
0.9120	0.074	2.853	0.9120	0.089	0.856
0.9416	0.120	2.858	0.9416	0.119	0.858

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.498 ALPHA 6.00 REYNOLDS 0.85*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.5147 CM = 0.0111 CDW = 0.0129

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-1.659	0.603	0.0119	0.930	0.960
0.0371	-1.562	0.614	0.0371	0.515	0.918
0.0623	-1.498	0.623	0.0623	0.315	0.869
0.0872	-1.182	0.670	0.0872	0.203	0.872
0.1122	-1.093	0.683	0.1122	0.127	0.861
0.1372	-0.994	0.697	0.1372	0.083	0.854
0.1623	-0.947	0.704	0.1623	0.043	0.848
0.1872	-0.888	0.714	0.1872	0.008	0.843
0.2122	-0.842	0.720	0.2122	-0.016	0.840
0.2372	-0.771	0.730	0.2372	-0.029	0.838
0.2620	-0.728	0.737	0.2620	-0.035	0.837
0.2872	-0.663	0.746	0.2872	-0.038	0.836
0.3122	-0.631	0.751	0.3122	-0.046	0.835
0.3375	-0.577	0.759	0.3375	-0.048	0.835
0.3618	-0.546	0.763	0.3618	-0.044	0.835
0.3873	-0.501	0.770	0.3873	-0.047	0.835
0.4124	-0.480	0.773	0.4124	-0.049	0.835
0.4371	-0.432	0.780	0.4371	-0.041	0.836
0.4621	-0.416	0.782	0.4621	-0.044	0.835
0.4871	-0.378	0.788	0.4871	-0.047	0.835
0.5120	-0.327	0.791	0.5120	-0.040	0.836
0.5371	-0.315	0.797	0.5371	-0.033	0.837
0.5621	-0.295	0.800	0.5621	-0.024	0.838
0.5870	-0.261	0.805	0.5870	-0.023	0.839
0.6122	-0.235	0.809	0.6122	-0.005	0.841
0.6371	-0.205	0.814	0.6371	-0.005	0.841
0.6661	-0.184	0.817	0.6661	0.000	0.842
0.6869	-0.156	0.821	0.6869	0.002	0.842
0.7121	-0.140	0.823	0.7121	0.009	0.843
0.7369	-0.105	0.828	0.7369	0.019	0.845
0.7620	-0.089	0.831	0.7620	0.024	0.846
0.7870	-0.028	0.835	0.7870	0.035	0.847
0.8117	-0.033	0.839	0.8117	0.044	0.849
0.8370	-0.003	0.843	0.8370	0.056	0.850
0.8619	0.022	0.847	0.8619	0.060	0.851
0.8869	0.056	0.852	0.8869	0.083	0.854
0.9120	0.076	0.855	0.9120	0.093	0.856
0.9416	0.125	0.862	0.9416	0.121	0.858

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.501 ALPHA 7.00 REYNOLDS 0.86*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.5942 CM = 0.0142 CDW = 0.0123

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-2.037	0.540	0.0119	0.997	0.990
0.0371	-1.814	0.573	0.0371	0.617	0.934
0.0623	-1.740	0.584	0.0623	0.414	0.904
0.0872	-1.298	0.650	0.0872	0.292	0.886
0.1122	-1.189	0.666	0.1122	0.212	0.875
0.1372	-1.060	0.684	0.1372	0.162	0.867
0.1623	-1.019	0.691	0.1623	0.109	0.859
0.1872	-0.939	0.703	0.1872	0.077	0.855
0.2122	-0.897	0.709	0.2122	0.046	0.852
0.2372	-0.816	0.721	0.2372	0.030	0.847
0.2623	-0.772	0.728	0.2623	0.021	0.846
0.2872	-0.699	0.739	0.2872	0.014	0.845
0.3122	-0.667	0.743	0.3122	0.005	0.844
0.3375	-0.602	0.753	0.3375	0.001	0.843
0.3618	-0.571	0.758	0.3618	0.002	0.843
0.3873	-0.520	0.765	0.3873	-0.003	0.842
0.4124	-0.499	0.768	0.4124	-0.006	0.842
0.4371	-0.446	0.776	0.4371	-0.001	0.843
0.4621	-0.434	0.779	0.4621	-0.008	0.841
0.4871	-0.389	0.785	0.4871	-0.010	0.841
0.5120	-0.369	0.788	0.5120	-0.006	0.842
0.5371	-0.322	0.795	0.5371	0.000	0.843
0.5621	-0.304	0.798	0.5621	0.008	0.844
0.5873	-0.264	0.804	0.5873	0.005	0.843
0.6122	-0.241	0.807	0.6122	0.022	0.846
0.6371	-0.204	0.812	0.6371	0.020	0.845
0.6661	-0.189	0.815	0.6661	0.024	0.846
0.6869	-0.155	0.820	0.6869	0.024	0.846
0.7121	-0.144	0.822	0.7121	0.030	0.847
0.7369	-0.103	0.828	0.7369	0.035	0.847
0.7620	-0.090	0.830	0.7620	0.041	0.848
0.7870	-0.002	0.835	0.7870	0.051	0.850
0.8117	-0.033	0.838	0.8117	0.059	0.851
0.8370	0.004	0.844	0.8370	0.068	0.852
0.8619	0.025	0.847	0.8619	0.068	0.852
0.8869	0.064	0.853	0.8869	0.090	0.855
0.9120	0.077	0.854	0.9120	0.098	0.857
0.9416	0.130	0.862	0.9416	0.124	0.861

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 8.00 REYNOLDS 0.86×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.6841 CM = 0.0182 CDW = 0.0132

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-2.477	0.478	0.0119	1.036	0.996
0.0371	-2.085	0.535	0.0371	0.698	0.946
0.0623	-1.973	0.552	0.0623	0.496	0.916
0.0872	-1.875	0.566	0.0872	0.375	0.898
0.1122	-1.154	0.673	0.1122	0.287	0.885
0.1372	-1.146	0.674	0.1372	0.233	0.877
0.1623	-1.088	0.683	0.1623	0.177	0.869
0.1872	-1.015	0.693	0.1872	0.143	0.864
0.2122	-0.953	0.702	0.2122	0.102	0.858
0.2372	-0.875	0.714	0.2372	0.085	0.855
0.2620	-0.816	0.723	0.2620	0.064	0.852
0.2872	-0.744	0.733	0.2872	0.062	0.852
0.3122	-0.702	0.740	0.3122	0.051	0.853
0.3375	-0.644	0.748	0.3375	0.046	0.849
0.3618	-0.602	0.754	0.3618	0.043	0.849
0.3873	-0.554	0.761	0.3873	0.037	0.848
0.4124	-0.523	0.766	0.4124	0.030	0.847
0.4371	-0.473	0.773	0.4371	0.031	0.847
0.4621	-0.449	0.777	0.4621	0.024	0.846
0.4871	-0.411	0.782	0.4871	0.022	0.846
0.5120	-0.383	0.787	0.5120	0.024	0.846
0.5371	-0.343	0.792	0.5371	0.028	0.847
0.5621	-0.313	0.797	0.5621	0.033	0.848
0.5870	-0.281	0.802	0.5870	0.031	0.847
0.6122	-0.248	0.806	0.6122	0.045	0.849
0.6371	-0.218	0.811	0.6371	0.042	0.849
0.6661	-0.191	0.815	0.6661	0.041	0.849
0.6869	-0.165	0.819	0.6869	0.042	0.849
0.7121	-0.141	0.822	0.7121	0.046	0.849
0.7369	-0.108	0.827	0.7369	0.051	0.850
0.7620	-0.084	0.831	0.7620	0.054	0.850
0.7870	-0.056	0.835	0.7870	0.063	0.852
0.8117	-0.025	0.839	0.8117	0.060	0.851
0.8370	0.002	0.843	0.8370	0.075	0.853
0.8619	0.032	0.848	0.8619	0.075	0.853
0.8869	0.061	0.852	0.8869	0.094	0.856
0.9120	0.082	0.855	0.9120	0.098	0.857
0.9416	0.121	0.861	0.9416	0.121	0.860

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.599 ALPHA 0.00 REY 0.86×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0000 CM = 0.0000

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.142	0.812	0.0119	0.142	0.812
0.0371	-0.269	0.727	0.0371	-0.289	0.727
0.0623	-0.438	0.698	0.0623	-0.438	0.698
0.0872	-0.455	0.694	0.0872	-0.455	0.694
0.1122	-0.473	0.691	0.1122	-0.473	0.691
0.1372	-0.458	0.694	0.1372	-0.458	0.694
0.1623	-0.475	0.691	0.1623	-0.475	0.691
0.1872	-0.466	0.692	0.1872	-0.466	0.692
0.2122	-0.471	0.691	0.2122	-0.471	0.691
0.2372	-0.442	0.697	0.2372	-0.442	0.697
0.2623	-0.431	0.699	0.2623	-0.431	0.699
0.2872	-0.397	0.706	0.2872	-0.397	0.706
0.3122	-0.387	0.708	0.3122	-0.387	0.708
0.3375	-0.359	0.714	0.3375	-0.359	0.714
0.3618	-0.342	0.717	0.3618	-0.342	0.717
0.3873	-0.320	0.721	0.3873	-0.320	0.721
0.4124	-0.312	0.723	0.4124	-0.312	0.723
0.4371	-0.281	0.729	0.4371	-0.281	0.729
0.4621	-0.277	0.730	0.4621	-0.277	0.730
0.4871	-0.254	0.734	0.4871	-0.254	0.734
0.5120	-0.241	0.737	0.5120	-0.241	0.737
0.5371	-0.213	0.742	0.5371	-0.213	0.742
0.5621	-0.199	0.745	0.5621	-0.199	0.745
0.5874	-0.179	0.749	0.5874	-0.179	0.749
0.6122	-0.155	0.754	0.6122	-0.155	0.754
0.6371	-0.139	0.757	0.6371	-0.139	0.757
0.6661	-0.124	0.760	0.6661	-0.124	0.760
0.6869	-0.108	0.763	0.6869	-0.108	0.763
0.7121	-0.094	0.766	0.7121	-0.094	0.766
0.7369	-0.072	0.770	0.7369	-0.072	0.770
0.7620	-0.059	0.773	0.7620	-0.059	0.773
0.7870	-0.037	0.777	0.7870	-0.037	0.777
0.8117	-0.016	0.781	0.8117	-0.016	0.781
0.8370	0.006	0.785	0.8370	0.006	0.785
0.8619	0.030	0.790	0.8619	0.030	0.790
0.8869	0.054	0.795	0.8869	0.054	0.795
0.9120	0.079	0.800	0.9120	0.079	0.800
0.9416	0.119	0.808	0.9416	0.119	0.808

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.051 ALPHA 0.00 REY 0.83×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0000 CM = 0.0000

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.169	2.793	0.0119	0.169	0.790
0.0371	-0.284	2.689	0.0371	-0.284	0.659
0.0623	-0.458	2.650	0.0623	-0.458	0.650
0.0872	-0.474	2.647	0.0872	-0.474	0.647
0.1122	-0.500	2.641	0.1122	-0.500	0.641
0.1372	-0.486	2.644	0.1372	-0.486	0.644
0.1623	-0.500	2.641	0.1623	-0.500	0.641
0.1872	-0.497	2.642	0.1872	-0.497	0.642
0.2122	-0.498	2.642	0.2122	-0.498	0.642
0.2372	-0.470	2.648	0.2372	-0.470	0.648
0.2620	-0.455	2.652	0.2620	-0.455	0.652
0.2872	-0.422	2.659	0.2872	-0.422	0.659
0.3122	-0.409	2.662	0.3122	-0.409	0.662
0.3375	-0.381	2.668	0.3375	-0.381	0.668
0.3618	-0.362	2.673	0.3618	-0.362	0.673
0.3873	-0.339	2.678	0.3873	-0.339	0.678
0.4124	-0.326	2.680	0.4124	-0.326	0.680
0.4371	-0.296	2.687	0.4371	-0.296	0.687
0.4621	-0.290	2.689	0.4621	-0.290	0.689
0.4871	-0.265	2.694	0.4871	-0.265	0.694
0.5120	-0.249	2.698	0.5120	-0.249	0.698
0.5371	-0.223	2.703	0.5371	-0.223	0.703
0.5621	-0.205	2.708	0.5621	-0.205	0.708
0.5870	-0.187	2.711	0.5870	-0.187	0.711
0.6122	-0.161	2.717	0.6122	-0.161	0.717
0.6371	-0.144	2.721	0.6371	-0.144	0.721
0.6661	-0.126	2.725	0.6661	-0.126	0.725
0.6869	-0.111	2.728	0.6869	-0.111	0.728
0.7121	-0.094	2.732	0.7121	-0.094	0.732
0.7369	-0.073	2.737	0.7369	-0.073	0.737
0.7620	-0.056	2.740	0.7620	-0.056	0.740
0.7870	-0.036	2.745	0.7870	-0.036	0.745
0.8117	-0.012	2.750	0.8117	-0.012	0.750
0.8370	0.006	2.754	0.8370	0.006	0.754
0.8619	0.034	2.761	0.8619	0.034	0.761
0.8869	0.058	2.766	0.8869	0.058	0.766
0.9120	0.085	2.772	0.9120	0.085	0.772
0.9416	0.124	2.781	0.9416	0.124	0.781

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.676 ALPHA 0.00 REY 0.86*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.0000 CM = -0.0000

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.186	2.780	0.0119	0.186	0.780
0.0371	-0.279	0.670	0.0371	-0.279	0.670
0.0623	-0.472	0.625	0.0623	-0.472	0.625
0.0872	-0.492	0.620	0.0872	-0.492	0.620
0.1122	-0.522	0.613	0.1122	-0.522	0.613
0.1372	-0.509	0.616	0.1372	-0.509	0.616
0.1623	-0.527	0.612	0.1623	-0.527	0.612
0.1872	-0.525	0.612	0.1872	-0.525	0.612
0.2122	-0.527	0.612	0.2122	-0.527	0.612
0.2372	-0.495	0.619	0.2372	-0.495	0.619
0.2620	-0.477	0.624	0.2620	-0.477	0.624
0.2872	-0.441	0.632	0.2872	-0.441	0.632
0.3122	-0.429	0.635	0.3122	-0.429	0.635
0.3375	-0.398	0.642	0.3375	-0.398	0.642
0.3618	-0.377	0.647	0.3618	-0.377	0.647
0.3873	-0.353	0.653	0.3873	-0.353	0.653
0.4124	-0.340	0.656	0.4124	-0.340	0.656
0.4371	-0.309	0.663	0.4371	-0.309	0.663
0.4621	-0.301	0.665	0.4621	-0.301	0.665
0.4871	-0.275	0.671	0.4871	-0.275	0.671
0.5120	-0.257	0.675	0.5120	-0.257	0.675
0.5371	-0.231	0.682	0.5371	-0.231	0.682
0.5621	-0.211	0.687	0.5621	-0.211	0.687
0.5870	-0.192	0.691	0.5870	-0.192	0.691
0.6122	-0.166	0.697	0.6122	-0.166	0.697
0.6371	-0.149	0.701	0.6371	-0.149	0.701
0.6651	-0.131	0.705	0.6651	-0.131	0.705
0.6869	-0.115	0.709	0.6869	-0.115	0.709
0.7121	-0.097	0.713	0.7121	-0.097	0.713
0.7369	-0.075	0.718	0.7369	-0.075	0.718
0.7620	-0.058	0.723	0.7620	-0.058	0.723
0.7870	-0.036	0.728	0.7870	-0.036	0.728
0.8117	-0.012	0.733	0.8117	-0.012	0.733
0.8370	0.008	0.738	0.8370	0.008	0.738
0.8619	0.034	0.744	0.8619	0.034	0.744
0.8869	0.060	0.750	0.8869	0.060	0.750
0.9120	0.086	0.756	0.9120	0.086	0.756
0.9416	0.125	0.766	0.9416	0.125	0.766

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.731 ALPHA 0.00 REY 0.83×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.3030 CM = -0.0000 CDW = 0.0117

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.197	0.769	0.0119	0.197	0.769
0.0371	-0.278	0.651	0.0371	-0.278	0.651
0.0623	-0.485	0.600	0.0623	-0.485	0.600
0.0872	-0.514	0.593	0.0872	-0.514	0.593
0.1122	-0.548	0.584	0.1122	-0.548	0.584
0.1372	-0.540	0.586	0.1372	-0.540	0.586
0.1623	-0.556	0.582	0.1623	-0.556	0.582
0.1872	-0.560	0.581	0.1872	-0.560	0.581
0.2122	-0.559	0.581	0.2122	-0.559	0.581
0.2372	-0.528	0.589	0.2372	-0.528	0.589
0.2620	-0.506	0.595	0.2620	-0.506	0.595
0.2872	-0.467	0.604	0.2872	-0.467	0.604
0.3122	-0.459	0.611	0.3122	-0.459	0.611
0.3375	-0.420	0.616	0.3375	-0.420	0.616
0.3618	-0.399	0.621	0.3618	-0.399	0.621
0.3873	-0.373	0.628	0.3873	-0.373	0.628
0.4124	-0.356	0.632	0.4124	-0.356	0.632
0.4371	-0.322	0.640	0.4371	-0.322	0.640
0.4621	-0.307	0.644	0.4621	-0.307	0.644
0.4871	-0.287	0.649	0.4871	-0.287	0.649
0.5120	-0.268	0.654	0.5120	-0.268	0.654
0.5371	-0.259	0.661	0.5371	-0.259	0.661
0.5621	-0.219	0.666	0.5621	-0.219	0.666
0.5870	-0.200	0.671	0.5870	-0.200	0.671
0.6122	-0.172	0.678	0.6122	-0.172	0.678
0.6371	-0.154	0.682	0.6371	-0.154	0.682
0.6661	-0.134	0.687	0.6661	-0.134	0.687
0.6869	-0.117	0.691	0.6869	-0.117	0.691
0.7121	-0.100	0.695	0.7121	-0.100	0.695
0.7369	-0.076	0.701	0.7369	-0.076	0.701
0.7620	-0.058	0.706	0.7620	-0.058	0.706
0.7870	-0.035	0.711	0.7870	-0.035	0.711
0.8117	-0.011	0.717	0.8117	-0.011	0.717
0.8370	0.011	0.723	0.8370	0.011	0.723
0.8619	0.039	0.730	0.8619	0.039	0.730
0.8869	0.065	0.736	0.8869	0.065	0.736
0.9124	0.093	0.743	0.9120	0.093	0.743
0.9416	0.133	0.753	0.9416	0.133	0.753

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.699 ALPHA 1.00 REYNOLDS 0.84*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0969 CM = 0.0028 CDW = 0.0144

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.022	0.726	0.0119	0.359	0.811
0.0371	-0.445	0.611	0.0371	-0.114	0.694
0.0623	-0.671	0.555	0.0623	-0.323	0.642
0.0872	-0.661	0.558	0.0872	-0.367	0.631
0.1122	-0.689	0.551	0.1122	-0.418	0.619
0.1372	-0.668	0.558	0.1372	-0.416	0.619
0.1623	-0.666	0.556	0.1623	-0.446	0.612
0.1872	-0.657	0.559	0.1872	-0.453	0.610
0.2122	-0.648	0.561	0.2122	-0.465	0.607
0.2372	-0.601	0.572	0.2372	-0.438	0.614
0.2620	-0.571	0.580	0.2620	-0.429	0.616
0.2872	-0.521	0.592	0.2872	-0.394	0.625
0.3122	-0.502	0.597	0.3122	-0.390	0.626
0.3375	-0.464	0.606	0.3375	-0.363	0.633
0.3619	-0.440	0.612	0.3619	-0.347	0.635
0.3873	-0.407	0.620	0.3873	-0.320	0.643
0.4124	-0.390	0.625	0.4124	-0.313	0.645
0.4371	-0.350	0.635	0.4371	-0.278	0.653
0.4621	-0.341	0.637	0.4621	-0.278	0.653
0.4871	-0.309	0.644	0.4871	-0.253	0.650
0.5120	-0.291	0.649	0.5120	-0.239	0.650
0.5371	-0.258	0.657	0.5371	-0.208	0.671
0.5621	-0.239	0.652	0.5621	-0.196	0.674
0.5870	-0.214	0.668	0.5870	-0.173	0.679
0.6122	-0.188	0.675	0.6122	-0.153	0.684
0.6371	-0.165	0.680	0.6371	-0.132	0.689
0.6661	-0.146	0.685	0.6661	-0.119	0.693
0.6869	-0.127	0.690	0.6869	-0.103	0.697
0.7121	-0.108	0.694	0.7121	-0.089	0.699
0.7369	-0.080	0.701	0.7369	-0.062	0.706
0.7620	-0.062	0.706	0.7620	-0.050	0.709
0.7870	-0.038	0.712	0.7870	-0.024	0.715
0.8117	-0.015	0.717	0.8117	-0.007	0.720
0.8370	0.011	0.724	0.8370	0.019	0.726
0.8619	0.037	0.730	0.8619	0.039	0.731
0.8869	0.067	0.738	0.8869	0.070	0.739
0.9120	0.093	0.744	0.9120	0.091	0.744
0.9416	0.134	0.754	0.9416	0.136	0.755

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.699 ALPHA 2.00 REY 0.84×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1930 CM = 0.0074 CDW = 0.0123

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.175	0.678	0.0119	0.506	0.846
0.0371	-0.626	0.557	0.0371	0.024	0.727
0.0623	-0.925	0.485	0.0623	-0.174	0.678
0.0872	-0.847	0.512	0.0872	-0.244	0.663
0.1122	-0.823	0.511	0.1122	-0.293	0.647
0.1372	-0.803	0.523	0.1372	-0.314	0.643
0.1623	-0.784	0.528	0.1623	-0.346	0.635
0.1872	-0.765	0.532	0.1872	-0.364	0.631
0.2122	-0.745	0.537	0.2122	-0.375	0.628
0.2372	-0.680	0.553	0.2372	-0.361	0.631
0.2620	-0.636	0.564	0.2620	-0.355	0.633
0.2872	-0.579	0.578	0.2872	-0.331	0.637
0.3122	-0.554	0.584	0.3122	-0.328	0.639
0.3375	-0.509	0.596	0.3375	-0.306	0.645
0.3618	-0.480	0.603	0.3618	-0.296	0.647
0.3873	-0.444	0.612	0.3873	-0.277	0.652
0.4124	-0.420	0.618	0.4124	-0.270	0.654
0.4371	-0.380	0.627	0.4371	-0.243	0.660
0.4621	-0.365	0.631	0.4621	-0.241	0.661
0.4871	-0.334	0.639	0.4871	-0.222	0.666
0.5120	-0.311	0.645	0.5120	-0.209	0.669
0.5371	-0.276	0.653	0.5371	-0.184	0.675
0.5621	-0.255	0.658	0.5621	-0.171	0.678
0.5870	-0.228	0.665	0.5870	-0.153	0.683
0.6122	-0.200	0.672	0.6122	-0.132	0.685
0.6371	-0.177	0.678	0.6371	-0.116	0.692
0.6661	-0.153	0.683	0.6661	-0.104	0.695
0.6869	-0.133	0.688	0.6869	-0.088	0.699
0.7121	-0.114	0.693	0.7121	-0.076	0.702
0.7369	-0.086	0.700	0.7369	-0.057	0.706
0.7620	-0.067	0.705	0.7620	-0.042	0.710
0.7870	-0.040	0.711	0.7870	-0.019	0.716
0.8117	-0.016	0.717	0.8117	-0.001	0.723
0.8370	0.010	0.724	0.8370	0.019	0.725
0.8619	0.038	0.731	0.8619	0.040	0.731
0.8869	0.068	0.738	0.8869	0.067	0.737
0.9120	0.094	0.744	0.9120	0.088	0.742
0.9416	0.138	0.755	0.9416	0.133	0.753

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.751 ALPHA 3.00 REYNOLDS 0.84×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2954 CM = 0.0115 CDW = 0.0127

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.361	0.630	0.0119	0.638	0.879
0.0371	-0.779	0.527	0.0371	0.161	0.760
0.0623	-1.207	0.420	0.0623	-0.040	0.711
0.0872	-1.143	0.435	0.0872	-0.122	0.690
0.1122	-1.113	0.444	0.1122	-0.186	0.675
0.1372	-1.023	0.456	0.1372	-0.211	0.668
0.1623	-0.944	0.456	0.1623	-0.249	0.659
0.1872	-0.852	0.501	0.1872	-0.270	0.654
0.2122	-0.814	0.518	0.2122	-0.290	0.649
0.2372	-0.755	0.537	0.2372	-0.282	0.651
0.2620	-0.687	0.549	0.2620	-0.281	0.651
0.2872	-0.624	0.565	0.2872	-0.264	0.655
0.3122	-0.593	0.573	0.3122	-0.265	0.655
0.3375	-0.547	0.584	0.3375	-0.249	0.659
0.3613	-0.513	0.593	0.3613	-0.240	0.661
0.3873	-0.473	0.602	0.3873	-0.227	0.664
0.4124	-0.446	0.609	0.4124	-0.224	0.665
0.4371	-0.406	0.619	0.4371	-0.200	0.671
0.4621	-0.386	0.624	0.4621	-0.201	0.671
0.4871	-0.353	0.632	0.4871	-0.184	0.675
0.5120	-0.328	0.638	0.5120	-0.176	0.677
0.5371	-0.292	0.647	0.5371	-0.153	0.683
0.5621	-0.266	0.654	0.5621	-0.142	0.685
0.5870	-0.241	0.660	0.5870	-0.127	0.689
0.6122	-0.209	0.668	0.6122	-0.108	0.694
0.6371	-0.192	0.672	0.6371	-0.094	0.697
0.6661	-0.160	0.680	0.6661	-0.083	0.702
0.6869	-0.146	0.685	0.6869	-0.070	0.703
0.7121	-0.119	0.690	0.7121	-0.059	0.706
0.7369	-0.090	0.697	0.7369	-0.043	0.710
0.7620	-0.058	0.703	0.7620	-0.027	0.714
0.7870	-0.042	0.709	0.7870	-0.008	0.719
0.8117	-0.015	0.716	0.8117	0.010	0.723
0.8370	0.011	0.722	0.8370	0.028	0.727
0.8619	0.040	0.730	0.8619	0.046	0.732
0.8869	0.070	0.737	0.8869	0.072	0.738
0.9120	0.097	0.744	0.9120	0.093	0.744
0.9416	0.139	0.754	0.9416	0.131	0.753

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.701 ALPHA 4.00 REYNOLDS 0.84×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3932 CM = 0.2167 CDW = 0.0127

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.546	2.585	0.0119	0.750	0.906
0.0371	-0.935	2.489	0.0371	0.283	0.793
0.0623	-1.311	2.396	0.0623	0.075	0.739
0.0872	-1.330	2.391	0.0872	-0.019	0.715
0.1122	-1.374	2.351	0.1122	-0.083	0.720
0.1372	-1.332	2.391	0.1372	-0.118	0.691
0.1623	-1.288	2.402	0.1623	-0.161	0.650
0.1872	-1.215	2.420	0.1872	-0.187	0.674
0.2122	-1.158	2.435	0.2122	-0.213	0.668
0.2372	-0.793	2.525	0.2372	-0.211	0.668
0.2620	-0.650	2.560	0.2620	-0.214	0.667
0.2872	-0.585	2.577	0.2872	-0.201	0.673
0.3122	-0.572	2.589	0.3122	-0.237	0.669
0.3375	-0.536	2.589	0.3375	-0.195	0.672
0.3618	-0.515	2.594	0.3618	-0.190	0.673
0.3873	-0.477	2.603	0.3873	-0.179	0.675
0.4124	-0.451	2.610	0.4124	-0.179	0.676
0.4371	-0.407	2.621	0.4371	-0.158	0.681
0.4621	-0.391	2.625	0.4621	-0.161	0.680
0.4871	-0.357	2.633	0.4871	-0.148	0.684
0.5123	-0.333	2.639	0.5120	-0.142	0.685
0.5371	-0.295	2.649	0.5371	-0.121	0.690
0.5621	-0.271	2.655	0.5621	-0.113	0.692
0.5870	-0.241	2.662	0.5870	-0.099	0.696
0.6122	-0.213	2.669	0.6122	-0.081	0.700
0.6371	-0.185	2.676	0.6371	-0.069	0.703
0.6601	-0.163	2.681	0.6601	-0.062	0.705
0.6869	-0.137	2.688	0.6869	-0.054	0.707
0.7121	-0.118	2.693	0.7121	-0.040	0.710
0.7369	-0.087	2.701	0.7369	-0.025	0.714
0.7620	-0.067	2.706	0.7620	-0.011	0.718
0.7870	-0.036	2.713	0.7870	0.006	0.722
0.8117	-0.012	2.719	0.8117	0.022	0.726
0.8370	0.018	2.726	0.8370	0.039	0.730
0.8619	0.044	2.733	0.8619	0.051	0.733
0.8869	0.076	2.741	0.8869	0.080	0.740
0.9120	0.101	2.747	0.9120	0.099	0.745
0.9416	0.145	2.758	0.9416	0.136	0.754

NACA-0012 243.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.699 ALPHA 5.00 REYNOLDS 0.84*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4987 CM = 0.0204 CDN = 0.0177

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.721	0.542	0.0119	0.836	0.928
0.0371	-1.097	0.449	0.0371	0.387	0.817
0.0623	-1.385	0.378	0.0623	0.183	0.766
0.0872	-1.431	0.366	0.0872	0.279	0.741
0.1122	-1.503	0.348	0.1122	0.212	0.724
0.1372	-1.482	0.354	0.1372	-0.033	0.713
0.1623	-1.498	0.350	0.1623	-0.076	0.702
0.1872	-1.433	0.365	0.1872	-0.106	0.695
0.2122	-1.407	0.370	0.2122	-0.133	0.689
0.2372	-1.353	0.384	0.2372	-0.135	0.686
0.2620	-1.173	0.428	0.2620	-0.144	0.686
0.2872	-0.803	0.520	0.2872	-0.136	0.688
0.3122	-0.617	0.566	0.3122	-0.144	0.686
0.3375	-0.524	0.599	0.3375	-0.135	0.688
0.3618	-0.476	0.601	0.3618	-0.133	0.689
0.3873	-0.437	0.611	0.3873	-0.126	0.690
0.4124	-0.415	0.617	0.4124	-0.120	0.690
0.4371	-0.383	0.625	0.4371	-0.111	0.694
0.4621	-0.369	0.628	0.4621	-0.118	0.693
0.4871	-0.343	0.635	0.4871	-0.109	0.695
0.5123	-0.318	0.641	0.5123	-0.103	0.697
0.5371	-0.285	0.649	0.5371	-0.086	0.701
0.5621	-0.259	0.655	0.5621	-0.078	0.703
0.5870	-0.234	0.662	0.5870	-0.069	0.705
0.6122	-0.203	0.669	0.6122	-0.052	0.709
0.6371	-0.179	0.675	0.6371	-0.047	0.710
0.6651	-0.154	0.681	0.6651	-0.036	0.713
0.6869	-0.134	0.686	0.6869	-0.031	0.714
0.7121	-0.111	0.692	0.7121	-0.020	0.717
0.7359	-0.083	0.699	0.7359	-0.007	0.720
0.7620	-0.060	0.705	0.7620	0.025	0.723
0.7870	-0.033	0.711	0.7870	0.019	0.727
0.8117	-0.006	0.718	0.8117	0.034	0.731
0.8370	0.019	0.724	0.8370	0.048	0.734
0.8619	0.047	0.731	0.8619	0.059	0.737
0.8869	0.076	0.739	0.8869	0.085	0.743
0.9120	0.105	0.746	0.9120	0.103	0.748
0.9416	0.144	0.755	0.9416	0.136	0.756

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.721 ALPHA 0.00 REYNOLDS 0.85*10⁶

INTEGRATED FORCE COEFFICIENTS

CV = -0.0000 CM = -0.0000

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.218	0.764	0.0119	0.218	0.764
0.0371	-0.259	0.641	0.0371	-0.259	0.641
0.0623	-0.462	0.553	0.0623	-0.482	0.583
0.0872	-0.512	0.575	0.0872	-0.512	0.575
0.1122	-0.560	0.563	0.1122	-0.560	0.563
0.1372	-0.554	0.565	0.1372	-0.554	0.565
0.1623	-0.578	0.558	0.1623	-0.578	0.553
0.1872	-0.582	0.558	0.1872	-0.582	0.558
0.2122	-0.585	0.557	0.2122	-0.585	0.557
0.2372	-0.549	0.556	0.2372	-0.549	0.566
0.2620	-0.526	0.572	0.2620	-0.526	0.572
0.2872	-0.460	0.584	0.2872	-0.483	0.584
0.3122	-0.467	0.587	0.3122	-0.467	0.587
0.3375	-0.432	0.596	0.3375	-0.432	0.596
0.3618	-0.408	0.602	0.3618	-0.408	0.602
0.3873	-0.380	0.609	0.3873	-0.380	0.609
0.4124	-0.364	0.613	0.4124	-0.364	0.613
0.4371	-0.323	0.623	0.4371	-0.328	0.623
0.4621	-0.319	0.625	0.4621	-0.319	0.625
0.4871	-0.291	0.632	0.4871	-0.291	0.632
0.5120	-0.274	0.637	0.5120	-0.274	0.637
0.5371	-0.244	0.645	0.5371	-0.244	0.645
0.5621	-0.223	0.650	0.5621	-0.223	0.650
0.5870	-0.201	0.656	0.5870	-0.201	0.656
0.6122	-0.175	0.662	0.6122	-0.175	0.662
0.6371	-0.155	0.667	0.6371	-0.155	0.667
0.6661	-0.135	0.672	0.6661	-0.135	0.672
0.6869	-0.119	0.677	0.6869	-0.119	0.677
0.7121	-0.101	0.681	0.7121	-0.101	0.681
0.7369	-0.076	0.688	0.7369	-0.076	0.688
0.7620	-0.058	0.692	0.7620	-0.058	0.692
0.7870	-0.034	0.698	0.7870	-0.034	0.698
0.8117	-0.008	0.705	0.8117	-0.008	0.705
0.8370	0.013	0.711	0.8370	0.013	0.711
0.8619	0.037	0.717	0.8619	0.037	0.717
0.8869	0.066	0.724	0.8869	0.066	0.724
0.9120	0.094	0.731	0.9120	0.094	0.731
0.9416	0.134	0.742	0.9416	0.134	0.742

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.739 ALPHA 0.00 REY 0.87×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0000 CM = 0.0000

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.231	0.757	0.0119	0.231	0.757
0.0371	-0.247	0.630	0.0371	-0.247	0.630
0.0623	-0.486	0.566	0.0623	-0.486	0.566
0.0872	-0.517	0.558	0.0872	-0.517	0.558
0.1122	-0.575	0.542	0.1122	-0.575	0.542
0.1372	-0.571	0.543	0.1372	-0.571	0.543
0.1623	-0.600	0.535	0.1623	-0.600	0.535
0.1872	-0.608	0.533	0.1872	-0.608	0.533
0.2122	-0.616	0.532	0.2122	-0.616	0.532
0.2372	-0.572	0.544	0.2372	-0.572	0.544
0.2620	-0.548	0.550	0.2620	-0.548	0.550
0.2872	-0.498	0.563	0.2872	-0.498	0.563
0.3122	-0.464	0.567	0.3122	-0.464	0.567
0.3375	-0.445	0.578	0.3375	-0.445	0.578
0.3618	-0.421	0.584	0.3618	-0.421	0.584
0.3873	-0.390	0.592	0.3873	-0.390	0.592
0.4124	-0.373	0.597	0.4124	-0.373	0.597
0.4371	-0.356	0.606	0.4371	-0.356	0.606
0.4621	-0.327	0.609	0.4621	-0.327	0.609
0.4871	-0.298	0.617	0.4871	-0.298	0.617
0.5120	-0.278	0.622	0.5120	-0.278	0.622
0.5371	-0.248	0.631	0.5371	-0.248	0.631
0.5621	-0.227	0.636	0.5621	-0.227	0.636
0.5870	-0.205	0.642	0.5870	-0.205	0.642
0.6122	-0.177	0.649	0.6122	-0.177	0.649
0.6371	-0.157	0.655	0.6371	-0.157	0.655
0.6661	-0.137	0.660	0.6661	-0.137	0.660
0.6869	-0.119	0.665	0.6869	-0.119	0.665
0.7121	-0.101	0.669	0.7121	-0.101	0.669
0.7369	-0.075	0.676	0.7369	-0.075	0.676
0.7620	-0.057	0.681	0.7620	-0.057	0.681
0.7870	-0.033	0.687	0.7870	-0.033	0.687
0.8117	-0.008	0.694	0.8117	-0.008	0.694
0.8370	0.015	0.700	0.8370	0.015	0.700
0.8619	0.037	0.706	0.8619	0.037	0.706
0.8869	0.070	0.715	0.8869	0.070	0.715
0.9120	0.098	0.722	0.9120	0.098	0.722
0.9416	0.137	0.733	0.9416	0.137	0.733

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.750 ALPHA 0.00 REYNOLDS 0.83*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0000 CM = 0.0000

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.255	0.752	0.0119	0.255	0.752
0.0371	-0.229	0.618	0.0371	-0.229	0.618
0.0623	-0.489	0.547	0.0623	-0.489	0.547
0.0872	-0.522	0.537	0.0872	-0.522	0.537
0.1122	-0.593	0.518	0.1122	-0.593	0.518
0.1372	-0.600	0.516	0.1372	-0.600	0.516
0.1623	-0.538	0.506	0.1623	-0.638	0.506
0.1872	-0.667	0.497	0.1872	-0.667	0.497
0.2122	-0.700	0.488	0.2122	-0.700	0.488
0.2372	-0.653	0.501	0.2372	-0.653	0.501
0.2620	-0.604	0.515	0.2620	-0.604	0.515
0.2872	-0.536	0.534	0.2872	-0.536	0.534
0.3122	-0.509	0.541	0.3122	-0.509	0.541
0.3375	-0.467	0.553	0.3375	-0.467	0.553
0.3618	-0.439	0.560	0.3618	-0.439	0.560
0.3873	-0.408	0.569	0.3873	-0.408	0.569
0.4124	-0.386	0.575	0.4124	-0.386	0.575
0.4371	-0.347	0.586	0.4371	-0.347	0.586
0.4621	-0.335	0.589	0.4621	-0.335	0.589
0.4871	-0.305	0.597	0.4871	-0.305	0.597
0.5120	-0.284	0.603	0.5120	-0.284	0.603
0.5371	-0.253	0.612	0.5371	-0.253	0.612
0.5621	-0.231	0.618	0.5621	-0.231	0.618
0.5870	-0.209	0.624	0.5870	-0.209	0.624
0.6122	-0.180	0.632	0.6122	-0.180	0.632
0.6371	-0.159	0.638	0.6371	-0.159	0.638
0.6661	-0.136	0.644	0.6661	-0.136	0.644
0.6869	-0.120	0.648	0.6869	-0.120	0.648
0.7121	-0.099	0.654	0.7121	-0.099	0.654
0.7369	-0.075	0.661	0.7369	-0.075	0.661
0.7620	-0.054	0.667	0.7620	-0.054	0.667
0.7870	-0.031	0.673	0.7870	-0.031	0.673
0.8117	-0.004	0.681	0.8117	-0.004	0.681
0.8370	0.019	0.687	0.8370	0.019	0.687
0.8619	0.042	0.693	0.8619	0.042	0.693
0.8869	0.074	0.702	0.8869	0.074	0.702
0.9120	0.104	0.710	0.9120	0.104	0.710
0.9416	0.143	0.721	0.9416	0.143	0.721

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.781 ALPHA 0.00 REY 0.84*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0000 CM = 0.0000

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.281	0.748	0.0119	0.281	0.748
0.0371	-0.203	0.610	0.0371	-0.203	0.610
0.0623	-0.482	0.531	0.0623	-0.482	0.531
0.0872	-0.512	0.522	0.0872	-0.512	0.522
0.1122	-0.592	0.499	0.1122	-0.592	0.499
0.1372	-0.608	0.495	0.1372	-0.608	0.495
0.1623	-0.643	0.485	0.1623	-0.643	0.485
0.1872	-0.679	0.474	0.1872	-0.679	0.474
0.2122	-0.745	0.455	0.2122	-0.745	0.455
0.2372	-0.766	0.449	0.2372	-0.766	0.449
0.2620	-0.802	0.439	0.2620	-0.802	0.439
0.2872	-0.785	0.444	0.2872	-0.785	0.444
0.3122	-0.700	0.468	0.3122	-0.700	0.468
0.3375	-0.480	0.531	0.3375	-0.480	0.531
0.3618	-0.419	0.548	0.3618	-0.419	0.548
0.3873	-0.393	0.556	0.3873	-0.393	0.556
0.4124	-0.378	0.560	0.4124	-0.378	0.560
0.4371	-0.345	0.570	0.4371	-0.345	0.570
0.4621	-0.336	0.572	0.4621	-0.336	0.572
0.4871	-0.307	0.581	0.4871	-0.307	0.581
0.5120	-0.285	0.587	0.5120	-0.285	0.587
0.5371	-0.255	0.595	0.5371	-0.255	0.595
0.5621	-0.232	0.602	0.5621	-0.232	0.602
0.5870	-0.209	0.609	0.5870	-0.209	0.609
0.6122	-0.174	0.619	0.6122	-0.174	0.619
0.6371	-0.159	0.623	0.6371	-0.159	0.623
0.6661	-0.134	0.630	0.6661	-0.134	0.630
0.6869	-0.119	0.634	0.6869	-0.119	0.634
0.7121	-0.097	0.640	0.7121	-0.097	0.640
0.7369	-0.073	0.647	0.7369	-0.073	0.647
0.7620	-0.050	0.653	0.7620	-0.050	0.653
0.7870	-0.028	0.660	0.7870	-0.028	0.660
0.8117	0.000	0.666	0.8117	0.000	0.666
0.8370	0.023	0.674	0.8370	0.023	0.674
0.8619	0.047	0.681	0.8619	0.047	0.681
0.8869	0.080	0.690	0.8869	0.080	0.690
0.9120	0.110	0.699	0.9120	0.110	0.699
0.9416	0.150	0.710	0.9416	0.150	0.710

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.830 ALPHA 0.00 REY 0.84×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3000 CM = 0.0000 CDW = 0.0146

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.3000	0.744	0.0119	0.3000	0.744
0.0371	-0.176	0.604	0.0371	-0.176	0.604
0.0623	-0.465	0.519	0.0623	-0.465	0.519
0.0872	-0.491	0.511	0.0872	-0.491	0.511
0.1122	-0.575	0.487	0.1122	-0.575	0.487
0.1372	-0.600	0.479	0.1372	-0.600	0.479
0.1623	-0.634	0.469	0.1623	-0.634	0.469
0.1872	-0.663	0.461	0.1872	-0.663	0.461
0.2122	-0.723	0.442	0.2122	-0.728	0.442
0.2372	-0.756	0.433	0.2372	-0.756	0.433
0.2620	-0.796	0.422	0.2620	-0.796	0.422
0.2872	-0.805	0.419	0.2872	-0.805	0.419
0.3122	-0.829	0.412	0.3122	-0.829	0.412
0.3375	-0.822	0.414	0.3375	-0.822	0.414
0.3618	-0.820	0.415	0.3618	-0.820	0.415
0.3873	-0.679	0.456	0.3873	-0.679	0.456
0.4124	-0.408	0.536	0.4124	-0.408	0.536
0.4371	-0.306	0.566	0.4371	-0.306	0.566
0.4621	-0.262	0.573	0.4621	-0.262	0.573
0.4871	-0.264	0.578	0.4871	-0.264	0.578
0.5120	-0.251	0.582	0.5120	-0.251	0.582
0.5371	-0.226	0.589	0.5371	-0.226	0.589
0.5621	-0.209	0.594	0.5621	-0.209	0.594
0.5870	-0.190	0.600	0.5870	-0.190	0.600
0.6122	-0.165	0.607	0.6122	-0.165	0.607
0.6371	-0.143	0.614	0.6371	-0.143	0.614
0.6661	-0.125	0.619	0.6661	-0.125	0.619
0.6869	-0.106	0.625	0.6869	-0.106	0.625
0.7121	-0.088	0.630	0.7121	-0.088	0.630
0.7369	-0.061	0.638	0.7369	-0.061	0.633
0.7620	-0.043	0.643	0.7620	-0.043	0.643
0.7870	-0.017	0.651	0.7870	-0.017	0.651
0.8117	0.007	0.658	0.8117	0.007	0.658
0.8370	0.033	0.665	0.8370	0.033	0.665
0.8619	0.055	0.672	0.8619	0.055	0.672
0.8869	0.089	0.682	0.8869	0.089	0.682
0.9120	0.118	0.690	0.9120	0.118	0.690
0.9416	0.159	0.703	0.9416	0.159	0.703

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.799 ALPHA 1.00 REYNOLDS 0.84×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1227 CM = 0.0002 CDW = 0.0171

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.190	0.712	0.0119	0.408	0.775
0.0371	-0.272	0.576	0.0371	-0.071	0.634
0.0623	-0.643	0.467	0.0623	-0.329	0.558
0.0872	-0.597	0.461	0.0872	-0.382	0.543
0.1122	-0.669	0.450	0.1122	-0.461	0.519
0.1372	-0.700	0.451	0.1372	-0.487	0.512
0.1623	-0.741	0.439	0.1623	-0.535	0.497
0.1872	-0.760	0.433	0.1872	-0.581	0.484
0.2122	-0.815	0.417	0.2122	-0.644	0.465
0.2372	-0.836	0.411	0.2372	-0.667	0.458
0.2623	-0.876	0.399	0.2623	-0.699	0.449
0.2872	-0.883	0.397	0.2872	-0.686	0.453
0.3122	-0.913	0.386	0.3122	-0.643	0.466
0.3375	-0.908	0.389	0.3375	-0.480	0.514
0.3618	-0.928	0.384	0.3618	-0.400	0.537
0.3873	-0.925	0.384	0.3873	-0.368	0.547
0.4124	-0.903	0.391	0.4124	-0.356	0.550
0.4371	-0.579	0.406	0.4371	-0.323	0.560
0.4621	-0.373	0.547	0.4621	-0.315	0.562
0.4871	-0.283	0.573	0.4871	-0.298	0.570
0.5120	-0.268	0.586	0.5120	-0.269	0.576
0.5371	-0.203	0.597	0.5371	-0.236	0.585
0.5621	-0.182	0.603	0.5621	-0.216	0.591
0.5870	-0.165	0.608	0.5870	-0.193	0.598
0.6122	-0.143	0.614	0.6122	-0.167	0.606
0.6371	-0.128	0.619	0.6371	-0.144	0.613
0.6661	-0.110	0.624	0.6661	-0.122	0.619
0.6869	-0.095	0.628	0.6869	-0.105	0.624
0.7121	-0.070	0.633	0.7121	-0.086	0.630
0.7369	-0.055	0.640	0.7369	-0.063	0.637
0.7620	-0.036	0.645	0.7620	-0.042	0.643
0.7870	-0.012	0.653	0.7870	-0.016	0.650
0.8117	0.013	0.660	0.8117	0.009	0.658
0.8370	0.036	0.667	0.8370	0.033	0.665
0.8619	0.058	0.673	0.8619	0.054	0.671
0.8869	0.092	0.683	0.8869	0.090	0.681
0.9120	0.119	0.691	0.9120	0.117	0.689
0.9416	0.159	0.703	0.9416	0.160	0.702

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.830 ALPHA 2.00 REY 0.84×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2354 CM = -0.0012 CDW = 0.0216

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.082	0.680	0.0119	0.516	0.837
0.0371	-0.358	0.551	0.0371	0.037	0.666
0.0623	-0.751	0.435	0.0623	-0.196	0.597
0.0872	-0.741	0.438	0.0872	-0.271	0.575
0.1122	-0.775	0.428	0.1122	-0.348	0.552
0.1372	-0.792	0.423	0.1372	-0.381	0.543
0.1623	-0.837	0.410	0.1623	-0.435	0.527
0.1872	-0.860	0.403	0.1872	-0.479	0.514
0.2122	-0.902	0.391	0.2122	-0.527	0.499
0.2372	-0.920	0.385	0.2372	-0.527	0.499
0.2623	-0.959	0.374	0.2623	-0.516	0.502
0.2872	-0.962	0.373	0.2872	-0.471	0.516
0.3122	-0.991	0.364	0.3122	-0.442	0.524
0.3375	-0.987	0.366	0.3375	-0.411	0.533
0.3618	-1.007	0.360	0.3618	-0.385	0.541
0.3873	-1.005	0.360	0.3873	-0.355	0.550
0.4124	-1.032	0.352	0.4124	-0.336	0.555
0.4371	-0.990	0.365	0.4371	-0.302	0.565
0.4621	-0.743	0.437	0.4621	-0.293	0.568
0.4871	-0.476	0.516	0.4871	-0.266	0.576
0.5123	-0.359	0.550	0.5123	-0.247	0.582
0.5371	-0.259	0.577	0.5371	-0.218	0.592
0.5621	-0.207	0.595	0.5621	-0.199	0.596
0.5873	-0.164	0.608	0.5873	-0.177	0.602
0.6122	-0.127	0.618	0.6122	-0.150	0.610
0.6371	-0.128	0.624	0.6371	-0.130	0.616
0.6661	-0.089	0.630	0.6661	-0.110	0.622
0.6869	-0.076	0.634	0.6869	-0.095	0.627
0.7121	-0.057	0.639	0.7121	-0.076	0.632
0.7369	-0.040	0.644	0.7369	-0.054	0.639
0.7620	-0.015	0.651	0.7620	-0.033	0.645
0.7873	0.005	0.657	0.7873	-0.011	0.651
0.8117	0.029	0.664	0.8117	0.013	0.659
0.8370	0.048	0.670	0.8370	0.037	0.666
0.8619	0.070	0.677	0.8619	0.058	0.672
0.8869	0.098	0.685	0.8869	0.090	0.681
0.9120	0.125	0.693	0.9120	0.119	0.690
0.9416	0.162	0.703	0.9416	0.160	0.702

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 0.00 REYNOLDS 0.43*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.0000 CM = -0.0000

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.098	0.857	0.0119	0.098	0.857
0.0371	-0.325	0.794	0.0371	-0.325	0.794
0.0623	-0.436	0.778	0.0623	-0.436	0.778
0.0872	-0.431	0.779	0.0872	-0.431	0.779
0.1122	-0.454	0.775	0.1122	-0.454	0.775
0.1372	-0.431	0.779	0.1372	-0.431	0.779
0.1623	-0.448	0.776	0.1623	-0.448	0.776
0.1872	-0.431	0.779	0.1872	-0.431	0.779
0.2122	-0.434	0.777	0.2122	-0.434	0.777
0.2372	-0.408	0.781	0.2372	-0.408	0.781
0.2620	-0.400	0.782	0.2620	-0.400	0.782
0.2872	-0.364	0.788	0.2872	-0.364	0.788
0.3122	-0.357	0.789	0.3122	-0.357	0.789
0.3375	-0.326	0.793	0.3375	-0.326	0.793
0.3618	-0.314	0.795	0.3618	-0.314	0.795
0.3873	-0.292	0.798	0.3873	-0.292	0.798
0.4124	-0.287	0.799	0.4124	-0.287	0.799
0.4371	-0.256	0.804	0.4371	-0.256	0.804
0.4621	-0.252	0.804	0.4621	-0.252	0.804
0.4871	-0.226	0.808	0.4871	-0.226	0.808
0.5120	-0.216	0.809	0.5120	-0.216	0.809
0.5371	-0.188	0.813	0.5371	-0.188	0.813
0.5621	-0.179	0.814	0.5621	-0.179	0.814
0.5870	-0.161	0.817	0.5870	-0.161	0.817
0.6122	-0.146	0.819	0.6122	-0.146	0.819
0.6371	-0.134	0.821	0.6371	-0.134	0.821
0.6661	-0.115	0.824	0.6661	-0.115	0.824
0.6869	-0.109	0.825	0.6869	-0.109	0.825
0.7121	-0.088	0.828	0.7121	-0.088	0.828
0.7369	-0.075	0.830	0.7369	-0.075	0.830
0.7620	-0.058	0.833	0.7620	-0.058	0.833
0.7870	-0.038	0.836	0.7870	-0.038	0.836
0.8117	-0.021	0.838	0.8117	-0.021	0.838
0.8370	0.000	0.841	0.8370	0.000	0.841
0.8619	0.024	0.845	0.8619	0.024	0.845
0.8869	0.044	0.848	0.8869	0.044	0.848
0.9120	0.073	0.852	0.9120	0.073	0.852
0.9416	0.107	0.857	0.9416	0.107	0.857

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 1.00 REY 0.43×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0768 CM = 0.0025

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.117	0.826	0.0119	0.273	0.884
0.0371	-0.490	0.771	0.0371	-0.164	0.820
0.0623	-0.570	0.759	0.0623	-0.297	0.820
0.0872	-0.543	0.763	0.0872	-0.316	0.798
0.1122	-0.552	0.762	0.1122	-0.350	0.792
0.1372	-0.509	0.768	0.1372	-0.339	0.794
0.1623	-0.521	0.766	0.1623	-0.362	0.791
0.1872	-0.492	0.770	0.1872	-0.356	0.792
0.2122	-0.494	0.770	0.2122	-0.372	0.789
0.2372	-0.456	0.776	0.2372	-0.345	0.793
0.2623	-0.451	0.777	0.2623	-0.350	0.792
0.2872	-0.405	0.783	0.2872	-0.320	0.797
0.3122	-0.399	0.784	0.3122	-0.319	0.797
0.3375	-0.363	0.789	0.3375	-0.294	0.801
0.3618	-0.353	0.791	0.3618	-0.291	0.801
0.3873	-0.323	0.795	0.3873	-0.264	0.805
0.4124	-0.319	0.796	0.4124	-0.260	0.806
0.4371	-0.283	0.801	0.4371	-0.228	0.810
0.4621	-0.279	0.802	0.4621	-0.226	0.811
0.4871	-0.253	0.806	0.4871	-0.207	0.813
0.5123	-0.245	0.807	0.5123	-0.198	0.815
0.5371	-0.211	0.812	0.5371	-0.171	0.819
0.5621	-0.205	0.813	0.5621	-0.164	0.820
0.5870	-0.179	0.817	0.5870	-0.156	0.821
0.6122	-0.166	0.819	0.6122	-0.130	0.825
0.6371	-0.150	0.821	0.6371	-0.119	0.826
0.6661	-0.132	0.824	0.6661	-0.102	0.829
0.6869	-0.120	0.825	0.6869	-0.096	0.830
0.7121	-0.101	0.828	0.7121	-0.081	0.832
0.7359	-0.082	0.831	0.7359	-0.064	0.834
0.7620	-0.067	0.833	0.7620	-0.050	0.837
0.7870	-0.044	0.836	0.7870	-0.030	0.839
0.8117	-0.028	0.839	0.8117	-0.012	0.842
0.8370	-0.002	0.843	0.8370	0.004	0.844
0.8619	0.021	0.846	0.8619	0.025	0.847
0.8869	0.046	0.850	0.8869	0.047	0.851
0.9120	0.074	0.854	0.9120	0.074	0.855
0.9416	0.110	0.859	0.9416	0.104	0.859

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499

ALPHA

2.00

REY

0.44×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1573 CM = 0.0055

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.353	0.791	0.0119	0.434	0.907
0.0371	-0.680	0.743	0.0371	-0.023	0.843
0.0623	-0.733	0.735	0.0623	-0.170	0.819
0.0872	-0.671	0.744	0.0872	-0.209	0.813
0.1122	-0.662	0.746	0.1122	-0.257	0.806
0.1372	-0.606	0.754	0.1372	-0.256	0.806
0.1623	-0.610	0.753	0.1623	-0.288	0.802
0.1872	-0.568	0.760	0.1872	-0.288	0.802
0.2122	-0.567	0.750	0.2122	-0.307	0.799
0.2372	-0.519	0.767	0.2372	-0.266	0.802
0.2620	-0.506	0.769	0.2620	-0.294	0.801
0.2872	-0.460	0.775	0.2872	-0.269	0.804
0.3122	-0.451	0.777	0.3122	-0.276	0.803
0.3375	-0.410	0.783	0.3375	-0.249	0.827
0.3618	-0.399	0.785	0.3618	-0.254	0.807
0.3873	-0.363	0.790	0.3873	-0.224	0.811
0.4124	-0.356	0.791	0.4124	-0.226	0.811
0.4371	-0.317	0.797	0.4371	-0.200	0.814
0.4621	-0.313	0.797	0.4621	-0.211	0.813
0.4871	-0.251	0.802	0.4871	-0.183	0.817
0.5120	-0.250	0.804	0.5120	-0.186	0.817
0.5371	-0.234	0.809	0.5371	-0.160	0.820
0.5621	-0.227	0.810	0.5621	-0.161	0.820
0.5870	-0.198	0.814	0.5870	-0.141	0.823
0.6122	-0.184	0.816	0.6122	-0.118	0.827
0.6371	-0.156	0.820	0.6371	-0.107	0.828
0.6661	-0.144	0.822	0.6661	-0.096	0.830
0.6869	-0.129	0.824	0.6869	-0.085	0.831
0.7121	-0.113	0.826	0.7121	-0.071	0.833
0.7369	-0.090	0.829	0.7369	-0.058	0.835
0.7620	-0.073	0.832	0.7620	-0.043	0.837
0.7870	-0.046	0.836	0.7870	-0.026	0.842
0.8117	-0.031	0.838	0.8117	-0.009	0.842
0.8370	-0.002	0.842	0.8370	0.011	0.845
0.8619	0.018	0.845	0.8619	0.028	0.843
0.8869	0.048	0.850	0.8869	0.051	0.851
0.9120	0.073	0.853	0.9120	0.074	0.855
0.9416	0.113	0.859	0.9416	0.107	0.859

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.502 ALPHA 3.00 REYNOLDS 0.44*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2336 CM = 0.0106

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.617	2.751	0.0119	0.592	0.930
0.0371	-0.878	2.712	0.0371	0.119	0.661
0.0623	-0.891	2.711	0.0623	-0.050	0.836
0.0872	-0.601	2.724	0.0872	-0.115	0.826
0.1122	-0.775	2.728	0.1122	-0.158	0.820
0.1372	-0.700	2.759	0.1372	-0.173	0.813
0.1623	-0.692	2.740	0.1623	-0.211	0.802
0.1872	-0.639	2.748	0.1872	-0.215	0.812
0.2122	-0.555	2.750	0.2122	-0.212	0.808
0.2372	-0.501	2.758	0.2372	-0.228	0.810
0.2623	-0.504	2.760	0.2623	-0.239	0.820
0.2872	-0.511	2.768	0.2872	-0.220	0.811
0.3122	-0.500	2.770	0.3122	-0.229	0.810
0.3375	-0.452	2.777	0.3375	-0.207	0.813
0.3618	-0.439	2.779	0.3618	-0.214	0.812
0.3873	-0.397	2.785	0.3873	-0.188	0.816
0.4124	-0.386	2.786	0.4124	-0.192	0.815
0.4371	-0.346	2.792	0.4371	-0.173	0.818
0.4621	-0.343	2.793	0.4621	-0.180	0.817
0.4871	-0.306	2.798	0.4871	-0.158	0.820
0.5120	-0.294	2.800	0.5120	-0.158	0.820
0.5371	-0.255	2.806	0.5371	-0.145	0.822
0.5621	-0.245	2.807	0.5621	-0.136	0.823
0.5870	-0.213	2.812	0.5870	-0.126	0.825
0.6122	-0.196	2.814	0.6122	-0.118	0.826
0.6371	-0.167	2.819	0.6371	-0.104	0.826
0.6661	-0.156	2.820	0.6661	-0.093	0.830
0.6869	-0.139	2.823	0.6869	-0.085	0.831
0.7121	-0.117	2.826	0.7121	-0.074	0.832
0.7369	-0.095	2.829	0.7369	-0.064	0.834
0.7620	-0.074	2.832	0.7620	-0.056	0.835
0.7870	-0.051	2.836	0.7870	-0.043	0.837
0.8117	-0.031	2.839	0.8117	-0.028	0.839
0.8370	-0.002	2.843	0.8370	0.002	0.844
0.8619	0.021	2.846	0.8619	0.025	0.847
0.8869	0.051	2.851	0.8869	0.055	0.852
0.9120	0.080	2.855	0.9120	0.081	0.855
0.9416	0.118	2.861	0.9416	0.111	0.860

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.501 ALPHA 4.00 REY 0.44×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3093 CM = 0.0192

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.899	0.708	0.0119	0.712	0.948
0.0371	-1.375	0.682	0.0371	0.254	0.879
0.0623	-1.045	0.667	0.0623	0.074	0.853
0.0872	-0.918	0.706	0.0872	-0.004	0.841
0.1122	-0.866	0.713	0.1122	-0.064	0.832
0.1372	-0.774	0.727	0.1372	-0.091	0.828
0.1623	-0.762	0.729	0.1623	-0.129	0.823
0.1872	-0.703	0.738	0.1872	-0.144	0.823
0.2122	-0.687	0.740	0.2122	-0.172	0.816
0.2372	-0.625	0.749	0.2372	-0.158	0.818
0.2623	-0.605	0.752	0.2623	-0.175	0.810
0.2872	-0.546	0.761	0.2872	-0.158	0.810
0.3122	-0.535	0.763	0.3122	-0.172	0.816
0.3375	-0.481	0.771	0.3375	-0.152	0.819
0.3618	-0.465	0.773	0.3618	-0.163	0.818
0.3873	-0.418	0.780	0.3873	-0.148	0.820
0.4124	-0.410	0.781	0.4124	-0.153	0.819
0.4371	-0.364	0.788	0.4371	-0.137	0.821
0.4621	-0.358	0.789	0.4621	-0.138	0.821
0.4871	-0.318	0.795	0.4871	-0.125	0.823
0.5123	-0.307	0.796	0.5123	-0.123	0.824
0.5371	-0.266	0.803	0.5371	-0.108	0.826
0.5621	-0.258	0.804	0.5621	-0.104	0.826
0.5873	-0.222	0.809	0.5873	-0.091	0.828
0.6122	-0.207	0.811	0.6122	-0.086	0.829
0.6371	-0.172	0.816	0.6371	-0.072	0.831
0.6661	-0.161	0.818	0.6661	-0.068	0.832
0.6869	-0.140	0.821	0.6869	-0.055	0.834
0.7121	-0.125	0.823	0.7121	-0.052	0.834
0.7369	-0.094	0.828	0.7369	-0.038	0.836
0.7623	-0.079	0.830	0.7623	-0.034	0.837
0.7873	-0.046	0.835	0.7873	-0.019	0.839
0.8117	-0.027	0.838	0.8117	-0.018	0.839
0.8373	0.002	0.842	0.8373	-0.004	0.841
0.8619	0.024	0.846	0.8619	0.000	0.842
0.8869	0.059	0.851	0.8869	0.017	0.844
0.9120	0.082	0.854	0.9120	0.037	0.847
0.9416	0.123	0.860	0.9416	0.093	0.856

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 5.00 REY 0.44×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3853 CM = 0.0175

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-1.190	0.668	0.0119	0.827	0.965
0.0371	-1.277	0.655	0.0371	0.571	0.898
0.0623	-1.199	0.666	0.0623	0.170	0.858
0.0872	-1.326	0.692	0.0872	0.090	0.856
0.1122	-0.960	0.702	0.1122	0.031	0.848
0.1372	-0.855	0.717	0.1372	-0.015	0.841
0.1623	-0.829	0.721	0.1623	-0.050	0.836
0.1872	-0.763	0.731	0.1872	-0.081	0.831
0.2122	-0.744	0.734	0.2122	-0.102	0.828
0.2372	-0.677	0.743	0.2372	-0.113	0.826
0.2620	-0.647	0.748	0.2620	-0.118	0.826
0.2872	-2.587	0.757	0.2872	-0.119	0.825
0.3122	-0.565	0.760	0.3122	-0.121	0.825
0.3375	-0.512	0.768	0.3375	-0.122	0.825
0.3618	-0.495	0.770	0.3618	-0.118	0.826
0.3873	-0.447	0.777	0.3873	-0.117	0.826
0.4124	-0.434	0.779	0.4124	-0.112	0.827
0.4371	-0.365	0.786	0.4371	-0.109	0.827
0.4621	-0.377	0.788	0.4621	-0.105	0.828
0.4871	-0.337	0.794	0.4871	-0.100	0.828
0.5120	-0.322	0.796	0.5120	-0.093	0.829
0.5371	-0.278	0.802	0.5371	-0.083	0.831
0.5621	-0.264	0.804	0.5621	-0.074	0.832
0.5870	-0.223	0.810	0.5870	-0.070	0.833
0.6122	-0.213	0.812	0.6122	-0.062	0.834
0.6371	-0.178	0.817	0.6371	-0.053	0.835
0.6661	-0.167	0.819	0.6661	-0.043	0.837
0.6869	-0.146	0.822	0.6869	-0.038	0.837
0.7121	-0.124	0.825	0.7121	-0.031	0.839
0.7369	-0.096	0.829	0.7369	-0.023	0.840
0.7620	-0.079	0.832	0.7620	-0.012	0.841
0.7870	-0.048	0.836	0.7870	-0.006	0.842
0.8117	-0.027	0.839	0.8117	0.006	0.844
0.8370	0.002	0.843	0.8370	0.013	0.845
0.8619	0.027	0.847	0.8619	0.022	0.846
0.8869	0.056	0.851	0.8869	0.028	0.847
0.9120	0.082	0.855	0.9120	0.040	0.849
0.9416	0.121	0.861	0.9416	0.060	0.852

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 6.00 REY 0.44×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.4966 CM = 0.0156

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-1.615	0.606	0.0119	0.901	0.976
0.0371	-1.570	0.612	0.0371	0.482	0.914
0.0623	-1.436	0.632	0.0623	0.279	0.864
0.0872	-1.221	0.664	0.0872	0.193	0.871
0.1122	-1.102	0.681	0.1122	0.118	0.860
0.1372	-0.970	0.701	0.1372	0.073	0.854
0.1623	-0.938	0.705	0.1623	0.028	0.847
0.1872	-0.861	0.717	0.1872	-0.004	0.842
0.2122	-0.833	0.722	0.2122	-0.028	0.840
0.2372	-0.756	0.733	0.2372	-0.043	0.837
0.2623	-0.723	0.738	0.2623	-0.053	0.836
0.2872	-0.653	0.748	0.2872	-0.058	0.835
0.3122	-0.628	0.752	0.3122	-0.062	0.835
0.3375	-0.567	0.751	0.3375	-0.064	0.834
0.3618	-0.549	0.764	0.3618	-0.065	0.834
0.3873	-0.491	0.772	0.3873	-0.064	0.834
0.4124	-0.479	0.774	0.4124	-0.065	0.834
0.4371	-0.424	0.782	0.4371	-0.062	0.835
0.4621	-0.417	0.784	0.4621	-0.063	0.835
0.4871	-0.369	0.791	0.4871	-0.056	0.836
0.5120	-0.355	0.793	0.5120	-0.050	0.837
0.5371	-0.310	0.799	0.5371	-0.045	0.838
0.5621	-0.297	0.801	0.5621	-0.041	0.838
0.5870	-0.255	0.807	0.5870	-0.037	0.839
0.6122	-0.242	0.809	0.6122	-0.028	0.840
0.6371	-0.200	0.815	0.6371	-0.024	0.841
0.6661	-0.193	0.817	0.6661	-0.016	0.842
0.6869	-0.154	0.822	0.6869	-0.011	0.843
0.7121	-0.147	0.823	0.7121	-0.003	0.844
0.7369	-0.114	0.828	0.7369	0.004	0.845
0.7620	-0.098	0.830	0.7620	0.012	0.846
0.7870	-0.063	0.835	0.7870	0.024	0.848
0.8117	-0.046	0.838	0.8117	0.028	0.849
0.8370	-0.013	0.843	0.8370	0.041	0.850
0.8619	0.009	0.846	0.8619	0.050	0.852
0.8869	0.042	0.851	0.8869	0.056	0.853
0.9120	0.064	0.854	0.9120	0.063	0.854
0.9416	0.101	0.860	0.9416	0.073	0.855

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 7.00 REYNOLDS 0.44*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.6062 CM = 0.0084

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-2.046	0.542	0.0119	0.994	0.990
0.0371	-1.812	0.577	0.0371	0.804	0.932
0.0623	-1.706	0.592	0.0623	0.597	0.921
0.0872	-1.386	0.639	0.0872	0.292	0.886
0.1122	-1.205	0.666	0.1122	0.204	0.873
0.1372	-1.068	0.686	0.1372	0.160	0.866
0.1623	-1.029	0.692	0.1623	0.112	0.859
0.1872	-0.941	0.705	0.1872	0.073	0.854
0.2122	-0.902	0.711	0.2122	0.046	0.850
0.2372	-0.817	0.723	0.2372	0.026	0.847
0.2620	-0.763	0.728	0.2620	0.015	0.845
0.2872	-0.704	0.740	0.2872	0.006	0.844
0.3122	-0.680	0.743	0.3122	0.000	0.843
0.3375	-0.610	0.754	0.3375	-0.009	0.842
0.3618	-0.569	0.757	0.3618	-0.009	0.841
0.3873	-0.529	0.766	0.3873	-0.013	0.841
0.4124	-0.516	0.767	0.4124	-0.015	0.841
0.4371	-0.456	0.776	0.4371	-0.015	0.841
0.4621	-0.446	0.778	0.4621	-0.012	0.841
0.4871	-0.395	0.785	0.4871	-0.015	0.841
0.5120	-0.383	0.787	0.5120	-0.009	0.841
0.5371	-0.352	0.794	0.5371	-0.006	0.842
0.5621	-0.322	0.796	0.5621	0.000	0.843
0.5870	-0.276	0.803	0.5870	0.002	0.843
0.6122	-0.261	0.805	0.6122	0.006	0.844
0.6371	-0.217	0.811	0.6371	0.011	0.844
0.6661	-0.206	0.813	0.6661	0.019	0.846
0.6869	-0.169	0.818	0.6869	0.023	0.846
0.7121	-0.161	0.820	0.7121	0.031	0.847
0.7369	-0.128	0.825	0.7369	0.036	0.848
0.7620	-0.112	0.827	0.7620	0.046	0.850
0.7870	-0.077	0.832	0.7870	0.051	0.850
0.8117	-0.058	0.835	0.8117	0.062	0.852
0.8370	-0.025	0.840	0.8370	0.073	0.854
0.8619	-0.009	0.842	0.8619	0.081	0.855
0.8869	0.025	0.847	0.8869	0.092	0.856
0.9120	0.042	0.850	0.9120	0.099	0.857
0.9416	0.079	0.855	0.9416	0.107	0.859

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.501 ALPHA 8.00 REYNOLDS 0.44*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.6711 CM = 0.0143

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-2.424	0.483	0.0119	1.016	0.993
0.0371	-2.067	0.536	0.0371	0.676	0.943
0.0623	-2.033	0.541	0.0623	0.471	0.912
0.0872	-1.460	0.626	0.0872	0.365	0.897
0.1122	-1.281	0.652	0.1122	0.273	0.883
0.1372	-1.143	0.673	0.1372	0.224	0.876
0.1623	-1.209	0.681	0.1623	0.161	0.867
0.1872	-0.996	0.694	0.1872	0.134	0.863
0.2122	-0.950	0.701	0.2122	0.102	0.858
0.2372	-0.855	0.715	0.2372	0.081	0.855
0.2620	-0.809	0.722	0.2620	0.065	0.852
0.2872	-0.733	0.733	0.2872	0.053	0.851
0.3122	-0.701	0.738	0.3122	0.043	0.849
0.3375	-0.634	0.748	0.3375	0.034	0.848
0.3618	-0.602	0.753	0.3618	0.031	0.847
0.3873	-0.545	0.761	0.3873	0.023	0.846
0.4124	-0.523	0.764	0.4124	0.022	0.846
0.4371	-0.464	0.773	0.4371	0.019	0.846
0.4621	-0.445	0.776	0.4621	0.015	0.845
0.4871	-0.398	0.783	0.4871	0.019	0.846
0.5120	-0.379	0.786	0.5120	0.019	0.846
0.5371	-0.333	0.793	0.5371	0.019	0.846
0.5621	-0.313	0.796	0.5621	0.022	0.846
0.5870	-0.271	0.802	0.5870	0.023	0.846
0.6122	-0.253	0.805	0.6122	0.031	0.847
0.6371	-0.211	0.811	0.6371	0.032	0.848
0.6661	-0.195	0.813	0.6661	0.034	0.848
0.6869	-0.162	0.818	0.6869	0.038	0.849
0.7121	-0.144	0.821	0.7121	0.043	0.849
0.7369	-0.118	0.825	0.7369	0.049	0.850
0.7620	-0.093	0.828	0.7620	0.053	0.851
0.7870	-0.064	0.833	0.7870	0.064	0.852
0.8117	-0.042	0.836	0.8117	0.071	0.853
0.8370	-0.014	0.840	0.8370	0.079	0.855
0.8619	0.006	0.843	0.8619	0.090	0.856
0.8869	0.035	0.847	0.8869	0.098	0.857
0.9120	0.054	0.850	0.9120	0.108	0.859
0.9416	0.085	0.855	0.9416	0.117	0.860

NACA-0012 243.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.701 ALPHA 0.00 REY 0.43*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0000 CM = 0.0000

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.174	0.763	0.0119	0.174	0.763
0.0371	-0.317	0.641	0.0371	-0.317	0.641
0.0623	-0.489	0.598	0.0623	-0.489	0.598
0.0872	-0.515	0.592	0.0872	-0.515	0.592
0.1122	-0.557	0.581	0.1122	-0.557	0.581
0.1372	-0.544	0.585	0.1372	-0.544	0.585
0.1623	-0.566	0.579	0.1623	-0.566	0.579
0.1872	-0.555	0.582	0.1872	-0.555	0.582
0.2122	-0.562	0.579	0.2122	-0.562	0.579
0.2372	-0.527	0.587	0.2372	-0.527	0.587
0.2620	-0.522	0.589	0.2620	-0.522	0.589
0.2872	-0.481	0.599	0.2872	-0.481	0.599
0.3122	-0.473	0.601	0.3122	-0.473	0.601
0.3375	-0.437	0.610	0.3375	-0.437	0.610
0.3618	-0.433	0.611	0.3618	-0.433	0.611
0.3873	-0.398	0.619	0.3873	-0.398	0.619
0.4124	-0.394	0.621	0.4124	-0.394	0.621
0.4371	-0.365	0.628	0.4371	-0.365	0.628
0.4621	-0.361	0.629	0.4621	-0.361	0.629
0.4871	-0.327	0.637	0.4871	-0.327	0.637
0.5120	-0.302	0.643	0.5120	-0.302	0.643
0.5371	-0.242	0.658	0.5371	-0.242	0.658
0.5621	-0.218	0.664	0.5621	-0.218	0.664
0.5870	-0.187	0.672	0.5870	-0.187	0.672
0.6122	-0.166	0.677	0.6122	-0.166	0.677
0.6371	-0.142	0.683	0.6371	-0.142	0.683
0.6661	-0.129	0.687	0.6661	-0.129	0.687
0.6869	-0.111	0.691	0.6869	-0.111	0.691
0.7121	-0.098	0.694	0.7121	-0.098	0.694
0.7369	-0.077	0.699	0.7369	-0.077	0.699
0.7620	-0.056	0.705	0.7620	-0.056	0.705
0.7870	-0.034	0.710	0.7870	-0.034	0.710
0.8117	-0.016	0.715	0.8117	-0.016	0.715
0.8370	0.008	0.721	0.8370	0.008	0.721
0.8619	0.033	0.727	0.8619	0.033	0.727
0.8869	0.061	0.734	0.8869	0.061	0.734
0.9120	0.087	0.740	0.9120	0.087	0.740
0.9416	0.132	0.752	0.9416	0.132	0.752

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.700 ALPHA 1.00 REYNOLDS 0.43*10⁶

INTEGRATED FORCE COEFFICIENTS

C_D = 0.0913 C_M = 0.0051

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	C _P	P/H	X/C	C _P	P/H
0.0119	0.016	0.725	0.0119	0.344	0.805
0.0371	-0.492	0.599	0.0371	-0.153	0.682
0.0623	-0.662	0.557	0.0623	-0.328	0.639
0.0872	-0.670	0.555	0.0872	-0.377	0.627
0.1122	-0.698	0.548	0.1122	-0.427	0.614
0.1372	-0.674	0.554	0.1372	-0.427	0.614
0.1623	-0.686	0.551	0.1623	-0.457	0.607
0.1872	-0.662	0.557	0.1872	-0.459	0.606
0.2122	-0.662	0.557	0.2122	-0.474	0.603
0.2372	-0.618	0.568	0.2372	-0.451	0.608
0.2620	-0.606	0.571	0.2620	-0.448	0.609
0.2872	-0.562	0.582	0.2872	-0.416	0.617
0.3122	-0.549	0.585	0.3122	-0.408	0.619
0.3375	-0.508	0.595	0.3375	-0.378	0.626
0.3618	-0.493	0.599	0.3618	-0.373	0.628
0.3873	-0.441	0.611	0.3873	-0.346	0.634
0.4124	-0.415	0.618	0.4124	-0.340	0.636
0.4371	-0.365	0.630	0.4371	-0.315	0.642
0.4621	-0.342	0.636	0.4621	-0.307	0.644
0.4871	-0.299	0.647	0.4871	-0.284	0.650
0.5120	-0.276	0.652	0.5120	-0.278	0.651
0.5371	-0.243	0.660	0.5371	-0.253	0.657
0.5621	-0.229	0.664	0.5621	-0.233	0.662
0.5870	-0.204	0.670	0.5870	-0.195	0.672
0.6122	-0.184	0.675	0.6122	-0.167	0.679
0.6371	-0.157	0.682	0.6371	-0.140	0.685
0.6661	-0.141	0.686	0.6661	-0.118	0.691
0.6869	-0.125	0.690	0.6869	-0.107	0.694
0.7121	-0.108	0.694	0.7121	-0.087	0.698
0.7369	-0.088	0.699	0.7369	-0.068	0.703
0.7620	-0.064	0.705	0.7620	-0.047	0.708
0.7870	-0.041	0.711	0.7870	-0.026	0.713
0.8117	-0.016	0.717	0.8117	-0.007	0.718
0.8370	0.008	0.723	0.8370	0.013	0.723
0.8619	0.035	0.729	0.8619	0.038	0.729
0.8869	0.063	0.736	0.8869	0.060	0.735
0.9120	0.087	0.742	0.9120	0.090	0.742
0.9416	0.136	0.754	0.9416	0.127	0.751

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.700 ALPHA 2.00 REYNOLDS 0.43*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1770 CM = 0.0108

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.176	0.677	0.0119	0.486	0.845
0.0371	-0.675	0.553	0.0371	-0.013	0.716
0.0623	-0.850	0.510	0.0623	-0.191	0.672
0.0872	-0.853	0.514	0.0872	-0.252	0.657
0.1122	-0.850	0.510	0.1122	-0.314	0.642
0.1372	-0.796	0.523	0.1372	-0.322	0.640
0.1623	-0.796	0.523	0.1623	-0.361	0.630
0.1872	-0.754	0.533	0.1872	-0.369	0.628
0.2122	-0.749	0.535	0.2122	-0.394	0.622
0.2372	-0.698	0.547	0.2372	-0.374	0.627
0.2620	-0.683	0.551	0.2620	-0.380	0.625
0.2872	-0.635	0.563	0.2872	-0.351	0.632
0.3122	-0.623	0.566	0.3122	-0.351	0.632
0.3375	-0.571	0.579	0.3375	-0.326	0.638
0.3618	-0.541	0.586	0.3618	-0.325	0.639
0.3873	-0.458	0.607	0.3873	-0.297	0.646
0.4124	-0.398	0.622	0.4124	-0.297	0.646
0.4371	-0.335	0.637	0.4371	-0.266	0.653
0.4621	-0.332	0.638	0.4621	-0.264	0.654
0.4871	-0.305	0.645	0.4871	-0.240	0.660
0.5120	-0.290	0.648	0.5120	-0.238	0.660
0.5371	-0.256	0.657	0.5371	-0.213	0.667
0.5621	-0.241	0.661	0.5621	-0.212	0.667
0.5870	-0.216	0.667	0.5870	-0.185	0.674
0.6122	-0.194	0.672	0.6122	-0.172	0.677
0.6371	-0.156	0.679	0.6371	-0.143	0.684
0.6661	-0.150	0.683	0.6661	-0.127	0.688
0.6869	-0.127	0.689	0.6869	-0.107	0.693
0.7121	-0.112	0.692	0.7121	-0.090	0.697
0.7369	-0.087	0.699	0.7369	-0.067	0.703
0.7620	-0.068	0.704	0.7620	-0.042	0.709
0.7870	-0.039	0.711	0.7870	-0.021	0.714
0.8117	-0.014	0.717	0.8117	-0.002	0.719
0.8370	0.013	0.724	0.8370	0.021	0.725
0.8619	0.037	0.730	0.8619	0.040	0.729
0.8869	0.069	0.737	0.8869	0.065	0.736
0.9120	0.091	0.743	0.9120	0.087	0.741
0.9416	0.142	0.755	0.9416	0.127	0.751

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.700 ALPHA 3.00 REY 0.43⁶ × 10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2689 CM = 0.0182

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.356	0.653	0.0119	0.615	0.873
0.0571	-0.349	0.511	0.0371	0.123	0.751
0.0623	-1.102	0.448	0.0523	-0.062	0.705
0.0872	-1.054	0.460	0.0872	-0.136	0.657
0.1122	-1.058	0.459	0.1122	-0.224	0.672
0.1372	-0.965	0.482	0.1372	-0.224	0.665
0.1623	-0.948	0.487	0.1623	-0.268	0.654
0.1872	-0.888	0.501	0.1872	-0.284	0.650
0.2122	-0.885	0.502	0.2122	-0.311	0.643
0.2372	-0.831	0.515	0.2372	-0.331	0.646
0.2623	-0.821	0.518	0.2623	-0.309	0.644
0.2872	-0.760	0.533	0.2872	-0.286	0.650
0.3122	-0.711	0.545	0.3122	-0.294	0.647
0.3375	-0.551	0.585	0.3375	-0.271	0.653
0.3618	-0.459	0.607	0.3618	-0.273	0.653
0.3873	-0.404	0.621	0.3873	-0.250	0.658
0.4124	-0.405	0.621	0.4124	-0.252	0.656
0.4371	-0.366	0.630	0.4371	-0.226	0.665
0.4621	-0.364	0.630	0.4621	-0.228	0.664
0.4871	-0.326	0.639	0.4871	-0.203	0.670
0.5120	-0.313	0.642	0.5120	-0.199	0.671
0.5371	-0.271	0.653	0.5371	-0.177	0.677
0.5621	-0.259	0.656	0.5621	-0.176	0.677
0.5870	-0.223	0.664	0.5870	-0.154	0.682
0.6122	-0.203	0.670	0.6122	-0.150	0.683
0.6371	-0.172	0.677	0.6371	-0.131	0.688
0.6661	-0.156	0.681	0.6661	-0.126	0.689
0.6869	-0.128	0.688	0.6869	-0.113	0.692
0.7121	-0.114	0.691	0.7121	-0.105	0.695
0.7369	-0.085	0.698	0.7369	-0.087	0.699
0.7620	-0.063	0.703	0.7620	-0.067	0.704
0.7870	-0.032	0.711	0.7870	-0.036	0.711
0.8117	-0.012	0.716	0.8117	-0.007	0.719
0.8370	0.021	0.724	0.8370	0.021	0.726
0.8619	0.044	0.730	0.8619	0.045	0.732
0.8869	0.060	0.739	0.8869	0.072	0.738
0.9120	0.098	0.743	0.9120	0.088	0.742
0.9416	0.152	0.757	0.9416	0.128	0.752

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.702 ALPHA 4.00 REY 0.43*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.3603 CM = 0.0253

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.530	0.588	0.0119	0.124	0.900
0.0371	-0.984	0.475	0.0371	0.244	0.781
0.0623	-1.260	0.406	0.0623	0.055	0.734
0.0872	-1.276	0.403	0.0872	-0.039	0.711
0.1122	-1.286	0.400	0.1122	-0.102	0.695
0.1372	-1.194	0.423	0.1372	-0.132	0.688
0.1623	-1.165	0.430	0.1623	-0.180	0.676
0.1872	-1.103	0.445	0.1872	-0.201	0.671
0.2122	-1.097	0.447	0.2122	-0.233	0.663
0.2372	-1.034	0.463	0.2372	-0.229	0.664
0.2620	-0.997	0.472	0.2620	-0.240	0.661
0.2872	-0.799	0.521	0.2872	-0.224	0.665
0.3122	-0.607	0.569	0.3122	-0.233	0.663
0.3375	-0.485	0.599	0.3375	-0.217	0.667
0.3618	-0.463	0.605	0.3618	-0.221	0.666
0.3873	-0.429	0.613	0.3873	-0.203	0.672
0.4124	-0.421	0.615	0.4124	-0.204	0.670
0.4371	-0.379	0.625	0.4371	-0.186	0.674
0.4621	-0.372	0.627	0.4621	-0.188	0.674
0.4871	-0.336	0.636	0.4871	-0.165	0.680
0.5120	-0.319	0.640	0.5120	-0.164	0.682
0.5371	-0.277	0.651	0.5371	-0.144	0.685
0.5621	-0.260	0.655	0.5621	-0.140	0.686
0.5870	-0.226	0.663	0.5870	-0.124	0.690
0.6122	-0.205	0.669	0.6122	-0.119	0.691
0.6371	-0.171	0.677	0.6371	-0.110	0.693
0.6661	-0.153	0.681	0.6661	-0.100	0.696
0.6869	-0.123	0.689	0.6869	-0.092	0.698
0.7121	-0.112	0.692	0.7121	-0.083	0.700
0.7369	-0.080	0.700	0.7369	-0.074	0.702
0.7620	-0.058	0.705	0.7620	-0.067	0.704
0.7870	-0.026	0.710	0.7870	-0.056	0.707
0.8117	-0.002	0.719	0.8117	-0.036	0.712
0.8370	0.027	0.726	0.8370	-0.008	0.718
0.8619	0.051	0.732	0.8619	0.026	0.727
0.8869	0.085	0.741	0.8869	0.064	0.736
0.9120	0.102	0.745	0.9120	0.088	0.742
0.9416	0.132	0.757	0.9416	0.129	0.752

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.698 ALPHA 5.00 REY 0.43⁶ × 10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4446 CM = 0.0311

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	-0.704	0.549	0.0119	0.813	0.922
0.0371	-1.119	0.446	0.0371	0.349	0.867
0.0623	-1.398	0.377	0.0623	0.153	0.759
0.0872	-1.410	0.374	0.0872	0.059	0.735
0.1122	-1.431	0.369	0.1122	-0.007	0.719
0.1372	-1.322	0.396	0.1372	-0.054	0.707
0.1623	-1.312	0.399	0.1623	-0.095	0.697
0.1872	-1.257	0.412	0.1872	-0.123	0.690
0.2122	-1.272	0.408	0.2122	-0.154	0.682
0.2372	-1.225	0.420	0.2372	-0.157	0.681
0.2620	-1.160	0.436	0.2620	-0.173	0.677
0.2872	-0.913	0.497	0.2872	-0.165	0.679
0.3122	-0.699	0.550	0.3122	-0.176	0.676
0.3375	-0.552	0.586	0.3375	-0.164	0.679
0.3618	-0.493	0.600	0.3618	-0.169	0.678
0.3873	-0.438	0.614	0.3873	-0.155	0.681
0.4124	-0.416	0.619	0.4124	-0.162	0.680
0.4371	-0.375	0.630	0.4371	-0.144	0.684
0.4621	-0.362	0.633	0.4621	-0.150	0.683
0.4871	-0.328	0.641	0.4871	-0.131	0.687
0.5120	-0.309	0.646	0.5120	-0.128	0.688
0.5371	-0.272	0.655	0.5371	-0.118	0.691
0.5621	-0.255	0.659	0.5621	-0.112	0.692
0.5870	-0.221	0.668	0.5870	-0.103	0.694
0.6122	-0.199	0.673	0.6122	-0.093	0.697
0.6371	-0.166	0.681	0.6371	-0.082	0.700
0.6661	-0.145	0.686	0.6661	-0.071	0.702
0.6869	-0.121	0.692	0.6869	-0.064	0.704
0.7121	-0.101	0.697	0.7121	-0.057	0.706
0.7369	-0.076	0.703	0.7369	-0.047	0.708
0.7620	-0.051	0.709	0.7620	-0.038	0.710
0.7870	-0.024	0.716	0.7870	-0.029	0.713
0.8117	0.002	0.723	0.8117	-0.024	0.714
0.8370	0.029	0.729	0.8370	-0.015	0.716
0.8619	0.054	0.735	0.8619	-0.007	0.718
0.8869	0.084	0.743	0.8869	0.010	0.722
0.9120	0.098	0.746	0.9120	0.040	0.730
0.9416	0.142	0.757	0.9416	0.095	0.743

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.000 ALPHA 0.00 REYNOLDS 0.44*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0000 CM = 0.0000

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.289	0.741	0.0119	0.289	0.741
0.0371	-0.213	0.593	0.0371	-0.213	0.593
0.0623	-0.424	0.531	0.0623	-0.424	0.531
0.0872	-0.488	0.513	0.0872	-0.488	0.513
0.1122	-0.573	0.487	0.1122	-0.573	0.487
0.1372	-0.591	0.482	0.1372	-0.591	0.482
0.1623	-0.641	0.468	0.1623	-0.641	0.468
0.1872	-0.666	0.460	0.1872	-0.666	0.460
0.2122	-0.732	0.441	0.2122	-0.732	0.441
0.2372	-0.742	0.438	0.2372	-0.742	0.438
0.2620	-0.755	0.434	0.2620	-0.755	0.434
0.2872	-0.694	0.452	0.2872	-0.694	0.452
0.3122	-0.675	0.458	0.3122	-0.675	0.458
0.3375	-0.634	0.469	0.3375	-0.634	0.469
0.3619	-0.641	0.468	0.3619	-0.641	0.468
0.3873	-0.618	0.474	0.3873	-0.618	0.474
0.4124	-0.628	0.471	0.4124	-0.628	0.471
0.4371	-0.604	0.478	0.4371	-0.604	0.478
0.4621	-0.614	0.476	0.4621	-0.614	0.476
0.4871	-0.579	0.486	0.4871	-0.579	0.486
0.5120	-0.458	0.521	0.5120	-0.458	0.521
0.5371	-0.235	0.587	0.5371	-0.235	0.587
0.5621	-0.145	0.613	0.5621	-0.145	0.613
0.5870	-0.128	0.613	0.5870	-0.128	0.613
0.6122	-0.116	0.622	0.6122	-0.116	0.622
0.6371	-0.106	0.625	0.6371	-0.106	0.625
0.6661	-0.095	0.628	0.6661	-0.095	0.628
0.6869	-0.086	0.631	0.6869	-0.086	0.631
0.7121	-0.073	0.635	0.7121	-0.073	0.635
0.7369	-0.051	0.641	0.7369	-0.051	0.641
0.7620	-0.032	0.647	0.7620	-0.032	0.647
0.7870	-0.010	0.653	0.7870	-0.010	0.653
0.8117	0.013	0.650	0.8117	0.013	0.650
0.8370	0.036	0.667	0.8370	0.036	0.667
0.8619	0.061	0.674	0.8619	0.061	0.674
0.8869	0.090	0.682	0.8869	0.090	0.682
0.9120	0.110	0.688	0.9120	0.110	0.688
0.9416	0.158	0.702	0.9416	0.158	0.702

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.840 ALPHA 1.00 REYNOLDS 0.44*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0983 CM = 0.0024

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.174	0.707	0.0119	0.394	0.772
0.0371	-0.325	0.561	0.0371	-0.099	0.627
0.0623	-0.541	0.497	0.0623	-0.312	0.564
0.0872	-0.590	0.483	0.0872	-0.377	0.545
0.1122	-0.617	0.457	0.1122	-0.468	0.519
0.1372	-0.697	0.421	0.1372	-0.483	0.514
0.1623	-0.746	0.437	0.1623	-0.546	0.495
0.1872	-0.763	0.432	0.1872	-0.573	0.487
0.2122	-0.819	0.415	0.2122	-0.644	0.465
0.2372	-0.824	0.414	0.2372	-0.646	0.465
0.2620	-0.830	0.412	0.2620	-0.658	0.461
0.2872	-0.756	0.431	0.2872	-0.591	0.451
0.3122	-0.746	0.437	0.3122	-0.576	0.455
0.3375	-0.714	0.446	0.3375	-0.527	0.502
0.3618	-0.723	0.444	0.3618	-0.530	0.499
0.3873	-0.702	0.450	0.3873	-0.491	0.512
0.4124	-0.714	0.446	0.4124	-0.498	0.528
0.4371	-0.698	0.451	0.4371	-0.465	0.518
0.4621	-0.714	0.446	0.4621	-0.474	0.514
0.4871	-0.702	0.450	0.4871	-0.442	0.523
0.5120	-0.553	0.493	0.5120	-0.443	0.523
0.5371	-0.317	0.563	0.5371	-0.345	0.552
0.5621	-0.195	0.599	0.5621	-0.228	0.589
0.5870	-0.143	0.614	0.5870	-0.129	0.616
0.6122	-0.113	0.623	0.6122	-0.113	0.622
0.6371	-0.394	0.628	0.6371	-0.097	0.625
0.6661	-0.075	0.634	0.6661	-0.099	0.625
0.6869	-0.068	0.636	0.6869	-0.085	0.629
0.7121	-0.050	0.641	0.7121	-0.073	0.632
0.7369	-0.032	0.647	0.7369	-0.051	0.639
0.7620	-0.013	0.652	0.7620	-0.038	0.643
0.7870	0.007	0.658	0.7870	-0.009	0.651
0.8117	0.029	0.665	0.8117	0.004	0.655
0.8370	0.050	0.671	0.8370	0.035	0.664
0.8619	0.075	0.678	0.8619	0.055	0.670
0.8869	0.101	0.686	0.8869	0.090	0.680
0.9120	0.122	0.692	0.9120	0.103	0.684
0.9416	0.164	0.704	0.9416	0.157	0.703

NACA-0012 203.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.802 ALPHA 2.00 REY 0.44*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1675 CM = 0.0012

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0119	0.385	0.688	0.0119	0.503	0.884
0.0371	-0.403	0.536	0.0371	-0.003	0.656
0.0623	-0.638	0.467	0.0623	-0.197	0.599
0.0872	-0.674	0.456	0.0872	-0.280	0.574
0.1122	-0.752	0.453	0.1122	-0.361	0.551
0.1372	-0.770	0.427	0.1372	-0.390	0.542
0.1623	-0.825	0.411	0.1623	-0.450	0.525
0.1872	-0.838	0.408	0.1872	-0.465	0.514
0.2122	-0.868	0.393	0.2122	-0.534	0.500
0.2372	-0.864	0.400	0.2372	-0.529	0.501
0.2623	-0.542	0.406	0.2623	-0.528	0.502
0.2872	-0.765	0.423	0.2872	-0.489	0.513
0.3122	-0.788	0.422	0.3122	-0.481	0.515
0.3375	-0.760	0.430	0.3375	-0.444	0.526
0.3618	-0.773	0.427	0.3618	-0.435	0.529
0.3873	-0.752	0.453	0.3873	-0.400	0.539
0.4124	-0.771	0.427	0.4124	-0.392	0.541
0.4371	-0.763	0.430	0.4371	-0.362	0.550
0.4621	-0.790	0.421	0.4621	-0.356	0.552
0.4871	-0.787	0.446	0.4871	-0.334	0.558
0.5123	-0.590	0.460	0.5123	-0.331	0.559
0.5371	-0.455	0.520	0.5371	-0.310	0.566
0.5621	-0.370	0.545	0.5621	-0.310	0.566
0.5870	-0.266	0.570	0.5870	-0.283	0.573
0.6122	-0.233	0.586	0.6122	-0.253	0.583
0.6371	-0.175	0.603	0.6371	-0.172	0.606
0.6661	-0.129	0.616	0.6661	-0.110	0.624
0.6869	-0.099	0.625	0.6869	-0.089	0.632
0.7121	-0.068	0.633	0.7121	-0.066	0.637
0.7369	-0.041	0.641	0.7369	-0.053	0.641
0.7623	-0.012	0.652			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 0.00 REYNOLDS 0.86×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0214 CM = 0.0036 CDW = 0.0109

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.195	0.814	0.0757	-0.454	0.775
0.0787	-0.485	0.771	0.1294	-0.485	0.771
0.1299	-0.469	0.773	0.1830	-0.449	0.776
0.1813	-0.457	0.775	0.2368	-0.428	0.779
0.2307	-0.432	0.779	0.2864	-0.392	0.784
0.2836	-0.389	0.785	0.3383	-0.350	0.791
0.3337	-0.360	0.789	0.3904	-0.321	0.795
0.3858	-0.325	0.795	0.4376	-0.281	0.801
0.4358	-0.292	0.799	0.4870	-0.253	0.805
0.4860	-0.259	0.804	0.5371	-0.215	0.811
0.5380	-0.213	0.811	0.5879	-0.183	0.815
0.5882	-0.174	0.817	0.6371	-0.149	0.820
0.6395	-0.148	0.821	0.6874	-0.126	0.824
0.6894	-0.115	0.826	0.7359	-0.094	0.826
0.7443	-0.081	0.831			
0.7954	-0.043	0.836			
0.8414	-0.006	0.842			
0.8891	0.036	0.848			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.501 ALPHA 1.00 REYNOLDS 0.67×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0945 CM = 0.0012 CDW = 0.0134

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.445	0.777	0.0757	-0.309	0.79
0.0787	-0.636	0.748	0.1294	-0.379	0.78
0.1299	-0.580	0.756	0.1830	-0.365	0.78
0.1813	-0.544	0.762	0.2368	-0.356	0.79
0.2307	-0.504	0.768	0.2864	-0.333	0.79
0.2836	-0.448	0.776	0.3383	-0.298	0.79
0.3337	-0.411	0.781	0.3904	-0.277	0.82
0.3858	-0.367	0.788	0.4376	-0.244	0.83
0.4358	-0.328	0.794	0.4870	-0.220	0.81
0.4860	-0.292	0.799	0.5371	-0.187	0.81
0.5380	-0.240	0.807	0.5879	-0.159	0.81
0.5862	-0.198	0.813	0.6371	-0.130	0.82
0.6395	-0.166	0.818	0.6874	-0.110	0.82
0.6894	-0.129	0.823	0.7359	-0.083	0.83
0.7443	-0.090	0.829			
0.7954	-0.048	0.836			
0.8414	-0.008	0.841			
0.8891	0.035	0.848			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 2.00 REYNOLDS 0.86×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1932 CH = 0.0026 CDW = 0.0144

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.724	0.736	0.0757	-0.171	0.817
0.0787	-0.795	0.726	0.1294	-0.274	0.802
0.1299	-0.697	0.740	0.1830	-0.279	0.801
0.1813	-0.636	0.749	0.2368	-0.288	0.800
0.2307	-0.561	0.757	0.2864	-0.273	0.802
0.2836	-0.511	0.768	0.3383	-0.248	0.806
0.3337	-0.463	0.774	0.3904	-0.234	0.808
0.3858	-0.410	0.782	0.4376	-0.207	0.812
0.4358	-0.365	0.789	0.4870	-0.188	0.815
0.4860	-0.322	0.795	0.5371	-0.160	0.819
0.5360	-0.266	0.803	0.5879	-0.137	0.822
0.5862	-0.219	0.810	0.6371	-0.109	0.826
0.6395	-0.184	0.816	0.6874	-0.095	0.828
0.6894	-0.143	0.822	0.7359	-0.069	0.832
0.7443	-0.100	0.828			
0.7954	-0.054	0.835			
0.8414	-0.012	0.841			
0.8891	0.035	0.848			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 3.00 REY 0.86×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2949 CM = 0.0043 CDW = 0.0150

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.030	0.690	0.0757	-0.040	0.836
0.0787	-0.955	0.701	0.1294	-0.171	0.817
0.1299	-0.814	0.722	0.1830	-0.197	0.813
0.1813	-0.728	0.735	0.2368	-0.219	0.810
0.2307	-0.657	0.746	0.2864	-0.212	0.811
0.2836	-0.573	0.758	0.3383	-0.198	0.813
0.3337	-0.516	0.766	0.3904	-0.190	0.814
0.3858	-0.455	0.775	0.4376	-0.168	0.817
0.4358	-0.401	0.783	0.4870	-0.155	0.819
0.4860	-0.352	0.790	0.5371	-0.131	0.823
0.5380	-0.290	0.800	0.5879	-0.111	0.826
0.5882	-0.239	0.807	0.6371	-0.090	0.829
0.6395	-0.200	0.813	0.6874	-0.077	0.831
0.6894	-0.155	0.820	0.7359	-0.056	0.834
0.7443	-0.107	0.827			
0.7954	-0.059	0.834			
0.8414	-0.015	0.840			
0.8891	0.035	0.848			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.501 ALPHA 4.00 REY 0.87×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3855 CM = 7.0064 CDW = 0.0140

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.365	0.639	0.0757	0.075	0.853
0.0787	-1.122	0.676	0.1294	-0.076	0.831
0.1299	-0.937	0.703	0.1830	-0.118	0.824
0.1813	-0.820	0.720	0.2358	-0.152	0.819
0.2307	-0.730	0.734	0.2864	-0.154	0.819
0.2836	-0.630	0.749	0.3383	-0.147	0.820
0.3337	-0.562	0.759	0.3904	-0.145	0.820
0.3858	-0.492	0.769	0.4376	-0.129	0.823
0.4358	-0.433	0.778	0.4870	-0.116	0.825
0.4850	-0.378	0.786	0.5371	-0.103	0.827
0.5350	-0.312	0.795	0.5879	-0.087	0.829
0.5852	-0.257	0.804	0.6371	-0.067	0.832
0.6395	-0.213	0.810	0.6874	-0.060	0.833
0.6894	-0.165	0.817	0.7359	-0.042	0.836
0.7443	-0.113	0.825			
0.7954	-0.063	0.833			
0.8414	-0.015	0.840			
0.8891	0.036	0.847			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 5.00 REYNOLDS 0.86*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4765 CM = 0.0094 CDW = 0.0155

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.718	0.590	0.0757	0.189	0.872
0.0767	-1.273	0.625	0.1294	0.016	0.846
0.1299	-1.043	0.689	0.1830	-0.039	0.838
0.1613	-0.902	0.710	0.2368	-0.084	0.832
0.2307	-0.794	0.726	0.2864	-0.096	0.830
0.2836	-0.681	0.743	0.3383	-0.095	0.830
0.3337	-0.601	0.754	0.3904	-0.100	0.829
0.3858	-0.526	0.766	0.4376	-0.091	0.831
0.4358	-0.463	0.776	0.4870	-0.087	0.831
0.4860	-0.400	0.785	0.5371	-0.072	0.833
0.5380	-0.331	0.795	0.5879	-0.062	0.835
0.5882	-0.272	0.804	0.6371	-0.045	0.837
0.6395	-0.223	0.811	0.6874	-0.042	0.838
0.6894	-0.171	0.819	0.7359	-0.029	0.840
0.7443	-0.118	0.827			
0.7954	-0.065	0.834			
0.8414	-0.015	0.842			
0.8891	0.035	0.849			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 6.00 REYNOLDS 2.86×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.5771 CM = 0.0126 CDW = 0.0150

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-2.143	2.528	0.0757	0.295	0.887
0.0787	-1.468	2.625	0.1294	0.107	0.859
0.1299	-1.146	2.675	0.1830	0.033	0.848
0.1813	-1.000	2.696	0.2368	-0.022	0.840
0.2307	-0.872	2.715	0.2864	-0.038	0.838
0.2836	-0.746	2.734	0.3383	-0.045	0.837
0.3337	-0.657	2.747	0.3904	-0.055	0.835
0.3828	-0.569	2.760	0.4376	-0.051	0.836
0.4358	-0.497	2.770	0.4870	-0.051	0.836
0.4860	-0.428	2.781	0.5371	-0.042	0.837
0.5380	-0.352	2.792	0.5879	-0.035	0.838
0.5862	-0.287	2.801	0.6371	-0.025	0.840
0.6395	-0.233	2.809	0.6874	-0.022	0.840
0.6894	-0.180	2.817	0.7359	-0.014	0.841
0.7443	-0.121	2.826			
0.7954	-0.066	2.834			
0.8414	-0.014	2.841			
0.8891	0.038	2.849			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 7.00 REYNOLDS 0.86×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.6740 CM = 0.0153 CDW = 0.0101

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-2.482	2.478	0.0757	0.391	0.921
0.0767	-1.708	0.592	0.1294	0.188	0.871
0.1299	-1.273	0.656	0.1830	0.103	0.859
0.1813	-1.088	0.683	0.2368	0.043	0.850
0.2307	-0.941	0.705	0.2864	0.019	0.846
0.2836	-0.800	0.726	0.3383	0.002	0.844
0.3357	-0.699	0.740	0.3904	-0.012	0.842
0.3858	-0.603	0.755	0.4376	-0.015	0.841
0.4358	-0.523	0.766	0.4870	-0.015	0.841
0.4860	-0.448	0.777	0.5371	-0.014	0.841
0.5380	-0.367	0.789	0.5879	-0.008	0.842
0.5882	-0.300	0.799	0.6371	-0.002	0.843
0.6395	-0.244	0.807	0.6874	-0.004	0.843
0.6894	-0.187	0.816	0.7359	0.003	0.844
0.7443	-0.125	0.825			
0.7954	-0.067	0.833			
0.8414	-0.015	0.841			
0.8891	0.038	0.849			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 8.00 REYNOLDS 0.86*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.7212 CM = 0.0177 CDW = 0.0125

UPPER SURFACE VALUES			LOWER SURFACE VALUES		
X/C	CP	P/H	X/C	CP	P/H
0.0207	-2.796	0.432	0.0757	0.467	0.913
0.0787	-1.590	0.609	0.1294	0.256	0.882
0.1299	-1.344	0.646	0.1830	0.164	0.868
0.1813	-1.144	0.675	0.2368	0.093	0.858
0.2307	-0.983	0.699	0.2864	0.061	0.853
0.2836	-0.831	0.721	0.3383	0.041	0.850
0.3337	-0.722	0.737	0.3904	0.024	0.848
0.3858	-0.622	0.752	0.4376	0.016	0.846
0.4358	-0.536	0.765	0.4870	0.010	0.846
0.4860	-0.456	0.777	0.5371	0.009	0.845
0.5360	-0.371	0.789	0.5879	0.012	0.846
0.5882	-0.302	0.799	0.6371	0.015	0.846
0.6395	-0.241	0.808	0.6874	0.009	0.846
0.6894	-0.182	0.817	0.7359	0.012	0.846
0.7443	-0.119	0.826			
0.7954	-0.062	0.835			
0.8414	-0.013	0.842			
0.8891	0.035	0.849			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.621 ALPHA 0.00 REY 0.81×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0197 CM = 0.0040 CDW = 0.0115

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.176	0.748	0.0757	-0.484	0.688
0.0787	-0.515	0.681	0.1294	-0.528	0.679
0.1299	-0.508	0.683	0.1830	-0.493	0.686
0.1813	-0.503	0.684	0.2368	-0.469	0.691
0.2307	-0.471	0.690	0.2864	-0.428	0.699
0.2836	-0.427	0.699	0.3383	-0.386	0.707
0.3337	-0.392	0.706	0.3904	-0.352	0.714
0.3818	-0.355	0.713	0.4376	-0.311	0.722
0.4358	-0.315	0.721	0.4870	-0.275	0.729
0.4860	-0.261	0.728	0.5371	-0.237	0.737
0.5380	-0.229	0.738	0.5879	-0.199	0.744
0.5882	-0.190	0.746	0.6371	-0.163	0.751
0.6395	-0.158	0.752	0.6874	-0.135	0.757
0.6894	-0.122	0.759	0.7359	-0.101	0.763
0.7443	-0.083	0.767			
0.7954	-0.043	0.775			
0.8414	-0.005	0.783			
0.8891	0.040	0.792			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.600 ALPHA 1.00 REYNOLDS 0.81×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0927 CM = 0.0016 CDW = 0.0116

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.415	0.702	0.0757	-0.338	0.716
0.0787	-0.671	0.651	0.1294	-0.418	0.720
0.1299	-0.626	0.660	0.1830	-0.405	0.733
0.1813	-0.595	0.666	0.2366	-0.397	0.735
0.2307	-0.549	0.675	0.2864	-0.368	0.710
0.2836	-0.490	0.667	0.3363	-0.334	0.717
0.3337	-0.445	0.696	0.3904	-0.308	0.722
0.3858	-0.397	0.705	0.4376	-0.271	0.729
0.4358	-0.350	0.714	0.4870	-0.244	0.735
0.4863	-0.309	0.723	0.5371	-0.210	0.742
0.5380	-0.253	0.734	0.5879	-0.177	0.748
0.5882	-0.209	0.742	0.6371	-0.145	0.754
0.6395	-0.175	0.749	0.6874	-0.122	0.759
0.6894	-0.134	0.757	0.7359	-0.092	0.765
0.7443	-0.093	0.765			
0.7954	-0.049	0.774			
0.8414	-0.008	0.782			
0.8891	0.040	0.792			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.631 ALPHA 2.00 REYNOLDS 6.81 $\times 10^6$

INTEGRATED FORCE COEFFICIENTS

CN = 0.2024 CM = 0.0045 CDW = 0.0142

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.704	0.643	0.0757	-0.185	0.745
0.0787	-0.864	0.611	0.1294	-0.299	0.722
0.1299	-0.771	0.630	0.1830	-0.310	0.720
0.1813	-0.759	0.642	0.2368	-0.319	0.718
0.2307	-0.640	0.656	0.2864	-0.302	0.722
0.2836	-0.563	0.671	0.3383	-0.279	0.726
0.3337	-0.505	0.682	0.3904	-0.261	0.730
0.3858	-0.448	0.694	0.4376	-0.232	0.736
0.4358	-0.342	0.704	0.4870	-0.211	0.740
0.4860	-0.344	0.714	0.5371	-0.182	0.746
0.5380	-0.282	0.726	0.5879	-0.154	0.751
0.5882	-0.253	0.736	0.6371	-0.125	0.757
0.6395	-0.191	0.744	0.6874	-0.106	0.760
0.6894	-0.143	0.753	0.7359	-0.079	0.766
0.7443	-0.101	0.762			
0.7954	-0.053	0.771			
0.8414	-0.009	0.780			
0.8891	0.041	0.790			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.598 ALPHA 3.00 REYNOLDS 0.81*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.3032 CM = 0.0074 CDW = 0.0153

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.021	0.584	0.0757	-0.051	0.775
0.0787	-1.057	0.577	0.1294	-0.188	0.748
0.1299	-0.900	0.608	0.1830	-0.220	0.741
0.1813	-0.806	0.626	0.2368	-0.243	0.737
0.2307	-0.712	0.645	0.2864	-0.238	0.738
0.2836	-0.622	0.662	0.3383	-0.223	0.741
0.3337	-0.554	0.676	0.3904	-0.212	0.743
0.3858	-0.485	0.689	0.4376	-0.190	0.747
0.4358	-0.424	0.701	0.4870	-0.174	0.750
0.4860	-0.370	0.712	0.5371	-0.151	0.755
0.5380	-0.324	0.725	0.5879	-0.128	0.759
0.5882	-0.249	0.736	0.6371	-0.104	0.764
0.6395	-0.204	0.745	0.6874	-0.089	0.767
0.6894	-0.157	0.754	0.7359	-0.065	0.771
0.7443	-0.107	0.764			
0.7954	-0.057	0.774			
0.8414	-0.009	0.783			
0.8891	0.041	0.793			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.620 ALPHA 4.00 REY 0.81*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4016 CH = 0.0109 COW = 0.0132

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.356	0.516	0.0757	0.068	0.798
0.0787	-1.267	0.534	0.1294	-0.069	0.767
0.1299	-1.041	0.578	0.1830	-0.136	0.757
0.1813	-0.898	0.606	0.2360	-0.172	0.750
0.2307	-0.786	0.628	0.2864	-0.175	0.750
0.2836	-0.682	0.649	0.3383	-0.169	0.751
0.3337	-0.599	0.665	0.3904	-0.166	0.751
0.3858	-0.523	0.681	0.4375	-0.150	0.755
0.4358	-0.454	0.694	0.4870	-0.138	0.757
0.4850	-0.393	0.706	0.5371	-0.120	0.760
0.5360	-0.321	0.721	0.5879	-0.101	0.764
0.5862	-0.263	0.732	0.6371	-0.082	0.765
0.6395	-0.213	0.742	0.6874	-0.072	0.770
0.6894	-0.163	0.752	0.7359	-0.052	0.774
0.7443	-0.109	0.762			
0.7954	-0.057	0.773			
0.8414	-0.007	0.782			
0.8891	0.044	0.793			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.599 ALPHA 5.00 REY 0.81×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.4981 CM = 0.0155 CDW = 0.0169

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.746	0.440	0.0757	0.180	0.820
0.0787	-1.527	0.483	0.1294	0.003	0.785
0.1299	-1.120	0.564	0.1830	-0.058	0.773
0.1813	-0.984	0.590	0.2360	-0.105	0.763
0.2307	-0.854	0.616	0.2864	-0.115	0.761
0.2836	-0.732	0.640	0.3383	-0.119	0.761
0.3337	-0.640	0.658	0.3904	-0.122	0.760
0.3858	-0.556	0.675	0.4375	-0.111	0.762
0.4358	-0.479	0.690	0.4870	-0.105	0.763
0.4860	-0.411	0.703	0.5371	-0.092	0.766
0.5380	-0.335	0.718	0.5879	-0.078	0.769
0.5882	-0.274	0.730	0.6371	-0.063	0.772
0.6395	-0.220	0.741	0.6874	-0.056	0.773
0.6894	-0.167	0.751	0.7359	-0.034	0.777
0.7443	-0.109	0.763			
0.7954	-0.055	0.773			
0.8414	-0.005	0.783			
0.8891	0.045	0.793			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.599 ALPHA 6.00 REYNOLDS 0.81×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.6052 CM = 0.0215 CDW = 0.0181

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-2.112	0.367	0.0757	0.280	0.839
0.0787	-1.782	0.432	0.1294	0.091	0.831
0.1299	-1.628	0.462	0.1830	0.016	0.786
0.1813	-1.321	0.582	0.2368	-0.039	0.775
0.2307	-0.903	0.606	0.2864	-0.057	0.772
0.2836	-0.777	0.631	0.3383	-0.066	0.770
0.3337	-0.677	0.650	0.3904	-0.077	0.768
0.3858	-0.584	0.669	0.4376	-0.071	0.769
0.4358	-0.501	0.684	0.4870	-0.071	0.769
0.4860	-0.429	0.699	0.5371	-0.060	0.771
0.5380	-0.349	0.715	0.5879	-0.052	0.773
0.5882	-0.285	0.727	0.6371	-0.040	0.775
0.6395	-0.227	0.739	0.6874	-0.038	0.775
0.6894	-0.171	0.750	0.7359	-0.025	0.778
0.7443	-0.112	0.762			
0.7954	-0.054	0.773			
0.8414	-0.005	0.783			
0.8891	0.045	0.793			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.599 ALPHA 7.00 REYNOLDS 0.81*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.6950 CM = 0.0274 CDW = 0.0176

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-2.320	0.326	0.0757	0.367	0.856
0.0787	-2.098	0.369	0.1294	0.168	0.817
0.1299	-1.984	0.392	0.1830	0.082	0.800
0.1813	-1.117	0.563	0.2368	0.021	0.768
0.2307	-0.910	0.604	0.2864	-0.005	0.783
0.2836	-0.786	0.629	0.3383	-0.021	0.779
0.3337	-0.687	0.648	0.3904	-0.035	0.777
0.3858	-0.594	0.667	0.4376	-0.035	0.777
0.4356	-0.509	0.683	0.4870	-0.038	0.776
0.4860	-0.434	0.698	0.5371	-0.035	0.777
0.5380	-0.353	0.714	0.5879	-0.027	0.778
0.5882	-0.286	0.727	0.6371	-0.020	0.780
0.6395	-0.228	0.738	0.6874	-0.020	0.780
0.6894	-0.170	0.750	0.7359	-0.012	0.781
0.7443	-0.111	0.762			
0.7954	-0.053	0.773			
0.8414	-0.006	0.782			
0.8891	0.042	0.792			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.650 ALPHA 0.00 REYNOLDS 0.82*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0230 CM = 0.0045 CDW = 0.0139

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.162	0.716	0.0757	-0.215	0.635
0.0787	-0.548	0.629	0.1294	-0.267	0.624
0.1299	-0.549	0.629	0.1830	-0.235	0.631
0.1813	-0.547	0.630	0.2368	-0.207	0.637
0.2307	-0.512	0.637	0.2864	-0.162	0.647
0.2836	-0.463	0.648	0.3383	-0.115	0.658
0.3337	-0.426	0.657	0.3904	-0.077	0.666
0.3858	-0.381	0.667	0.4376	-0.032	0.676
0.4358	-0.338	0.675	0.4870	-0.0294	0.685
0.4860	-0.300	0.684	0.5371	-0.0253	0.694
0.5360	-0.244	0.696	0.5879	-0.0211	0.703
0.5882	-0.201	0.706	0.6371	-0.0174	0.712
0.6395	-0.167	0.714	0.6874	-0.0142	0.719
0.6894	-0.130	0.722	0.7359	-0.0106	0.727
0.7443	-0.089	0.731			
0.7954	-0.047	0.741			
0.8414	-0.005	0.750			
0.8891	0.041	0.760			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.651 ALPHA 1.00 REYNOLDS 0.81×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1070 CM = 0.0028 CDW = 0.0109

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.406	0.561	0.0757	-0.341	0.675
0.0787	-0.729	0.589	0.1294	-0.434	0.655
0.1299	-0.689	0.598	0.1830	-0.429	0.656
0.1813	-0.655	0.606	0.2368	-0.422	0.657
0.2307	-0.599	0.618	0.2864	-0.393	0.664
0.2836	-0.533	0.633	0.3383	-0.357	0.672
0.3337	-0.462	0.644	0.3904	-0.328	0.678
0.3858	-0.428	0.656	0.4376	-0.289	0.687
0.4358	-0.377	0.668	0.4870	-0.259	0.694
0.4860	-0.331	0.678	0.5371	-0.223	0.702
0.5380	-0.270	0.692	0.5879	-0.188	0.709
0.5882	-0.222	0.702	0.6371	-0.154	0.717
0.6395	-0.183	0.711	0.6874	-0.129	0.723
0.6894	-0.141	0.720	0.7359	-0.095	0.730
0.7443	-0.097	0.730			
0.7954	-0.050	0.741			
0.8414	-0.006	0.751			
0.8891	0.344	0.762			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.650 ALPHA 2.00 REY 0.82×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2112 CM = 0.0059 CDW = 0.0146

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.673	0.603	0.0757	-0.199	0.727
0.0787	-0.939	0.543	0.1294	-0.318	0.681
0.1299	-0.834	0.567	0.1830	-0.335	0.677
0.1813	-0.770	0.581	0.2368	-0.342	0.675
0.2307	-0.689	0.599	0.2864	-0.327	0.679
0.2836	-0.600	0.619	0.3383	-0.301	0.655
0.3337	-0.537	0.633	0.3904	-0.281	0.659
0.3858	-0.472	0.647	0.4376	-0.250	0.696
0.4358	-0.414	0.660	0.4870	-0.227	0.741
0.4860	-0.350	0.672	0.5371	-0.195	0.748
0.5360	-0.293	0.687	0.5879	-0.166	0.715
0.5862	-0.241	0.699	0.6371	-0.134	0.722
0.6395	-0.197	0.708	0.6874	-0.116	0.726
0.6894	-0.151	0.719	0.7359	-0.084	0.733
0.7443	-0.103	0.729			
0.7954	-0.052	0.741			
0.8414	-0.007	0.751			
0.8891	0.045	0.762			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.651 ALPHA 3.00 REY 0.81×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3176 CM = 0.0111 CDW = 0.0161

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.956	0.538	0.0757	-0.058	0.740
0.0787	-1.212	0.481	0.1294	-0.201	0.709
0.1299	-0.983	0.532	0.1830	-0.237	0.701
0.1813	-0.868	0.558	0.2368	-0.261	0.695
0.2307	-0.758	0.583	0.2864	-0.255	0.697
0.2836	-0.657	0.605	0.3383	-0.240	0.700
0.3357	-0.578	0.623	0.3904	-0.229	0.702
0.3858	-0.506	0.639	0.4376	-0.205	0.708
0.4358	-0.441	0.654	0.4870	-0.168	0.712
0.4860	-0.381	0.667	0.5371	-0.163	0.717
0.5380	-0.310	0.683	0.5879	-0.139	0.723
0.5852	-0.254	0.696	0.6371	-0.115	0.728
0.6395	-0.205	0.707	0.6874	-0.098	0.732
0.6894	-0.156	0.718	0.7359	-0.072	0.738
0.7443	-0.103	0.729			
0.7954	-0.052	0.741			
0.8414	-0.004	0.751			
0.8891	0.046	0.763			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.650 ALPHA 4.00 REY 0.81×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.4334 CM = 0.0178 CDW = 0.0157

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.256	0.473	0.0757	0.059	0.765
0.0787	-1.587	0.399	0.1294	-0.101	0.729
0.1299	-1.430	0.434	0.1830	-0.152	0.718
0.1813	-0.911	0.549	0.2368	-0.189	0.709
0.2307	-0.819	0.570	0.2864	-0.193	0.708
0.2836	-0.710	0.594	0.3383	-0.187	0.710
0.3337	-0.622	0.614	0.3904	-0.184	0.711
0.3858	-0.542	0.632	0.4376	-0.167	0.714
0.4358	-0.469	0.647	0.4870	-0.153	0.717
0.4860	-0.402	0.662	0.5371	-0.134	0.722
0.5380	-0.328	0.679	0.5879	-0.114	0.726
0.5882	-0.266	0.693	0.6371	-0.093	0.731
0.6395	-0.214	0.704	0.6874	-0.081	0.733
0.6894	-0.162	0.716	0.7359	-0.061	0.738
0.7443	-0.106	0.728			
0.7954	-0.051	0.741			
0.8414	-0.001	0.752			
0.8891	0.049	0.763			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.649 ALPHA 5.10 REYNOLDS 0.81*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.5291 CM = 0.0203 CDW = 0.0191

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.557	0.407	0.0757	0.173	0.792
0.0787	-1.652	0.386	0.1294	-0.006	0.752
0.1299	-1.654	0.386	0.1830	-0.068	0.738
0.1813	-1.275	0.470	0.2368	-0.116	0.728
0.2307	-0.840	0.567	0.2864	-0.128	0.725
0.2836	-0.722	0.593	0.3383	-0.129	0.725
0.3337	-0.637	0.612	0.3904	-0.133	0.724
0.3858	-0.556	0.630	0.4376	-0.120	0.727
0.4358	-0.481	0.646	0.4870	-0.114	0.728
0.4860	-0.414	0.661	0.5371	-0.100	0.731
0.5383	-0.356	0.679	0.5879	-0.084	0.735
0.5882	-0.272	0.693	0.6371	-0.068	0.738
0.6395	-0.218	0.705	0.6874	-0.061	0.740
0.6894	-0.165	0.717	0.7359	-0.042	0.744
0.7443	-0.107	0.730			
0.7954	-0.051	0.742			
0.8414	-0.003	0.753			
0.8891	0.048	0.764			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.649 ALPHA 6.00 REYNOLDS 0.81*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.6315 CM = 0.0248 CDW = 0.0246

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.771	0.359	0.0757	0.268	0.813
0.0787	-1.814	0.349	0.1294	0.079	0.771
0.1299	-1.770	0.359	0.1830	0.004	0.754
0.1813	-1.687	0.378	0.2366	-0.050	0.742
0.2307	-1.102	0.508	0.2864	-0.072	0.737
0.2836	-0.808	0.573	0.3383	-0.079	0.735
0.3357	-0.653	0.608	0.3904	-0.088	0.733
0.3858	-0.556	0.629	0.4376	-0.082	0.735
0.4350	-0.479	0.646	0.4870	-0.081	0.735
0.4860	-0.411	0.662	0.5371	-0.069	0.737
0.5380	-0.338	0.678	0.5879	-0.059	0.739
0.5882	-0.272	0.693	0.6371	-0.044	0.743
0.6395	-0.218	0.705	0.6874	-0.044	0.743
0.6894	-0.163	0.717	0.7359	-0.030	0.746
0.7443	-0.106	0.729			
0.7954	-0.051	0.742			
0.8414	-0.003	0.753			
0.8891	0.047	0.763			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.076 ALPHA 0.00 REYNOLDS 0.82 $\times 10^6$

INTEGRATED FORCE COEFFICIENTS

CN = 0.0258 CM = 0.0047 CDW = 0.0136

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.153	0.700	0.0757	-0.532	0.611
0.0787	-0.506	0.602	0.1294	-0.585	0.598
0.1299	-0.575	0.600	0.1830	-0.556	0.605
0.1813	-0.574	0.601	0.2368	-0.527	0.612
0.2307	-0.537	0.609	0.2864	-0.482	0.622
0.2836	-0.464	0.622	0.3383	-0.431	0.634
0.3337	-0.443	0.632	0.3904	-0.391	0.644
0.3858	-0.396	0.642	0.4376	-0.343	0.655
0.4358	-0.352	0.653	0.4870	-0.303	0.665
0.4860	-0.310	0.663	0.5371	-0.259	0.675
0.5360	-0.252	0.677	0.5879	-0.217	0.685
0.5882	-0.207	0.687	0.6371	-0.178	0.694
0.6395	-0.173	0.695	0.6874	-0.148	0.701
0.6894	-0.132	0.705	0.7359	-0.107	0.711
0.7443	-0.090	0.715			
0.7954	-0.046	0.725			
0.8414	-0.004	0.735			
0.8891	0.044	0.746			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.076 ALPHA 1.00 REY 0.82×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1112 CM = 0.0038 CDW = 0.0122

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.389	2.644	0.0757	-0.359	0.651
0.0787	-0.759	2.556	0.1294	-0.452	0.629
0.1299	-0.723	2.565	0.1830	-0.450	0.629
0.1813	-0.692	2.572	0.2368	-0.442	0.631
0.2307	-0.651	2.587	0.2864	-0.412	0.633
0.2836	-0.557	2.624	0.3383	-0.373	0.648
0.3337	-0.503	2.617	0.3904	-0.343	0.655
0.3858	-0.445	2.631	0.4376	-0.304	0.664
0.4358	-0.390	2.643	0.4870	-0.271	0.672
0.4850	-0.341	2.655	0.5371	-0.233	0.681
0.5350	-0.277	2.670	0.5879	-0.196	0.689
0.5882	-0.223	2.682	0.6371	-0.159	0.698
0.6395	-0.168	2.691	0.6874	-0.133	0.714
0.6894	-0.144	2.701	0.7359	-0.098	0.712
0.7443	-0.097	2.712			
0.7954	-0.049	2.724			
0.8414	-0.004	2.734			
0.8891	0.046	2.746			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.676 ALPHA 2.00 REYNOLDS 0.82*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2236 CM = 0.0083 CDW = 0.0135

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.648	0.584	0.0751	-0.202	0.689
0.0787	-1.002	0.500	0.1294	-0.326	0.660
0.1279	-0.880	0.529	0.1830	-0.346	0.655
0.1813	-0.809	0.546	0.2368	-0.355	0.653
0.2307	-0.716	0.568	0.2864	-0.339	0.657
0.2836	-0.622	0.590	0.3383	-0.312	0.663
0.3337	-0.554	0.606	0.3904	-0.291	0.668
0.3858	-0.466	0.622	0.4370	-0.259	0.675
0.4358	-0.424	0.637	0.4870	-0.233	0.682
0.4863	-0.368	0.650	0.5371	-0.203	0.689
0.5380	-0.299	0.666	0.5879	-0.171	0.696
0.5882	-0.245	0.679	0.6371	-0.141	0.703
0.6395	-0.199	0.690	0.6874	-0.119	0.709
0.6894	-0.152	0.701	0.7359	-0.089	0.715
0.7443	-0.103	0.713			
0.7954	-0.051	0.725			
0.8414	-0.003	0.736			
0.8891	0.047	0.748			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.674 ALPHA 3.00 REYNOLDS 0.82*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.3564 CM = 0.0177 CDW = 0.0147

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.902	0.526	0.0757	-0.065	0.723
0.0787	-1.376	0.415	0.1294	-0.212	0.688
0.1299	-1.178	0.461	0.1830	-0.249	0.680
0.1813	-0.887	0.529	0.2366	-0.275	0.673
0.2307	-0.784	0.554	0.2864	-0.269	0.675
0.2836	-0.678	0.579	0.3383	-0.253	0.679
0.3337	-0.602	0.596	0.3904	-0.241	0.681
0.3858	-0.521	0.615	0.4376	-0.217	0.687
0.4358	-0.451	0.632	0.4870	-0.197	0.692
0.4850	-0.368	0.647	0.5371	-0.171	0.698
0.5380	-0.315	0.664	0.5879	-0.145	0.704
0.5882	-0.257	0.677	0.6371	-0.119	0.710
0.6395	-0.206	0.689	0.6874	-0.102	0.714
0.6894	-0.157	0.701	0.7359	-0.076	0.720
0.7443	-0.103	0.714			
0.7954	-0.050	0.726			
0.8414	-0.002	0.737			
0.8891	0.050	0.749			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.074 ALPHA 4.00 REYNOLDS 0.82×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.4468 CM = 0.0182 CDW = 0.0176

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.151	0.466	0.0757	0.053	0.749
0.0787	-1.481	0.388	0.1294	-0.109	0.710
0.1279	-1.555	0.371	0.1830	-0.162	0.698
0.1813	-1.425	0.401	0.2368	-0.198	0.689
0.2307	-0.835	0.540	0.2864	-0.203	0.688
0.2836	-0.608	0.575	0.3383	-0.196	0.690
0.3357	-0.608	0.593	0.3904	-0.193	0.691
0.3858	-0.534	0.611	0.4376	-0.174	0.695
0.4358	-0.463	0.627	0.4870	-0.161	0.698
0.4860	-0.398	0.642	0.5371	-0.143	0.703
0.5360	-0.324	0.663	0.5879	-0.119	0.728
0.5862	-0.263	0.674	0.6371	-0.096	0.713
0.6395	-0.212	0.686	0.6874	-0.084	0.716
0.6894	-0.159	0.699	0.7359	-0.060	0.722
0.7443	-0.103	0.712			
0.7954	-0.048	0.725			
0.8414	-0.003	0.735			
0.8891	0.052	0.748			

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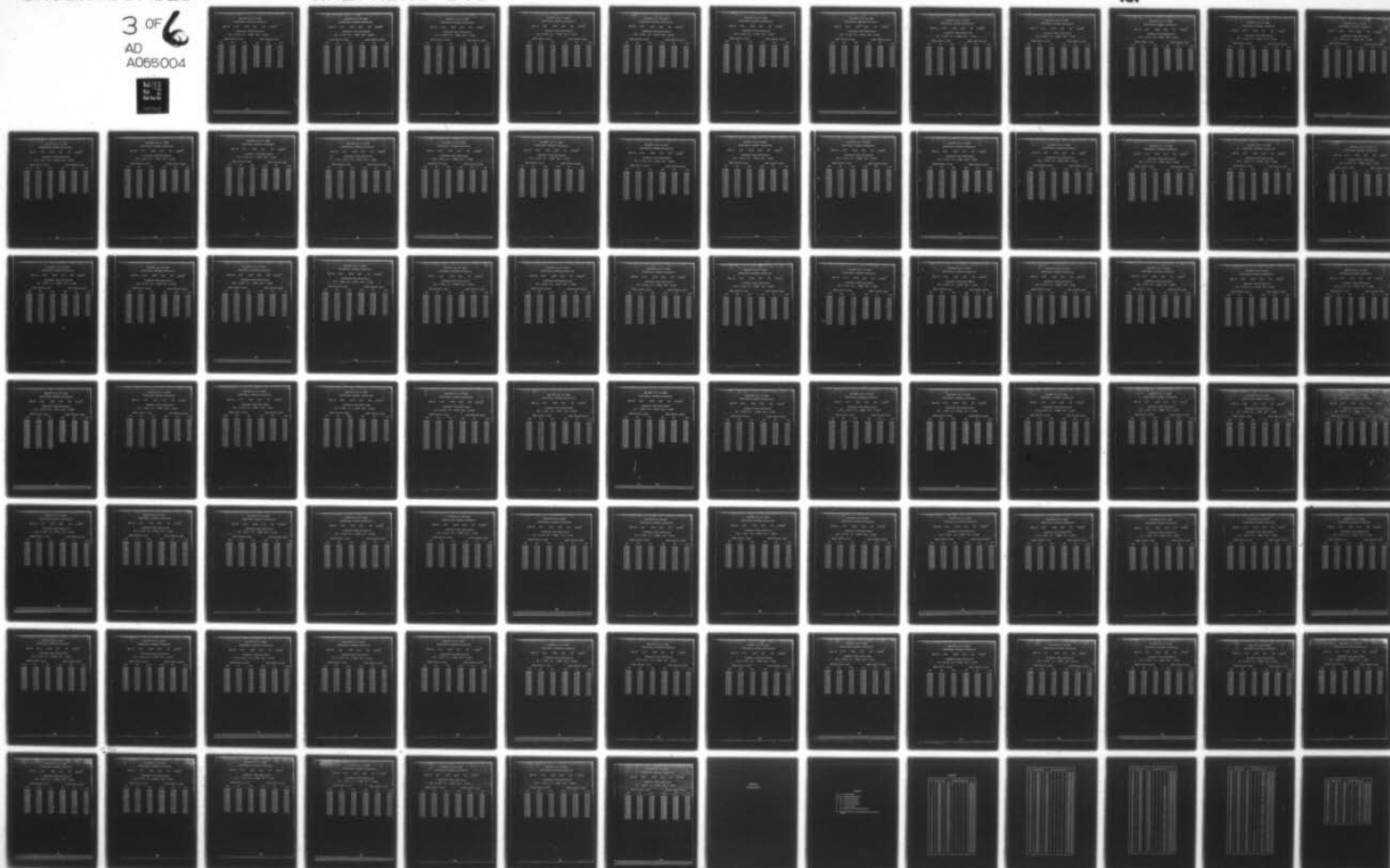
AERONAUTICAL RESEARCH LABS MELBOURNE (AUSTRALIA)
TRANSONIC WIND TUNNEL TESTS ON A NACA 0012 AEROFOIL, (U)
APR 77 N POLLOCK, B D FAIRLIE
ARL/AERO-148

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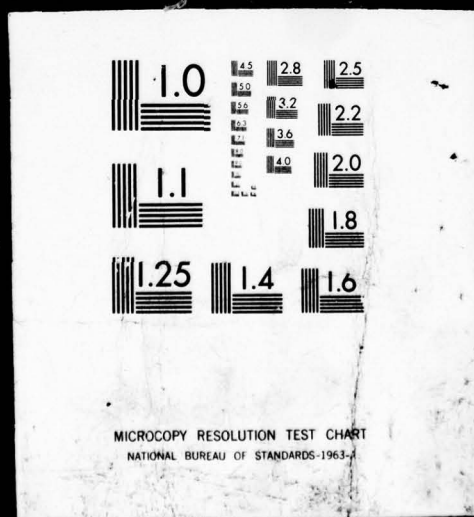


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NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.675 ALPHA 5.00 REY 0.83×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.5504 CM = 0.0215 CDW = 0.0240

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.399	0.407	0.0757	0.105	0.775
0.0787	-1.495	0.385	0.1294	-0.015	0.733
0.1299	-1.649	0.357	0.1830	-0.077	0.718
0.1813	-1.629	0.354	0.2500	-0.125	0.707
0.2307	-1.318	0.427	0.2864	-0.137	0.704
0.2836	-0.812	0.546	0.3383	-0.139	0.704
0.3337	-0.640	0.586	0.3904	-0.141	0.703
0.3858	-0.533	0.611	0.4376	-0.128	0.706
0.4358	-0.457	0.629	0.4870	-0.121	0.708
0.4860	-0.394	0.644	0.5371	-0.106	0.712
0.5380	-0.322	0.661	0.5879	-0.089	0.716
0.5882	-0.262	0.675	0.6371	-0.073	0.720
0.6395	-0.210	0.687	0.6874	-0.063	0.722
0.6894	-0.158	0.699	0.7359	-0.044	0.726
0.7443	-0.111	0.713			
0.7954	-0.048	0.725			
0.8414	0.001	0.737			
0.8891	0.050	0.748			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.701 ALPHA 0.00 REY 0.85-10

INTEGRATED FORCE COEFFICIENTS

CN = 0.0275 CM = 0.0048 CDW = 0.0124

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.138	0.686	0.0757	-0.545	0.585
0.0787	-0.580	0.576	0.1294	-0.608	0.570
0.1299	-0.596	0.572	0.1830	-0.565	0.575
0.1813	-0.600	0.572	0.2360	-0.554	0.583
0.2307	-0.559	0.582	0.2864	-0.505	0.595
0.2836	-0.505	0.595	0.3383	-0.450	0.609
0.3337	-0.459	0.606	0.3904	-0.406	0.620
0.3858	-0.411	0.618	0.4376	-0.356	0.632
0.4358	-0.362	0.630	0.4870	-0.312	0.643
0.4860	-0.320	0.641	0.5371	-0.267	0.654
0.5380	-0.260	0.656	0.5879	-0.223	0.665
0.5882	-0.213	0.668	0.6371	-0.181	0.675
0.6395	-0.177	0.676	0.6874	-0.148	0.684
0.6894	-0.136	0.687	0.7359	-0.109	0.693
0.7443	-0.091	0.698			
0.7954	-0.046	0.709			
0.8414	-0.005	0.719			
0.8891	0.045	0.731			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.700 ALPHA 1.00 REYNOLDS 0.85×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.1189$ $C_M = 0.0054$ $C_{DW} = 0.0126$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	C_P	P/H	X/C	C_P	P/H
0.0207	-0.365	0.630	0.0757	-0.365	0.631
0.0767	-0.790	0.525	0.1294	-0.465	0.626
0.1299	-0.756	0.533	0.1830	-0.468	0.625
0.1813	-0.734	0.539	0.2368	-0.461	0.627
0.2307	-0.664	0.556	0.2864	-0.430	0.614
0.2836	-0.584	0.576	0.3383	-0.390	0.624
0.3337	-0.521	0.592	0.3904	-0.358	0.632
0.3858	-0.460	0.607	0.4376	-0.315	0.643
0.4358	-0.403	0.621	0.4870	-0.280	0.651
0.4863	-0.351	0.634	0.5371	-0.240	0.661
0.5380	-0.284	0.650	0.5879	-0.202	0.671
0.5882	-0.253	0.663	0.6371	-0.165	0.680
0.6395	-0.190	0.674	0.6874	-0.137	0.687
0.6894	-0.146	0.684	0.7359	-0.101	0.696
0.7443	-0.097	0.697			
0.7954	-0.047	0.709			
0.8414	-0.002	0.720			
0.8891	0.048	0.733			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.698 ALPHA 2.00 REYNOLDS 0.84×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2550 CM = 0.0149 CDW = 0.0143

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.599	0.574	0.0757	-0.209	0.670
0.0787	-1.110	0.448	0.1294	-0.337	0.639
0.1299	-0.933	0.492	0.1830	-0.359	0.633
0.1813	-0.875	0.506	0.2368	-0.371	0.630
0.2307	-0.753	0.536	0.2864	-0.353	0.635
0.2836	-0.647	0.562	0.3383	-0.327	0.641
0.3337	-0.569	0.581	0.3904	-0.305	0.646
0.3858	-0.499	0.598	0.4376	-0.271	0.655
0.4358	-0.434	0.615	0.4870	-0.243	0.662
0.4860	-0.373	0.630	0.5371	-0.212	0.669
0.5383	-0.302	0.647	0.5879	-0.176	0.678
0.5882	-0.246	0.661	0.6371	-0.146	0.686
0.6395	-0.198	0.673	0.6874	-0.121	0.692
0.6894	-0.152	0.684	0.7359	-0.091	0.699
0.7443	-0.098	0.697			
0.7954	-0.049	0.710			
0.8414	0.001	0.722			
0.8891	0.051	0.734			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.699 ALPHA 3.00 REYNOLDS 0.84×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3477 CM = 0.0158 CDW = 0.0169

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.826	0.518	0.0757	-0.070	0.724
0.0787	-1.309	0.398	0.1294	-0.222	0.667
0.1299	-1.349	0.388	0.1830	-0.263	0.656
0.1813	-1.263	0.410	0.2368	-0.288	0.650
0.2307	-0.811	0.521	0.2864	-0.282	0.652
0.2836	-0.666	0.557	0.3383	-0.266	0.656
0.3337	-0.593	0.575	0.3904	-0.253	0.659
0.3858	-0.519	0.593	0.4376	-0.226	0.666
0.4358	-0.452	0.610	0.4870	-0.205	0.671
0.4860	-0.390	0.625	0.5371	-0.178	0.677
0.5360	-0.315	0.644	0.5879	-0.151	0.684
0.5882	-0.255	0.658	0.6371	-0.124	0.691
0.6395	-0.206	0.671	0.6874	-0.105	0.695
0.6894	-0.155	0.683	0.7359	-0.078	0.702
0.7443	-0.100	0.697			
0.7954	-0.046	0.710			
0.8414	0.003	0.722			
0.8891	0.054	0.735			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.098 ALPHA 4.00 REY 0.84×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.4641 CM = 0.0194 CDW = 0.0199

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.260	0.460	0.0757	0.047	0.733
0.0707	-1.377	0.382	0.1294	-0.117	0.693
0.1299	-1.498	0.352	0.1830	-0.171	0.679
0.1813	-1.495	0.353	0.2368	-0.209	0.670
0.2307	-1.447	0.365	0.2864	-0.213	0.669
0.2836	-0.796	0.526	0.3383	-0.207	0.670
0.3337	-0.618	0.569	0.3904	-0.202	0.672
0.3858	-0.512	0.596	0.4376	-0.181	0.677
0.4358	-0.441	0.613	0.4870	-0.168	0.680
0.4360	-0.380	0.628	0.5371	-0.145	0.686
0.5380	-0.311	0.645	0.5879	-0.124	0.691
0.5882	-0.252	0.659	0.6371	-0.101	0.697
0.6395	-0.202	0.672	0.6874	-0.087	0.700
0.6894	-0.152	0.684	0.7359	-0.063	0.706
0.7443	-0.097	0.698			
0.7954	-0.044	0.711			
0.8414	0.005	0.723			
0.8891	0.055	0.735			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.698 ALPHA 5.00 REYNOLDS 0.84*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.5594 CM = 0.0194 CDW = 0.0301

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.258	0.412	0.0757	0.151	0.759
0.0787	-1.370	0.384	0.1294	-0.027	0.715
0.1299	-1.598	0.328	0.1830	-0.091	0.699
0.1813	-1.602	0.327	0.2368	-0.139	0.658
0.2307	-1.609	0.325	0.2864	-0.151	0.685
0.2836	-1.019	0.471	0.3383	-0.153	0.634
0.3337	-0.790	0.527	0.3904	-0.154	0.634
0.3858	-0.631	0.566	0.4376	-0.143	0.687
0.4358	-0.507	0.597	0.4870	-0.134	0.689
0.4863	-0.405	0.622	0.5371	-0.119	0.693
0.5380	-0.320	0.643	0.5879	-0.101	0.697
0.5882	-0.252	0.660	0.6371	-0.081	0.702
0.6395	-0.195	0.674	0.6874	-0.072	0.704
0.6894	-0.143	0.686	0.7359	-0.052	0.709
0.7443	-0.068	0.700			
0.7954	-0.039	0.712			
0.8414	0.007	0.724			
0.8891	0.053	0.735			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.720 ALPHA 0.00 REYNOLDS 0.80×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.0237$ $C_M = 0.0055$ $C_{DW} = 0.0130$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	C _P	P/H	X/C	C _P	P/H
0.0207	-0.112	0.679	0.0757	-0.562	0.563
0.0787	-0.580	0.559	0.1294	-0.638	0.544
0.1299	-0.606	0.552	0.1830	-0.618	0.549
0.1813	-0.624	0.547	0.2368	-0.563	0.553
0.2307	-0.561	0.558	0.2864	-0.530	0.572
0.2836	-0.524	0.573	0.3383	-0.472	0.586
0.3337	-0.471	0.557	0.3904	-0.425	0.598
0.3858	-0.422	0.559	0.4376	-0.369	0.613
0.4358	-0.369	0.613	0.4870	-0.323	0.625
0.4860	-0.325	0.624	0.5371	-0.275	0.637
0.5380	-0.262	0.640	0.5879	-0.230	0.649
0.5882	-0.214	0.653	0.6371	-0.186	0.660
0.6395	-0.175	0.663	0.6874	-0.150	0.669
0.6894	-0.136	0.673	0.7359	-0.109	0.680
0.7443	-0.091	0.684			
0.7954	-0.045	0.696			
0.8414	-0.002	0.707			
0.8891	0.048	0.720			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.720 ALPHA 1.00 REY 0.80×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1265 CM = 0.0080 CDW = 0.0131

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.326	0.624	0.0757	-0.375	0.610
0.0767	-0.819	0.497	0.1294	-0.489	0.581
0.1299	-0.788	0.504	0.1830	-0.496	0.579
0.1813	-0.794	0.503	0.2368	-0.488	0.581
0.2307	-0.701	0.527	0.2864	-0.453	0.590
0.2836	-0.608	0.501	0.3383	-0.413	0.600
0.3337	-0.537	0.569	0.3904	-0.375	0.610
0.3858	-0.471	0.586	0.4376	-0.332	0.621
0.4358	-0.409	0.601	0.4870	-0.285	0.633
0.4860	-0.356	0.615	0.5371	-0.251	0.642
0.5380	-0.286	0.633	0.5879	-0.211	0.652
0.5852	-0.234	0.647	0.6371	-0.170	0.662
0.6395	-0.189	0.658	0.6874	-0.140	0.670
0.6894	-0.145	0.670	0.7359	-0.103	0.680
0.7443	-0.095	0.682			
0.7954	-0.046	0.695			
0.8414	0.001	0.707			
0.8891	0.051	0.720			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.720 ALPHA 2.00 REYNOLDS 0.80*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2735 CM = 0.0177 CGM = 0.0145

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.547	0.567	0.0757	-0.212	0.654
0.0787	-1.101	0.424	0.1294	-0.346	0.622
0.1299	-1.094	0.426	0.1830	-0.373	0.613
0.1813	-0.998	0.451	0.2368	-0.385	0.610
0.2307	-0.801	0.501	0.2864	-0.367	0.614
0.2836	-0.669	0.536	0.3393	-0.340	0.621
0.3337	-0.576	0.559	0.3904	-0.314	0.628
0.3858	-0.503	0.570	0.4376	-0.281	0.636
0.4358	-0.434	0.596	0.4870	-0.250	0.644
0.4860	-0.376	0.612	0.5371	-0.217	0.653
0.5380	-0.302	0.631	0.5879	-0.183	0.661
0.5882	-0.246	0.645	0.6371	-0.150	0.670
0.6395	-0.197	0.657	0.6874	-0.123	0.677
0.6894	-0.150	0.669	0.7359	-0.091	0.685
0.7443	-0.097	0.683			
0.7954	-0.046	0.696			
0.8414	0.003	0.709			
0.8891	0.053	0.722			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.720 ALPHA 3.00 REY 0.50×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3648 CM = 0.0163 CDW = 0.0172

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.743	0.516	0.0757	-0.075	0.688
0.0787	-1.214	0.395	0.1294	-0.229	0.648
0.1299	-1.286	0.377	0.1830	-0.273	0.637
0.1813	-1.304	0.372	0.2368	-0.300	0.630
0.2307	-1.328	0.366	0.2864	-0.292	0.632
0.2836	-0.919	0.471	0.3383	-0.277	0.636
0.3337	-0.598	0.554	0.3904	-0.263	0.639
0.3858	-0.491	0.581	0.4376	-0.236	0.646
0.4358	-0.423	0.598	0.4870	-0.213	0.652
0.4860	-0.365	0.613	0.5371	-0.184	0.660
0.5360	-0.299	0.630	0.5879	-0.158	0.666
0.5882	-0.243	0.645	0.6371	-0.128	0.674
0.6395	-0.194	0.657	0.6874	-0.108	0.679
0.6894	-0.145	0.670	0.7359	-0.078	0.687
0.7443	-0.093	0.683			
0.7954	-0.040	0.697			
0.8414	0.006	0.709			
0.8891	0.058	0.722			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.719 ALPHA 4.00 REYNOLDS 0.80*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4683 CM = 0.0161 CDW = 0.0241

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.953	0.464	0.0757	0.035	0.718
0.0787	-1.279	0.380	0.1294	-0.129	0.476
0.1299	-1.407	0.348	0.1830	-0.183	0.302
0.1813	-1.427	0.342	0.2368	-0.222	0.352
0.2307	-1.454	0.335	0.2864	-0.225	0.651
0.2856	-1.295	0.376	0.3383	-0.220	0.653
0.3357	-0.750	0.516	0.3904	-0.213	0.634
0.3858	-0.502	0.559	0.4376	-0.194	0.659
0.4358	-0.462	0.590	0.4870	-0.178	0.663
0.4860	-0.377	0.612	0.5371	-0.155	0.669
0.5390	-0.298	0.632	0.5879	-0.133	0.675
0.5882	-0.238	0.648	0.6371	-0.109	0.681
0.6395	-0.186	0.661	0.6874	-0.093	0.685
0.6894	-0.137	0.674	0.7359	-0.067	0.692
0.7443	-0.085	0.687			
0.7954	-0.035	0.700			
0.8414	0.012	0.712			
0.8891	0.058	0.724			

VACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.740 ALPHA 0.00 REY 0.80*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0254 CM = 0.0063 CDW = 0.0127

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.094	0.670	0.0757	-0.578	0.540
0.0787	-0.594	0.536	0.1294	-0.668	0.516
0.1299	-0.634	0.526	0.1830	-0.662	0.518
0.1813	-0.672	0.515	0.2360	-0.622	0.528
0.2307	-0.624	0.528	0.2864	-0.564	0.544
0.2836	-0.559	0.546	0.3383	-0.496	0.562
0.3337	-0.496	0.562	0.3904	-0.441	0.577
0.3858	-0.440	0.577	0.4370	-0.384	0.592
0.4358	-0.385	0.592	0.4870	-0.333	0.606
0.4860	-0.336	0.605	0.5371	-0.282	0.619
0.5380	-0.269	0.623	0.5879	-0.235	0.632
0.5882	-0.220	0.636	0.6371	-0.192	0.643
0.6395	-0.178	0.647	0.6874	-0.153	0.654
0.6894	-0.138	0.658	0.7359	-0.110	0.665
0.7443	-0.091	0.670			
0.7954	-0.045	0.683			
0.8414	0.001	0.695			
0.8891	0.049	0.708			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.739 ALPHA 1.00 REYNOLDS 0.80*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1446 CM = 0.0108 CDW = 0.0127

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.294	0.617	0.0757	-0.378	0.594
0.0787	-0.852	0.468	0.1294	-0.499	0.562
0.1299	-0.803	0.481	0.1830	-0.512	0.559
0.1813	-0.883	0.460	0.2368	-0.505	0.561
0.2307	-0.809	0.480	0.2864	-0.469	0.570
0.2836	-0.638	0.525	0.3383	-0.427	0.581
0.3337	-0.552	0.548	0.3904	-0.387	0.592
0.3858	-0.481	0.567	0.4376	-0.348	0.625
0.4358	-0.416	0.584	0.4870	-0.298	0.616
0.4860	-0.360	0.599	0.5371	-0.254	0.628
0.5380	-0.289	0.618	0.5879	-0.214	0.638
0.5892	-0.235	0.632	0.6371	-0.175	0.649
0.6395	-0.169	0.645	0.6874	-0.141	0.658
0.6894	-0.145	0.657	0.7359	-0.103	0.668
0.7443	-0.093	0.670			
0.7954	-0.043	0.684			
0.8414	0.004	0.696			
0.8891	0.054	0.710			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.741 ALPHA 2.00 REY 0.80×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2720 CM = 2.0168 CDW = 0.0144

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.483	0.535	0.0757	-0.222	0.635
0.0787	-1.039	0.417	0.1294	-0.361	0.598
0.1299	-1.076	2.447	0.1830	-0.393	0.590
0.1813	-1.074	0.407	0.2366	-0.407	0.586
0.2307	-1.086	0.404	0.2864	-0.390	0.591
0.2836	-0.917	0.450	0.3383	-0.359	0.599
0.3337	-0.566	0.543	0.3904	-0.332	0.646
0.3858	-0.480	0.566	0.4376	-0.296	0.616
0.4358	-0.417	0.583	0.4870	-0.263	0.625
0.4860	-0.304	0.597	0.5371	-0.226	0.634
0.5363	-0.294	0.616	0.5879	-0.191	0.644
0.5882	-0.240	0.631	0.6371	-0.156	0.653
0.6395	-0.190	0.644	0.6874	-0.129	0.660
0.6894	-0.145	0.656	0.7359	-0.094	0.669
0.7443	-0.092	0.670			
0.7954	-0.041	0.684			
0.8414	0.007	0.697			
0.8891	0.057	0.710			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.740 ALPHA 3.00 REY 0.80×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3734 CM = 0.0148 CDW = 0.0202

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.660	0.519	0.0757	-0.089	0.671
0.0787	-1.139	0.391	0.1294	-0.246	0.629
0.1299	-1.212	0.371	0.1830	-0.291	0.617
0.1813	-1.240	0.364	0.2368	-0.318	0.610
0.2307	-1.287	0.351	0.2864	-0.312	0.611
0.2836	-1.271	0.355	0.3383	-0.296	0.616
0.3337	-0.955	0.437	0.3904	-0.279	0.620
0.3853	-0.571	0.542	0.4376	-0.251	0.628
0.4358	-0.443	0.577	0.4870	-0.226	0.634
0.4860	-0.358	0.599	0.5371	-0.196	0.642
0.5360	-0.284	0.619	0.5879	-0.167	0.650
0.5882	-0.228	0.634	0.6371	-0.137	0.658
0.6395	-0.179	0.647	0.6874	-0.114	0.664
0.6894	-0.133	0.659	0.7359	-0.083	0.673
0.7443	-0.061	0.673			
0.7954	-0.032	0.686			
0.8414	0.014	0.698			
0.8991	0.062	0.711			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.741 ALPHA 4.00 REYNOLDS 6.80×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.4646 CM = 0.0098 CDW = 0.0303

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.825	0.473	0.0757	0.016	0.698
0.0787	-1.174	0.380	0.1294	-0.154	0.653
0.1299	-1.304	0.346	0.1830	-0.211	0.638
0.1813	-1.353	0.338	0.2368	-0.251	0.627
0.2307	-1.375	0.326	0.2864	-0.253	0.626
0.2836	-1.374	0.327	0.3383	-0.247	0.628
0.3337	-0.890	0.406	0.3904	-0.239	0.630
0.3858	-0.657	0.518	0.4376	-0.217	0.636
0.4358	-0.567	0.542	0.4870	-0.199	0.641
0.4860	-0.465	0.564	0.5371	-0.175	0.647
0.5380	-0.401	0.587	0.5879	-0.153	0.653
0.5882	-0.313	0.610	0.6371	-0.128	0.660
0.6395	-0.231	0.632	0.6874	-0.111	0.664
0.6894	-0.161	0.651	0.7359	-0.086	0.671
0.7443	-0.094	0.669			
0.7954	-0.039	0.684			
0.8414	0.003	0.695			
0.8891	0.047	0.706			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.758 ALPHA 0.00 REYNOLDS 0.78*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0166 CM = 0.0045 CDW = 0.0124

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.076	2.662	0.0757	-0.586	0.520
0.0787	-0.600	2.513	0.1294	-0.688	0.492
0.1299	-0.653	2.503	0.1830	-0.747	0.476
0.1813	-0.746	2.478	0.2368	-0.686	0.493
0.2307	-0.707	2.489	0.2864	-0.614	0.513
0.2836	-0.606	2.516	0.3383	-0.527	0.537
0.3337	-0.530	2.537	0.3904	-0.463	0.554
0.3858	-0.461	2.556	0.4376	-0.399	0.572
0.4358	-0.400	2.572	0.4870	-0.344	0.587
0.4860	-0.346	2.587	0.5371	-0.290	0.602
0.5380	-0.277	2.607	0.5879	-0.240	0.615
0.5882	-0.225	2.621	0.6371	-0.194	0.628
0.6395	-0.182	2.633	0.6874	-0.154	0.639
0.6894	-0.140	2.644	0.7359	-0.110	0.651
0.7443	-0.091	2.658			
0.7954	-0.042	2.671			
0.8414	0.003	2.684			
0.8991	0.053	2.697			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.750 ALPHA 1.00 REY 0.78×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1530 CM = 0.0118 CDW = 0.0133

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.257	0.612	0.0757	-0.590	0.575
0.0787	-0.855	0.448	0.1294	-0.522	0.539
0.1299	-0.782	0.468	0.1830	-0.550	0.531
0.1813	-0.865	0.445	0.2366	-0.544	0.533
0.2307	-0.940	0.424	0.2864	-0.507	0.543
0.2836	-0.933	0.426	0.3383	-0.454	0.558
0.3337	-0.601	0.517	0.3904	-0.409	0.570
0.3858	-0.466	0.555	0.4376	-0.357	0.585
0.4358	-0.406	0.571	0.4870	-0.312	0.597
0.4860	-0.354	0.585	0.5371	-0.263	0.610
0.5380	-0.266	0.604	0.5879	-0.222	0.622
0.5882	-0.252	0.619	0.6371	-0.180	0.633
0.6395	-0.186	0.632	0.6874	-0.145	0.643
0.6894	-0.141	0.644	0.7359	-0.104	0.654
0.7443	-0.090	0.658			
0.7954	-0.040	0.672			
0.8414	0.007	0.685			
0.8891	0.057	0.699			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.761 ALPHA 2.00 REY 0.79⁶ × 10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2819 CM = 0.0143 CDW = 0.0166

UPPER SURFACE VALUES

X/C	CP	P/H
0.0207	-0.412	0.567
0.0787	-0.068	0.414
0.1299	-1.022	0.399
0.1813	-1.051	0.391
0.2307	-1.084	0.382
0.2836	-1.089	0.380
0.3337	-1.097	0.378
0.3858	-0.621	0.510
0.4358	-0.427	0.563
0.4860	-0.343	0.587
0.5380	-0.273	0.606
0.5882	-0.221	0.620
0.6395	-0.173	0.634
0.6894	-0.150	0.645
0.7443	-0.078	0.660
0.7954	-0.051	0.673
0.8414	0.016	0.686
0.8891	0.064	0.699

LOWER SURFACE VALUES

X/C	CP	P/H
0.0757	-0.232	0.617
0.1294	-0.381	0.576
0.1830	-0.421	0.565
0.2369	-0.436	0.561
0.2864	-0.417	0.566
0.3383	-0.386	0.575
0.3904	-0.356	0.583
0.4376	-0.315	0.594
0.4870	-0.277	0.605
0.5371	-0.236	0.616
0.5879	-0.201	0.626
0.6371	-0.163	0.636
0.6874	-0.133	0.645
0.7359	-0.097	0.654

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.759 ALPHA 3.00 REYNOLDS 0.78⁶ × 10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.3753 CM = 0.0101 CDW = 0.0246

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.579	0.523	0.0757	-0.106	0.654
0.0707	-1.062	0.390	0.1294	-0.266	0.610
0.1299	-1.142	0.368	0.1830	-0.316	0.596
0.1813	-1.177	0.359	0.2368	-0.345	0.589
0.2307	-1.230	0.344	0.2864	-0.340	0.590
0.2836	-1.234	0.343	0.3383	-0.322	0.595
0.3337	-1.259	0.336	0.3904	-0.303	0.600
0.3858	-0.775	0.469	0.4376	-0.272	0.608
0.4358	-0.538	0.535	0.4870	-0.244	0.616
0.4850	-0.425	0.567	0.5371	-0.212	0.625
0.5380	-0.331	0.592	0.5879	-0.181	0.634
0.5882	-0.254	0.614	0.6371	-0.148	0.643
0.6395	-0.186	0.632	0.6874	-0.126	0.649
0.6894	-0.129	0.648	0.7359	-0.094	0.657
0.7443	-0.070	0.664			
0.7954	-0.024	0.677			
0.8414	0.022	0.689			
0.8991	0.066	0.702			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.761 ALPHA 4.00 REYNOLDS 0.79*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4051 CM = 0.0016 CDW = 0.0444

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.697	0.408	0.0757	-0.023	0.675
0.0787	-1.007	0.301	0.1294	-0.197	0.627
0.1299	-1.206	0.348	0.1830	-0.257	0.610
0.1813	-1.240	0.338	0.2368	-0.297	0.599
0.2307	-1.202	0.327	0.2864	-0.303	0.597
0.2836	-1.137	0.307	0.3383	-0.297	0.599
0.3337	-0.702	0.407	0.3904	-0.289	0.601
0.3853	-0.579	0.521	0.4376	-0.265	0.638
0.4358	-0.527	0.535	0.4870	-0.245	0.613
0.4860	-0.491	0.545	0.5371	-0.220	0.620
0.5300	-0.442	0.559	0.5879	-0.195	0.627
0.5802	-0.303	0.575	0.6371	-0.170	0.634
0.6395	-0.317	0.593	0.6874	-0.154	0.638
0.6894	-0.257	0.610	0.7359	-0.130	0.645
0.7443	-0.196	0.627			
0.7954	-0.148	0.640			
0.8414	-0.095	0.655			
0.8891	-0.047	0.668			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.781 ALPHA 0.00 REY 0.80×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2033 CM = 0.0028 CDW = 0.0137

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.041	0.627	0.0757	-0.291	0.520
0.0787	-0.580	0.503	0.1294	-0.675	0.476
0.1299	-0.632	0.408	0.1830	-0.762	0.451
0.1813	-0.737	0.458	0.2368	-0.808	0.438
0.2307	-0.811	0.437	0.2864	-0.830	0.431
0.2836	-0.823	0.434	0.3383	-0.712	0.465
0.3337	-0.715	0.464	0.3904	-0.452	0.539
0.3858	-0.447	0.541	0.4376	-0.390	0.557
0.4358	-0.356	0.558	0.4870	-0.338	0.572
0.4860	-0.338	0.572	0.5371	-0.285	0.507
0.5380	-0.271	0.591	0.5879	-0.237	0.621
0.5882	-0.222	0.605	0.6371	-0.190	0.614
0.6395	-0.178	0.618	0.6874	-0.149	0.626
0.6894	-0.135	0.630	0.7357	-0.106	0.633
0.7443	-0.086	0.644			
0.7954	-0.037	0.658			
0.8414	0.009	0.671			
0.8891	0.059	0.685			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.781 ALPHA 1.00 REYNOLDS 0.80⁶ × 10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1497 CM = 0.0068 CDW = 0.0149

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.197	2.611	0.0757	-0.398	0.554
0.0787	-0.799	2.439	0.1294	-0.542	0.513
0.1299	-0.781	2.444	0.1830	-0.612	0.493
0.1813	-0.831	2.430	0.2368	-0.609	0.494
0.2307	-0.888	2.414	0.2864	-0.581	0.532
0.2836	-0.931	2.402	0.3383	-0.503	0.524
0.3337	-0.984	2.386	0.3904	-0.443	0.541
0.3858	-0.927	2.403	0.4376	-0.380	0.559
0.4358	-0.467	2.534	0.4870	-0.327	0.574
0.4860	-0.337	2.571	0.5371	-0.275	0.589
0.5380	-0.262	2.593	0.5879	-0.229	0.602
0.5882	-0.212	2.607	0.6371	-0.185	0.615
0.6395	-0.165	2.620	0.6874	-0.147	0.626
0.6894	-0.125	2.632	0.7359	-0.106	0.638
0.7443	-0.076	2.646			
0.7954	-0.027	2.660			
0.8414	0.018	2.673			
0.8891	0.066	2.687			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.784 ALPHA 2.00 REYNOLDS 0.80×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.2867$ $C_M = 0.0080$ $C_{DW} = 0.0200$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	C _P	P/H	X/C	C _P	P/H
0.0207	-0.347	0.570	0.0757	-0.250	0.597
0.0787	-0.900	0.412	0.1294	-0.409	0.552
0.1299	-0.961	0.395	0.1830	-0.458	0.538
0.1813	-0.998	0.384	0.2368	-0.478	0.532
0.2307	-1.046	0.370	0.2864	-0.459	0.538
0.2836	-1.049	0.369	0.3383	-0.422	0.546
0.3337	-1.054	0.360	0.3904	-0.386	0.559
0.3858	-1.065	0.359	0.4376	-0.338	0.572
0.4358	-0.674	0.477	0.4870	-0.295	0.584
0.4860	-0.429	0.546	0.5371	-0.252	0.597
0.5360	-0.323	0.577	0.5879	-0.214	0.608
0.5882	-0.243	0.599	0.6371	-0.175	0.619
0.6395	-0.174	0.619	0.6874	-0.145	0.627
0.6894	-0.122	0.634	0.7359	-0.106	0.639
0.7443	-0.065	0.650			
0.7954	-0.019	0.663			
0.8414	0.026	0.676			
0.8891	0.073	0.689			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.779 ALPHA 3.00 REY 0.80×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3387 CM = 0.0013 CD_M = 0.0395

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.474	0.535	0.0757	-0.142	0.628
0.0767	-0.972	0.393	0.1294	-0.309	0.583
0.1299	-1.052	0.370	0.1830	-0.371	0.563
0.1813	-1.043	0.359	0.2368	-0.408	0.552
0.2307	-1.150	0.342	0.2864	-0.403	0.554
0.2836	-1.160	0.340	0.3383	-0.381	0.560
0.3337	-1.102	0.333	0.3904	-0.361	0.565
0.3858	-0.721	0.405	0.4376	-0.324	0.576
0.4358	-0.515	0.522	0.4870	-0.291	0.585
0.4860	-0.449	0.541	0.5371	-0.257	0.595
0.5380	-0.393	0.556	0.5879	-0.225	0.624
0.5882	-0.347	0.570	0.6371	-0.192	0.613
0.6395	-0.291	0.586	0.6874	-0.168	0.620
0.6894	-0.235	0.602	0.7359	-0.139	0.628
0.7443	-0.168	0.621			
0.7954	-0.113	0.637			
0.8414	-0.064	0.651			
0.8891	-0.016	0.664			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.801 ALPHA 0.00 REY 0.81×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0023 CM = 0.0019 CDW = 0.0168

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.310	0.652	0.0757	-0.573	0.485
0.0787	-0.561	0.490	0.1294	-0.633	0.467
0.1299	-0.683	0.417	0.1830	-0.733	0.438
0.1813	-0.776	0.447	0.2368	-0.786	0.422
0.2307	-0.789	0.422	0.2864	-0.827	0.410
0.2836	-0.821	0.413	0.3383	-0.843	0.405
0.3357	-0.869	0.399	0.3934	-0.872	0.397
0.3858	-0.871	0.398	0.4576	-0.882	0.453
0.4358	-0.653	0.462	0.4872	-0.852	0.558
0.4853	-0.339	0.554	0.5371	-0.262	0.577
0.5360	-0.255	0.579	0.5879	-0.269	0.592
0.5882	-0.198	0.596	0.6371	-0.167	0.605
0.6395	-0.155	0.609	0.6874	-0.129	0.616
0.6894	-0.116	0.623	0.7359	-0.088	0.628
0.7443	-0.070	0.634			
0.7954	-0.024	0.647			
0.8414	0.020	0.650			
0.8891	0.069	0.675			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.800 ALPHA 1.03 REYNOLDS 0.81×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1390 CM = -0.0012 CDW = 0.0193

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.145	0.613	0.0757	-0.462	0.538
0.0787	-0.742	0.438	0.1294	-0.543	0.497
0.1299	-0.750	0.435	0.1830	-0.633	0.471
0.1813	-0.799	0.421	0.2368	-0.678	0.457
0.2307	-0.862	0.402	0.2864	-0.702	0.451
0.2836	-0.893	0.393	0.3383	-0.687	0.455
0.3337	-0.951	0.376	0.3904	-0.475	0.517
0.3859	-0.969	0.371	0.4376	-0.378	0.545
0.4358	-0.959	0.374	0.4870	-0.329	0.563
0.4860	-0.487	0.513	0.5371	-0.277	0.575
0.5383	-0.313	0.564	0.5879	-0.231	0.589
0.5882	-0.226	0.590	0.6371	-0.186	0.602
0.6395	-0.162	0.629	0.6874	-0.148	0.613
0.6894	-0.112	0.623	0.7359	-0.107	0.625
0.7443	-0.059	0.639			
0.7954	-0.014	0.652			
0.8414	0.028	0.664			
0.8891	0.076	0.679			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.830 ALPHA 2.00 REYNOLDS 0.81=10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2480 CM = 0.0025 CDW = 0.0282

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.269	0.577	0.0757	-0.273	0.576
0.0757	-0.823	0.414	0.1294	-0.438	0.528
0.1299	-0.888	0.395	0.1830	-0.523	0.504
0.1813	-0.930	0.383	0.2365	-0.553	0.494
0.2307	-0.981	0.368	0.2864	-0.556	0.493
0.2836	-0.993	0.364	0.3383	-0.498	0.513
0.3337	-1.031	0.353	0.3904	-0.546	0.526
0.3858	-1.043	0.350	0.4376	-0.584	0.544
0.4358	-0.950	0.384	0.4870	-0.533	0.559
0.4860	-0.477	0.517	0.5371	-0.283	0.573
0.5383	-0.369	0.548	0.5879	-0.241	0.586
0.5882	-0.295	0.570	0.6371	-0.220	0.598
0.6395	-0.227	0.590	0.6874	-0.170	0.607
0.6894	-0.166	0.608	0.7359	-0.131	0.618
0.7443	-0.100	0.627			
0.7954	-0.042	0.644			
0.8414	0.006	0.658			
0.8891	0.049	0.671			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.819 ALPHA 0.00 REY 0.81+10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0006 CM = 0.0020 CDW = 0.0233

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	0.014	0.648	0.0757	-0.550	0.478
0.0787	-0.542	0.480	0.1294	-0.597	0.464
0.1299	-0.567	0.472	0.1830	-0.702	0.432
0.1813	-0.674	0.440	0.2368	-0.759	0.415
0.2307	-0.760	0.414	0.2854	-0.803	0.402
0.2836	-0.796	0.403	0.3383	-0.825	0.395
0.3337	-0.848	0.387	0.3904	-0.865	0.383
0.3858	-0.862	0.383	0.4376	-0.874	0.383
0.4358	-0.892	0.375	0.4870	-0.784	0.407
0.4863	-0.718	0.423	0.5371	-0.556	0.537
0.5380	-0.337	0.542	0.5879	-0.237	0.573
0.5882	-0.226	0.576	0.6371	-0.168	0.594
0.6395	-0.156	0.597	0.6874	-0.116	0.609
0.6894	-0.106	0.612	0.7359	-0.074	0.622
0.7443	-0.055	0.628			
0.7954	-0.012	0.641			
0.8414	0.030	0.653			
0.8891	0.077	0.667			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.820 ALPHA 1.00 REY 0.81×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1034 CM = -0.0046 CDW = 0.0243

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.094	0.614	0.0757	-0.595	0.525
0.0787	-0.685	0.435	0.1294	-0.526	0.485
0.1299	-0.705	0.429	0.1830	-0.619	0.457
0.1813	-0.757	0.414	0.2368	-0.679	0.439
0.2307	-0.818	0.395	0.2864	-0.719	0.427
0.2836	-0.849	0.386	0.3383	-0.738	0.421
0.3337	-0.909	0.368	0.3904	-0.774	0.410
0.3858	-0.927	0.362	0.4376	-0.776	0.410
0.4358	-0.962	0.352	0.4870	-0.473	0.501
0.4860	-0.833	0.392	0.5371	-0.276	0.561
0.5360	-0.400	0.523	0.5879	-0.217	0.579
0.5882	-0.288	0.556	0.6371	-0.176	0.591
0.6395	-0.210	0.580	0.6874	-0.142	0.601
0.6894	-0.147	0.599	0.7359	-0.106	0.612
0.7443	-0.079	0.620			
0.7954	-0.026	0.636			
0.8414	0.023	0.650			
0.8891	0.069	0.664			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.820 ALPHA 2.00 REY 0.82×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1429 CM = 0.0077 CDW = 0.0343

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.187	0.566	0.0757	-0.288	0.555
0.0787	-0.742	0.417	0.1294	-0.447	0.506
0.1299	-0.809	0.397	0.1830	-0.546	0.477
0.1813	-0.853	0.384	0.2368	-0.599	0.460
0.2307	-0.908	0.367	0.2864	-0.638	0.449
0.2836	-0.925	0.362	0.3383	-0.664	0.441
0.3337	-0.967	0.349	0.3904	-0.704	0.429
0.3858	-0.974	0.347	0.4376	-0.713	0.426
0.4358	-0.685	0.435	0.4870	-0.586	0.464
0.4860	-0.395	0.523	0.5371	-0.315	0.547
0.5383	-0.336	0.541	0.5879	-0.255	0.565
0.5882	-0.295	0.553	0.6371	-0.220	0.575
0.6395	-0.257	0.565	0.6874	-0.195	0.583
0.6894	-0.221	0.576	0.7359	-0.163	0.592
0.7443	-0.174	0.590			
0.7954	-0.125	0.604			
0.8414	-0.087	0.616			
0.8891	-0.046	0.629			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 0.00 REYNOLDS 0.41 $\times 10^6$

INTEGRATED FORCE COEFFICIENTS

CN = 0.0159 CM = 0.0033 CDW = 0.0127

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.179	0.817	0.0757	-0.458	0.776
0.0787	-0.472	0.774	0.1294	-0.470	0.774
0.1299	-0.461	0.775	0.1830	-0.449	0.777
0.1813	-0.451	0.777	0.2368	-0.420	0.781
0.2307	-0.426	0.780	0.2864	-0.391	0.786
0.2836	-0.383	0.767	0.3383	-0.350	0.792
0.3337	-0.359	0.790	0.3904	-0.324	0.795
0.3858	-0.318	0.796	0.4376	-0.281	0.802
0.4358	-0.292	0.800	0.4870	-0.252	0.806
0.4860	-0.253	0.806	0.5371	-0.214	0.812
0.5387	-0.212	0.812	0.5879	-0.184	0.816
0.5852	-0.172	0.818	0.6371	-0.149	0.821
0.6395	-0.147	0.822	0.6874	-0.126	0.825
0.6894	-0.111	0.827	0.7359	-0.093	0.833
0.7443	-0.080	0.831			
0.7954	-0.044	0.837			
0.8414	-0.006	0.842			
0.8891	0.034	0.848			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 1.00 REY 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2908 CM = 0.0008 CDW = 0.0120

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.423	0.781	0.0757	-0.308	0.798
0.0787	-0.614	0.753	0.1294	-0.357	0.791
0.1299	-0.568	0.760	0.1830	-0.362	0.790
0.1813	-0.534	0.765	0.2368	-0.350	0.792
0.2307	-0.498	0.770	0.2864	-0.332	0.794
0.2836	-0.441	0.778	0.3383	-0.295	0.800
0.3337	-0.409	0.783	0.3904	-0.278	0.802
0.3858	-0.359	0.791	0.4376	-0.240	0.808
0.4358	-0.322	0.796	0.4870	-0.220	0.811
0.4860	-0.282	0.802	0.5371	-0.185	0.816
0.5350	-0.236	0.809	0.5879	-0.161	0.820
0.5882	-0.193	0.815	0.6371	-0.128	0.824
0.6395	-0.164	0.819	0.6874	-0.112	0.827
0.6894	-0.125	0.825	0.7359	-0.085	0.831
0.7443	-0.091	0.830			
0.7954	-0.051	0.836			
0.8414	-0.013	0.841			
0.8891	0.032	0.848			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.530 ALPHA 2.00 REY 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1844 CM = 0.0018 CDW = 0.0158

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.685	0.741	0.0757	-0.174	0.817
0.0787	-0.773	0.728	0.1294	-0.254	0.826
0.1299	-0.678	0.742	0.1830	-0.279	0.832
0.1813	-0.621	0.751	0.2368	-0.280	0.832
0.2307	-0.569	0.759	0.2864	-0.269	0.843
0.2836	-0.502	0.756	0.3383	-0.248	0.827
0.3337	-0.457	0.775	0.3904	-0.234	0.839
0.3858	-0.401	0.763	0.4376	-0.205	0.813
0.4358	-0.359	0.743	0.4870	-0.188	0.815
0.4860	-0.312	0.797	0.5371	-0.157	0.823
0.5383	-0.261	0.804	0.5879	-0.137	0.823
0.5882	-0.215	0.811	0.6371	-0.108	0.829
0.6395	-0.183	0.816	0.6874	-0.093	0.832
0.6894	-0.139	0.822	0.7359	-0.071	0.833
0.7443	-0.100	0.828			
0.7954	-0.058	0.834			
0.8414	-0.016	0.840			
0.8691	0.031	0.847			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.501 ALPHA 3.00 REY 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2815 CM = 0.0035 CDW = 0.0172

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.980	0.696	0.0757	-0.047	0.835
0.0797	-0.938	0.703	0.1294	-0.157	0.819
0.1299	-0.801	0.723	0.1830	-0.199	0.812
0.1813	-0.714	0.736	0.2368	-0.211	0.811
0.2307	-0.648	0.746	0.2864	-0.215	0.810
0.2836	-0.563	0.758	0.3383	-0.198	0.813
0.3337	-0.509	0.766	0.3904	-0.191	0.814
0.3858	-0.444	0.776	0.4376	-0.165	0.817
0.4358	-0.397	0.783	0.4870	-0.155	0.819
0.4860	-0.343	0.791	0.5371	-0.128	0.823
0.5380	-0.288	0.799	0.5879	-0.112	0.825
0.5882	-0.235	0.807	0.6371	-0.087	0.829
0.6395	-0.199	0.812	0.6874	-0.077	0.830
0.6894	-0.151	0.820	0.7359	-0.058	0.833
0.7443	-0.109	0.826			
0.7954	-0.063	0.833			
0.8414	-0.017	0.839			
0.8891	0.032	0.847			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 4.00 REYNOLDS 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3783 CM = 0.0056 CDW = 0.0157

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.310	0.649	0.0757	0.073	0.854
0.0787	-1.114	0.678	0.1294	-0.067	0.833
0.1299	-0.920	0.707	0.1830	-0.121	0.825
0.1813	-0.808	0.723	0.2368	-0.143	0.822
0.2307	-0.723	0.736	0.2864	-0.156	0.820
0.2836	-0.619	0.751	0.3383	-0.148	0.821
0.3357	-0.558	0.760	0.3904	-0.149	0.821
0.3858	-0.484	0.771	0.4376	-0.128	0.824
0.4358	-0.432	0.779	0.4870	-0.121	0.825
0.4862	-0.371	0.788	0.5371	-0.099	0.828
0.5353	-0.310	0.797	0.5879	-0.086	0.830
0.5862	-0.255	0.805	0.6371	-0.070	0.832
0.6395	-0.211	0.812	0.6874	-0.059	0.834
0.6894	-0.160	0.819	0.7359	-0.042	0.837
0.7443	-0.116	0.826			
0.7954	-0.068	0.833			
0.8414	-0.019	0.840			
0.8891	0.032	0.847			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.495 ALPHA 5.00 REY 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.4720 CM = 0.0084 CDW = 0.0177

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-1.657	0.600	0.0757	0.177	0.869
0.0787	-1.296	0.653	0.1294	0.018	0.845
0.1299	-1.057	0.668	0.1830	-0.043	0.836
0.1813	-0.901	0.711	0.2368	-0.085	0.830
0.2307	-0.784	0.729	0.2864	-0.098	0.823
0.2856	-0.678	0.744	0.3383	-0.102	0.828
0.3357	-0.606	0.755	0.3904	-0.108	0.827
0.3858	-0.528	0.766	0.4376	-0.095	0.829
0.4358	-0.463	0.775	0.4870	-0.089	0.829
0.4860	-0.398	0.785	0.5371	-0.079	0.831
0.5380	-0.331	0.795	0.5879	-0.060	0.833
0.5882	-0.273	0.803	0.6371	-0.051	0.835
0.6395	-0.223	0.811	0.6874	-0.041	0.836
0.6894	-0.173	0.818	0.7359	-0.029	0.838
0.7443	-0.126	0.825			
0.7954	-0.072	0.833			
0.8414	-0.018	0.841			
0.8891	0.032	0.848			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.498 ALPHA 6.00 REYNOLDS 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.5740$ $C_M = 0.0105$ $C_{DW} = 0.0137$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	C _p	P/H	X/C	C _p	P/H
0.0207	-2.249	0.543	0.0757	0.281	0.885
0.0787	-1.489	0.625	0.1294	0.107	0.860
0.1299	-1.270	0.657	0.1830	0.034	0.849
0.1813	-0.949	0.705	0.2368	-0.019	0.841
0.2307	-0.851	0.719	0.2864	-0.038	0.836
0.2836	-0.753	0.736	0.3383	-0.055	0.836
0.3337	-0.658	0.747	0.3934	-0.061	0.835
0.3858	-0.564	0.761	0.4376	-0.060	0.835
0.4358	-0.496	0.771	0.4870	-0.053	0.836
0.4860	-0.425	0.781	0.5371	-0.047	0.837
0.5380	-0.353	0.792	0.5879	-0.035	0.839
0.5882	-0.291	0.801	0.6371	-0.025	0.840
0.6395	-0.239	0.809	0.6874	-0.018	0.841
0.6894	-0.185	0.817	0.7359	-0.008	0.843
0.7443	-0.132	0.824			
0.7954	-0.081	0.832			
0.8414	-0.029	0.840			
0.8891	0.019	0.847			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 7.20 REY 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.6783 CM = 0.0105 CDW = 0.0099

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-2.406	0.428	0.0757	0.386	0.920
0.0787	-1.868	0.568	0.1294	0.194	0.872
0.1299	-1.153	0.673	0.1830	0.103	0.858
0.1813	-1.045	0.689	0.2366	0.049	0.850
0.2307	-0.937	0.705	0.2864	0.022	0.846
0.2836	-0.792	0.726	0.3383	0.001	0.843
0.3337	-0.700	0.740	0.3904	-0.011	0.841
0.3858	-0.605	0.754	0.4376	-0.013	0.841
0.4358	-0.530	0.765	0.4870	-0.011	0.841
0.4850	-0.454	0.776	0.5371	-0.008	0.842
0.5360	-0.380	0.787	0.5879	-0.002	0.843
0.5862	-0.313	0.797	0.6371	0.004	0.844
0.6395	-0.259	0.805	0.6874	0.010	0.845
0.6894	-0.201	0.814	0.7359	0.019	0.846
0.7443	-0.146	0.822			
0.7924	-0.089	0.830			
0.8414	-0.041	0.837			
0.8891	0.008	0.844			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 8.00 REYNOLDS 0.41*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.7589 CM = 0.0176 CDW = 0.0205

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-2.603	0.460	0.0757	0.459	0.911
0.0787	-2.295	0.505	0.1294	0.260	0.882
0.1299	-1.237	0.661	0.1830	0.162	0.867
0.1813	-1.113	0.679	0.2368	0.100	0.858
0.2307	-0.969	0.701	0.2864	0.068	0.853
0.2836	-0.824	0.722	0.3383	0.040	0.849
0.3337	-0.724	0.737	0.3904	0.025	0.847
0.3858	-0.621	0.752	0.4376	0.018	0.846
0.4358	-0.540	0.754	0.4870	0.016	0.846
0.4860	-0.461	0.775	0.5371	0.014	0.845
0.5380	-0.381	0.787	0.5879	0.017	0.846
0.5882	-0.313	0.797	0.6371	0.020	0.846
0.6395	-0.254	0.806	0.6874	0.024	0.847
0.6894	-0.198	0.814	0.7359	0.029	0.847
0.7443	-0.138	0.823			
0.7954	-0.087	0.830			
0.8414	-0.038	0.838			
0.8891	0.007	0.844			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.699 ALPHA 0.00 REY 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0100 CM = 0.0029 CDW = 0.0126

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.115	0.693	0.0757	-0.556	0.584
0.0787	-0.572	0.580	0.1294	-0.587	0.576
0.1299	-0.588	0.576	0.1830	-0.588	0.576
0.1813	-0.569	0.576	0.2368	-0.545	0.587
0.2307	-0.563	0.582	0.2864	-0.516	0.594
0.2836	-0.510	0.595	0.3383	-0.464	0.617
0.3337	-0.475	0.604	0.3904	-0.427	0.616
0.3858	-0.416	0.619	0.4376	-0.365	0.631
0.4358	-0.368	0.633	0.4870	-0.317	0.643
0.4863	-0.309	0.645	0.5371	-0.265	0.656
0.5380	-0.251	0.659	0.5879	-0.220	0.667
0.5882	-0.205	0.671	0.6371	-0.176	0.678
0.6395	-0.172	0.679	0.6874	-0.146	0.685
0.6894	-0.129	0.689	0.7359	-0.138	0.695
0.7443	-0.292	0.699			
0.7954	-0.345	0.710			
0.8414	-0.381	0.721			
0.8891	0.248	0.733			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.701 ALPHA 1.00 REYNOLDS 0.42×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1098 CM = 0.0029 CDN = 0.0126

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.351	0.658	0.0757	-0.579	0.626
0.0787	-0.782	0.526	0.1294	-0.450	0.608
0.1299	-0.758	0.532	0.1830	-0.476	0.602
0.1813	-0.722	0.541	0.2368	-0.456	0.607
0.2307	-0.674	0.553	0.2864	-0.442	0.610
0.2836	-0.609	0.569	0.3383	-0.400	0.621
0.3337	-0.556	0.582	0.3904	-0.371	0.628
0.3858	-0.458	0.606	0.4376	-0.318	0.641
0.4358	-0.393	0.622	0.4870	-0.282	0.650
0.4860	-0.337	0.636	0.5371	-0.237	0.661
0.5382	-0.278	0.651	0.5879	-0.203	0.669
0.5882	-0.227	0.664	0.6371	-0.160	0.680
0.6395	-0.191	0.672	0.6874	-0.135	0.686
0.6894	-0.142	0.685	0.7359	-0.096	0.695
0.7443	-0.089	0.695			
0.7954	-0.050	0.707			
0.8414	-0.004	0.719			
0.8891	0.049	0.732			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.699 ALPHA 2.00 REYNOLDS 0.42⁶ × 10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2482 CM = 0.0111 CDW = 0.0143

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.563	2.582	0.0757	-0.218	0.667
0.0787	-1.063	2.458	0.1294	-0.324	0.641
0.1299	-0.955	2.405	0.1830	-0.367	0.630
0.1813	-0.879	2.504	0.2368	-0.368	0.630
0.2307	-0.824	2.517	0.2864	-0.365	0.631
0.2836	-0.742	2.538	0.3383	-0.335	0.638
0.3337	-0.582	2.577	0.3904	-0.314	2.644
0.3858	-0.466	2.626	0.4376	-0.275	0.653
0.4358	-0.419	2.618	0.4870	-0.242	0.651
0.4860	-0.368	2.630	0.5371	-0.236	0.670
0.5380	-0.301	2.647	0.5879	-0.174	2.676
0.5882	-0.246	2.660	0.6371	-0.140	0.657
0.6395	-0.201	2.671	0.6874	-0.116	0.692
0.6894	-0.153	2.683	0.7359	-0.084	0.700
0.7443	-0.104	2.695			
0.7954	-0.053	2.708			
0.8414	-0.002	2.721			
0.8891	0.048	2.733			

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.701 ALPHA 3.00 REY 0.42×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3516 CM = 0.0145 CDW = 0.0168

UPPER SURFACE VALUES

X/C	CP	P/H
0.0207	-0.766	0.529
0.0767	-1.244	0.411
0.1299	-1.239	0.412
0.1813	-1.114	0.443
0.2307	-1.096	0.448
0.2836	-0.906	0.495
0.3337	-0.526	0.509
0.3858	-0.464	0.604
0.4358	-0.427	0.614
0.4860	-0.376	0.626
0.5380	-0.318	0.643
0.5882	-0.253	0.657
0.6395	-0.209	0.658
0.6894	-0.155	0.651
0.7443	-0.102	0.694
0.7954	-0.053	0.707
0.8414	-0.001	0.719
0.8891	0.051	0.732

LOWER SURFACE VALUES

X/C	CP	P/H
0.0757	-0.086	0.698
0.1294	-0.216	0.666
0.1830	-0.271	0.652
0.2360	-0.268	0.648
0.2864	-0.294	0.647
0.3383	-0.274	0.652
0.3904	-0.265	0.654
0.4376	-0.233	0.662
0.4870	-0.203	0.668
0.5371	-0.176	0.676
0.5879	-0.151	0.682
0.6371	-0.120	0.690
0.6874	-0.102	0.694
0.7359	-0.073	0.702

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.699 ALPHA 4.00 REYNOLDS 0.42*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4464 CM = 0.0177 CDW = 0.0197

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.982	0.478	0.0757	0.031	0.729
0.0787	-1.336	0.391	0.1294	-0.120	0.691
0.1299	-1.381	0.380	0.1830	-0.185	0.675
0.1813	-1.252	0.411	0.2368	-0.215	0.668
0.2307	-1.252	0.411	0.2864	-0.228	0.665
0.2836	-1.219	0.420	0.3383	-0.222	0.666
0.3337	-0.845	0.512	0.3904	-0.218	0.667
0.3858	-0.498	0.598	0.4376	-0.197	0.672
0.4358	-0.404	0.621	0.4870	-0.181	0.676
0.4860	-0.353	0.634	0.5371	-0.153	0.683
0.5363	-0.303	0.646	0.5879	-0.131	0.689
0.5882	-0.242	0.661	0.6371	-0.107	0.695
0.6395	-0.197	0.672	0.6874	-0.092	0.698
0.6894	-0.146	0.685	0.7359	-0.068	0.714
0.7443	-0.094	0.698			
0.7954	-0.045	0.710			
0.8414	0.006	0.722			
0.8891	0.052	0.734			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.699 ALPHA 5.00 REY 0.42×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.5140 CM = 0.0170 CDW = 0.0315

UPPER SURFACE VALUES

X/C	CP	P/H
0.0207	-1.161	0.434
0.0787	-1.377	0.381
0.1299	-1.379	0.381
0.1813	-1.306	0.399
0.2307	-1.322	0.395
0.2836	-1.196	0.426
0.3357	-0.923	0.493
0.3858	-0.654	0.560
0.4358	-0.496	0.599
0.4860	-0.399	0.623
0.5380	-0.319	0.643
0.5882	-0.251	0.659
0.6395	-0.196	0.673
0.6894	-0.144	0.686
0.7443	-0.092	0.699
0.7954	-0.047	0.710
0.8414	-0.001	0.721
0.8891	0.039	0.731

LOWER SURFACE VALUES

X/C	CP	P/H
0.0757	0.119	0.751
0.1294	-0.044	0.710
0.1830	-0.113	0.693
0.2360	-0.154	0.683
0.2864	-0.173	0.678
0.3383	-0.175	0.678
0.3904	-0.179	0.677
0.4376	-0.163	0.681
0.4870	-0.153	0.683
0.5371	-0.133	0.688
0.5879	-0.116	0.693
0.6371	-0.095	0.698
0.6874	-0.086	0.700
0.7359	-0.060	0.706

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.801 ALPHA 0.00 REYNOLDS 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.0012$ $C_M = 0.0015$ $C_{DW} = 0.0169$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	0.313	0.629	0.0757	-0.354	0.492
0.0787	-0.549	0.493	0.1294	-0.609	0.476
0.1299	-0.609	0.476	0.1830	-0.722	0.442
0.1813	-0.697	0.450	0.2368	-0.772	0.423
0.2307	-0.782	0.425	0.2864	-0.822	0.413
0.2836	-0.795	0.421	0.3383	-0.784	0.424
0.3337	-0.744	0.436	0.3904	-0.733	0.439
0.3858	-0.692	0.451	0.4376	-0.696	0.453
0.4358	-0.649	0.449	0.4870	-0.626	0.471
0.4860	-0.700	0.449	0.5371	-0.454	0.521
0.5350	-0.393	0.539	0.5879	-0.251	0.581
0.5852	-0.180	0.602	0.6371	-0.142	0.613
0.6395	-0.127	0.618	0.6874	-0.097	0.627
0.6894	-0.090	0.628	0.7359	-0.059	0.638
0.7443	-0.054	0.639			
0.7954	-0.014	0.651			
0.8414	0.030	0.664			
0.8891	0.077	0.678			

NACA-0012, 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.799 ALPHA 1.00 REYNOLDS 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1539 CM = 0.0024 CDW = 0.0195

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.115	0.623	0.0757	-0.413	0.537
0.0787	-0.695	0.452	0.1294	-0.524	0.524
0.1299	-0.753	0.441	0.1830	-0.628	0.474
0.1813	-0.766	0.426	0.2368	-0.671	0.461
0.2307	-0.857	0.405	0.2864	-0.684	0.457
0.2856	-0.855	0.406	0.3383	-0.688	0.480
0.3337	-0.797	0.423	0.3904	-0.565	0.492
0.3858	-0.766	0.432	0.4376	-0.513	0.528
0.4358	-0.780	0.428	0.4870	-0.435	0.530
0.4860	-0.708	0.426	0.5371	-0.309	0.567
0.5380	-0.554	0.495	0.5879	-0.225	0.592
0.5882	-0.279	0.575	0.6371	-0.164	0.612
0.6395	-0.159	0.611	0.6874	-0.127	0.621
0.6894	-0.093	0.630	0.7359	-0.091	0.631
0.7443	-0.043	0.645			
0.7954	0.000	0.657			
0.8414	0.044	0.670			
0.8891	0.087	0.683			

NACA-0012 101.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.798 ALPHA 2.00 REY 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2076 CM = -0.0023 CD = 0.0291

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0207	-0.229	0.590	0.0757	-0.293	0.571
0.0787	-0.778	0.429	0.1293	-0.428	0.532
0.1299	-0.869	0.403	0.1830	-0.523	0.504
0.1813	-0.907	0.391	0.2368	-0.557	0.494
0.2307	-0.931	0.385	0.2864	-0.568	0.491
0.2836	-0.847	0.409	0.3383	-0.522	0.504
0.3337	-0.832	0.414	0.3904	-0.483	0.516
0.3858	-0.820	0.417	0.4376	-0.436	0.530
0.4358	-0.834	0.413	0.4870	-0.389	0.543
0.4860	-0.767	0.427	0.5371	-0.336	0.559
0.5380	-0.558	0.494	0.5879	-0.271	0.578
0.5882	-0.370	0.549	0.6371	-0.205	0.597
0.6395	-0.258	0.587	0.6874	-0.154	0.612
0.6894	-0.153	0.612	0.7359	-0.112	0.624
0.7443	-0.077	0.635			
0.7954	-0.023	0.651			
0.8414	0.027	0.665			
0.8891	0.068	0.677			

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 0.00 REY 0.40×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0034 CM = -0.0006 CDW = 0.0145

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.160	0.819	0.0253	-0.167	0.818
0.0794	-0.427	0.779	0.0794	-0.427	0.779
0.1374	-0.444	0.777	0.1374	-0.450	0.776
0.1911	-0.412	0.782	0.1911	-0.416	0.781
0.2368	-0.397	0.784	0.2368	-0.402	0.783
0.2892	-0.373	0.787	0.2892	-0.373	0.787
0.3371	-0.357	0.793	0.3371	-0.342	0.792
0.3867	-0.286	0.800	0.3867	-0.287	0.800
0.4396	-0.276	0.801	0.4396	-0.278	0.800
0.4913	-0.248	0.806	0.4913	-0.249	0.805
0.5369	-0.216	0.810	0.5369	-0.221	0.809
0.5802	-0.194	0.814	0.5802	-0.195	0.813
0.6275	-0.166	0.818	0.6275	-0.165	0.817
0.6769	-0.123	0.824	0.6769	-0.121	0.824
0.7346	-0.087	0.829	0.7346	-0.087	0.829

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.497 ALPHA 0.50 REYNOLDS 0.40×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0663 CM = 0.0001 CDW = 0.0152

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.343	0.794	0.0253	-0.010	0.843
0.0794	-0.543	0.764	0.0794	-0.320	0.794
0.1374	-0.528	0.766	0.1374	-0.367	0.737
0.1911	-0.478	0.774	0.1911	-0.355	0.769
0.2368	-0.453	0.777	0.2368	-0.349	0.790
0.2892	-0.418	0.783	0.2892	-0.330	0.793
0.3371	-0.379	0.768	0.3371	-0.302	0.797
0.3867	-0.320	0.797	0.3867	-0.254	0.834
0.4396	-0.306	0.798	0.4396	-0.251	0.805
0.4913	-0.273	0.803	0.4913	-0.226	0.808
0.5369	-0.268	0.808	0.5369	-0.197	0.813
0.5802	-0.212	0.812	0.5802	-0.179	0.815
0.6275	-0.161	0.817	0.6275	-0.152	0.819
0.6769	-0.134	0.824	0.6769	-0.110	0.826
0.7346	-0.096	0.829	0.7346	-0.079	0.833

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.501 ALPHA 1.00 REYNOLDS 0.40x10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1364 CM = 0.0012 CDW = 0.0152

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.519	0.765	0.0253	0.123	0.861
0.0794	-0.654	0.745	0.0794	-0.230	0.839
0.1374	-0.607	0.752	0.1374	-0.293	0.799
0.1911	-0.541	0.762	0.1911	-0.296	0.799
0.2368	-0.504	0.767	0.2368	-0.249	0.798
0.2892	-0.461	0.774	0.2892	-0.287	0.800
0.3371	-0.413	0.781	0.3371	-0.267	0.803
0.3867	-0.350	0.793	0.3867	-0.224	0.813
0.4396	-0.331	0.793	0.4396	-0.225	0.809
0.4913	-0.296	0.798	0.4913	-0.205	0.812
0.5369	-0.256	0.804	0.5369	-0.179	0.816
0.5802	-0.228	0.808	0.5802	-0.162	0.819
0.6275	-0.193	0.813	0.6275	-0.140	0.822
0.6769	-0.145	0.821	0.6769	-0.100	0.828
0.7346	-0.103	0.827	0.7346	-0.070	0.832

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 1.50 REYNOLDS 0.40×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.2044$ $C_M = 0.0026$ $C_D = 0.0161$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.704	0.738	0.0253	0.255	0.880
0.0794	-0.765	0.729	0.0794	-0.135	0.823
0.1374	-0.664	0.741	0.1374	-0.223	0.810
0.1911	-0.600	0.754	0.1911	-0.240	0.837
0.2368	-0.552	0.751	0.2368	-0.250	0.826
0.2892	-0.501	0.758	0.2892	-0.247	0.806
0.3371	-0.447	0.776	0.3371	-0.231	0.809
0.3867	-0.380	0.766	0.3867	-0.192	0.814
0.4396	-0.325	0.790	0.4396	-0.198	0.813
0.4913	-0.315	0.796	0.4913	-0.181	0.816
0.5369	-0.272	0.802	0.5369	-0.158	0.819
0.5802	-0.241	0.807	0.5802	-0.144	0.821
0.6275	-0.205	0.812	0.6275	-0.126	0.824
0.6769	-0.154	0.820	0.6769	-0.087	0.830
0.7346	-0.110	0.826	0.7346	-0.062	0.834

NACA-0012 50.8 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.501 ALPHA 2.00 REYNOLDS 0.40×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2739 CM = 0.0036 CDW = 0.0182

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.901	0.708	0.0253	0.366	0.897
0.0794	-0.878	0.712	0.0794	-0.050	0.835
0.1374	-0.763	0.729	0.1374	-0.155	0.820
0.1911	-0.663	0.744	0.1911	-0.182	0.816
0.2368	-0.605	0.752	0.2368	-0.203	0.813
0.2892	-0.543	0.762	0.2892	-0.206	0.812
0.3371	-0.482	0.771	0.3371	-0.196	0.814
0.3867	-0.419	0.781	0.3867	-0.161	0.819
0.4396	-0.383	0.785	0.4396	-0.171	0.818
0.4913	-0.336	0.792	0.4913	-0.158	0.820
0.5369	-0.291	0.799	0.5369	-0.138	0.822
0.5802	-0.258	0.804	0.5802	-0.126	0.824
0.6275	-0.218	0.810	0.6275	-0.110	0.827
0.6769	-0.165	0.817	0.6769	-0.073	0.832
0.7346	-0.117	0.825	0.7346	-0.052	0.835

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.630 ALPHA 0.00 REY 0.39⁶ × 10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.0041 CM = -0.0006 CDW = 0.0127

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.142	0.755	0.0253	-0.156	0.754
0.0794	-0.455	0.693	0.0794	-0.455	0.695
0.1374	-0.481	0.688	0.1374	-0.488	0.688
0.1911	-0.453	0.694	0.1911	-0.456	0.694
0.2368	-0.438	0.697	0.2368	-0.443	0.698
0.2892	-0.408	0.703	0.2892	-0.410	0.704
0.3371	-0.373	0.710	0.3371	-0.373	0.711
0.3867	-0.311	0.722	0.3867	-0.314	0.722
0.4396	-0.302	0.724	0.4396	-0.304	0.724
0.4913	-0.271	0.733	0.4913	-0.273	0.731
0.5369	-0.235	0.737	0.5369	-0.241	0.737
0.5802	-0.210	0.742	0.5802	-0.213	0.742
0.6275	-0.179	0.748	0.6275	-0.181	0.749
0.6759	-0.132	0.758	0.6759	-0.131	0.759
0.7346	-0.093	0.765	0.7346	-0.093	0.766

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.599 ALPHA 0.50 REY 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.8790 CM = 0.0003 CDW = 0.0140

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.345	0.716	0.0253	0.032	0.791
0.0794	-0.595	0.607	0.0794	-0.332	0.719
0.1374	-0.585	0.669	0.1374	-0.388	0.728
0.1911	-0.534	0.679	0.1911	-0.381	0.729
0.2368	-0.504	0.685	0.2368	-0.375	0.710
0.2892	-0.465	0.693	0.2892	-0.358	0.714
0.3371	-0.418	0.702	0.3371	-0.329	0.720
0.3867	-0.352	0.715	0.3867	-0.276	0.730
0.4396	-0.334	0.718	0.4396	-0.271	0.731
0.4913	-0.300	0.725	0.4913	-0.248	0.735
0.5369	-0.259	0.733	0.5369	-0.214	0.742
0.5802	-0.230	0.739	0.5802	-0.194	0.746
0.6275	-0.196	0.746	0.6275	-0.166	0.752
0.6769	-0.145	0.756	0.6769	-0.119	0.761
0.7346	-0.102	0.764	0.7346	-0.084	0.766

NACA-0012 50.8 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.599 ALPHA 1.00 REY 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1624 CM = 0.0026 CDW = 0.0108

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.547	0.677	0.0253	0.183	0.820
0.0794	-0.734	0.640	0.0794	-0.214	0.742
0.1374	-0.680	0.651	0.1374	-0.298	0.725
0.1911	-0.608	0.665	0.1911	-0.309	0.723
0.2368	-0.562	0.674	0.2368	-0.314	0.722
0.2892	-0.512	0.684	0.2892	-0.306	0.723
0.3371	-0.456	0.695	0.3371	-0.286	0.727
0.3867	-0.386	0.709	0.3867	-0.243	0.737
0.4396	-0.361	0.713	0.4396	-0.243	0.737
0.4913	-0.322	0.721	0.4913	-0.220	0.740
0.5369	-0.277	0.730	0.5369	-0.192	0.746
0.5802	-0.246	0.736	0.5802	-0.175	0.749
0.6275	-0.208	0.744	0.6275	-0.150	0.754
0.6769	-0.155	0.754	0.6769	-0.108	0.753
0.7346	-0.108	0.763	0.7346	-0.074	0.769

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.606 ALPHA 1.50 REYNOLDS 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2590 CM = 0.0043 CDW = 0.0127

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.752	0.655	0.0253	0.313	0.846
0.0794	-0.877	0.610	0.0794	-0.112	0.761
0.1374	-0.777	0.630	0.1374	-0.216	0.741
0.1911	-0.653	0.649	0.1911	-0.241	0.736
0.2368	-0.624	0.660	0.2368	-0.258	0.732
0.2892	-0.562	0.673	0.2892	-0.258	0.732
0.3371	-0.498	0.685	0.3371	-0.242	0.736
0.3867	-0.420	0.701	0.3867	-0.203	0.743
0.4396	-0.358	0.707	0.4396	-0.209	0.742
0.4913	-0.344	0.716	0.4913	-0.194	0.745
0.5369	-0.295	0.725	0.5369	-0.168	0.750
0.5802	-0.260	0.732	0.5802	-0.154	0.753
0.6275	-0.218	0.741	0.6275	-0.132	0.757
0.6769	-0.164	0.751	0.6769	-0.093	0.765
0.7346	-0.115	0.751	0.7346	-0.065	0.771

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.598 ALPHA 2.00 REYNOLDS 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3138 CM = 0.0067 CDW = 0.0128

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.975	0.593	0.0253	0.423	0.867
0.0794	-1.022	0.583	0.0794	-0.023	0.778
0.1374	-0.874	0.612	0.1374	-0.142	0.755
0.1911	-0.747	0.657	0.1911	-0.180	0.747
0.2368	-0.678	0.651	0.2368	-0.206	0.742
0.2892	-0.603	0.666	0.2892	-0.215	0.740
0.3371	-0.534	0.680	0.3371	-0.205	0.742
0.3867	-0.449	0.696	0.3867	-0.169	0.749
0.4396	-0.411	0.703	0.4396	-0.180	0.747
0.4913	-0.362	0.713	0.4913	-0.168	0.750
0.5369	-0.313	0.723	0.5369	-0.148	0.754
0.5802	-0.273	0.731	0.5802	-0.136	0.756
0.6275	-0.230	0.739	0.6275	-0.118	0.760
0.6769	-0.172	0.750	0.6769	-0.080	0.767
0.7346	-0.120	0.761	0.7346	-0.056	0.772

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.651 ALPHA 0.00 REY 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0027 CM = -0.0009 CDW = 0.0160

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.130	0.722	0.0253	-0.138	0.722
0.0794	-0.402	0.644	0.0794	-0.481	0.646
0.1374	-0.517	0.636	0.1374	-0.524	0.636
0.1911	-0.492	0.641	0.1911	-0.492	0.643
0.2368	-0.473	0.646	0.2368	-0.474	0.647
0.2892	-0.442	0.653	0.2892	-0.442	0.654
0.3371	-0.401	0.662	0.3371	-0.401	0.664
0.3867	-0.336	0.676	0.3867	-0.337	0.678
0.4396	-0.321	0.679	0.4396	-0.321	0.681
0.4913	-0.290	0.686	0.4913	-0.290	0.688
0.5369	-0.250	0.695	0.5369	-0.254	0.696
0.5802	-0.223	0.701	0.5802	-0.224	0.703
0.6275	-0.189	0.709	0.6275	-0.187	0.711
0.6769	-0.138	0.720	0.6769	-0.136	0.722
0.7346	-0.097	0.729	0.7346	-0.096	0.731

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.652 ALPHA 0.50 REY 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0895 CM = 0.0015 CDW = 0.0155

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.347	0.673	0.0253	0.058	0.765
0.0794	-0.650	0.605	0.0794	-0.335	0.677
0.1374	-0.642	0.607	0.1374	-0.402	0.662
0.1911	-0.587	0.619	0.1911	-0.401	0.662
0.2368	-0.549	0.628	0.2368	-0.399	0.663
0.2892	-0.503	0.638	0.2892	-0.381	0.657
0.3371	-0.450	0.650	0.3371	-0.350	0.674
0.3867	-0.378	0.666	0.3867	-0.295	0.686
0.4396	-0.356	0.672	0.4396	-0.287	0.687
0.4913	-0.318	0.680	0.4913	-0.263	0.693
0.5369	-0.274	0.690	0.5369	-0.227	0.701
0.5802	-0.241	0.697	0.5802	-0.205	0.706
0.6275	-0.204	0.706	0.6275	-0.175	0.712
0.6769	-0.150	0.716	0.6769	-0.126	0.723
0.7346	-0.104	0.728	0.7346	-0.089	0.732

NACA-0012 50.8 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.649 ALPHA 1.00 REY 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1781 CM = 0.0039 CDN = 0.0134

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.556	2.650	0.0253	0.207	0.799
0.0794	-0.814	3.572	0.0794	-0.213	0.786
0.1374	-0.754	2.586	0.1374	-0.307	0.635
0.1911	-0.609	2.605	0.1911	-0.321	0.682
0.2368	-0.615	2.617	0.2368	-0.330	0.680
0.2892	-0.555	3.630	0.2892	-0.324	0.631
0.3371	-0.494	3.644	0.3371	-0.301	0.686
0.3867	-0.413	2.662	0.3867	-0.251	0.697
0.4396	-0.384	2.668	0.4396	-0.253	0.697
0.4913	-0.341	2.678	0.4913	-0.233	0.732
0.5369	-0.293	2.688	0.5369	-0.203	0.746
0.5802	-0.257	2.696	0.5802	-0.183	0.713
0.6275	-0.216	2.705	0.6275	-0.157	0.718
0.6769	-0.159	2.718	0.6769	-0.113	0.728
0.7346	-0.111	2.729	0.7346	-0.079	0.736

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.651 ALPHA 1.50 REYNOLDS 0.39*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2688 CM = 0.0080 CDW = 0.0168

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.773	0.579	0.0253	0.529	0.827
0.0794	-1.013	0.525	0.0794	-0.111	0.730
0.1374	-0.876	0.556	0.1374	-0.222	0.705
0.1911	-0.759	0.582	0.1911	-0.253	0.698
0.2368	-0.685	0.599	0.2368	-0.271	0.694
0.2892	-0.610	0.616	0.2892	-0.274	0.693
0.3371	-0.536	0.632	0.3371	-0.259	0.697
0.3867	-0.449	0.651	0.3867	-0.214	0.707
0.4396	-0.412	0.660	0.4396	-0.221	0.705
0.4913	-0.363	0.671	0.4913	-0.205	0.709
0.5369	-0.309	0.683	0.5369	-0.179	0.714
0.5802	-0.270	0.691	0.5802	-0.163	0.718
0.6275	-0.225	0.701	0.6275	-0.141	0.723
0.6769	-0.168	0.714	0.6769	-0.099	0.732
0.7346	-0.115	0.726	0.7346	-0.069	0.739

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.676 ALPHA 0.00 REYNOLDS 0.40×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0046 CM = -0.0006 CDW = 0.0178

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.121	0.708	0.0253	-0.128	0.708
0.0794	-0.496	0.620	0.0794	-0.493	0.622
0.1374	-0.534	0.611	0.1374	-0.543	0.610
0.1911	-0.509	0.617	0.1911	-0.516	0.616
0.2368	-0.490	0.621	0.2368	-0.497	0.621
0.2892	-0.459	0.628	0.2892	-0.461	0.629
0.3371	-0.413	0.639	0.3371	-0.418	0.639
0.3867	-0.347	0.655	0.3867	-0.350	0.655
0.4396	-0.331	0.659	0.4396	-0.334	0.659
0.4913	-0.299	0.667	0.4913	-0.300	0.667
0.5369	-0.257	0.677	0.5369	-0.262	0.676
0.5802	-0.229	0.683	0.5802	-0.230	0.673
0.6275	-0.193	0.692	0.6275	-0.193	0.692
0.6769	-0.141	0.704	0.6769	-0.140	0.704
0.7346	-0.097	0.714	0.7346	-0.098	0.714

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.675 ALPHA 0.50 REY 0.40×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.3951 CM = 0.0021 CDW = 0.0157

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.340	0.627	0.0253	0.071	0.754
0.0794	-0.678	0.577	0.0794	-0.335	0.659
0.1374	-0.674	0.578	0.1374	-0.408	0.642
0.1911	-0.616	0.592	0.1911	-0.411	0.641
0.2368	-0.576	0.601	0.2368	-0.411	0.641
0.2892	-0.525	0.613	0.2892	-0.394	0.645
0.3371	-0.468	0.626	0.3371	-0.363	0.652
0.3867	-0.392	0.644	0.3867	-0.304	0.666
0.4396	-0.306	0.653	0.4396	-0.297	0.668
0.4913	-0.327	0.659	0.4913	-0.270	0.674
0.5369	-0.260	0.670	0.5369	-0.235	0.682
0.5802	-0.248	0.678	0.5802	-0.210	0.686
0.6275	-0.208	0.688	0.6275	-0.179	0.695
0.6769	-0.153	0.701	0.6769	-0.129	0.737
0.7346	-0.104	0.712	0.7346	-0.091	0.716

NACA-0012 50.8 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.676 ALPHA 1.00 REY 0.40×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1909 CH = 0.0052 CDW = 0.0180

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.547	0.607	0.0253	0.236	0.791
0.0794	-0.872	0.531	0.0794	-0.202	0.698
0.1374	-0.804	0.547	0.1374	-0.305	0.664
0.1911	-0.710	0.569	0.1911	-0.328	0.659
0.2368	-0.651	0.563	0.2368	-0.337	0.656
0.2892	-0.584	0.599	0.2892	-0.334	0.657
0.3371	-0.515	0.615	0.3371	-0.309	0.663
0.3867	-0.431	0.635	0.3867	-0.261	0.674
0.4396	-0.401	0.642	0.4396	-0.261	0.674
0.4913	-0.351	0.654	0.4913	-0.241	0.679
0.5369	-0.300	0.666	0.5369	-0.208	0.687
0.5802	-0.263	0.675	0.5802	-0.189	0.691
0.6275	-0.219	0.685	0.6275	-0.162	0.698
0.6769	-0.161	0.699	0.6769	-0.116	0.739
0.7346	-0.111	0.711	0.7346	-0.081	0.717

NACA-0012 50.8 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.676 ALPHA 1.50 REYNOLDS 0.40*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2907 CM = 0.0111 CDW = 0.0143

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.749	2.559	0.0253	0.361	0.821
0.0794	-1.123	2.471	0.0794	-0.091	0.714
0.1374	-0.960	2.509	0.1374	-0.214	0.686
0.1911	-0.801	2.547	0.1911	-0.251	0.677
0.2368	-0.718	2.567	0.2368	-0.274	0.671
0.2892	-0.635	2.586	0.2892	-0.278	0.670
0.3371	-0.564	2.604	0.3371	-0.265	0.674
0.3867	-0.466	2.626	0.3867	-0.220	0.654
0.4396	-0.424	2.636	0.4396	-0.227	0.683
0.4913	-0.371	2.648	0.4913	-0.212	0.686
0.5369	-0.317	2.661	0.5369	-0.186	0.692
0.5802	-0.275	2.671	0.5802	-0.169	0.696
0.6275	-0.229	2.682	0.6275	-0.146	0.732
0.6769	-0.168	2.696	0.6769	-0.103	0.712
0.7346	-0.115	2.709	0.7346	-0.072	0.719

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.099 ALPHA 0.00 REY 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0099 CM = -0.0012 CDW = 0.0162

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.088	0.700	0.0253	-0.116	0.691
0.0794	-0.492	0.600	0.0794	-0.508	0.593
0.1374	-0.543	0.587	0.1374	-0.569	0.579
0.1911	-0.525	0.592	0.1911	-0.545	0.584
0.2368	-0.509	0.596	0.2368	-0.524	0.589
0.2892	-0.475	0.604	0.2892	-0.488	0.593
0.3371	-0.429	0.615	0.3371	-0.440	0.612
0.3867	-0.358	0.633	0.3867	-0.366	0.629
0.4396	-0.342	0.637	0.4396	-0.349	0.633
0.4913	-0.309	0.645	0.4913	-0.313	0.642
0.5469	-0.265	0.656	0.5469	-0.272	0.652
0.5802	-0.235	0.663	0.5802	-0.238	0.661
0.6275	-0.199	0.672	0.6275	-0.199	0.673
0.6769	-0.145	0.685	0.6769	-0.143	0.684
0.7346	-0.100	0.696	0.7346	-0.099	0.695

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.700 ALPHA 0.50 REYNOLDS 0.39*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1045 CM = 0.0029 CDW = 0.0169

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.323	0.640	0.0253	0.102	0.746
0.0794	-0.709	0.545	0.0794	-0.324	0.641
0.1374	-0.708	0.545	0.1374	-0.410	0.620
0.1911	-0.652	0.559	0.1911	-0.419	0.617
0.2368	-0.606	0.570	0.2368	-0.422	0.617
0.2892	-0.552	0.584	0.2892	-0.405	0.621
0.3371	-0.487	0.600	0.3371	-0.371	0.629
0.3867	-0.406	0.620	0.3867	-0.310	0.644
0.4396	-0.376	0.627	0.4396	-0.302	0.647
0.4913	-0.337	0.637	0.4913	-0.277	0.653
0.5369	-0.287	0.650	0.5369	-0.239	0.663
0.5802	-0.252	0.658	0.5802	-0.215	0.669
0.6275	-0.211	0.668	0.6275	-0.182	0.677
0.6769	-0.155	0.682	0.6769	-0.133	0.689
0.7346	-0.105	0.694	0.7346	-0.093	0.699

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.699 ALPHA 1.00 REY 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2100 CM = 0.0067 CDW = 0.0159

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.553	0.590	0.0253	0.259	0.785
0.0794	-0.986	0.482	0.0794	-0.188	0.675
0.1374	-0.873	0.506	0.1374	-0.299	0.647
0.1911	-0.777	0.529	0.1911	-0.327	0.643
0.2368	-0.697	0.549	0.2368	-0.341	0.637
0.2892	-0.618	0.569	0.2892	-0.341	0.637
0.3371	-0.539	0.588	0.3371	-0.316	0.643
0.3867	-0.443	0.610	0.3867	-0.266	0.655
0.4396	-0.413	0.623	0.4396	-0.265	0.656
0.4913	-0.393	0.631	0.4913	-0.247	0.660
0.5369	-0.368	0.645	0.5369	-0.213	0.668
0.5802	-0.268	0.655	0.5802	-0.195	0.673
0.6275	-0.224	0.666	0.6275	-0.166	0.680
0.6769	-0.164	0.680	0.6769	-0.120	0.692
0.7346	-0.111	0.694	0.7346	-0.082	0.701

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.720 ALPHA 0.00 REYNOLDS 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0030 CM = -0.0011 CDW = 0.0146

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.095	0.683	0.0253	-0.101	0.682
0.0794	-0.522	0.573	0.0794	-0.517	0.574
0.1374	-0.583	0.557	0.1374	-0.586	0.557
0.1911	-0.565	0.562	0.1911	-0.571	0.561
0.2368	-0.545	0.567	0.2368	-0.551	0.566
0.2892	-0.507	0.577	0.2892	-0.509	0.577
0.3371	-0.452	0.591	0.3371	-0.457	0.593
0.3867	-0.377	0.610	0.3867	-0.378	0.610
0.4396	-0.356	0.616	0.4396	-0.358	0.616
0.4913	-0.321	0.625	0.4913	-0.321	0.625
0.5369	-0.274	0.637	0.5369	-0.278	0.636
0.5802	-0.242	0.645	0.5802	-0.243	0.645
0.6275	-0.205	0.655	0.6275	-0.201	0.656
0.6769	-0.149	0.669	0.6769	-0.145	0.670
0.7346	-0.101	0.681	0.7346	-0.099	0.682

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.721 ALPHA 0.50 REYNOLDS 0.39*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1150 CH = 0.0039 CDW = 0.0147

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.360	0.622	0.0253	0.107	0.735
0.0794	-0.769	0.509	0.0794	-0.334	0.622
0.1374	-0.768	0.509	0.1374	-0.427	0.598
0.1911	-0.712	0.524	0.1911	-0.440	0.594
0.2368	-0.656	0.538	0.2368	-0.443	0.594
0.2892	-0.587	0.556	0.2892	-0.425	0.598
0.3371	-0.513	0.575	0.3371	-0.388	0.608
0.3867	-0.425	0.598	0.3867	-0.324	0.624
0.4396	-0.393	0.606	0.4396	-0.315	0.627
0.4913	-0.349	0.618	0.4913	-0.288	0.634
0.5369	-0.297	0.631	0.5369	-0.249	0.644
0.5802	-0.257	0.641	0.5802	-0.222	0.651
0.6275	-0.216	0.652	0.6275	-0.189	0.659
0.6769	-0.156	0.667	0.6769	-0.136	0.673
0.7346	-0.106	0.680	0.7346	-0.094	0.684

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.721 ALPHA 1.00 REYNOLDS 0.39*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2298 CM = 0.0092 CDW = 0.0168

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.522	0.573	0.0253	0.256	0.774
0.0794	-1.038	0.440	0.0794	-0.197	0.657
0.1374	-1.014	0.446	0.1374	-0.310	0.628
0.1911	-0.913	0.472	0.1911	-0.341	0.620
0.2368	-0.794	0.503	0.2368	-0.358	0.616
0.2892	-0.651	0.539	0.2892	-0.358	0.616
0.3371	-0.558	0.563	0.3371	-0.333	0.622
0.3867	-0.462	0.568	0.3867	-0.276	0.637
0.4396	-0.420	0.599	0.4396	-0.275	0.637
0.4913	-0.369	0.612	0.4913	-0.256	0.642
0.5369	-0.313	0.626	0.5369	-0.221	0.651
0.5802	-0.270	0.637	0.5802	-0.201	0.656
0.6275	-0.224	0.649	0.6275	-0.170	0.664
0.6769	-0.163	0.665	0.6769	-0.123	0.676
0.7346	-0.108	0.679	0.7346	-0.085	0.686

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.739 ALPHA 0.00 REY 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0104 CM = -0.0008 CDW = 0.0140

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.061	0.679	0.0253	-0.085	0.671
0.0794	-0.512	0.559	0.0794	-0.528	0.553
0.1374	-0.593	0.537	0.1374	-0.619	0.529
0.1911	-0.588	0.539	0.1911	-0.609	0.532
0.2368	-0.571	0.543	0.2368	-0.591	0.536
0.2892	-0.532	0.554	0.2892	-0.545	0.549
0.3371	-0.471	0.570	0.3371	-0.484	0.565
0.3867	-0.387	0.592	0.3867	-0.395	0.589
0.4396	-0.365	0.598	0.4396	-0.372	0.595
0.4913	-0.329	0.607	0.4913	-0.332	0.605
0.5359	-0.281	0.620	0.5359	-0.289	0.617
0.5802	-0.247	0.629	0.5802	-0.249	0.627
0.6275	-0.209	0.639	0.6275	-0.206	0.639
0.6769	-0.150	0.655	0.6769	-0.148	0.654
0.7346	-0.101	0.668	0.7346	-0.104	0.666

NACA-0012 50.8 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.738 ALPHA 0.50 REYNOLDS 0.39⁶ × 10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1285 CM = 0.0053 CDW = 0.0157

UPPER SURFACE VALUES

X/C	CP	P/H
0.0253	-0.303	0.615
0.0794	-0.802	0.483
0.1374	-0.815	0.479
0.1911	-0.790	0.486
0.2368	-0.731	0.502
0.2892	-0.631	0.528
0.3371	-0.535	0.554
0.3867	-0.436	0.560
0.4396	-0.401	0.569
0.4913	-0.355	0.602
0.5369	-0.301	0.616
0.5802	-0.262	0.626
0.6275	-0.215	0.639
0.6769	-0.157	0.654
0.7346	-0.104	0.668

LOWER SURFACE VALUES

X/C	CP	P/H
0.0253	0.136	0.731
0.0794	-0.320	0.610
0.1374	-0.425	0.582
0.1911	-0.446	0.576
0.2368	-0.454	0.574
0.2892	-0.441	0.578
0.3371	-0.402	0.588
0.3867	-0.332	0.607
0.4396	-0.322	0.609
0.4913	-0.296	0.616
0.5369	-0.255	0.627
0.5802	-0.227	0.635
0.6275	-0.192	0.644
0.6769	-0.138	0.658
0.7346	-0.095	0.670

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.739 ALPHA 1.00 REYNOLDS 0.39×10^6

INTEGRATED FORCE COEFFICIENTS
CN = 0.2658 CM = 0.0129 CDW = 0.0176

UPPER SURFACE VALUES

X/C	CP	P/H
0.0253	-0.483	0.567
0.0794	-0.994	0.431
0.1374	-1.086	0.406
0.1911	-1.051	0.416
0.2368	-1.032	0.421
0.2892	-0.846	0.470
0.3371	-0.580	0.541
0.3867	-0.457	0.574
0.4396	-0.402	0.588
0.4913	-0.354	0.601
0.5369	-0.303	0.614
0.5802	-0.262	0.625
0.6275	-0.216	0.637
0.6769	-0.157	0.653
0.7346	-0.103	0.668

LOWER SURFACE VALUES

X/C	CP	P/H
0.0253	0.287	0.772
0.0794	-0.174	0.650
0.1374	-0.297	0.619
0.1911	-0.335	0.607
0.2368	-0.355	0.602
0.2892	-0.358	0.601
0.3371	-0.337	0.607
0.3867	-0.277	0.623
0.4396	-0.277	0.623
0.4913	-0.259	0.628
0.5369	-0.225	0.637
0.5802	-0.203	0.643
0.6275	-0.171	0.651
0.6769	-0.122	0.664
0.7346	-0.085	0.674

NACA-0012 50.8 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.760 ALPHA 0.00 REY 0.40×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0126 CM = -0.0015 CDW = 0.0171

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.351	0.673	0.0253	-0.061	0.654
0.0794	-0.509	0.542	0.0794	-0.533	0.534
0.1374	-0.608	0.514	0.1374	-0.642	0.504
0.1911	-0.631	0.508	0.1911	-0.668	0.496
0.2368	-0.627	0.509	0.2368	-0.659	0.499
0.2892	-0.585	0.521	0.2892	-0.603	0.514
0.3371	-0.509	0.542	0.3371	-0.519	0.537
0.3867	-0.405	0.570	0.3867	-0.411	0.567
0.4396	-0.379	0.577	0.4396	-0.384	0.575
0.4913	-0.341	0.588	0.4913	-0.343	0.586
0.5369	-0.290	0.602	0.5369	-0.296	0.599
0.5802	-0.255	0.611	0.5802	-0.255	0.611
0.6275	-0.212	0.623	0.6275	-0.209	0.623
0.6769	-0.152	0.640	0.6769	-0.149	0.640
0.7346	-0.101	0.654	0.7346	-0.101	0.653

NACA-0012 50.8 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.757 ALPHA 0.50 REYNOLDS 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

$C_M = 0.1463$ $C_M = 0.0067$ $C_{DM} = 0.0174$

UPPER SURFACE VALUES

X/C	CP	P/H
0.0253	-0.263	0.611
0.0794	-0.800	0.464
0.1374	-0.839	0.453
0.1911	-0.864	0.446
0.2368	-0.866	0.440
0.2892	-0.792	0.466
0.3371	-0.571	0.527
0.3867	-0.436	0.564
0.4396	-0.394	0.575
0.4913	-0.350	0.587
0.5369	-0.299	0.601
0.5802	-0.260	0.612
0.6275	-0.212	0.625
0.6769	-0.154	0.641
0.7346	-0.100	0.656

LOWER SURFACE VALUES

X/C	CP	P/H
0.0253	0.168	0.728
0.0794	-0.300	0.598
0.1374	-0.418	0.566
0.1911	-0.453	0.556
0.2368	-0.468	0.552
0.2892	-0.458	0.555
0.3371	-0.419	0.566
0.3867	-0.340	0.587
0.4396	-0.331	0.590
0.4913	-0.302	0.598
0.5369	-0.261	0.610
0.5802	-0.232	0.618
0.6275	-0.195	0.628
0.6769	-0.138	0.643
0.7346	-0.095	0.655

NACA-0012: 50.8 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.761

ALPHA 1.00

REYNOLDS NO. 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.2887 CM = 0.0110 CDW = 0.0248

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.420	0.565	0.0253	0.297	0.765
0.0794	-0.925	0.425	0.0794	-0.171	0.636
0.1374	-1.042	0.393	0.1374	-0.302	0.600
0.1911	-1.041	0.393	0.1911	-0.345	0.588
0.2358	-1.073	0.385	0.2358	-0.370	0.581
0.2892	-1.083	0.362	0.2892	-0.376	0.580
0.3371	-1.031	0.396	0.3371	-0.353	0.586
0.3867	-0.615	0.511	0.3867	-0.288	0.604
0.4396	-0.429	0.562	0.4396	-0.287	0.604
0.4913	-0.363	0.569	0.4913	-0.267	0.610
0.5369	-0.277	0.605	0.5369	-0.232	0.620
0.5802	-0.237	0.616	0.5802	-0.210	0.626
0.6275	-0.192	0.628	0.6275	-0.178	0.634
0.6769	-0.140	0.642	0.6769	-0.126	0.649
0.7346	-0.088	0.657	0.7346	-0.087	0.663

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.780 ALPHA 0.00 REYNOLDS 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0188 CM = -0.0024 CDW = 0.0146

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	0.001	0.669	0.0253	-0.035	0.659
0.0794	-0.489	0.529	0.0794	-0.520	0.521
0.1374	-0.605	0.496	0.1374	-0.645	0.485
0.1911	-0.674	0.476	0.1911	-0.703	0.468
0.2368	-0.699	0.469	0.2368	-0.742	0.457
0.2892	-0.672	0.477	0.2892	-0.732	0.460
0.3371	-0.573	0.505	0.3371	-0.600	0.498
0.3867	-0.420	0.549	0.3867	-0.419	0.549
0.4396	-0.363	0.559	0.4396	-0.384	0.560
0.4913	-0.344	0.571	0.4913	-0.344	0.571
0.5369	-0.294	0.585	0.5369	-0.297	0.585
0.5802	-0.256	0.596	0.5802	-0.254	0.597
0.6275	-0.210	0.609	0.6275	-0.211	0.609
0.6769	-0.150	0.626	0.6769	-0.147	0.627
0.7346	-0.100	0.640	0.7346	-0.098	0.641

NACA-0012 50.8 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.781 ALPHA 0.50 REYNOLDS 0.39*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1738 CM = 0.0080 CDW = 0.0186

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.205	0.609	0.0253	0.168	0.716
0.0794	-0.746	0.455	0.0794	-0.338	0.580
0.1374	-0.801	0.439	0.1374	-0.441	0.542
0.1911	-0.840	0.428	0.1911	-0.490	0.528
0.2368	-0.893	0.414	0.2368	-0.513	0.521
0.2892	-0.940	0.397	0.2892	-0.513	0.521
0.3371	-0.977	0.389	0.3371	-0.470	0.534
0.3867	-0.823	0.439	0.3867	-0.369	0.562
0.4396	-0.446	0.540	0.4396	-0.351	0.567
0.4913	-0.330	0.573	0.4913	-0.319	0.576
0.5369	-0.208	0.591	0.5369	-0.276	0.589
0.5802	-0.201	0.602	0.5802	-0.241	0.599
0.6275	-0.187	0.614	0.6275	-0.202	0.613
0.6769	-0.155	0.629	0.6769	-0.142	0.627
0.7346	-0.085	0.643	0.7346	-0.097	0.640

NACA-0012 50.8 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.781 ALPHA 1.00 REYNOLDS 0.39×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.2745$ $C_M = 0.0066$ $C_{DW} = 0.0252$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.325	0.575	0.0253	0.266	0.744
0.0794	-0.838	0.429	0.0794	-0.206	0.609
0.1374	-0.956	0.395	0.1374	-0.344	0.572
0.1911	-0.972	0.391	0.1911	-0.397	0.555
0.2368	-1.009	0.380	0.2368	-0.426	0.546
0.2892	-1.036	0.372	0.2892	-0.434	0.544
0.3371	-1.081	0.359	0.3371	-0.408	0.551
0.3867	-1.016	0.378	0.3867	-0.328	0.574
0.4396	-0.670	0.477	0.4396	-0.321	0.576
0.4913	-0.408	0.552	0.4913	-0.298	0.583
0.5359	-0.307	0.581	0.5359	-0.257	0.594
0.5802	-0.235	0.601	0.5802	-0.228	0.603
0.6275	-0.172	0.619	0.6275	-0.191	0.613
0.6769	-0.124	0.633	0.6769	-0.136	0.629
0.7346	-0.069	0.649	0.7346	-0.094	0.641

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.799 ALPHA 0.00 REY 0.40×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0173 CM = -0.0006 CDW = 0.0171

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	0.014	0.660	0.0253	0.006	0.658
0.0794	-0.484	0.514	0.0794	-0.485	0.513
0.1374	-0.600	0.480	0.1374	-0.612	0.475
0.1911	-0.674	0.458	0.1911	-0.676	0.456
0.2368	-0.736	0.440	0.2368	-0.738	0.438
0.2892	-0.789	0.424	0.2892	-0.795	0.421
0.3371	-0.824	0.414	0.3371	-0.844	0.407
0.3867	-0.780	0.400	0.3867	-0.789	0.423
0.4396	-0.407	0.537	0.4396	-0.524	0.501
0.4913	-0.321	0.562	0.4913	-0.327	0.559
0.5369	-0.274	0.576	0.5369	-0.270	0.575
0.5802	-0.241	0.585	0.5802	-0.232	0.587
0.6275	-0.198	0.598	0.6275	-0.188	0.599
0.6769	-0.140	0.615	0.6769	-0.132	0.616
0.7346	-0.091	0.629	0.7346	-0.085	0.630

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.798 ALPHA 0.50 REY 6.40×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1523 CM = 0.0029 CDW = 0.0242

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.145	0.614	0.0253	0.143	0.698
0.0794	-0.684	0.456	0.0794	-0.346	0.554
0.1374	-0.751	0.437	0.1374	-0.485	0.513
0.1911	-0.795	0.424	0.1911	-0.560	0.491
0.2368	-0.846	0.408	0.2368	-0.606	0.478
0.2892	-0.903	0.392	0.2892	-0.642	0.467
0.3371	-0.954	0.377	0.3371	-0.681	0.479
0.3867	-0.965	0.382	0.3867	-0.433	0.529
0.4396	-0.825	0.415	0.4396	-0.371	0.547
0.4913	-0.435	0.529	0.4913	-0.337	0.557
0.5369	-0.298	0.569	0.5369	-0.290	0.571
0.5802	-0.227	0.590	0.5802	-0.253	0.581
0.6275	-0.172	0.606	0.6275	-0.206	0.595
0.6769	-0.119	0.622	0.6769	-0.147	0.613
0.7346	-0.070	0.636	0.7346	-0.098	0.627

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.800 ALPHA 1.00 REYNOLDS 0.40*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2511 CM = -0.0009 CDW = 0.0300

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.237	0.586	0.0253	0.229	0.724
0.0794	-0.759	0.433	0.0794	-0.250	0.584
0.1374	-0.866	0.401	0.1374	-0.392	0.542
0.1911	-0.886	0.395	0.1911	-0.437	0.529
0.2368	-0.933	0.381	0.2368	-0.497	0.511
0.2892	-0.966	0.372	0.2892	-0.517	0.505
0.3371	-1.021	0.355	0.3371	-0.484	0.515
0.3867	-1.008	0.359	0.3867	-0.376	0.507
0.4396	-0.973	0.370	0.4396	-0.354	0.554
0.4913	-0.643	0.467	0.4913	-0.326	0.562
0.5369	-0.396	0.540	0.5369	-0.281	0.575
0.5842	-0.288	0.571	0.5842	-0.249	0.585
0.6275	-0.193	0.599	0.6275	-0.207	0.597
0.6769	-0.129	0.618	0.6769	-0.149	0.614
0.7346	-0.064	0.637	0.7346	-0.105	0.627

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.820 ALPHA 0.00 REY 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

CN = -0.0155 CM = 0.0008 CDW = 0.0273

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	0.045	0.656	0.0253	0.042	0.655
0.0794	-0.453	0.505	0.0794	-0.452	0.505
0.1374	-0.567	0.471	0.1374	-0.574	0.468
0.1911	-0.640	0.449	0.1911	-0.641	0.448
0.2368	-0.705	0.429	0.2368	-0.707	0.428
0.2892	-0.767	0.410	0.2892	-0.767	0.409
0.3371	-0.820	0.394	0.3371	-0.824	0.392
0.3867	-0.805	0.399	0.3867	-0.810	0.396
0.4396	-0.806	0.398	0.4396	-0.816	0.394
0.4913	-0.735	0.420	0.4913	-0.793	0.401
0.5369	-0.304	0.526	0.5369	-0.466	0.501
0.5802	-0.250	0.567	0.5802	-0.269	0.560
0.6275	-0.175	0.590	0.6275	-0.186	0.586
0.6769	-0.116	0.607	0.6769	-0.117	0.606
0.7346	-0.066	0.623	0.7346	-0.063	0.623

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.820 ALPHA 0.50 REY 0.41×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.1014 CM = -0.0086 CDW = 0.0280

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.043	0.629	0.0253	0.125	0.682
0.0794	-0.576	0.468	0.0794	-0.565	0.534
0.1374	-0.658	0.443	0.1374	-0.501	0.493
0.1911	-0.702	0.430	0.1911	-0.585	0.469
0.2368	-0.769	0.409	0.2368	-0.646	0.449
0.2892	-0.823	0.393	0.2892	-0.701	0.433
0.3371	-0.877	0.377	0.3371	-0.749	0.418
0.3867	-0.873	0.378	0.3867	-0.728	0.424
0.4396	-0.880	0.375	0.4396	-0.670	0.442
0.4913	-0.871	0.378	0.4913	-0.412	0.520
0.5369	-0.559	0.473	0.5369	-0.274	0.561
0.5802	-0.319	0.545	0.5802	-0.232	0.574
0.6275	-0.209	0.579	0.6275	-0.190	0.587
0.6769	-0.129	0.603	0.6769	-0.136	0.603
0.7346	-0.062	0.623	0.7346	-0.089	0.617

NACA-0012 50.8 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.820 ALPHA 1.00 REYNOLDS 0.41*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1873 CM = -0.0073 CDW = 0.0336

UPPER SURFACE VALUES

LOWER SURFACE VALUES

X/C	CP	P/H	X/C	CP	P/H
0.0253	-0.151	0.603	0.0253	0.197	0.704
0.0794	-0.659	0.443	0.0794	-0.288	0.557
0.1374	-0.751	0.415	0.1374	-0.435	0.513
0.1911	-0.794	0.403	0.1911	-0.522	0.487
0.2368	-0.829	0.392	0.2368	-0.574	0.471
0.2892	-0.885	0.375	0.2892	-0.638	0.452
0.3371	-0.943	0.357	0.3371	-0.683	0.438
0.3867	-0.929	0.362	0.3867	-0.644	0.450
0.4396	-0.934	0.360	0.4396	-0.526	0.485
0.4913	-0.916	0.366	0.4913	-0.364	0.534
0.5369	-0.590	0.464	0.5369	-0.292	0.556
0.5802	-0.372	0.530	0.5802	-0.258	0.566
0.6275	-0.251	0.567	0.6275	-0.218	0.579
0.6769	-0.164	0.593	0.6769	-0.160	0.596
0.7346	-0.081	0.618	0.7346	-0.114	0.610

APPENDIX B
PLOTTED RESULTS

NOTATION

<i>AL</i>	(α)	Angle of incidence
<i>CM</i>	(C_m)	Pitching moment coefficient
<i>CN</i>	(C_N)	Normal force coefficient
<i>H</i>	(H)	Free stream total pressure
<i>M</i>	(M)	Free stream Mach number
<i>P</i>	(P)	Local static pressure
<i>R</i>	(R)	Reynolds number based on aerofoil chord

Note: Symbols in brackets are corresponding symbols used in main part of report.

CONTENTS

No.	Model Chord	M	Nominal α	Corrected α	R
1	203.2	0.50	0	0	1.68×10^6
2	203.2	0.50	1	1	1.68×10^6
3	203.2	0.50	2	2	1.68×10^6
4	203.2	0.50	3	3	1.68×10^6
5	203.2	0.50	4	4	1.68×10^6
6	203.2	0.50	5	5	1.68×10^6
7	203.2	0.50	6	6	1.68×10^6
8	203.2	0.50	7	7	1.68×10^6
9	203.2	0.50	8	8	1.68×10^6
10	203.2	0.60	0	0	1.68×10^6
11	203.2	0.60	1	1	1.68×10^6
12	203.2	0.60	2	2	1.68×10^6
13	203.2	0.60	3	3	1.68×10^6
14	203.2	0.60	4	4	1.68×10^6
15	203.2	0.60	5	5	1.68×10^6
16	203.2	0.60	6	6	1.68×10^6
17	203.2	0.60	7	7	1.68×10^6
18	203.2	0.65	0	0	1.68×10^6
19	203.2	0.65	1	1	1.68×10^6
20	203.2	0.65	2	2	1.68×10^6
21	203.2	0.65	3	3	1.68×10^6
22	203.2	0.65	4	4	1.68×10^6
23	203.2	0.65	5	5	1.68×10^6
24	203.2	0.65	6	6	1.68×10^6
25	203.2	0.675	0	0	1.68×10^6
26	203.2	0.675	1	1	1.68×10^6
27	203.2	0.675	2	2	1.68×10^6
28	203.2	0.675	3	3	1.68×10^6
29	203.2	0.675	4	4	1.68×10^6
30	203.2	0.675	5	5	1.68×10^6
31	203.2	0.70	0	0	1.68×10^6
32	203.2	0.70	1	1	1.68×10^6
33	203.2	0.70	2	2	1.68×10^6
34	203.2	0.70	3	3	1.68×10^6
35	203.2	0.70	4	4	1.68×10^6
36	203.2	0.70	5	5	1.68×10^6
37	203.2	0.72	0	0	1.68×10^6
38	203.2	0.72	1	1	1.68×10^6
39	203.2	0.72	2	2	1.68×10^6
40	203.2	0.72	3	3	1.68×10^6
41	203.2	0.72	4	4	1.68×10^6
42	203.2	0.74	0	0	1.68×10^6
43	203.2	0.74	1	1	1.68×10^6
44	203.2	0.74	2	2	1.68×10^6

No.	Model Chord	M	Nominal α	Corrected α	R
45	203-2	0.74	3	3	1.68×10^6
46	203-2	0.74	4	4	1.68×10^6
47	203-2	0.76	0	0	1.68×10^6
48	203-2	0.76	.	1	1.68×10^6
49	203-2	0.76	2	2	1.68×10^6
50	203-2	0.76	3	3	1.68×10^6
51	203-2	0.76	4	4	1.68×10^6
52	203-2	0.78	0	0	1.68×10^6
53	203-2	0.78	1	1	1.68×10^6
54	203-2	0.78	2	2	1.68×10^6
55	203-2	0.78	3	3	1.68×10^6
56	203-2	0.80	0	0	1.68×10^6
57	203-2	0.80	1	1	1.68×10^6
58	203-2	0.80	2	2	1.68×10^6
59	203-2	0.82	0	0	1.68×10^6
60	203-2	0.82	1	1	1.68×10^6
61	203-2	0.82	2	2	1.68×10^6
62	203-2	0.50	0	0	0.84×10^6
63	203-2	0.50	1	1	0.84×10^6
64	203-2	0.50	2	2	0.84×10^6
65	203-2	0.50	3	3	0.84×10^6
66	203-2	0.50	4	4	0.84×10^6
67	203-2	0.50	5	5	0.84×10^6
68	203-2	0.50	6	6	0.84×10^6
69	203-2	0.50	7	7	0.84×10^6
70	203-2	0.50	8	8	0.84×10^6
71	203-2	0.60	0	0	0.84×10^6
72	203-2	0.65	0	0	0.84×10^6
73	203-2	0.675	0	0	0.84×10^6
74	203-2	0.70	0	0	0.84×10^6
75	203-2	0.70	1	1	0.84×10^6
76	203-2	0.70	2	2	0.84×10^6
77	203-2	0.70	3	3	0.84×10^6
78	203-2	0.70	4	4	0.84×10^6
79	203-2	0.70	5	5	0.84×10^6
80	203-2	0.72	0	0	0.84×10^6
81	203-2	0.74	0	0	0.84×10^6
82	203-2	0.76	0	0	0.84×10^6
83	203-2	0.78	0	0	0.84×10^6
84	203-2	0.80	0	0	0.84×10^6
85	203-2	0.80	1	1	0.84×10^6
86	203-2	0.80	2	2	0.84×10^6
87	203-2	0.50	0	0	0.42×10^6
88	203-2	0.50	1	1	0.42×10^6
89	203-2	0.50	2	2	0.42×10^6
90	203-2	0.50	3	3	0.42×10^6
91	203-2	0.50	4	4	0.42×10^6
92	203-2	0.50	5	5	0.42×10^6
93	203-2	0.50	6	6	0.42×10^6
94	203-2	0.50	7	7	0.42×10^6
95	203-2	0.50	8	8	0.42×10^6

No.	Model Chord	M	Nominal α	Corrected α	R
96	203.2	0.70	0	0	0.42×10^6
97	203.2	0.70	1	1	0.42×10^6
98	203.2	0.70	2	2	0.42×10^6
99	203.2	0.70	3	3	0.42×10^6
100	203.2	0.70	4	4	0.42×10^6
101	203.2	0.70	5	5	0.42×10^6
102	203.2	0.80	0	0	0.42×10^6
103	203.2	0.80	1	1	0.42×10^6
104	203.2	0.80	2	2	0.42×10^6
105	101.6	0.50	0	0.01	0.84×10^6
106	101.6	0.50	1	1.02	0.84×10^6
107	101.6	0.50	2	2.03	0.84×10^6
108	101.6	0.50	3	3.05	0.84×10^6
109	101.6	0.50	4	4.06	0.84×10^6
110	101.6	0.50	5	5.07	0.84×10^6
111	101.6	0.50	6	6.09	0.84×10^6
112	101.6	0.50	7	7.10	0.84×10^6
113	101.6	0.50	8	8.11	0.84×10^6
114	101.6	0.60	0	0.01	0.84×10^6
115	101.6	0.60	1	1.03	0.84×10^6
116	101.6	0.60	2	2.04	0.84×10^6
117	101.6	0.60	3	3.06	0.84×10^6
118	101.6	0.60	4	4.08	0.84×10^6
119	101.6	0.60	5	5.09	0.84×10^6
120	101.6	0.60	6	6.12	0.84×10^6
121	101.6	0.60	7	7.15	0.84×10^6
122	101.6	0.65	0	0.01	0.84×10^6
123	101.6	0.65	1	1.03	0.84×10^6
124	101.6	0.65	2	2.05	0.84×10^6
125	101.6	0.65	3	3.07	0.84×10^6
126	101.6	0.65	4	4.10	0.84×10^6
127	101.6	0.65	5	5.12	0.84×10^6
128	101.6	0.65	6	6.14	0.84×10^6
129	101.6	0.675	0	0.01	0.84×10^6
130	101.6	0.675	1	1.03	0.84×10^6
131	101.6	0.675	2	2.06	0.84×10^6
132	101.6	0.675	3	3.09	0.84×10^6
133	101.6	0.675	4	4.11	0.84×10^6
134	101.6	0.675	5	5.13	0.84×10^6
135	101.6	0.70	0	0.01	0.84×10^6
136	101.6	0.70	1	1.04	0.84×10^6
137	101.6	0.70	2	2.06	0.84×10^6
138	101.6	0.70	3	3.10	0.84×10^6
139	101.6	0.70	4	4.12	0.84×10^6
140	101.6	0.70	5	5.14	0.84×10^6
141	101.6	0.72	0	0.01	0.84×10^6
142	101.6	0.72	1	1.04	0.84×10^6
143	101.6	0.72	2	2.07	0.84×10^6
144	101.6	0.72	3	3.10	0.84×10^6
145	101.6	0.72	4	4.12	0.84×10^6
146	101.6	0.74	0	0.01	0.84×10^6

No.	Model Chord	M	Nominal α	Corrected α	R
147	101.6	0.74	1	1.04	0.84×10^6
148	101.6	0.74	2	2.08	0.84×10^6
149	101.6	0.74	3	3.10	0.84×10^6
150	101.6	0.74	4	4.11	0.84×10^6
151	101.6	0.76	0	0.01	0.84×10^6
152	101.6	0.76	1	1.05	0.84×10^6
153	101.6	0.76	2	2.08	0.84×10^6
154	101.6	0.76	3	3.09	0.84×10^6
155	101.6	0.76	4	4.09	0.84×10^6
156	101.6	0.78	0	0	0.84×10^6
157	101.6	0.78	1	1.04	0.84×10^6
158	101.6	0.78	2	2.07	0.84×10^6
159	101.6	0.78	3	3.07	0.84×10^6
160	101.6	0.80	0	0	0.84×10^6
161	101.6	0.80	1	1.03	0.84×10^6
162	101.6	0.80	2	2.05	0.84×10^6
163	101.6	0.82	0	0	0.84×10^6
164	101.6	0.82	1	1.02	0.84×10^6
165	101.6	0.82	2	2.05	0.84×10^6
166	101.6	0.50	0	0	0.42×10^6
167	101.6	0.50	1	1.01	0.42×10^6
168	101.6	0.50	2	2.02	0.42×10^6
169	101.6	0.50	3	3.03	0.42×10^6
170	101.6	0.50	4	4.03	0.42×10^6
171	101.6	0.50	5	5.04	0.42×10^6
172	101.6	0.50	6	6.04	0.42×10^6
173	101.6	0.50	7	7.05	0.42×10^6
174	101.6	0.50	8	8.06	0.42×10^6
175	101.6	0.70	0	0	0.42×10^6
176	101.6	0.70	1	1.02	0.42×10^6
177	101.6	0.70	2	2.03	0.42×10^6
178	101.6	0.70	3	3.05	0.42×10^6
179	101.6	0.70	4	4.06	0.42×10^6
180	101.6	0.70	5	5.07	0.42×10^6
181	101.6	0.80	0	0	0.42×10^6
182	101.6	0.80	1	1.01	0.42×10^6
183	101.6	0.80	2	2.02	0.42×10^6
184	50.8	0.50	0	0.02	0.42×10^6
185	50.8	0.50	0.5	0.61	0.42×10^6
186	50.8	0.50	1.0	1.25	0.42×10^6
187	50.8	0.50	1.5	1.89	0.42×10^6
188	50.8	0.50	2.0	2.52	0.42×10^6
189	50.8	0.60	0	0.02	0.42×10^6
190	50.8	0.60	0.5	0.72	0.42×10^6
191	50.8	0.60	1.0	1.36	0.42×10^6
192	50.8	0.60	1.5	2.05	0.42×10^6
193	50.8	0.60	2.0	2.75	0.42×10^6
194	50.8	0.65	0	0.03	0.42×10^6
195	50.8	0.65	0.5	0.72	0.42×10^6
196	50.8	0.65	1.0	1.45	0.42×10^6
197	50.8	0.65	1.5	2.24	0.42×10^6

No.	Model Chord	M	Nominal α	Corrected α	R
198	50.8	0.675	0	-0.02	0.42×10^6
199	50.8	0.675	0.5	0.75	0.42×10^6
200	50.8	0.675	1.0	1.53	0.42×10^6
201	50.8	0.675	1.5	2.40	0.42×10^6
202	50.8	0.70	0	-0.05	0.42×10^6
203	50.8	0.70	0.5	0.80	0.42×10^6
204	50.8	0.70	1.0	1.62	0.42×10^6
205	50.8	0.72	0	-0.03	0.42×10^6
206	50.8	0.72	0.5	0.85	0.42×10^6
207	50.8	0.72	1.0	1.75	0.42×10^6
208	50.8	0.74	0	-0.04	0.42×10^6
209	50.8	0.74	0.5	0.93	0.42×10^6
210	50.8	0.74	1.0	1.93	0.42×10^6
211	50.8	0.76	0	-0.07	0.42×10^6
212	50.8	0.76	0.5	1.01	0.42×10^6
213	50.8	0.76	1.0	1.95	0.42×10^6
214	50.8	0.78	0	-0.10	0.42×10^6
215	50.8	0.78	0.5	1.11	0.42×10^6
216	50.8	0.78	1.0	1.80	0.42×10^6
217	50.8	0.80	0	-0.05	0.42×10^6
218	50.8	0.80	0.5	0.93	0.42×10^6
219	50.8	0.80	1.0	1.54	0.42×10^6
220	50.8	0.82	0	-0.01	0.42×10^6
221	50.8	0.82	0.5	0.45	0.42×10^6
222	50.8	0.82	1.0	1.21	0.42×10^6

AD-A055 004

AERONAUTICAL RESEARCH LABS MELBOURNE (AUSTRALIA)
TRANSONIC WIND TUNNEL TESTS ON A NACA 0012 AEROFOIL, (U)
APR 77 N POLLOCK, B D FAIRLIE

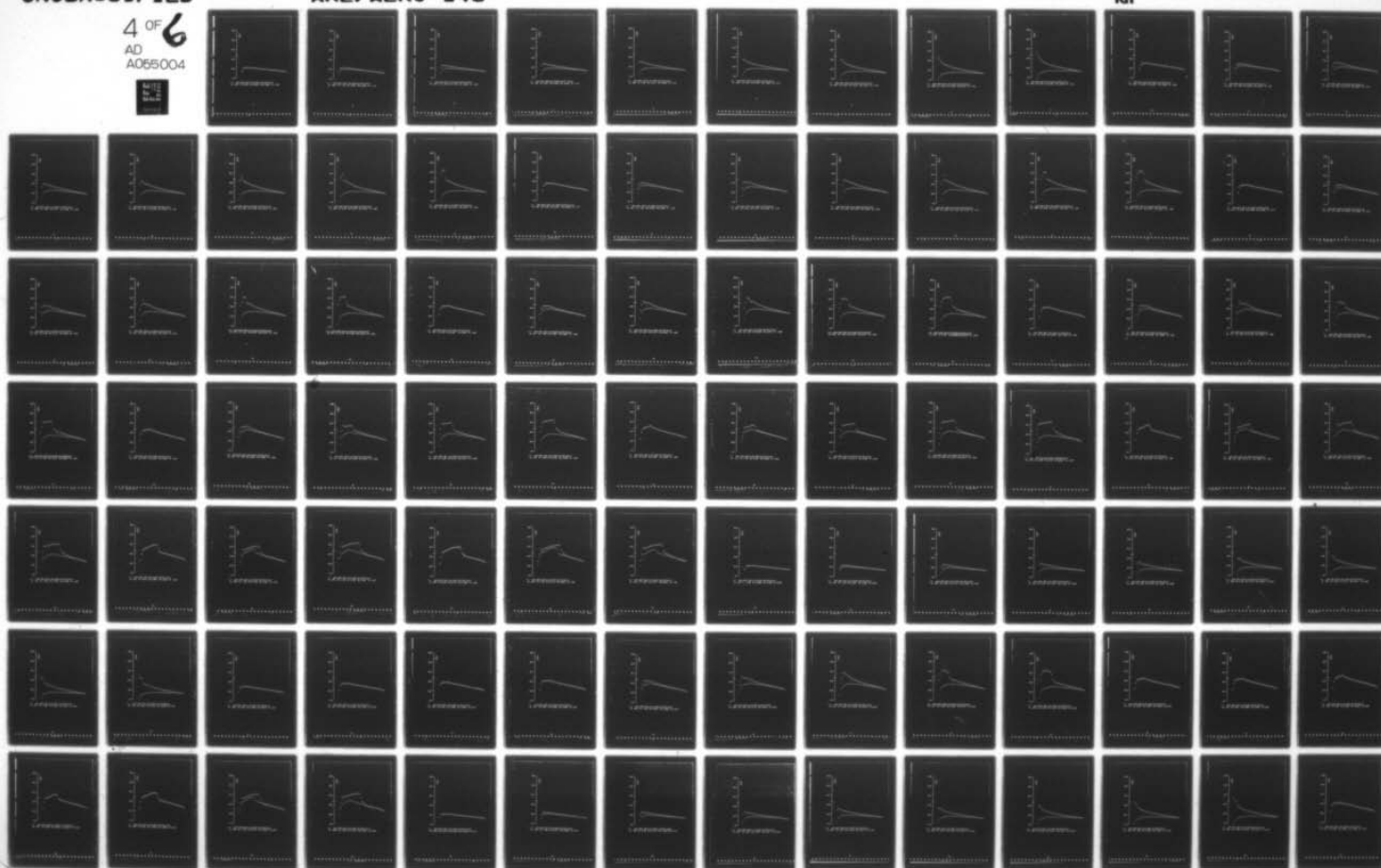
F/G 20/4

UNCLASSIFIED

ARL/AERO-148

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4 OF 6
AD
A055004

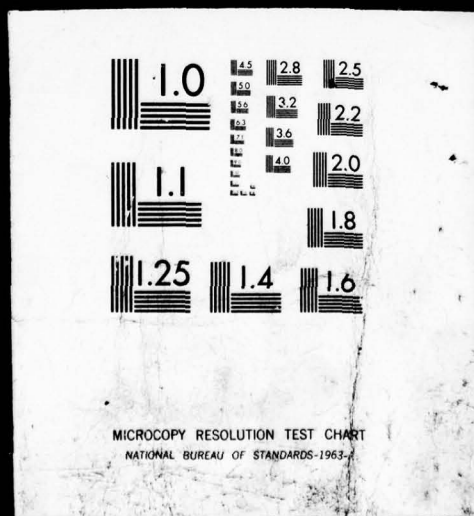


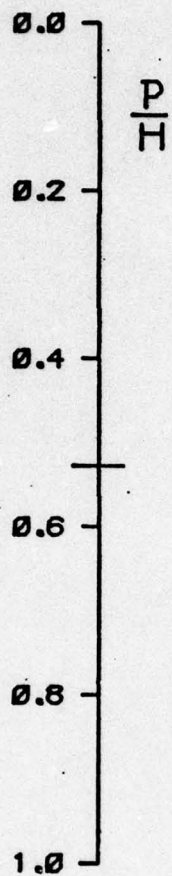
SIFTED

4 OF 6

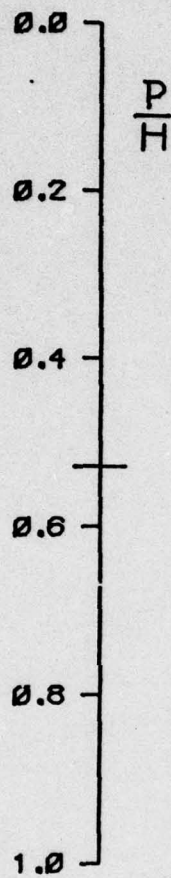
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A055004

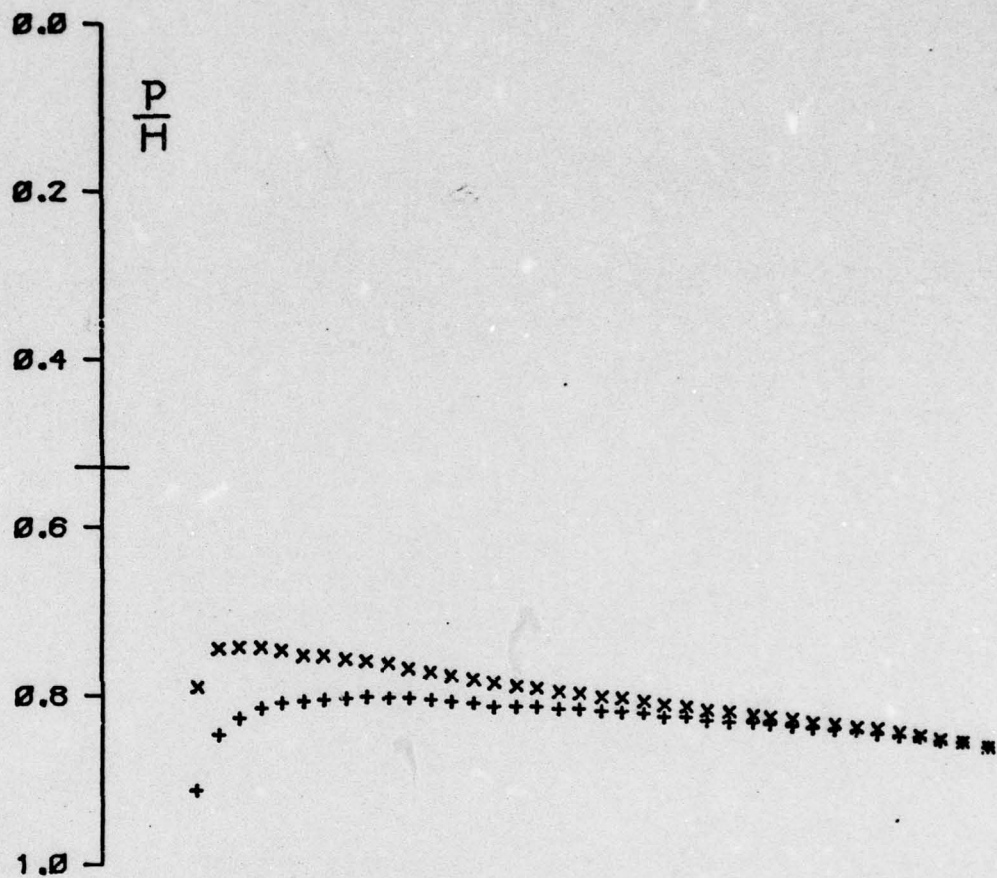




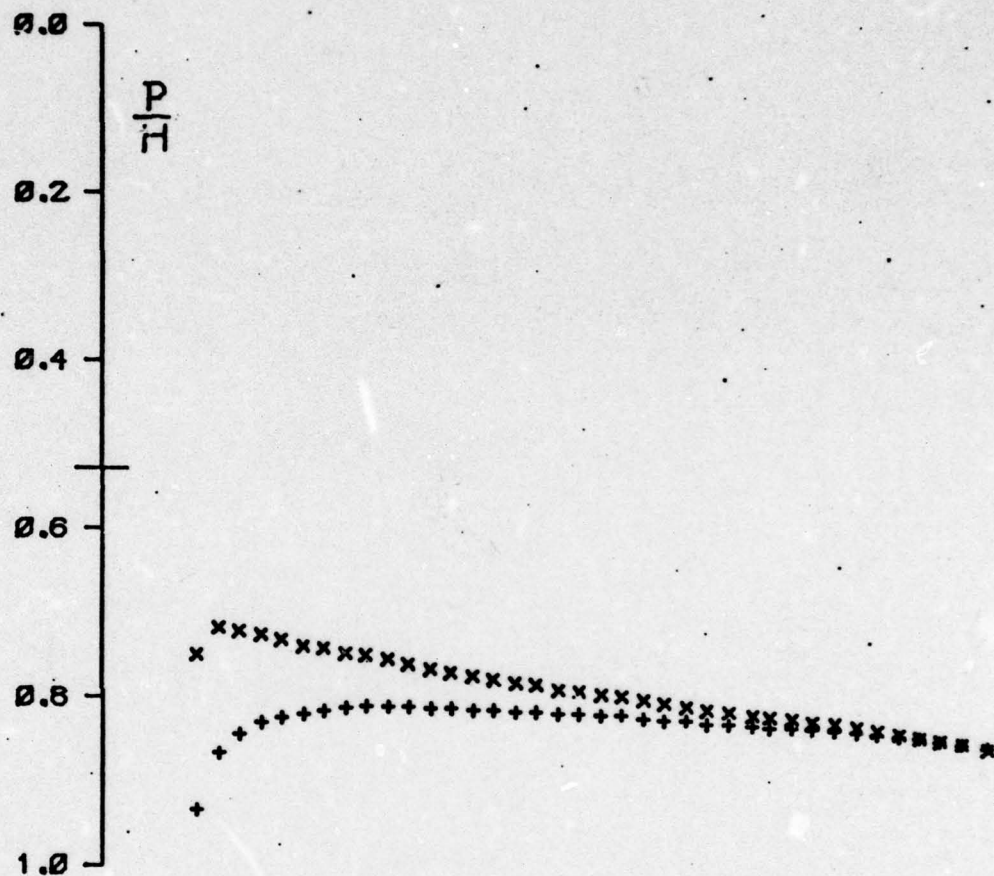
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .501$ $AL = 0.00$ $CN = -0.003$ $CN = 0.000$ $R = 1.730$



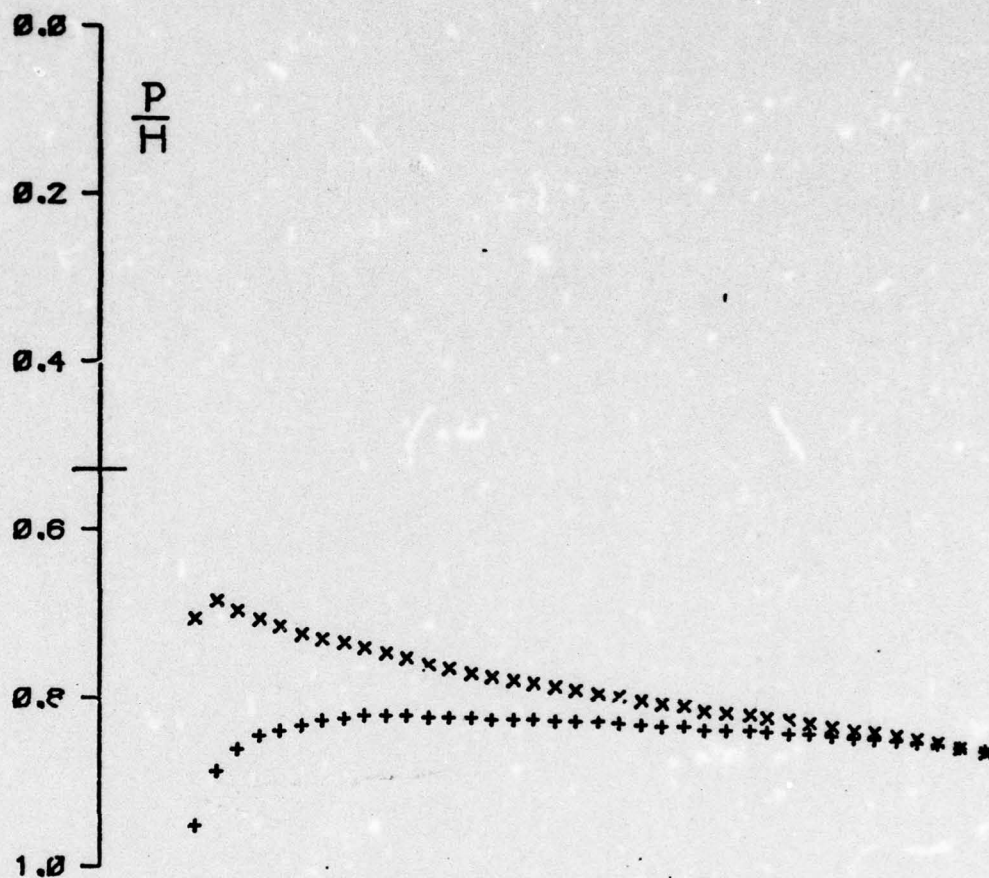
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .500$ $AL = 1.00$ $CN = 0.084$ $CM = 0.002$ $R = 1.719$



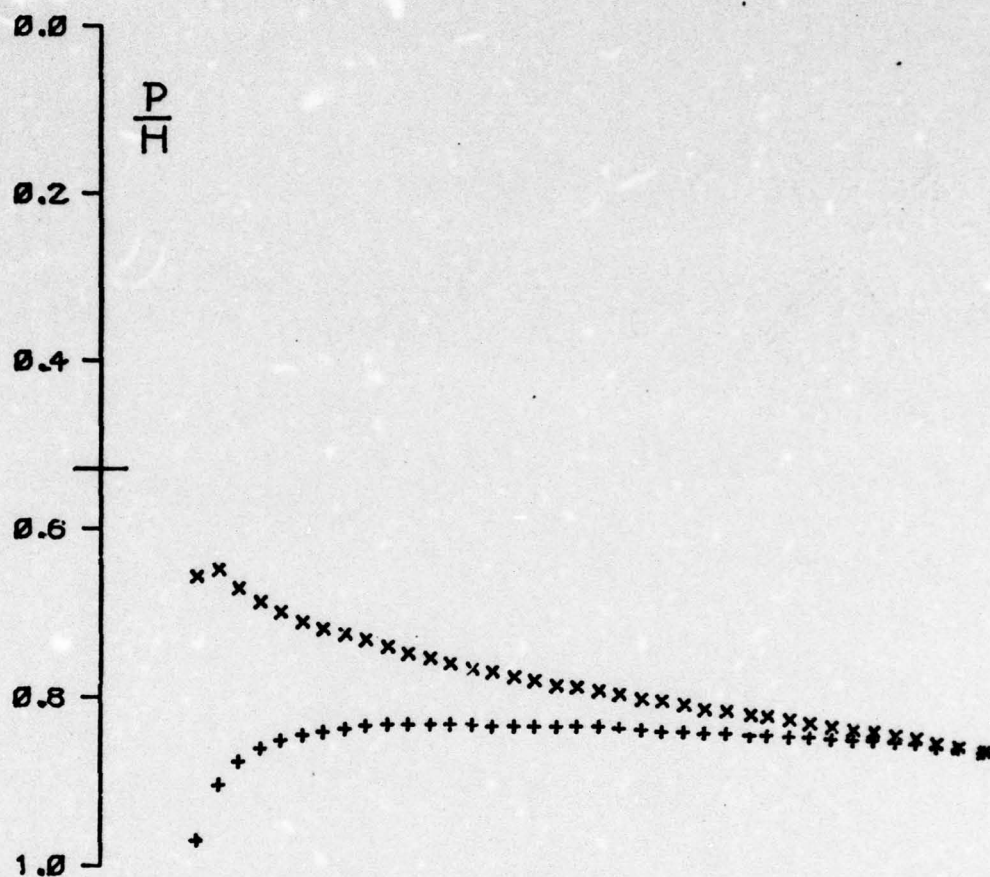
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .502$ $AL = 2.00$ $CN = 0.170$ $CM = 0.003$ $R = 1.722$



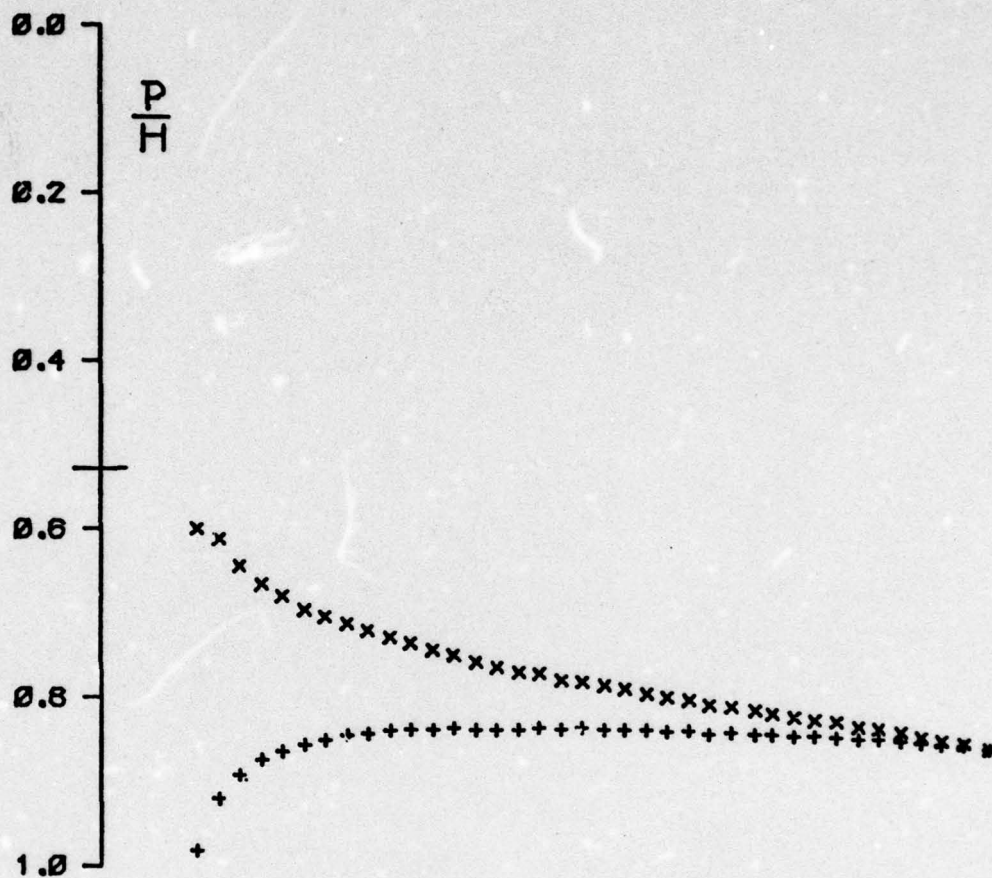
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .499$ $AL = 3.00$ $CN = 0.258$ $CN = 0.004$ $R = 1.715$



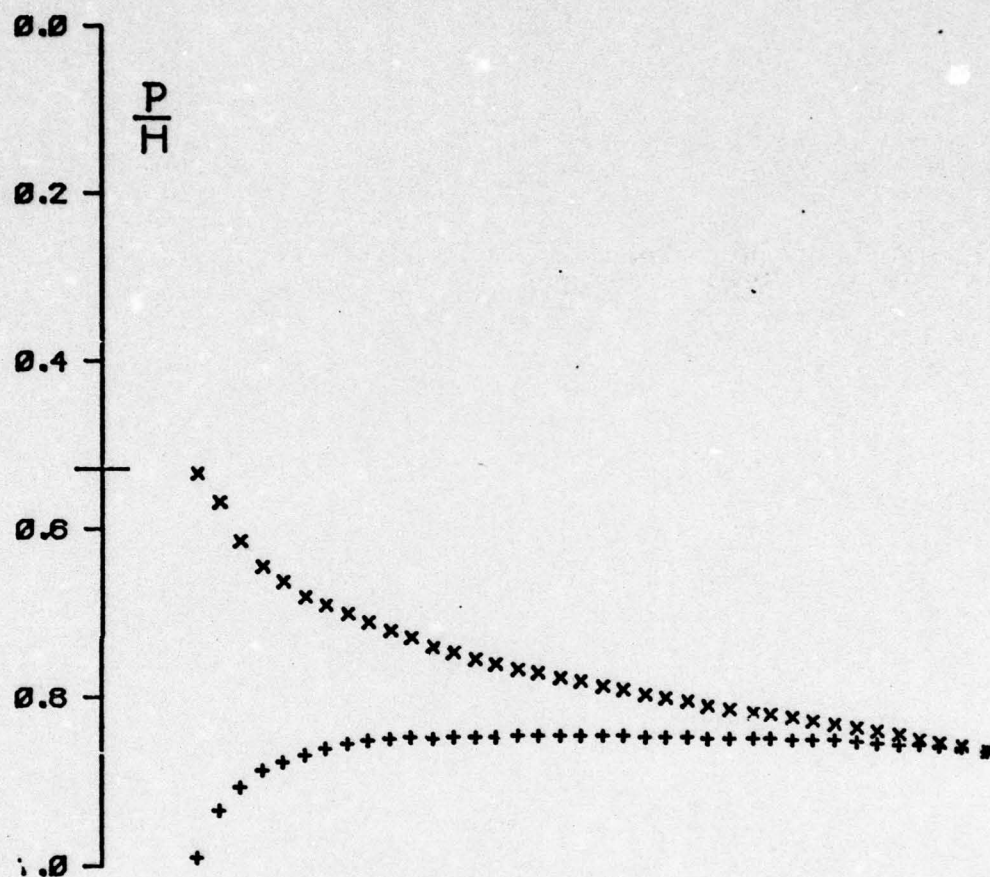
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .500$ $AL = 4.00$ $CN = 0.345$ $CM = 0.003$ $R = 1.727$



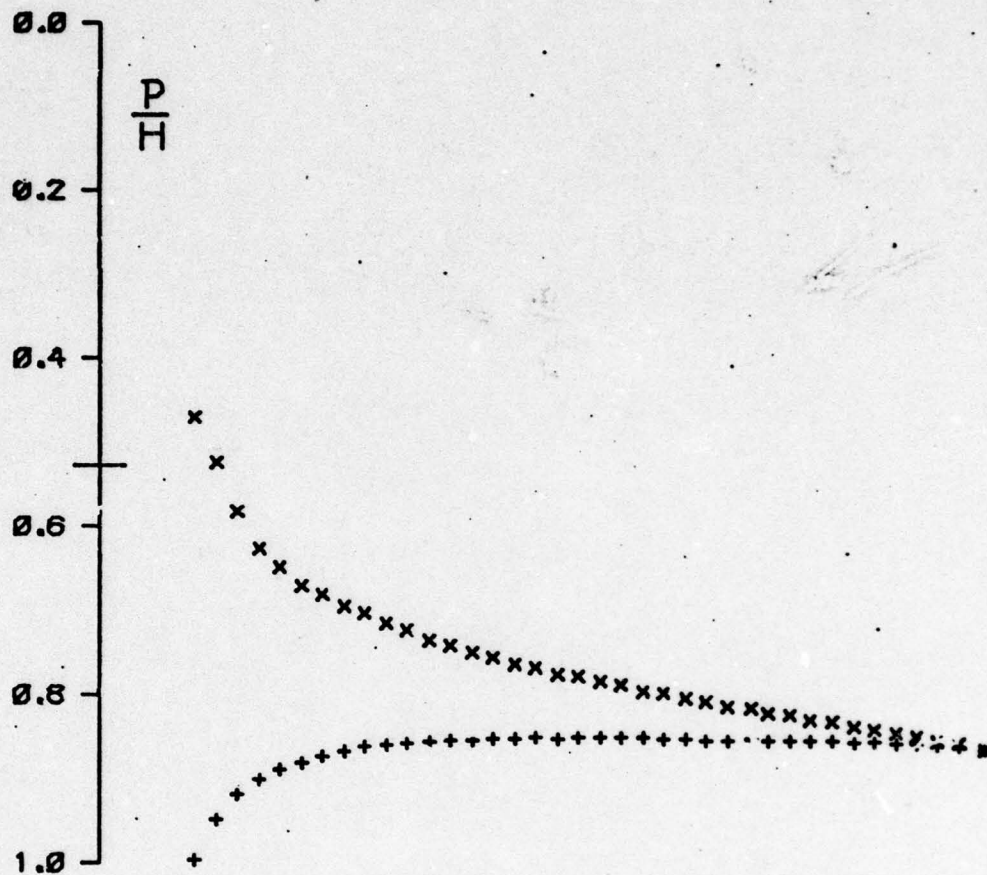
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .499$ $AL = 5.00$ $CN = 0.432$ $CM = 2.008$ $R = 1.719$



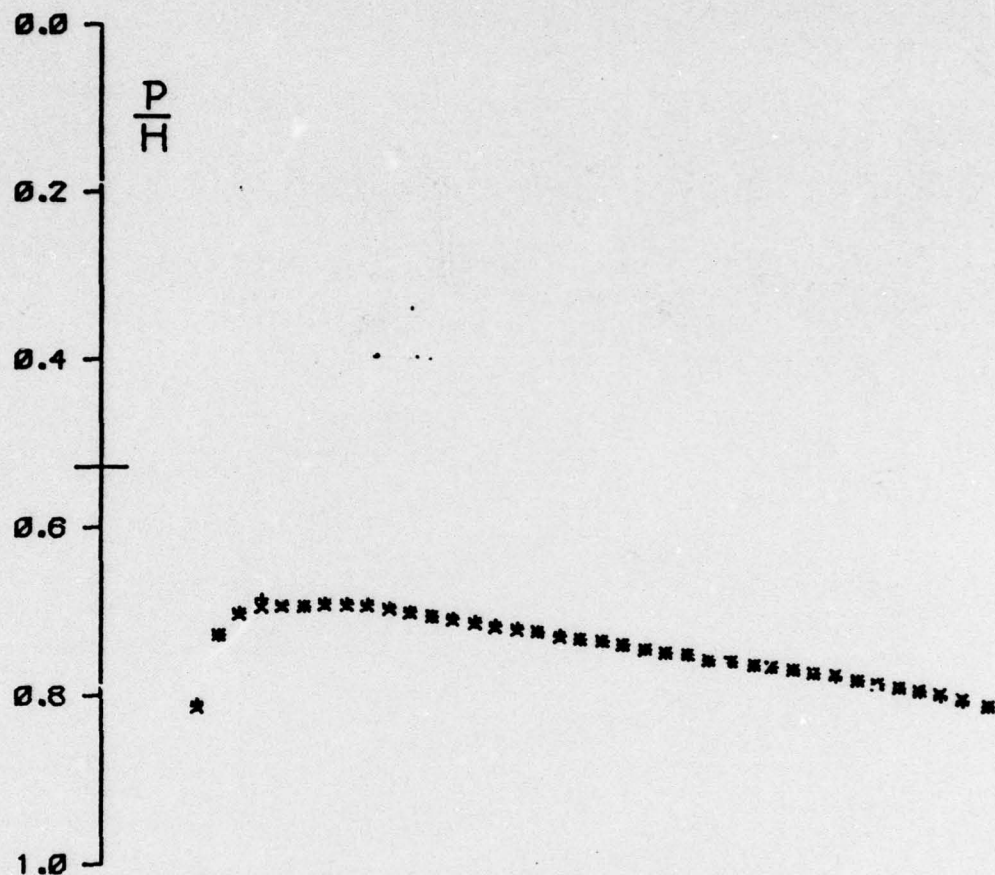
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .499 AL= 6.00 CN= 0.518 CM= 0.010 R= 1.719



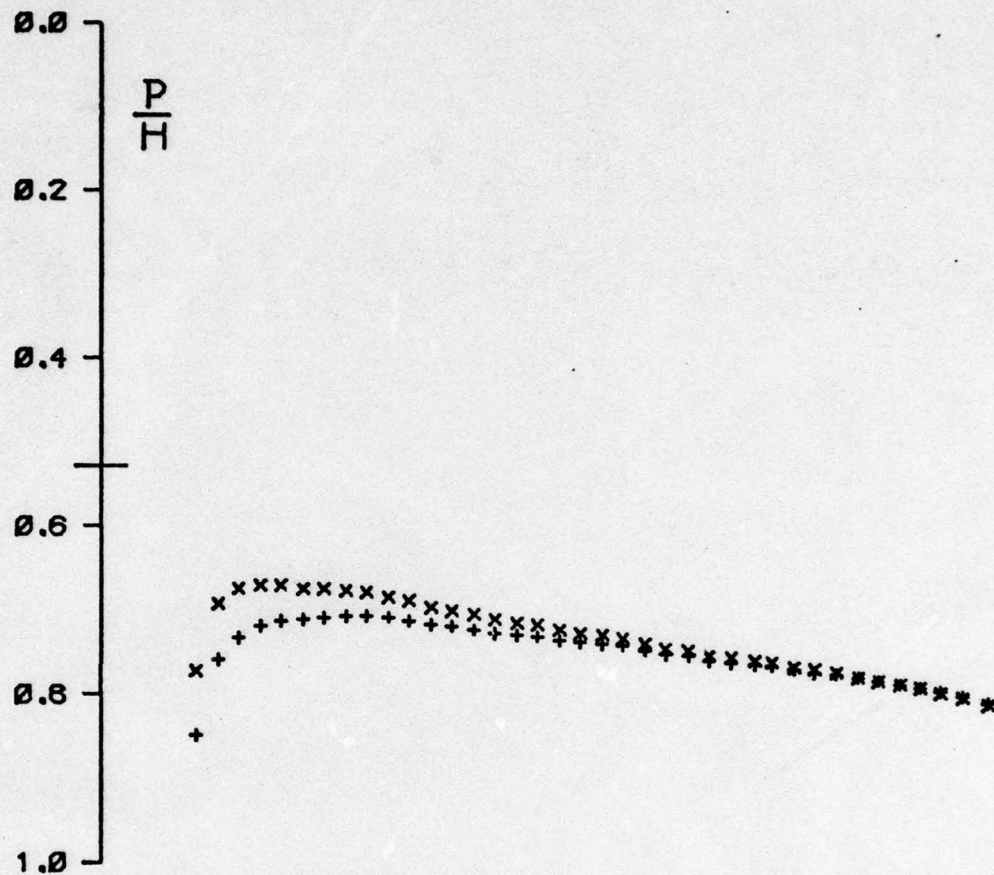
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .500 AL= 7.00 CN= 0.605 CM= 0.012 R= 1.724



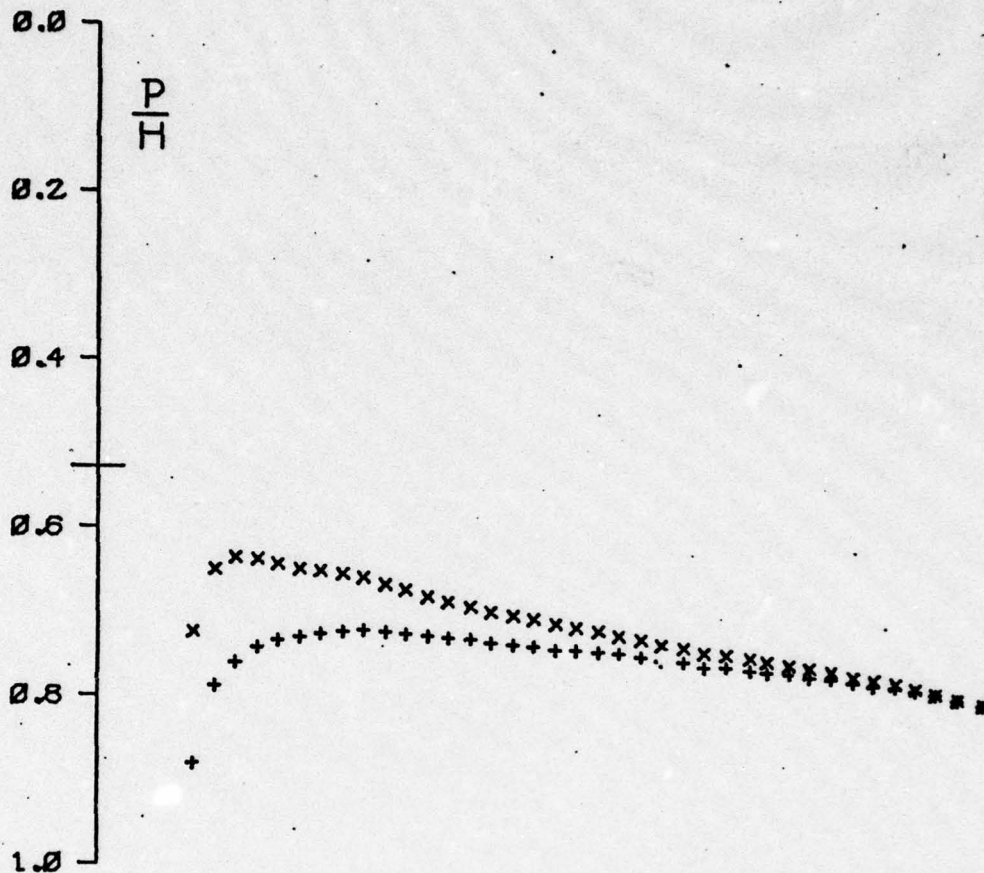
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .499 AL= 8.00 CN= 0.687 CM= 0.015 R= 1.715



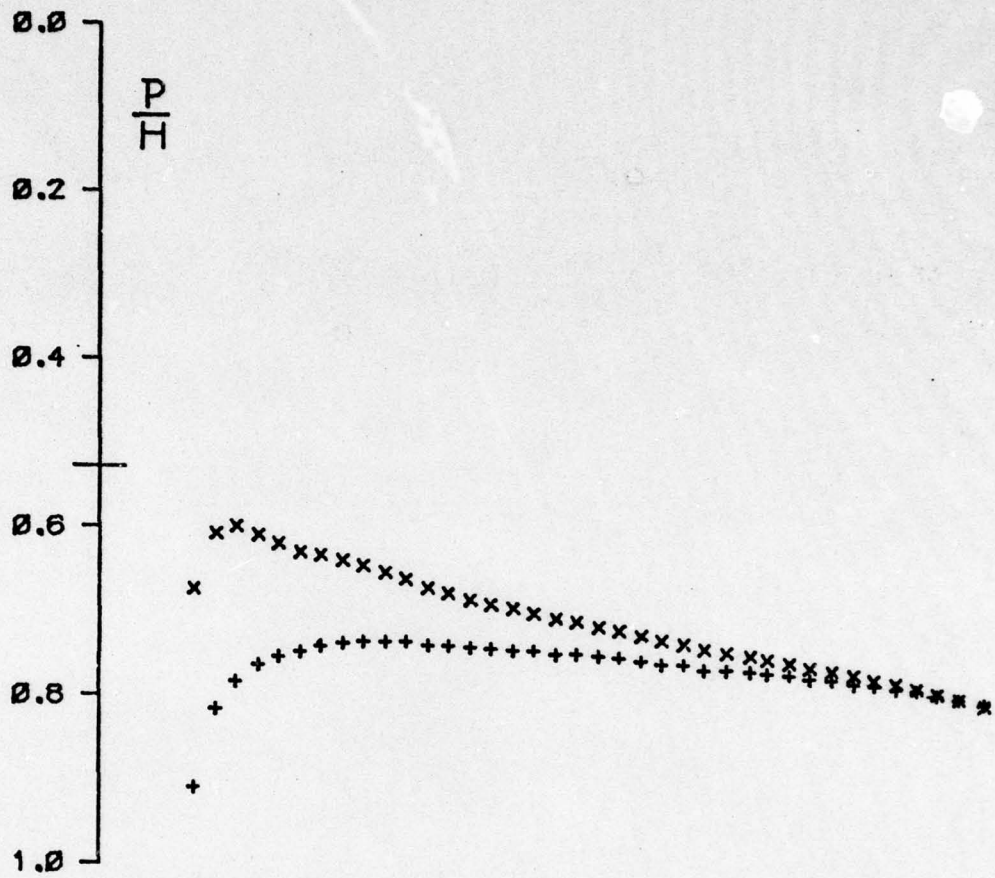
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .599 AL= 0.00 CN=-0.001 CM= 0.000 R= 1.698



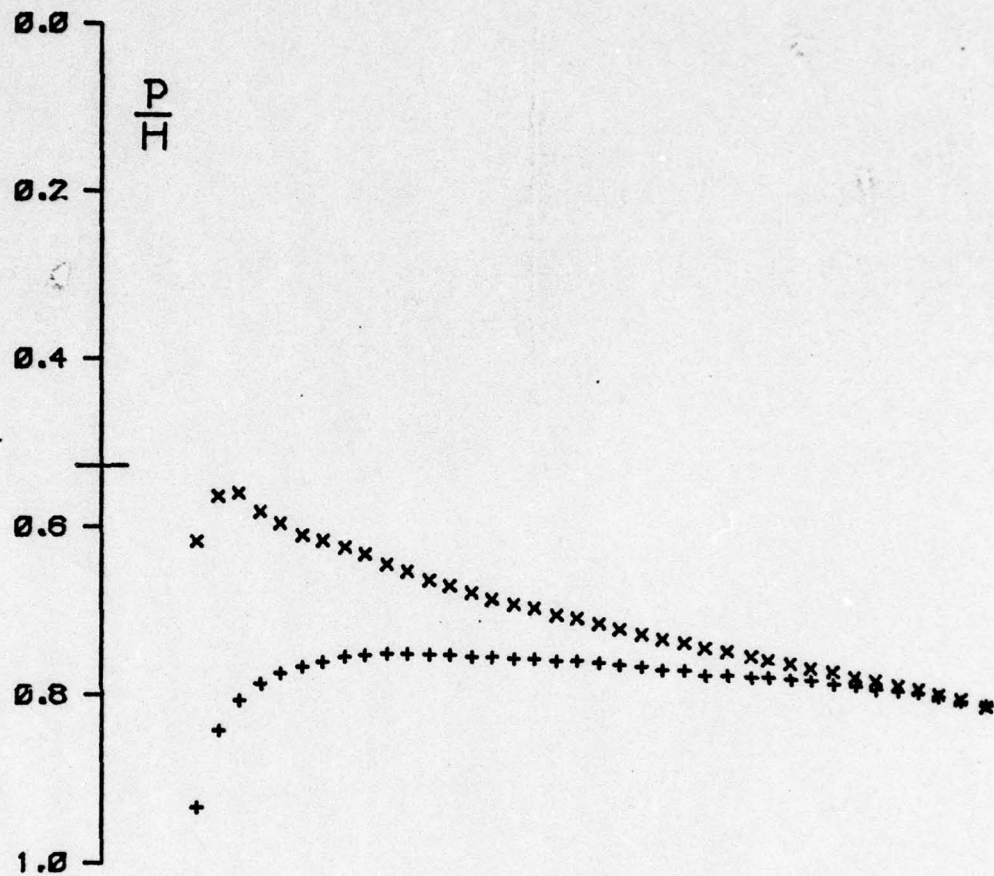
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .599$ $AL = 1.00$ $CN = 0.088$ $CM = 0.003$ $R = 1.697$



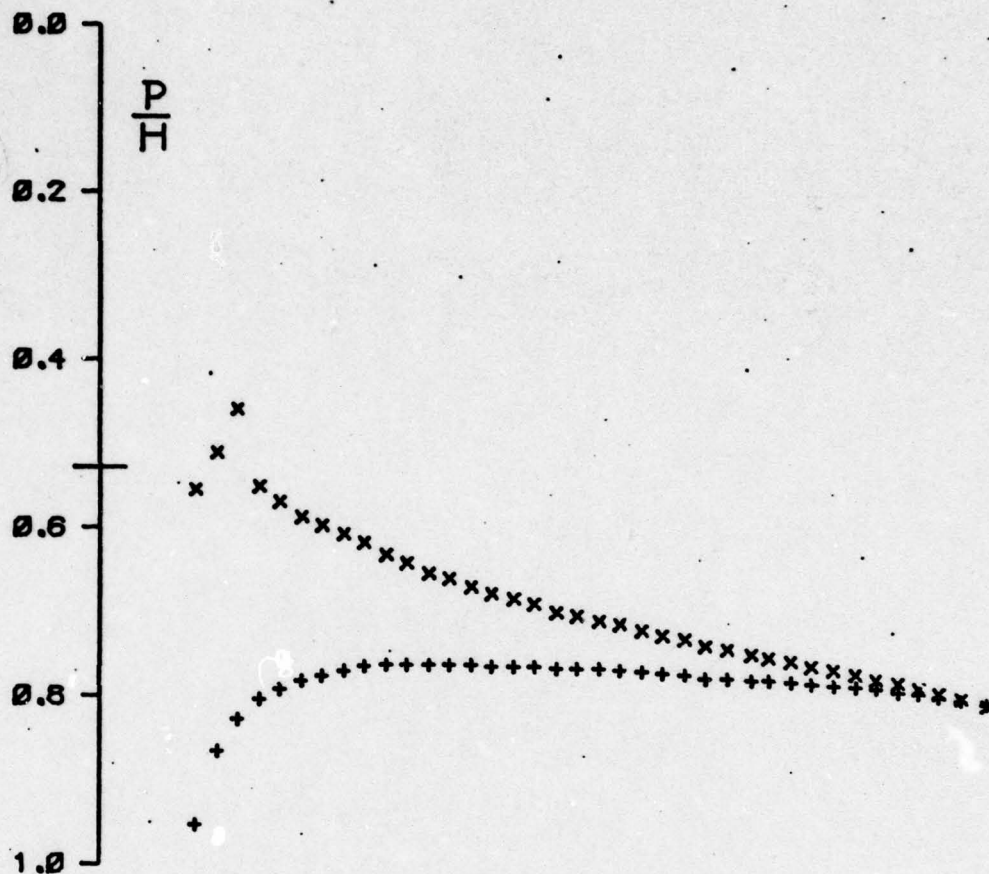
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .601 AL= 2.00 CN= 0.179 CM= 0.004 R= 1.710



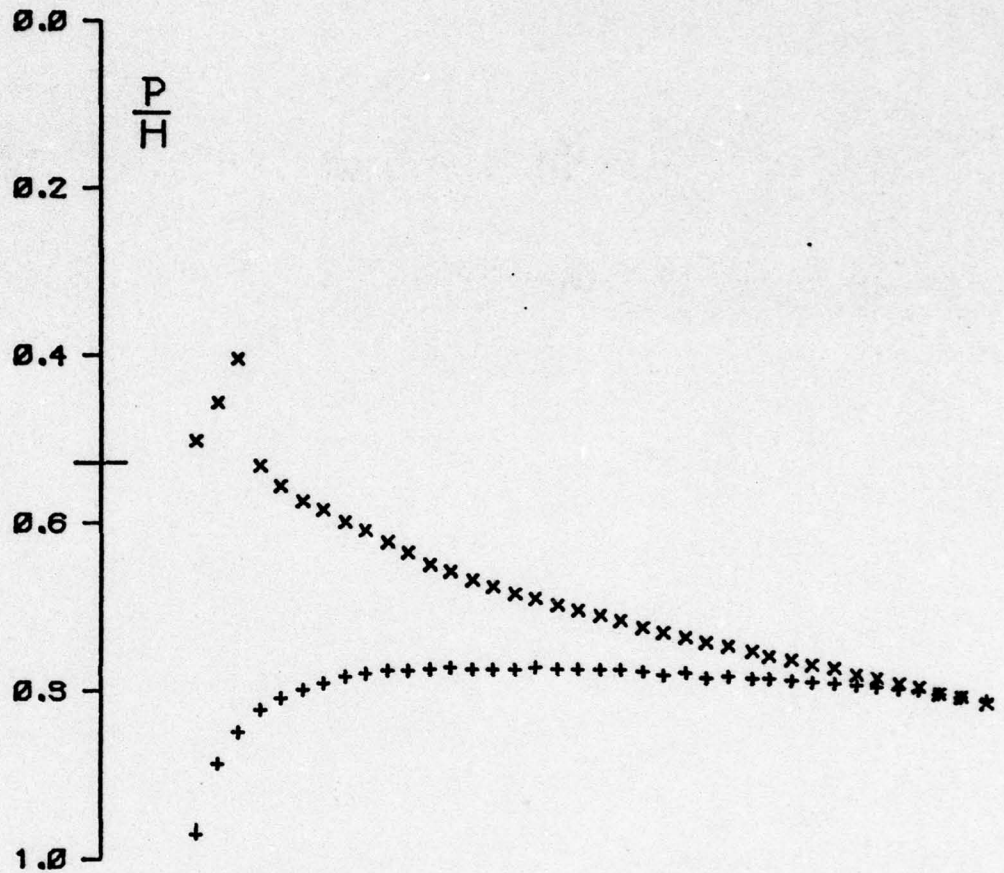
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .600 AL= 3.00 CN= 0.271 CM= 0.006 R= 1.699



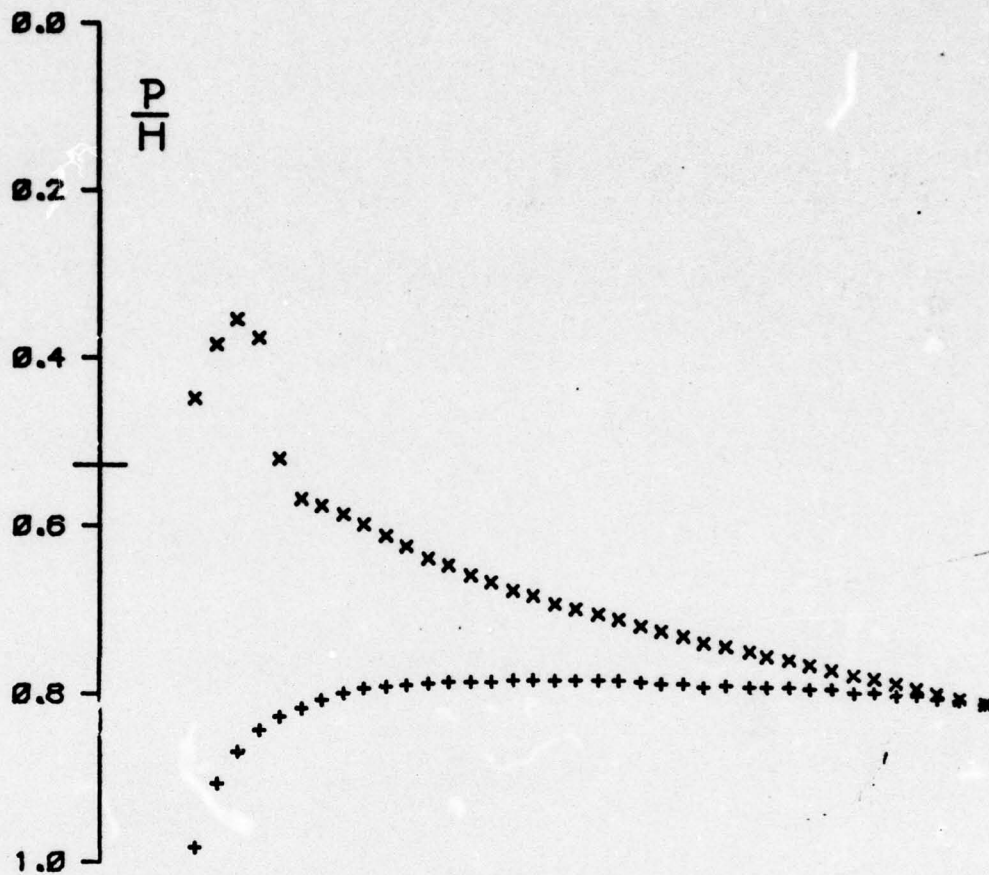
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .599$ $AL = 4.00$ $CN = 0.360$ $CM = 0.009$ $R = 1.701$



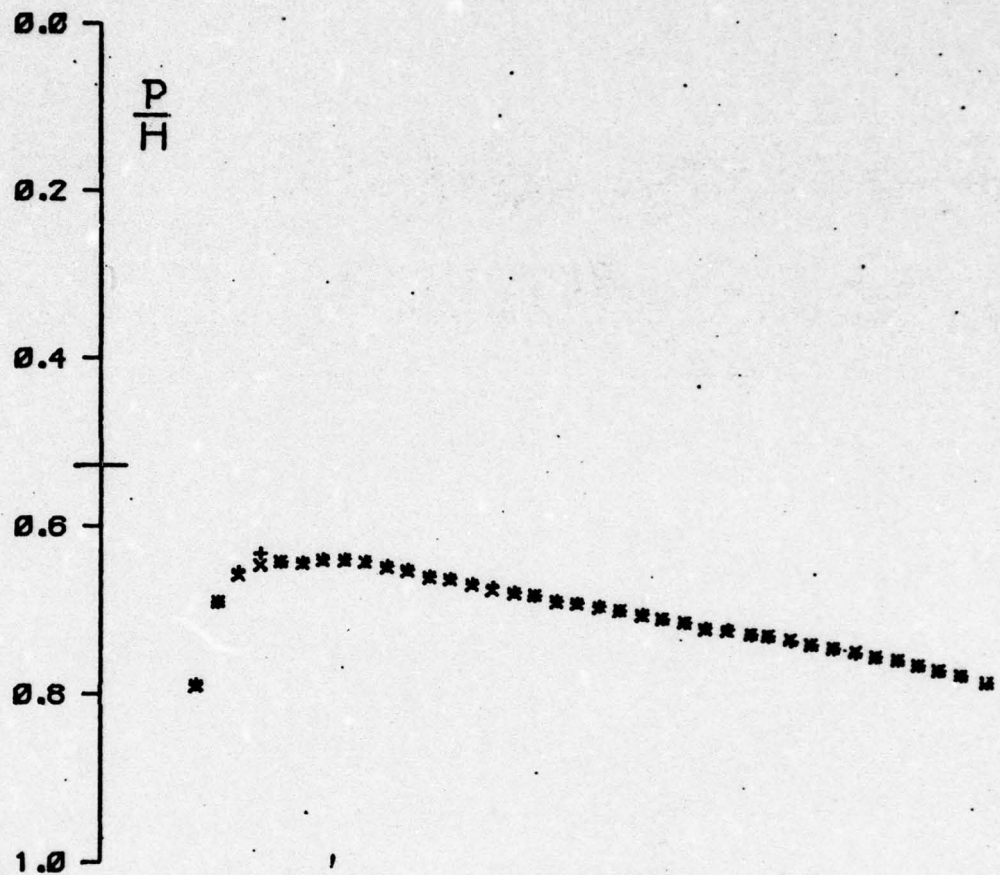
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .598 AL= 5.00 CN= 0.456 CM= 0.013 R= 1.697



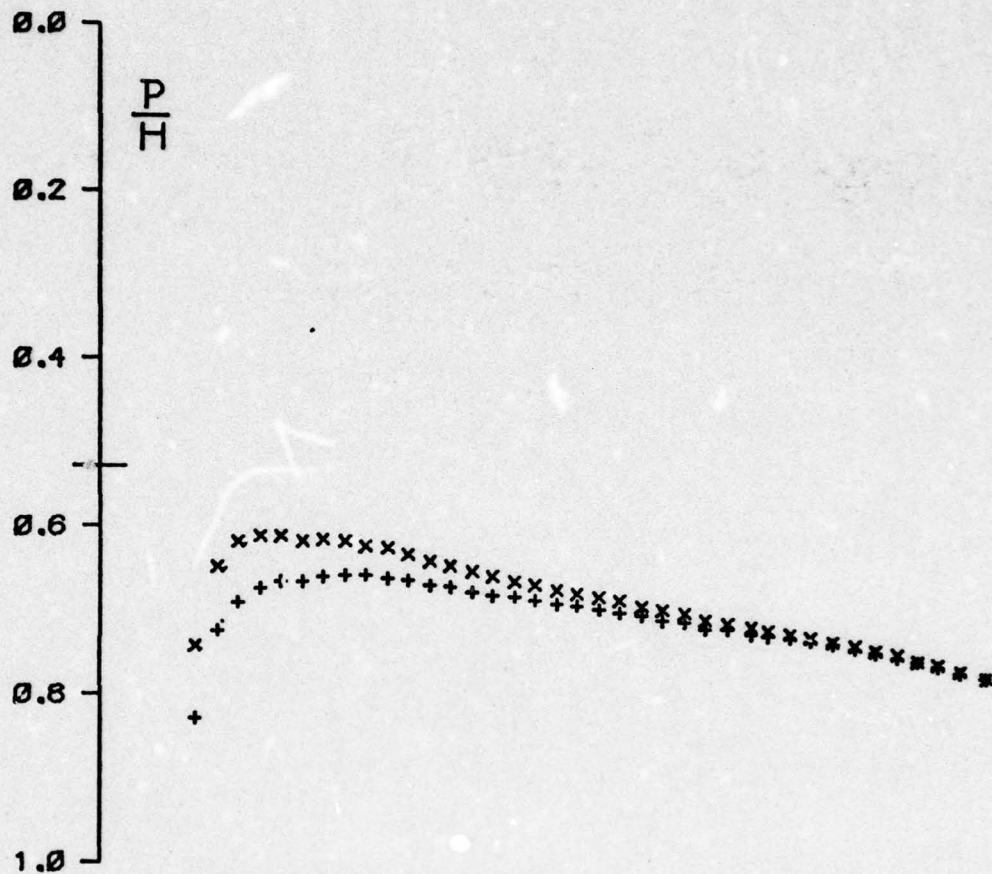
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .598$ $AL = 6.00$ $CN = 0.539$ $CM = 0.016$ $R = 1.697$



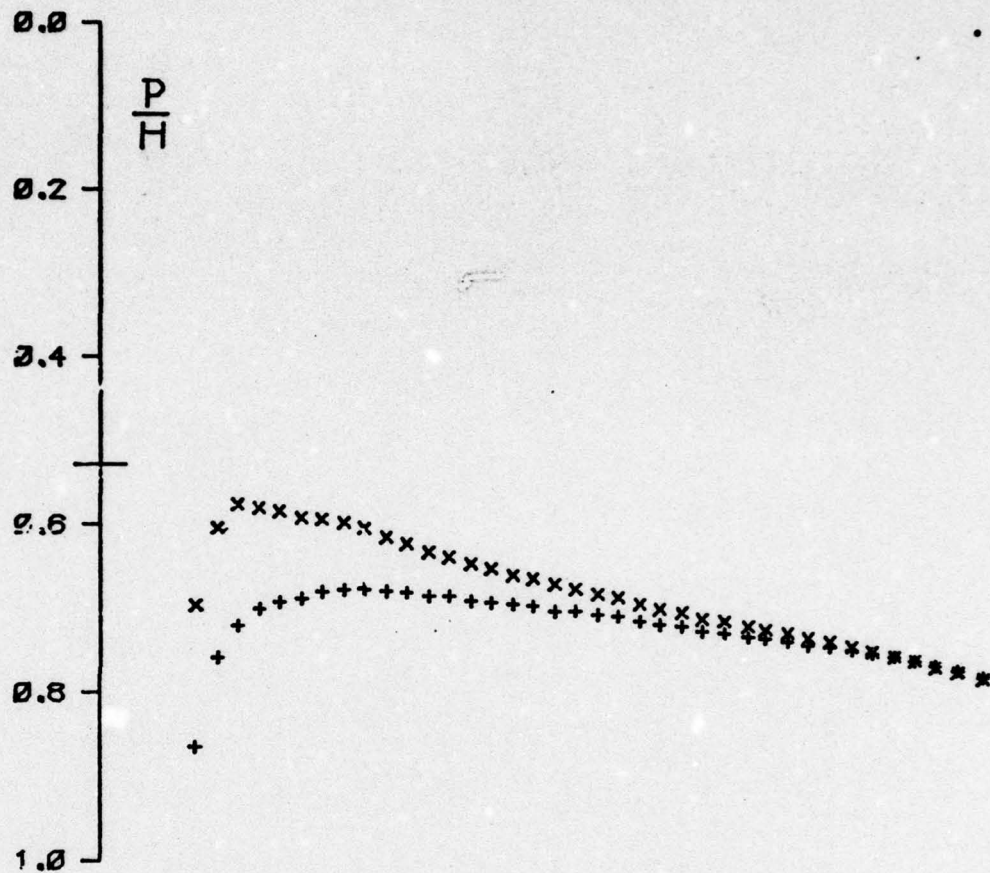
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .601$ $AL = 7.00$ $CN = 0.631$ $CM = 0.021$ $R = 1.704$



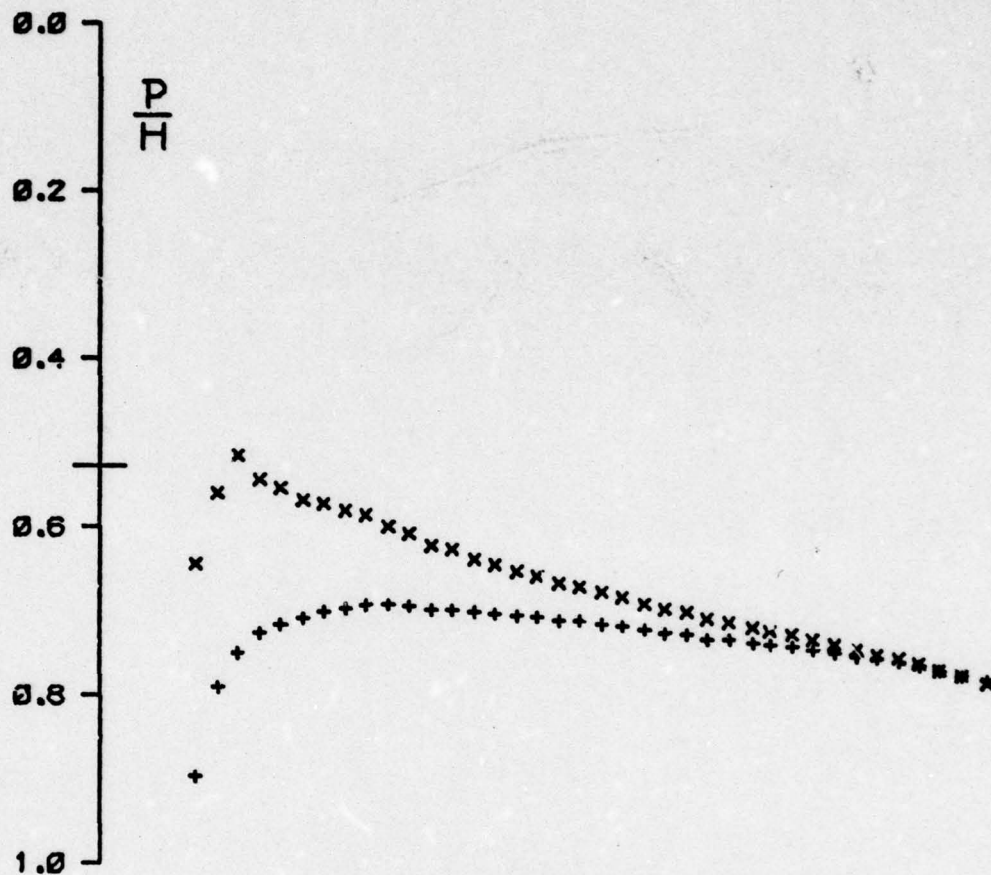
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .649 AL= 0.00 CN=-0.001 CM=-0.000 R= 1.717



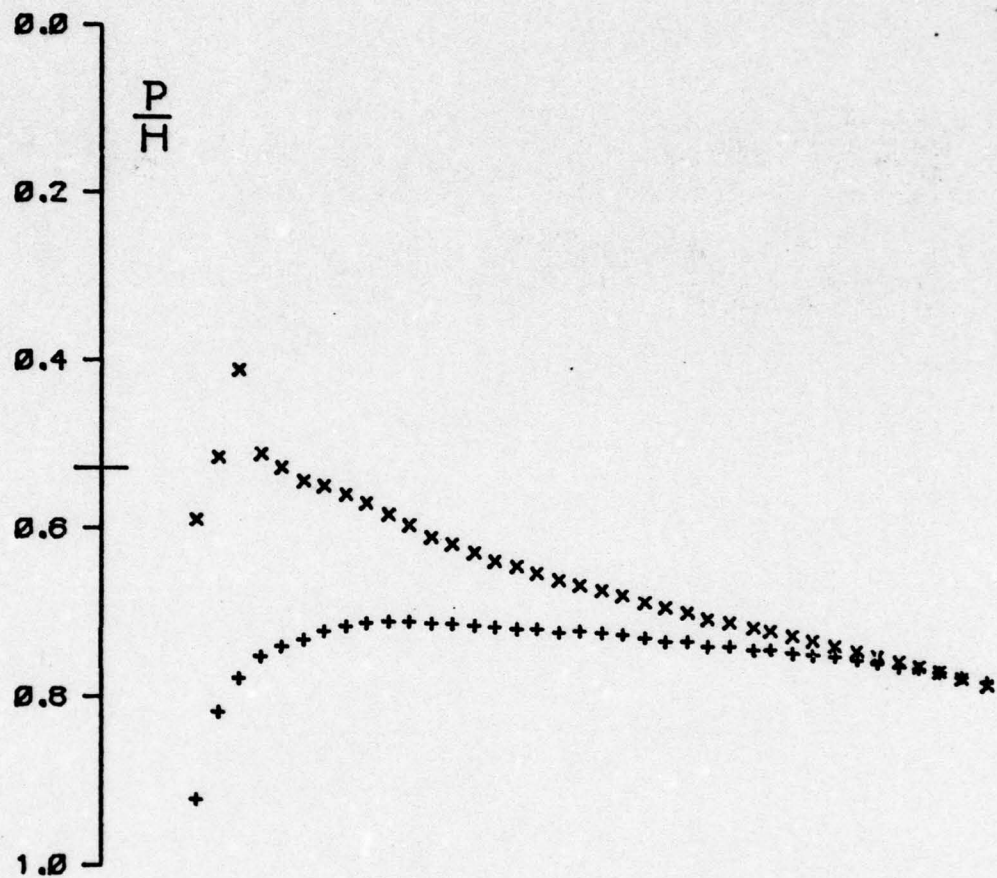
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .651$ $AL = 1.00$ $CN = 0.093$ $CM = 0.003$ $P = 1.725$



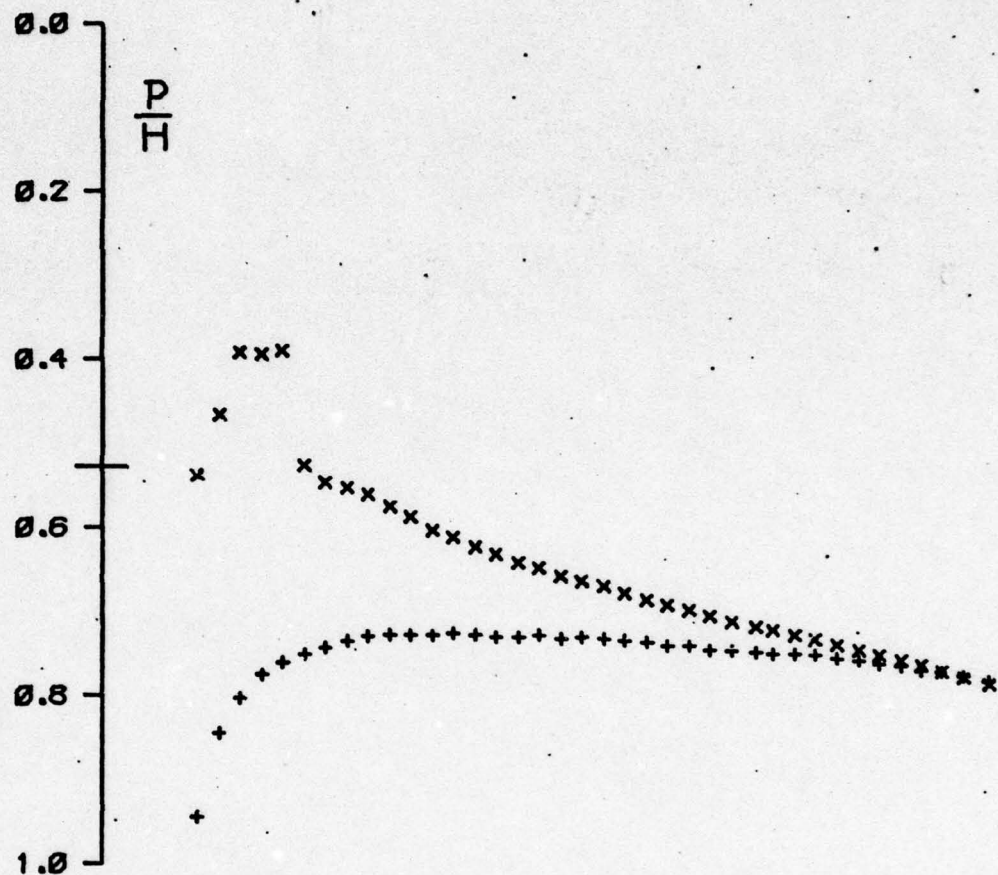
+ x NACA 0012 253.2 MM CHORD SLOTTED WALLS
 $M = .651$ $AL = 2.00$ $CN = 0.182$ $CM = 0.005$ $R = 1.725$



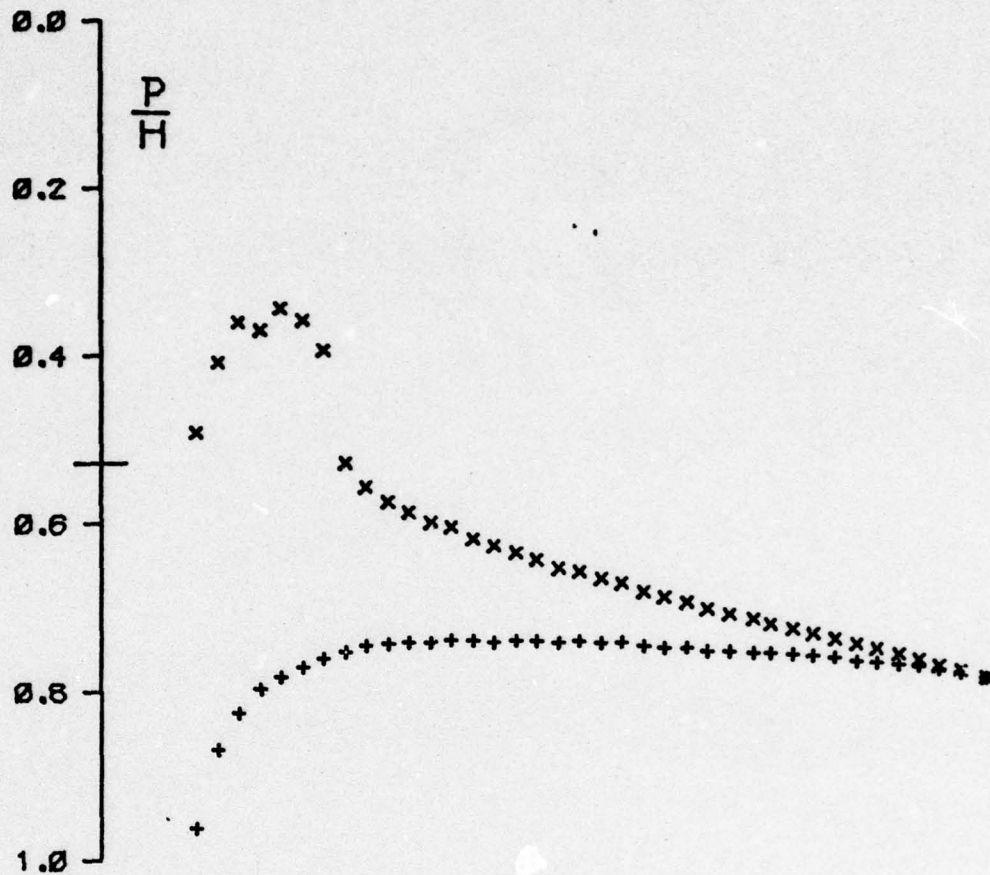
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .649$ $AL = 3.00$ $CN = 0.278$ $CM = 0.009$ $R = 1.720$



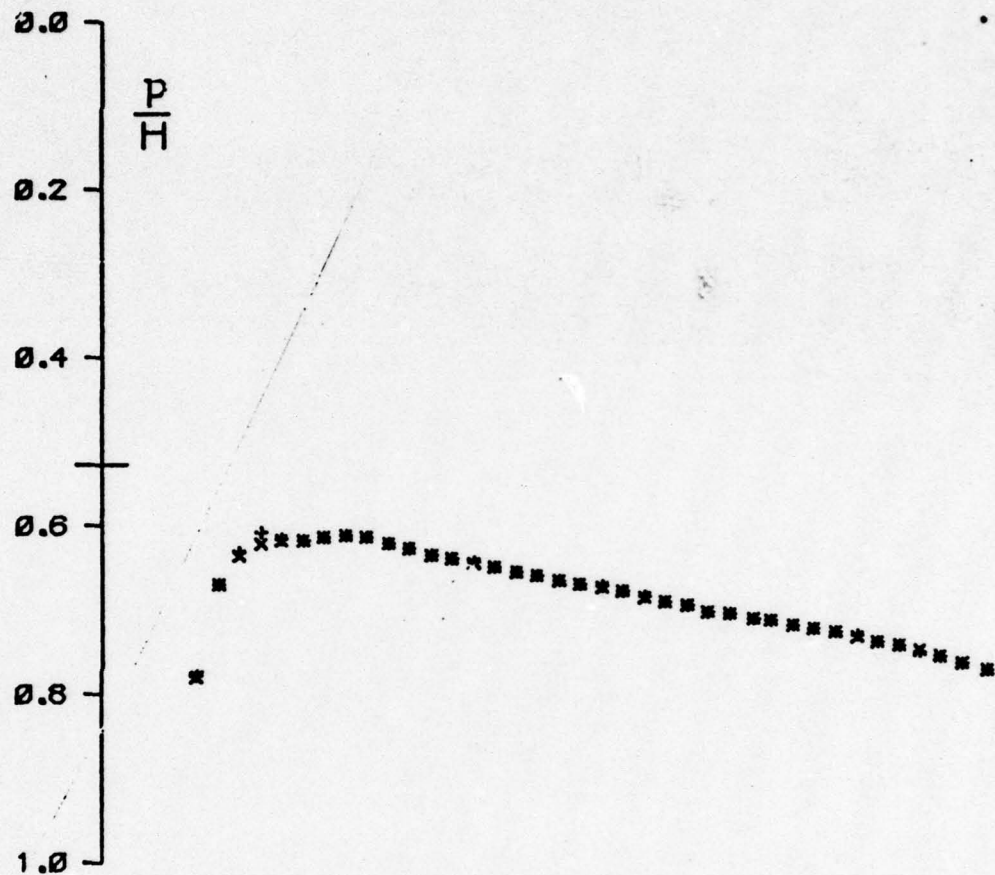
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M = .649 AL = 4.00 CN = 0.373 CM = 0.012 R = 1.717



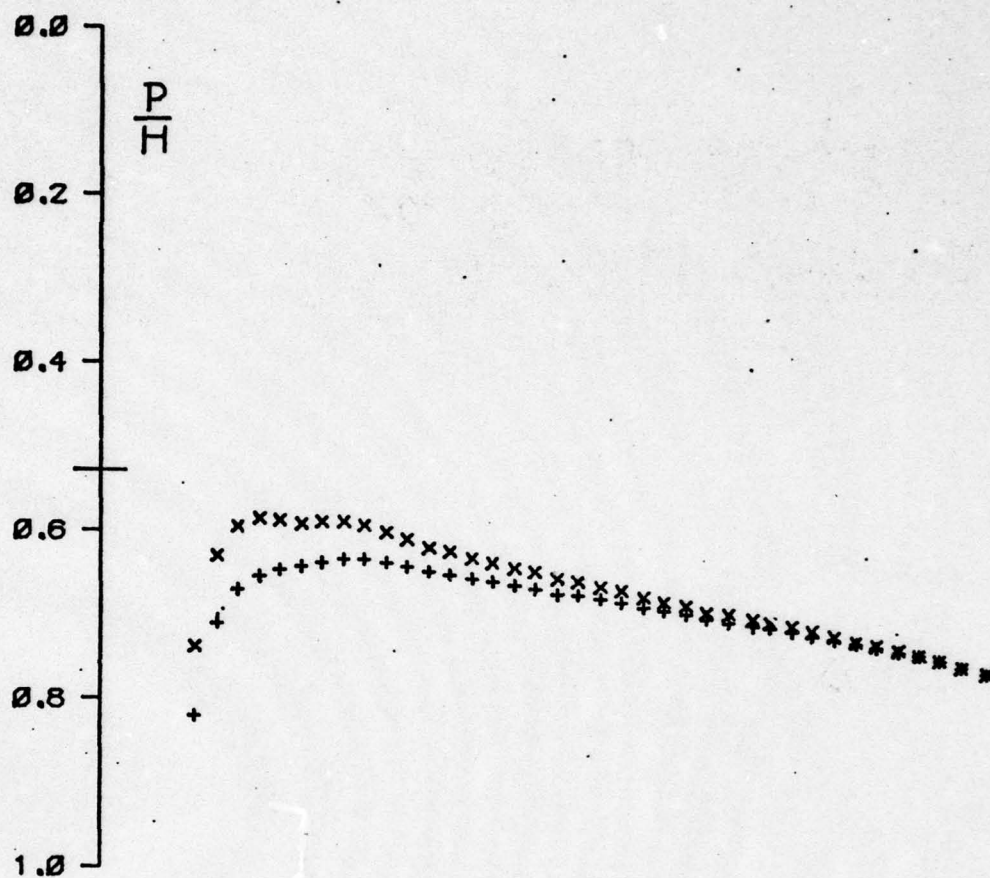
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .650$ $AL = 5.00$ $CN = 0.471$ $CM = 0.017$ $R = 1.722$



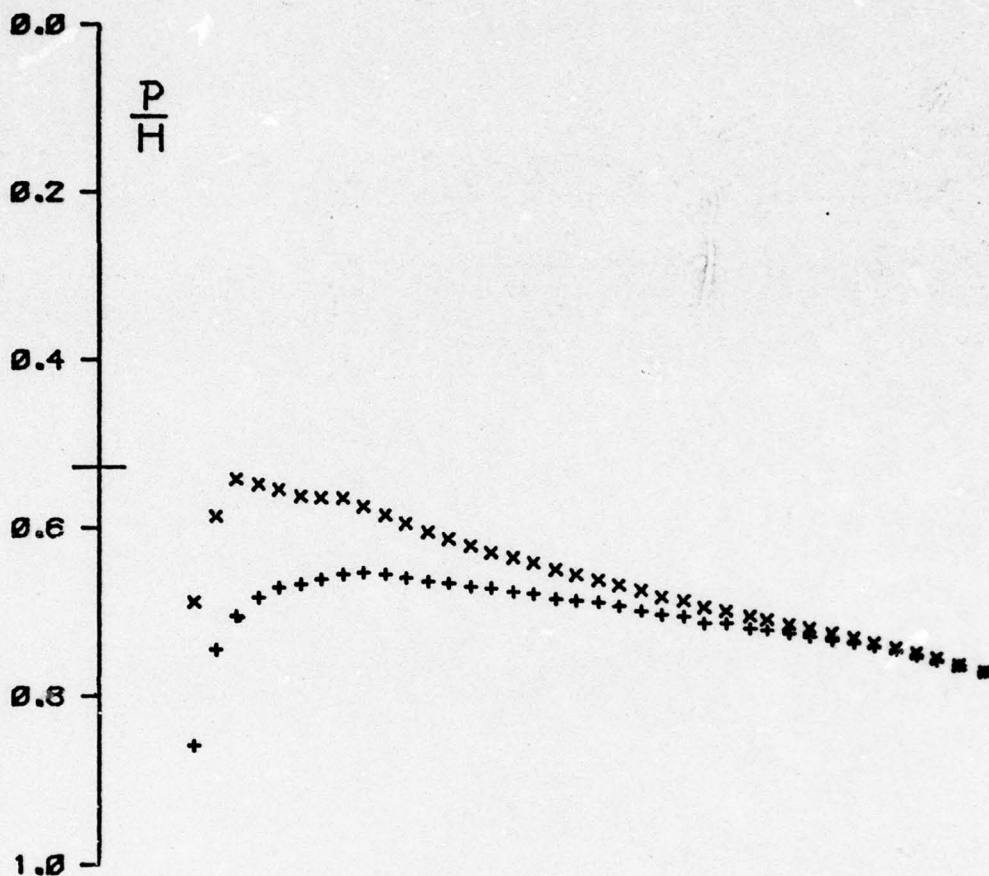
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .652 AL= 6.00 CN= 0.573 CM= 0.021 R= .728



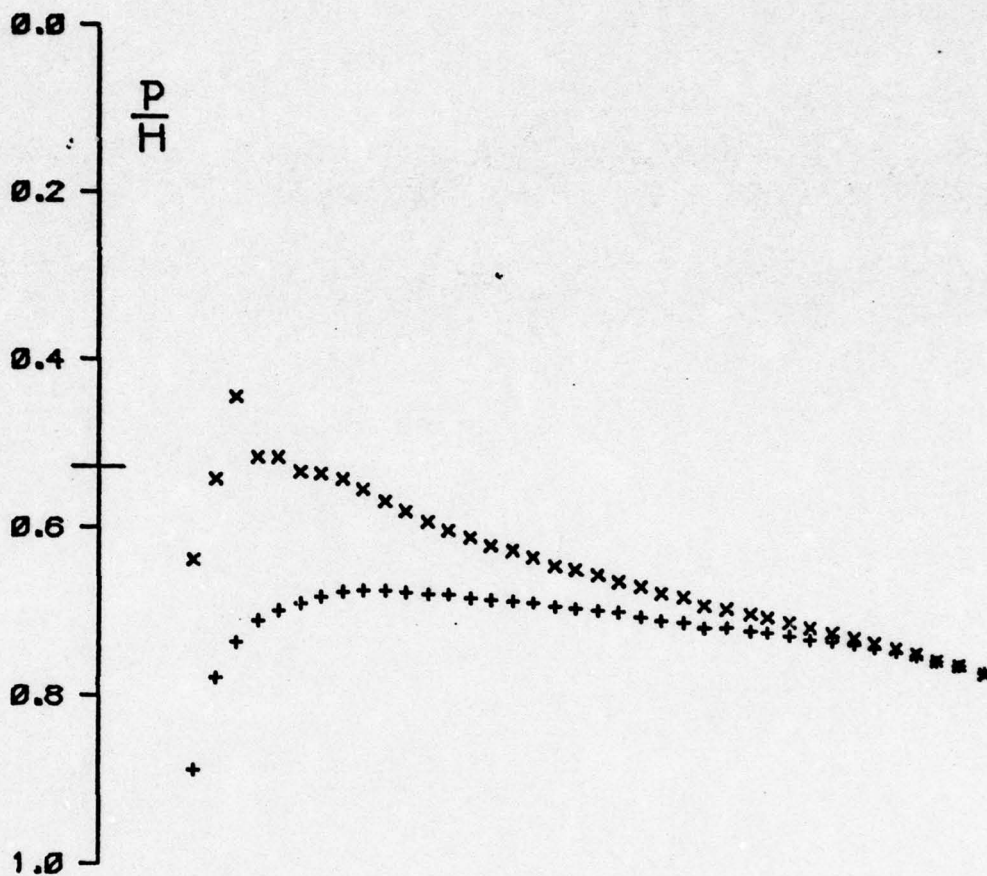
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .675$ $AL = 0.00$ $CN = -0.006$ $CM = 0.000$ $R = 1.615$



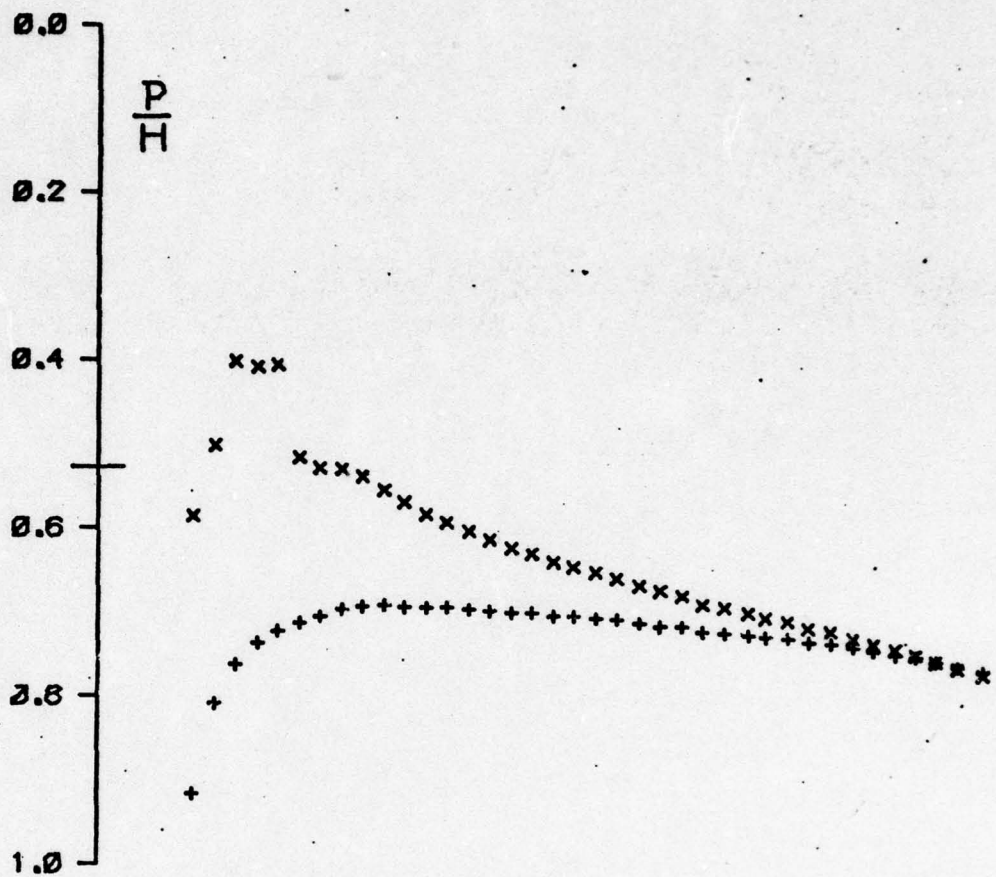
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .674 AL= 1.00 CN= 0.094 CM= 0.003 R= 1.614



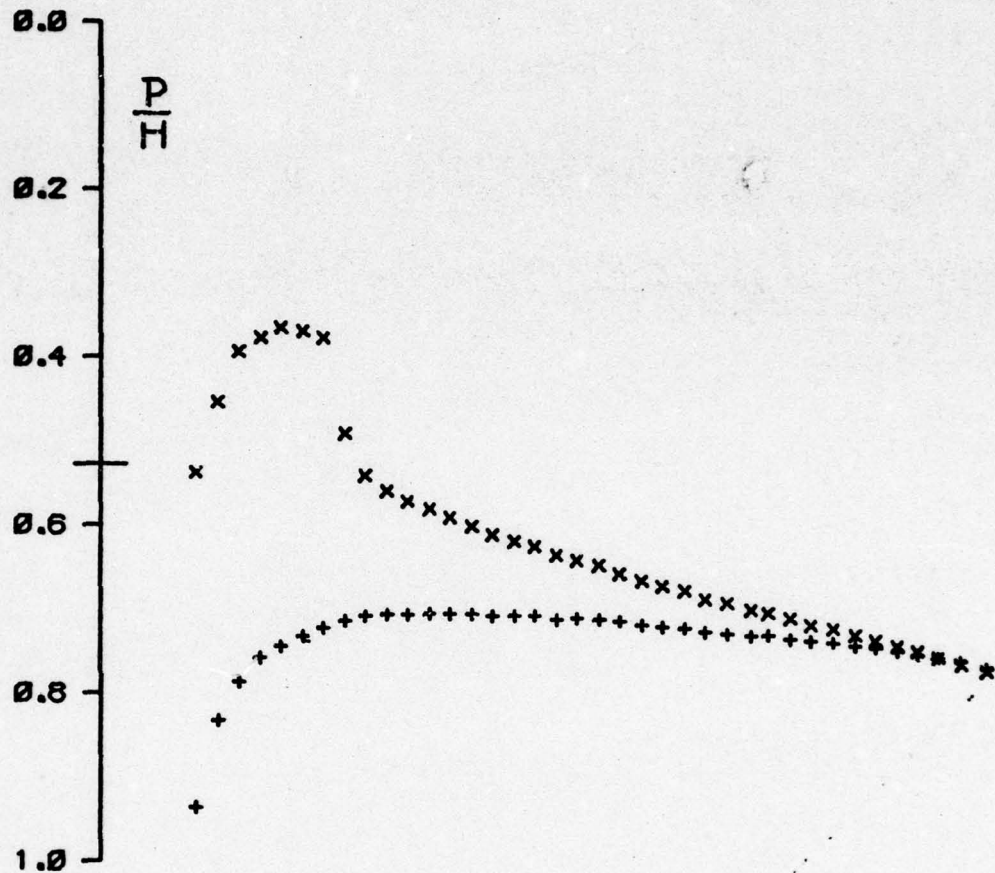
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .676 AL= 2.00 CN= 0.187 CM= 0.006 R= 1.619



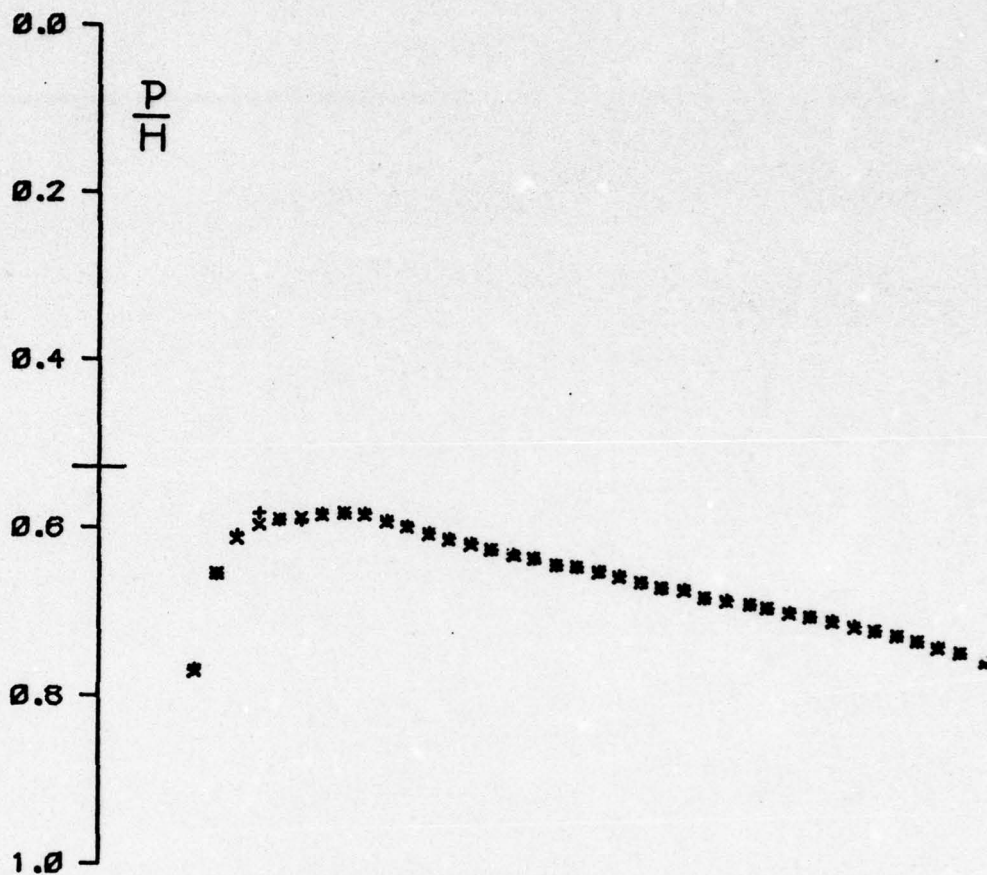
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .674 AL= 3.00 CN= 0.284 CM= 0.010 R= 1.617



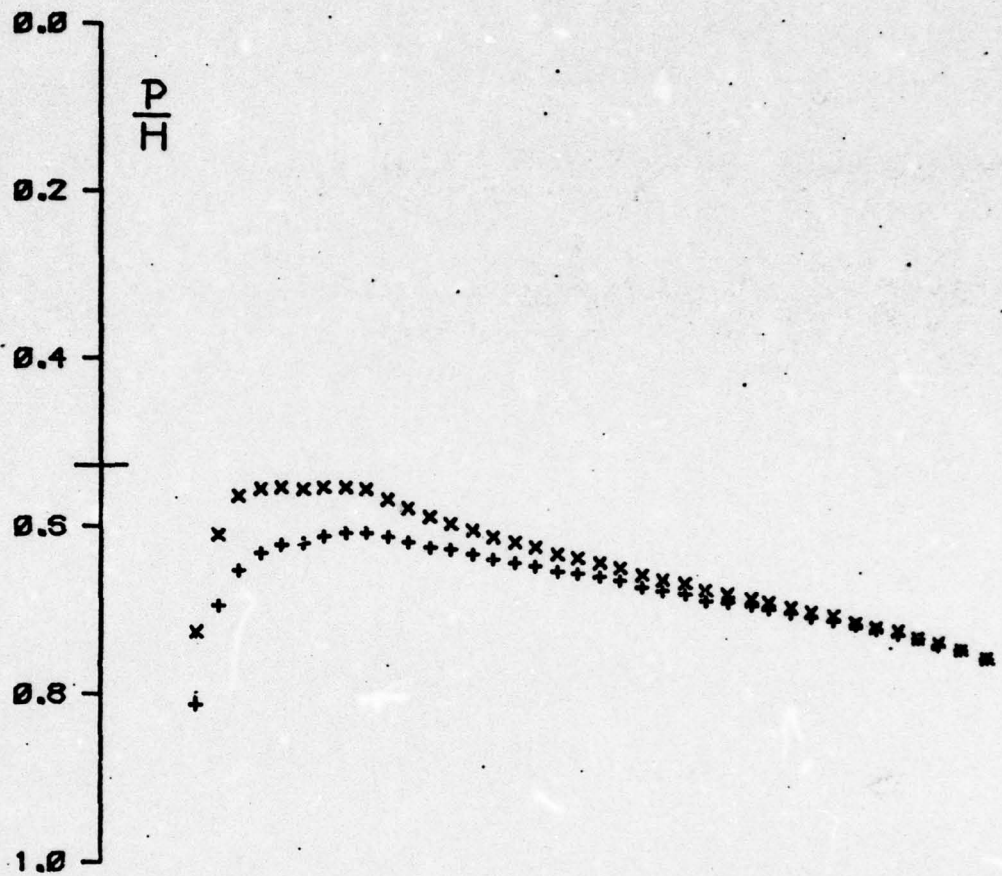
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .675 AL= 4.00 CN= 0.383 CM= 0.015 R= 1.619



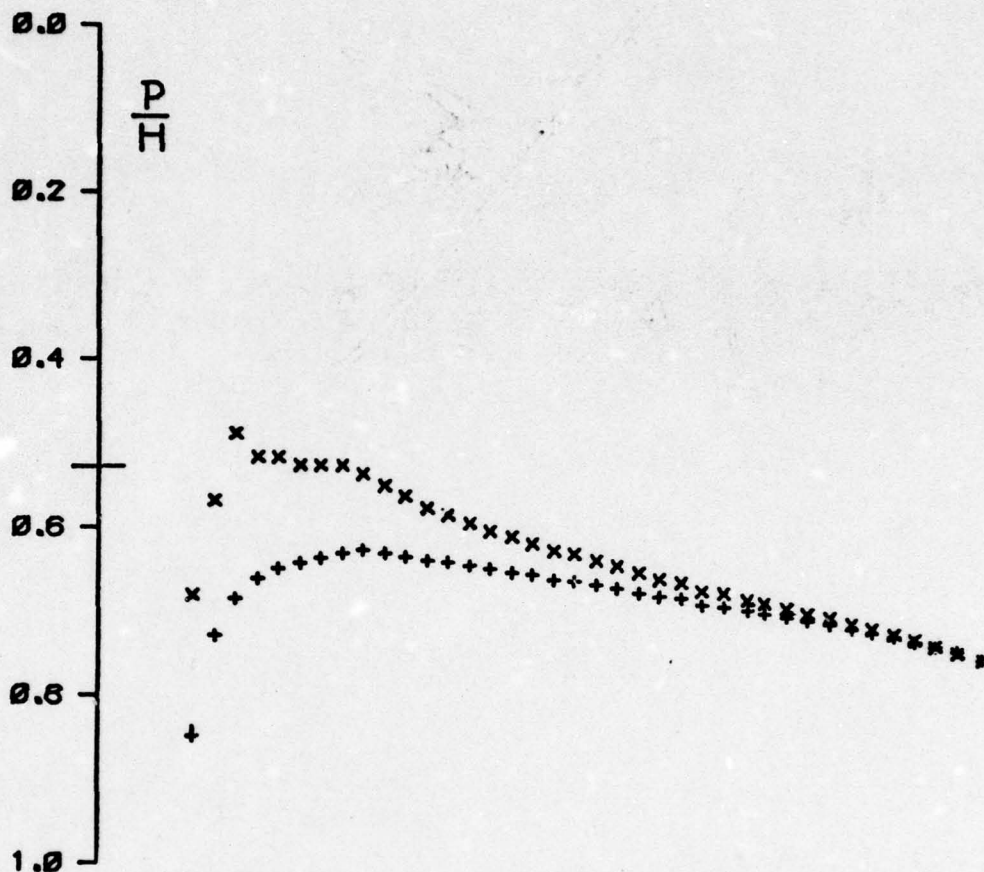
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .674$ $AL = 5.00$ $CH = 0.482$ $CM = 0.019$ $R = 1.620$



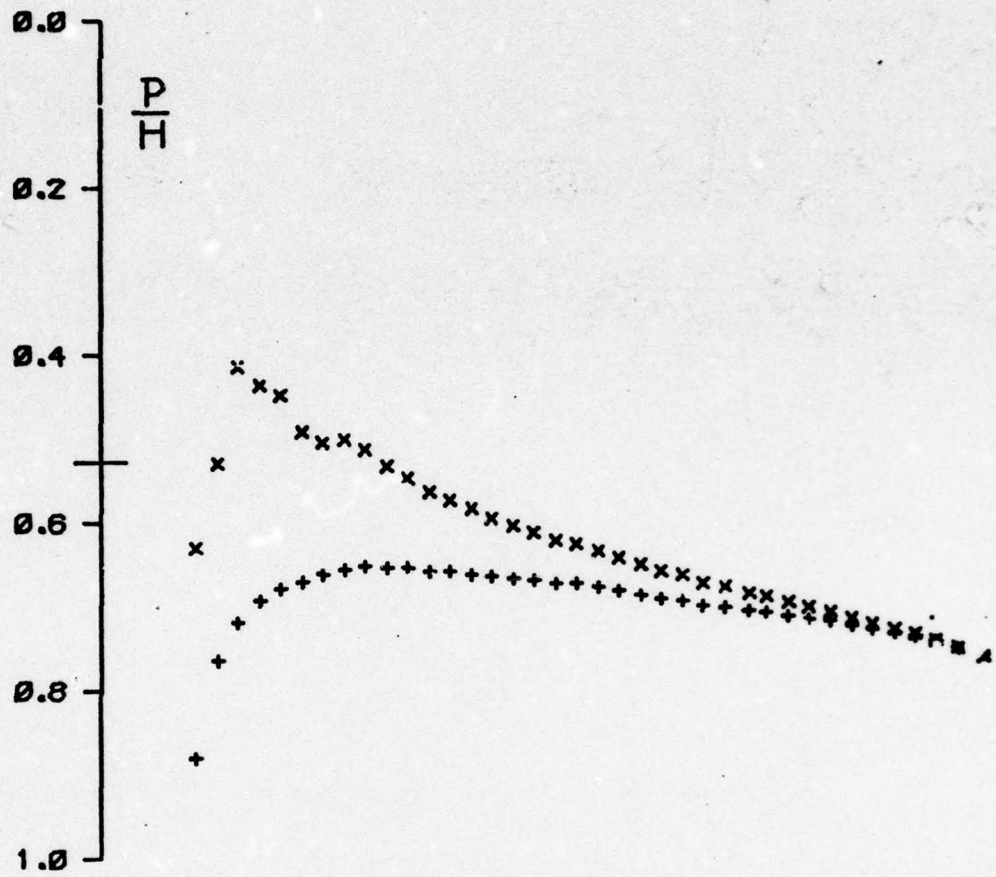
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .700 AL= 0.00 CN=-0.005 CM=-0.000 R= 1.656



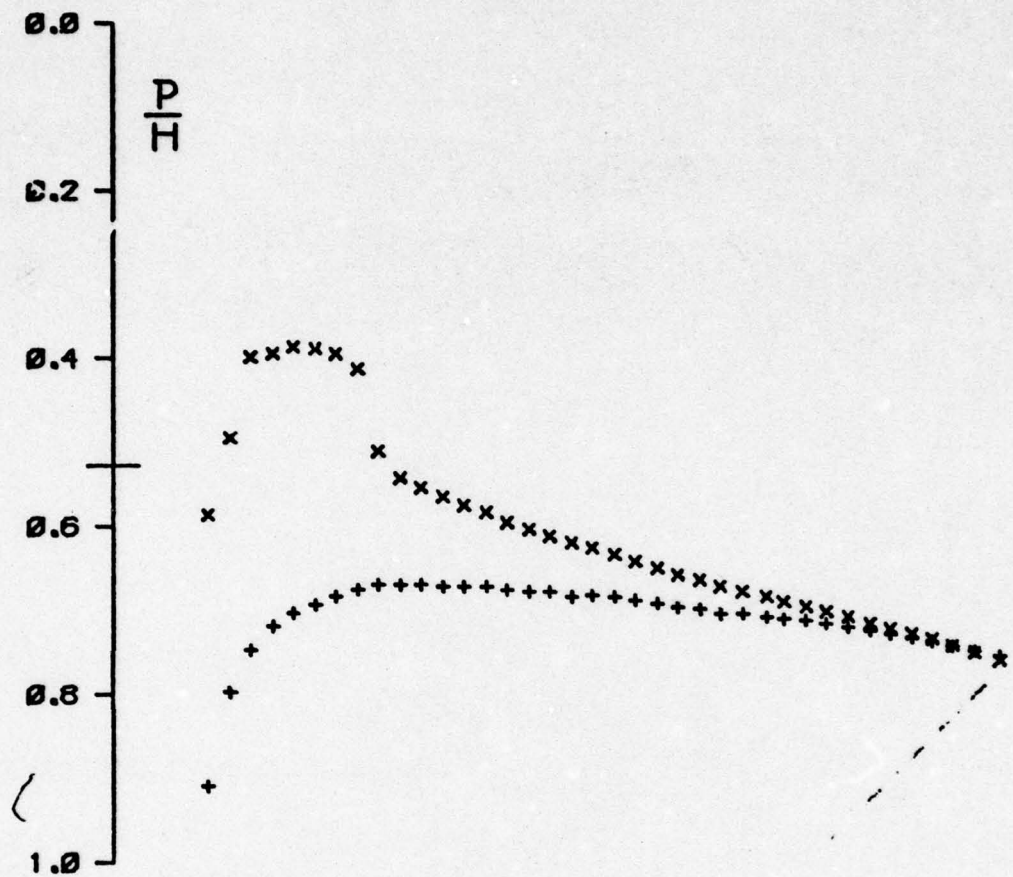
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .701$ $AL = 1.00$ $CN = 0.097$ $CM = 0.003$ $R = 1.655$



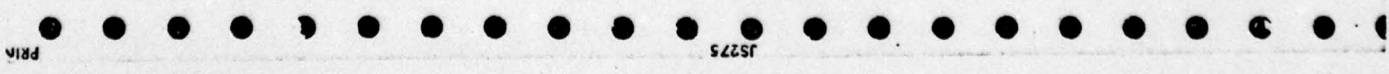
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .700 AL= 2.00 CN= 0.194 CM= 0.028 R= 1.658

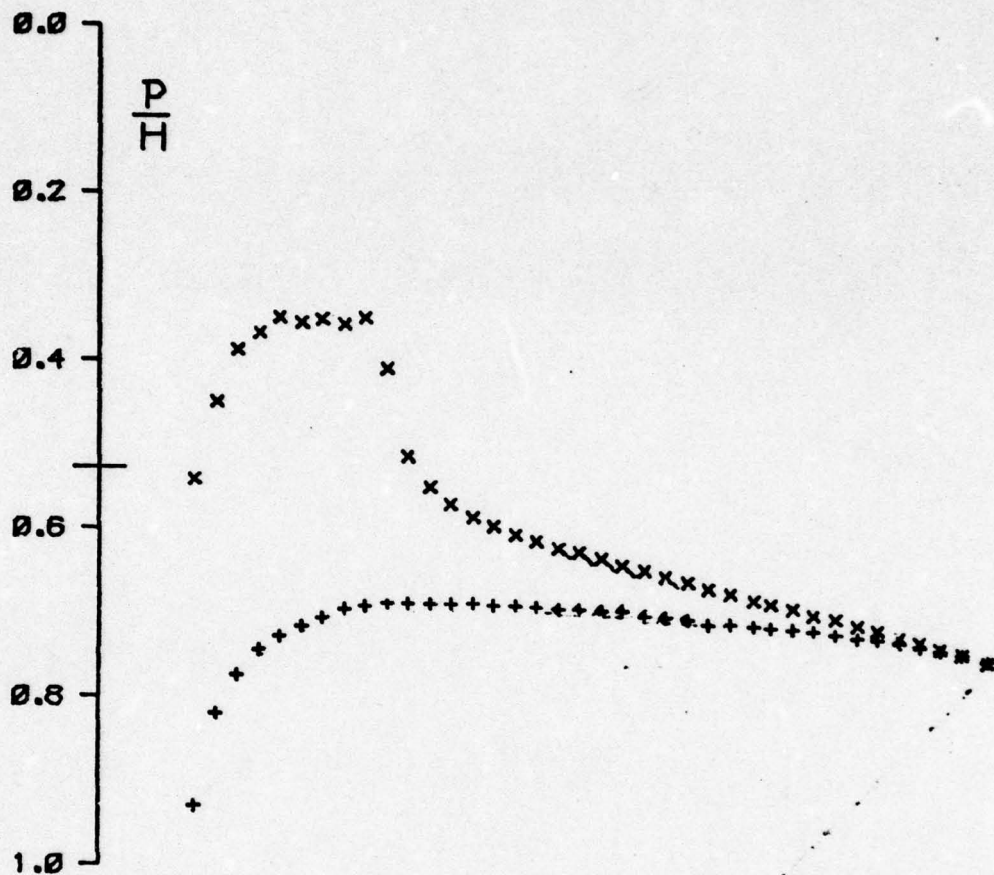


+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .702 AL= 3.00 CN= 0.293 CM= 0.012 R= 1.660

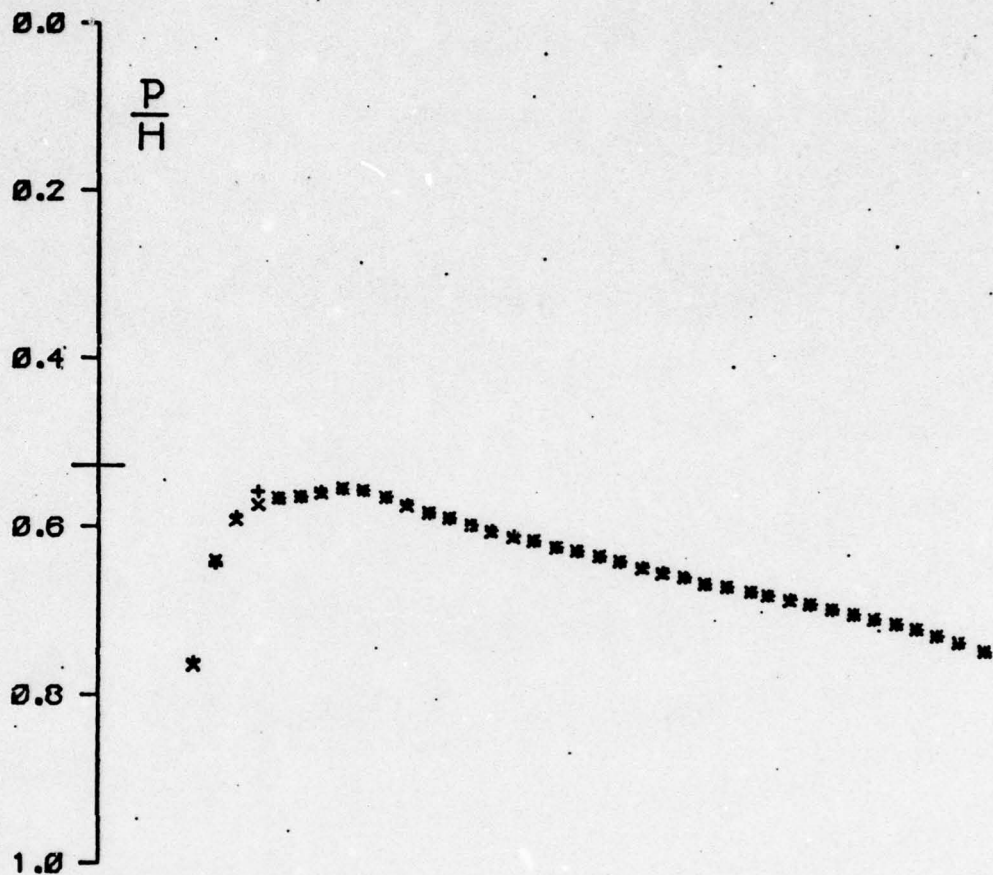


+ x NACA 0012 223.2 MM CHORD SLOTTED WALKS
 M= .698 AL= 4.00 CN= 0.394 CM= 0.017 R= 1.651

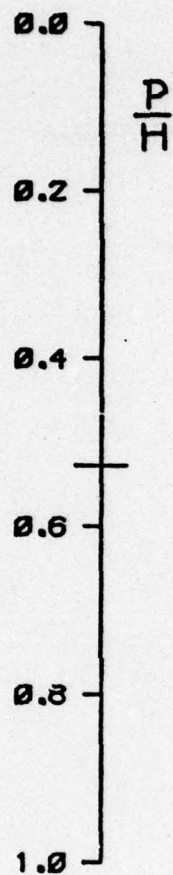




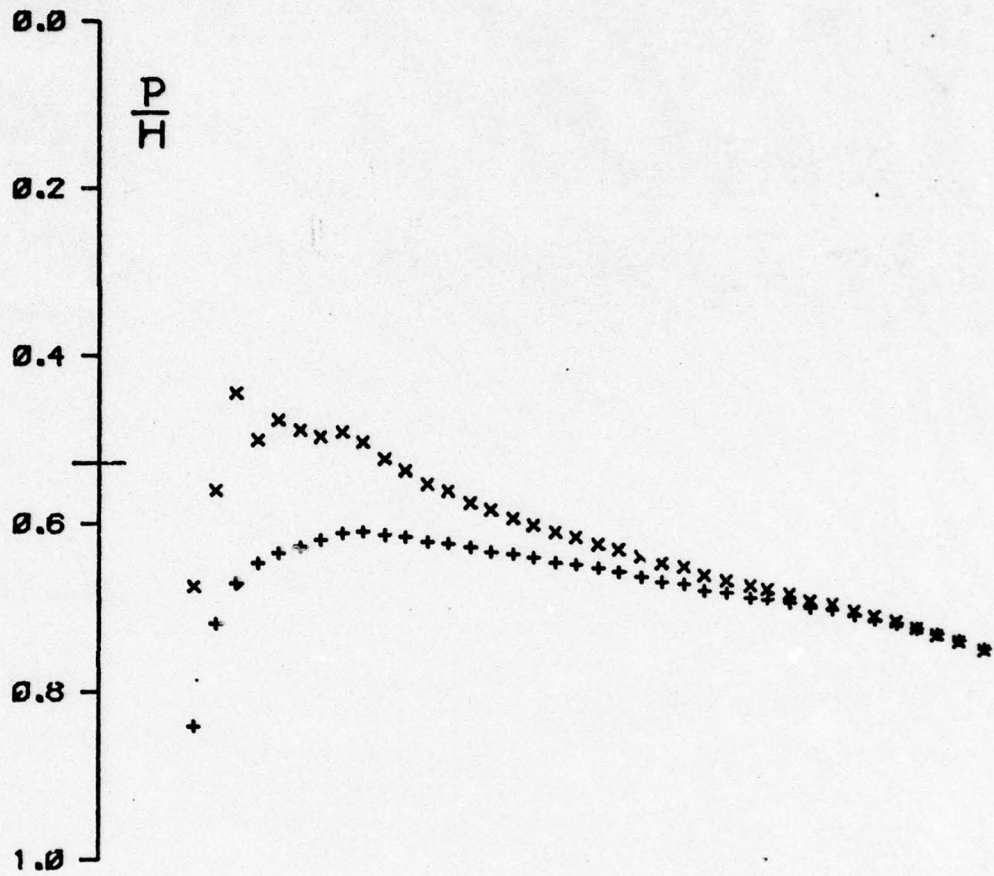
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .698$ $AL = 5.00$ $CN = 0.500$ $CA = 0.019$ $R = 1.655$



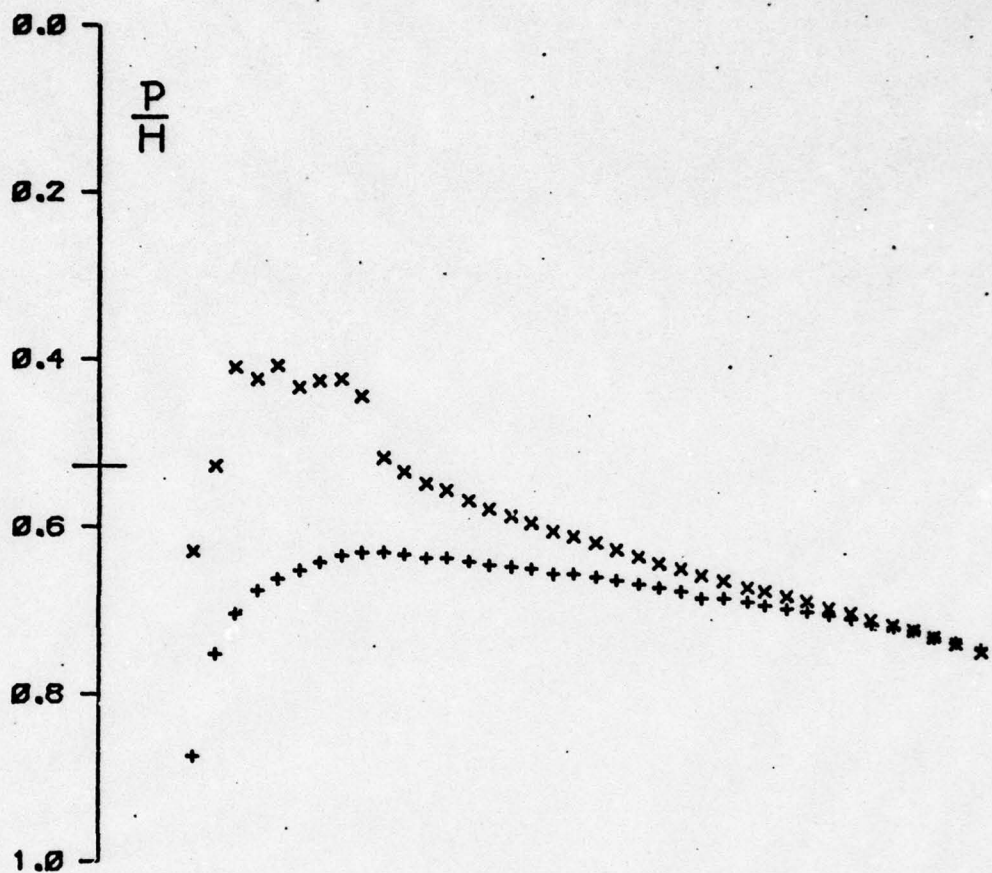
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .721$ $AL = 0.00$ $CN = -0.007$ $CM = 0.000$ $R = 1.660$



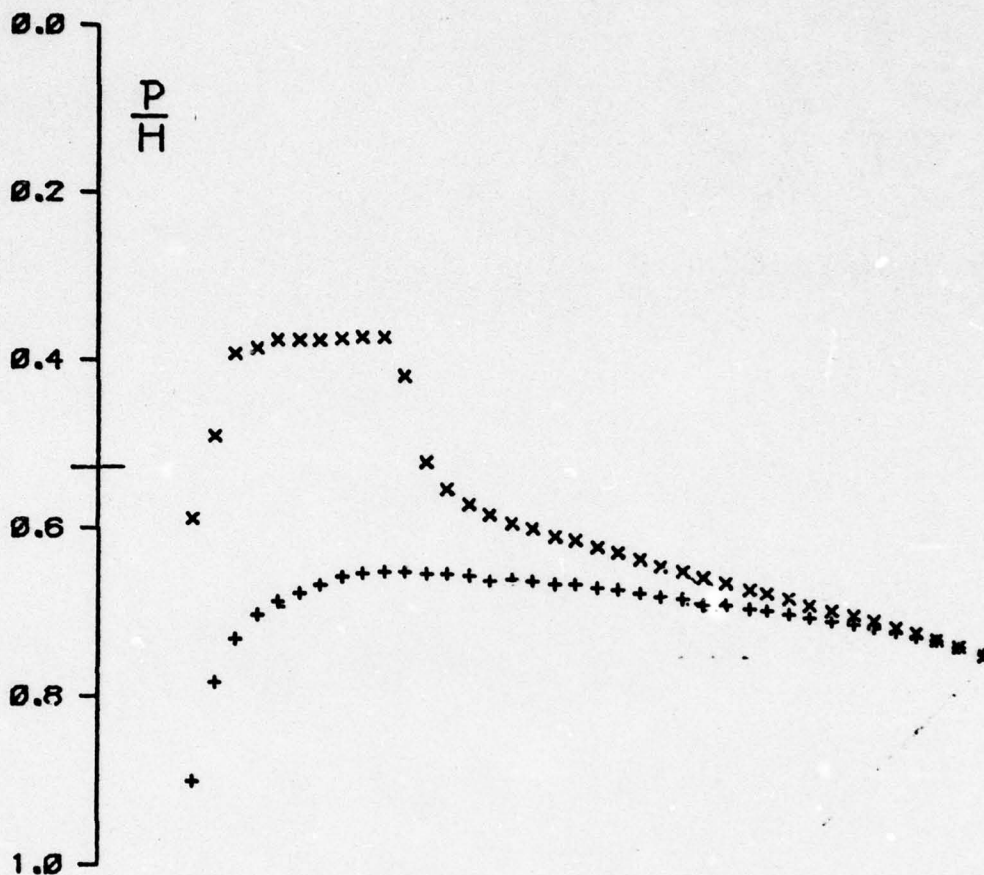
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M = .720 AL = 1.00 CN = 0.098 CM = 0.004 R = 1.653



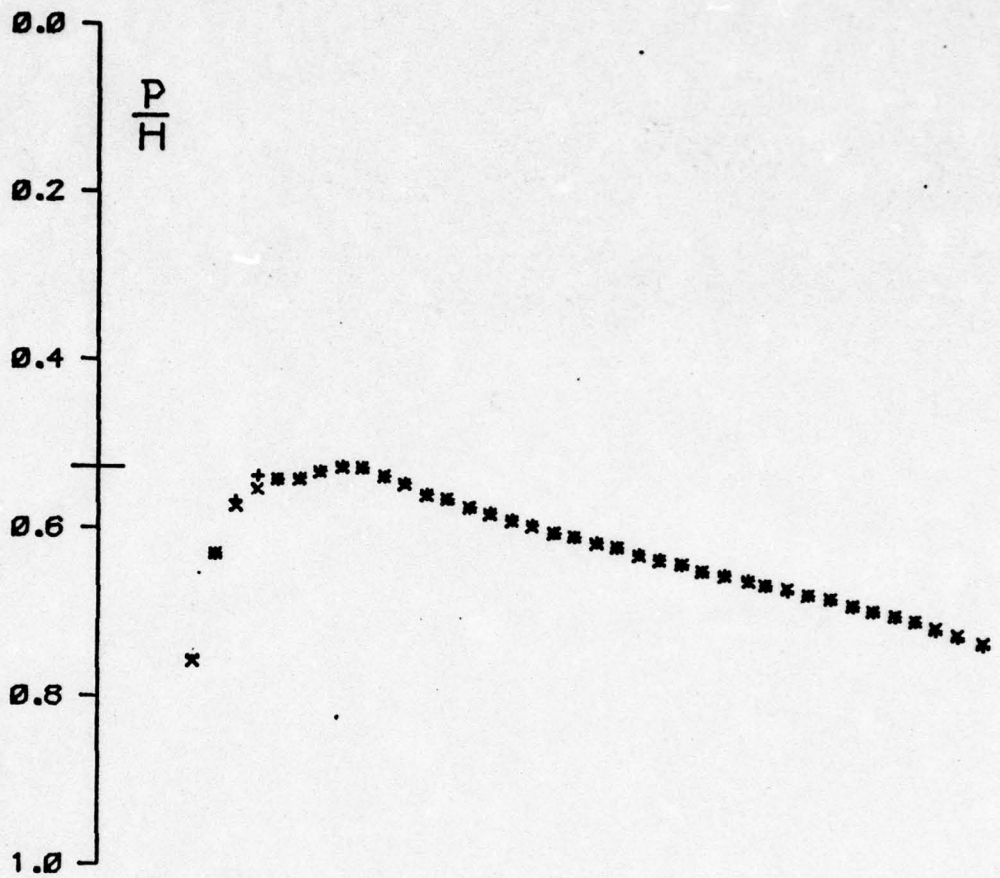
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .721 AL= 2.00 CN= 0.198 CM= 0.028 R= 1.680



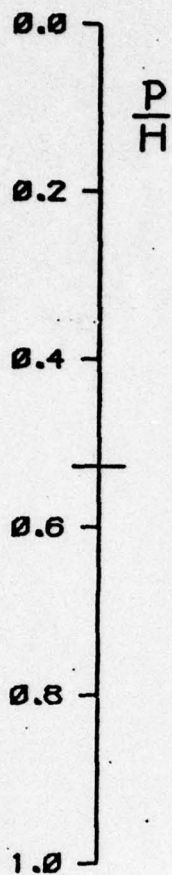
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .715 AL= 3.00 CN= 0.303 CM= 0.013 R= 1.655



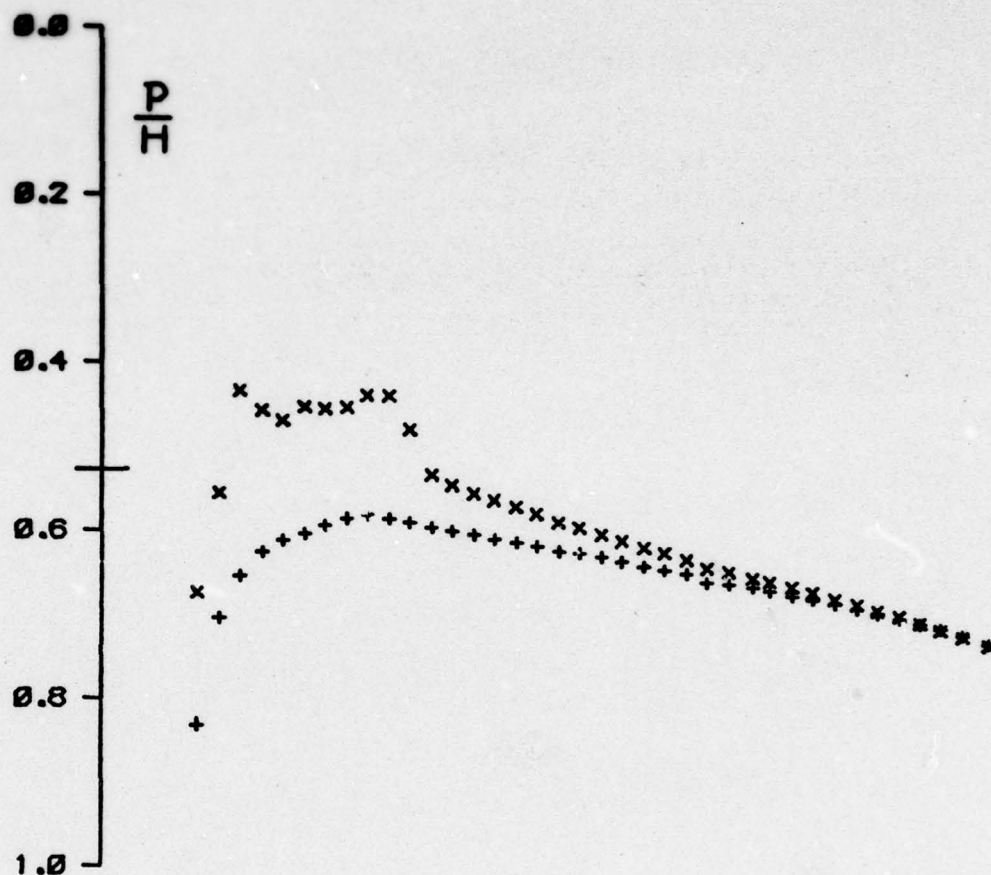
x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M = .719 AL = 4.00 CN = 0.410 CM = 0.012 R = 1.663



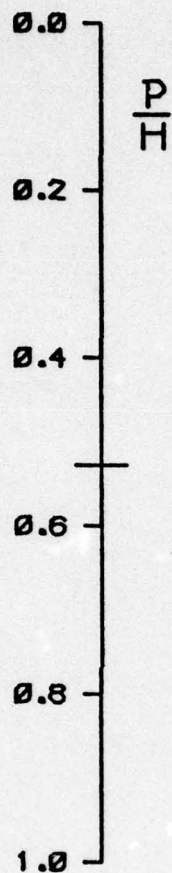
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .740 AL= 0.00 CN=-0.006 CM= 0.000 R= 1.646



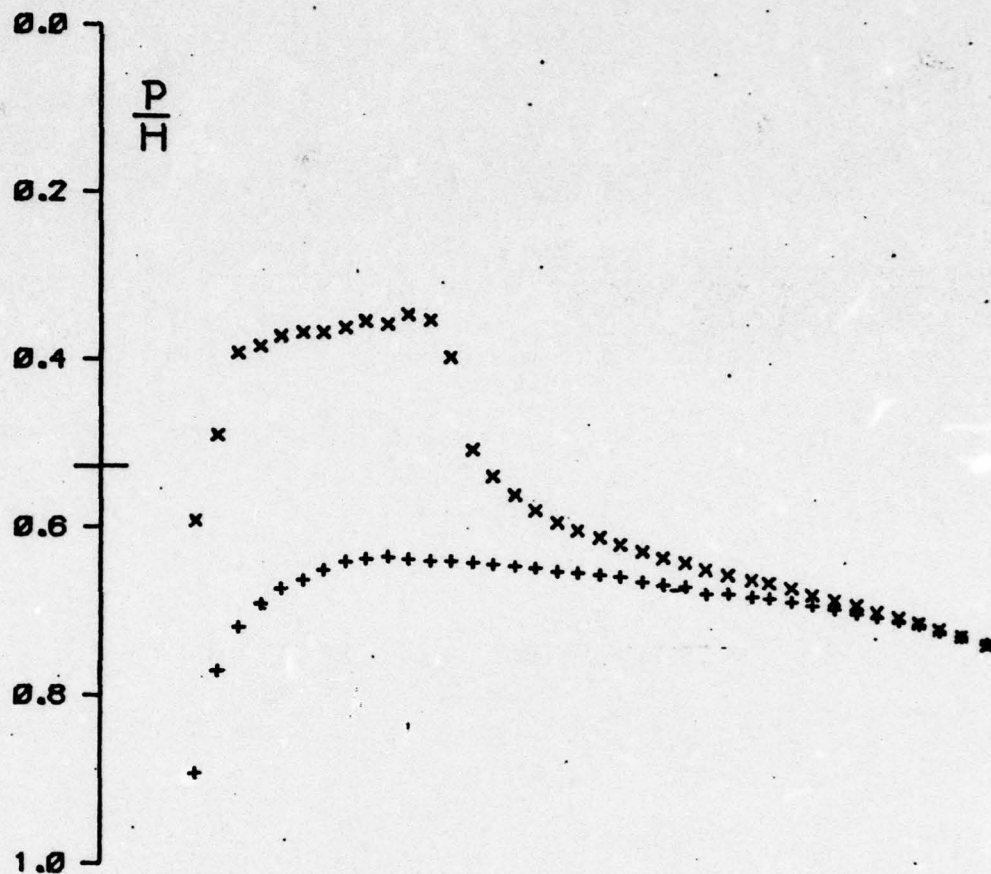
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .742 AL= 1.00 CN= 0.099 CM= 0.004 R= 1.650



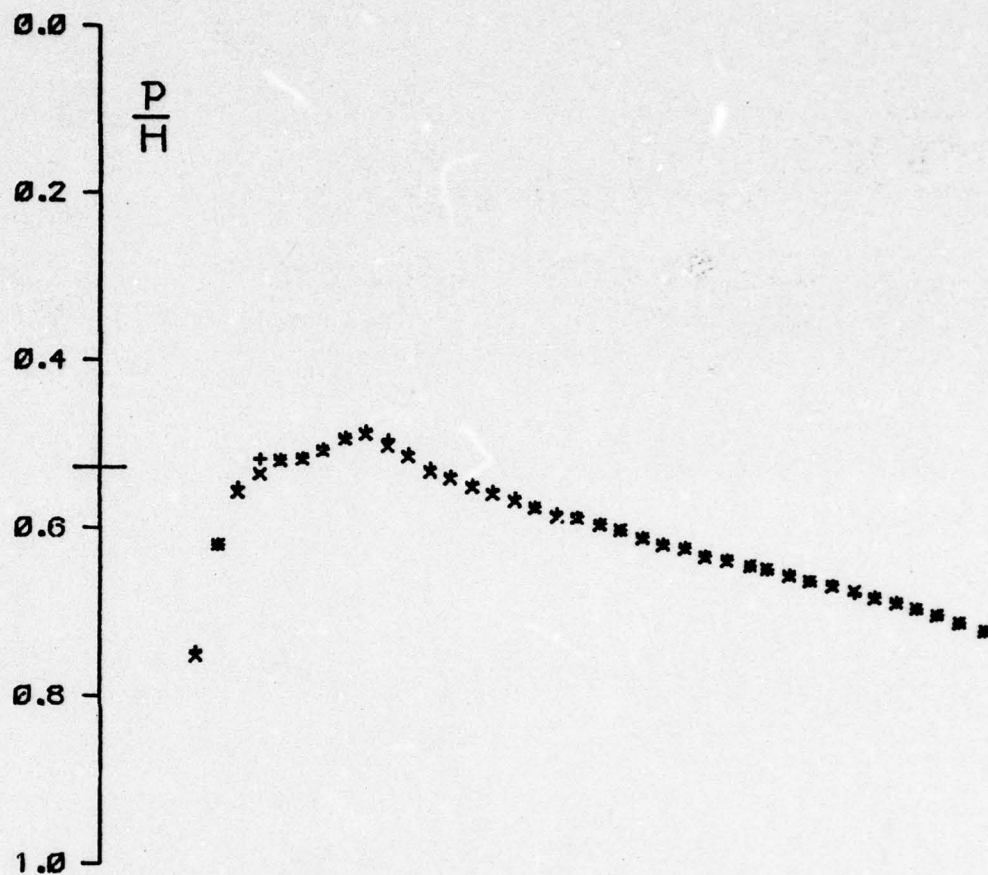
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .741 AL= 2.00 CN= 0.205 CM= 0.010 R= 1.650



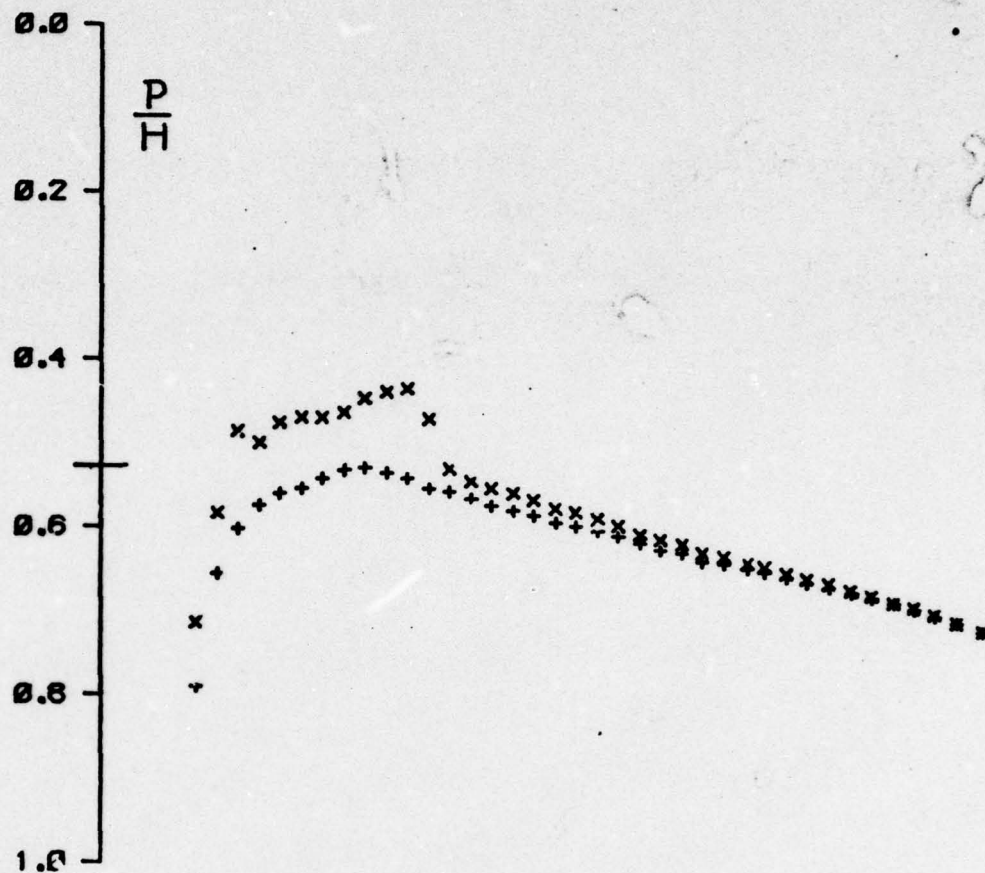
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .740$ $AL = 3.00$ $CN = 0.315$ $CN = 0.014$ $R = 1.646$



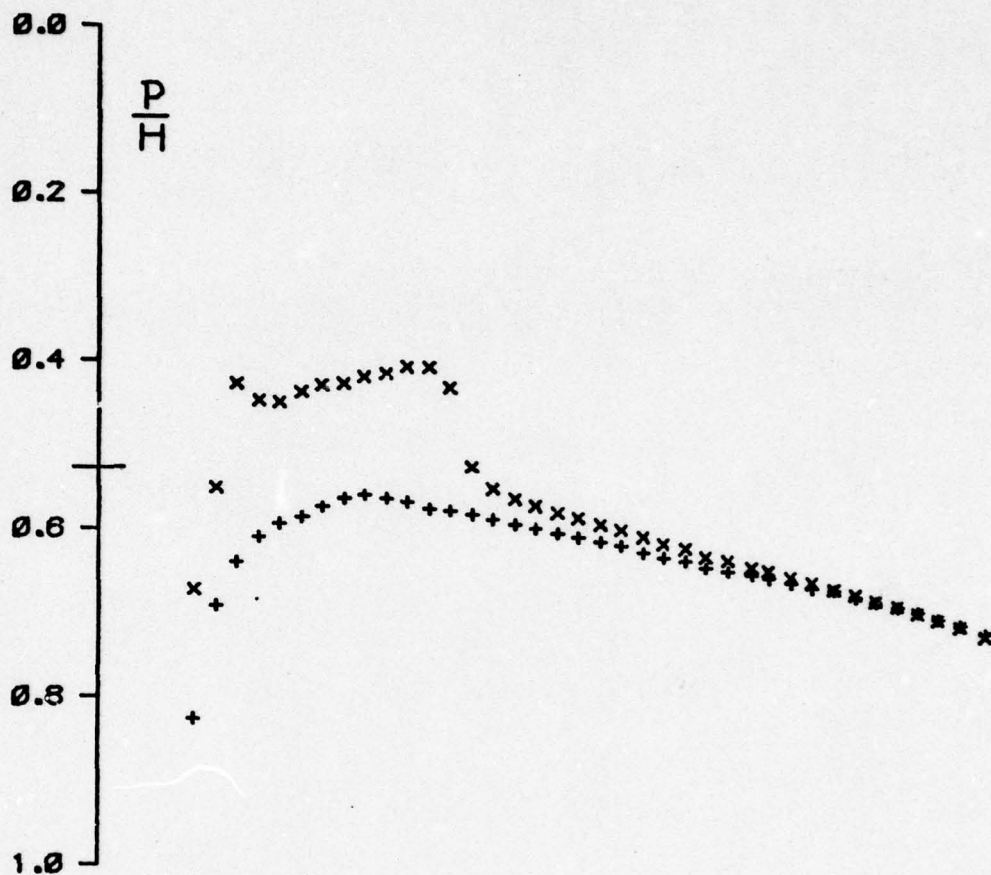
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .739$ $AL = 4.00$ $CN = 0.422$ $CN = 0.015$ $R = 1.648$



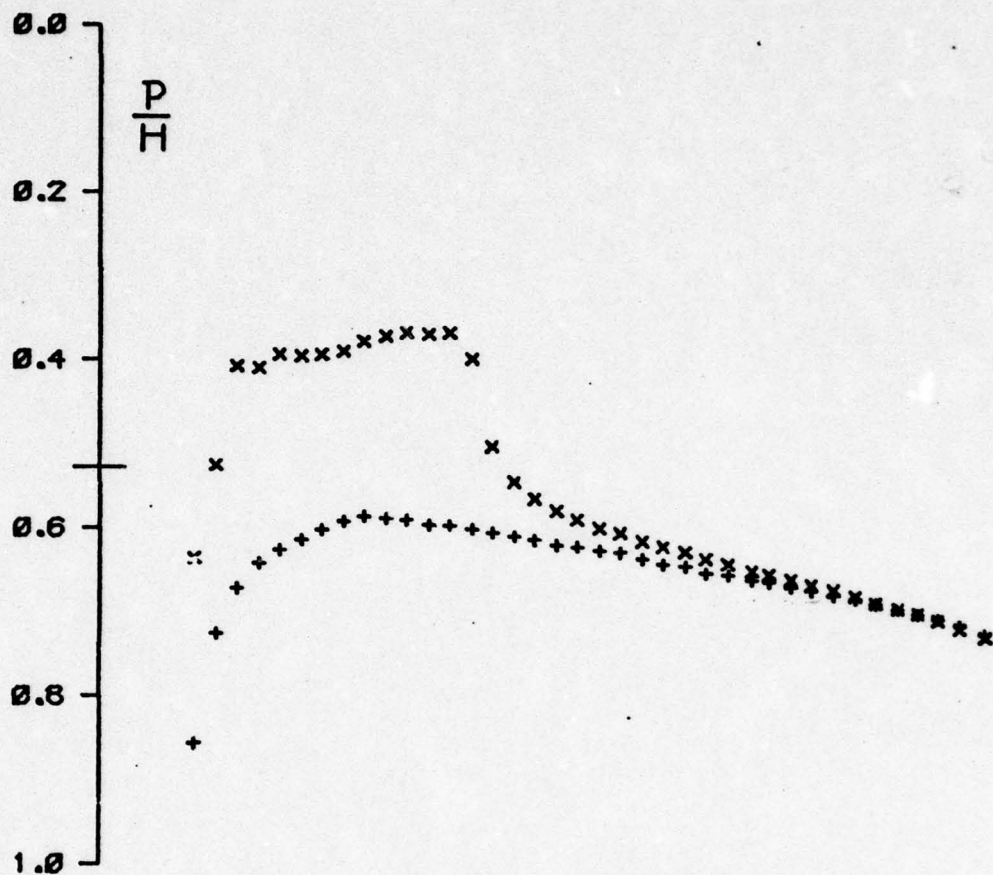
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .759 AL= 0.00 CN=-0.007 CM=-0.000 R= 1.643



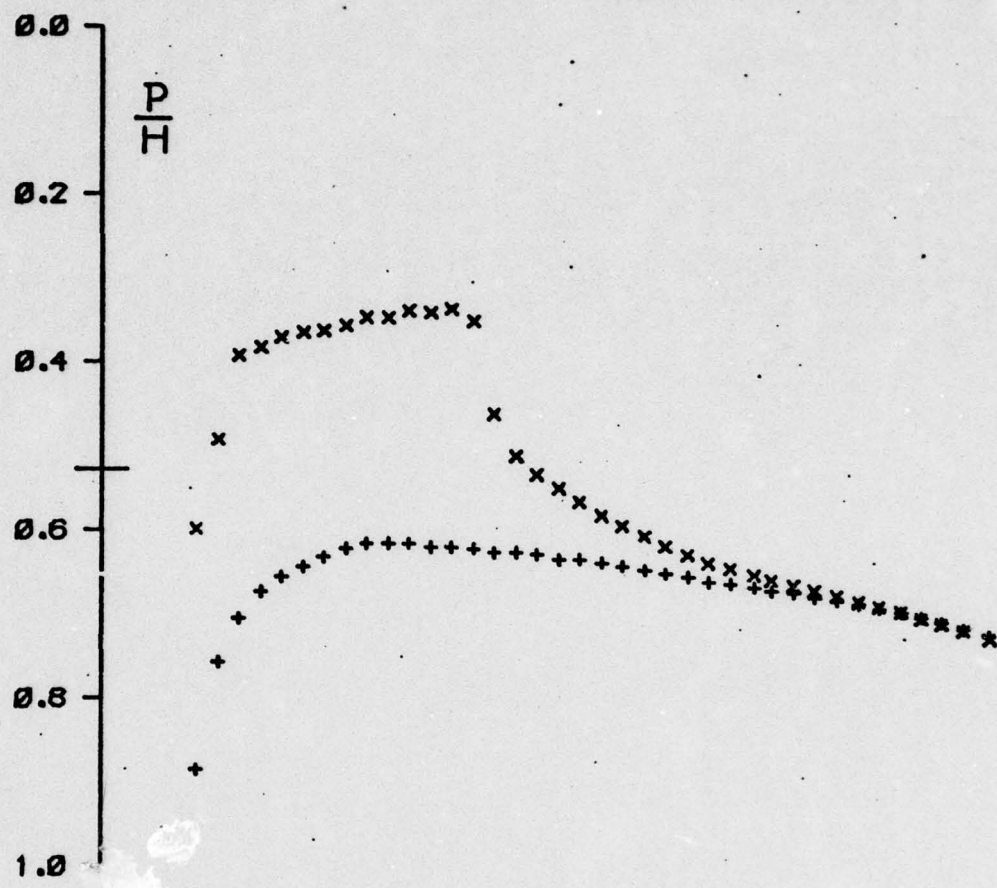
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .761 AL= 1.20 CN= 0.107 CM= 0.005 R= 1.646



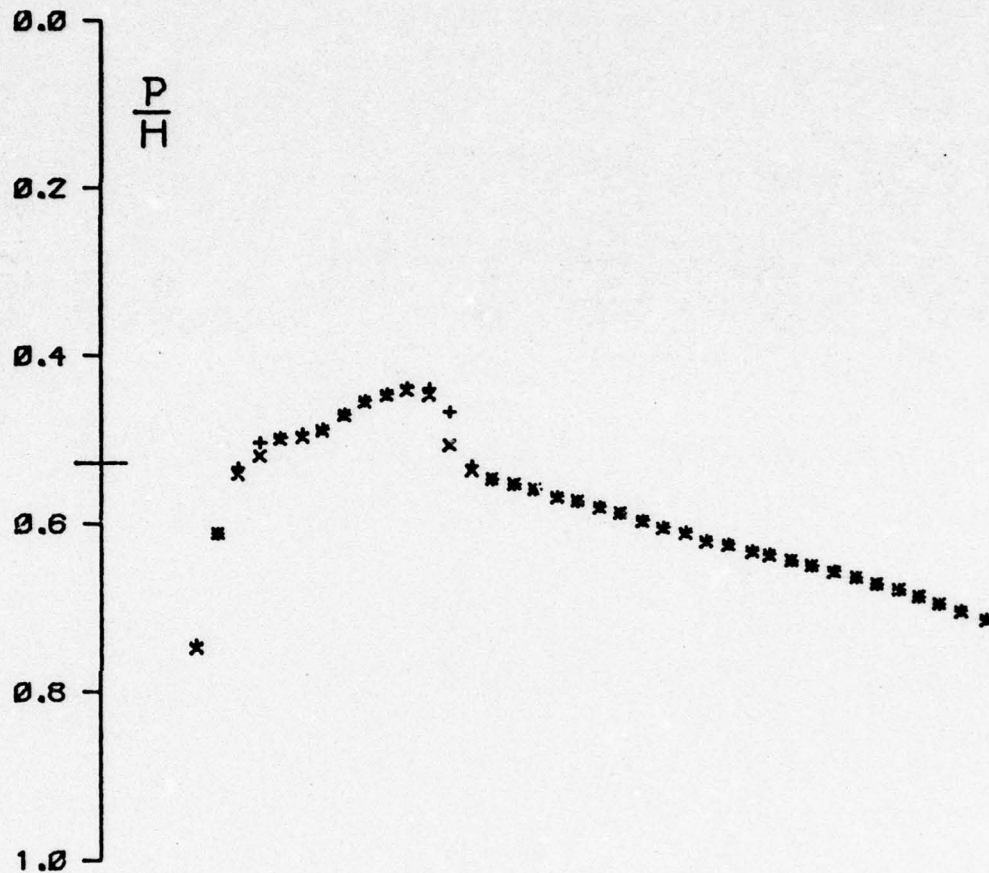
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .761$ $AL = 2.00$ $CN = 0.214$ $CM = 0.009$ $R = 1.646$



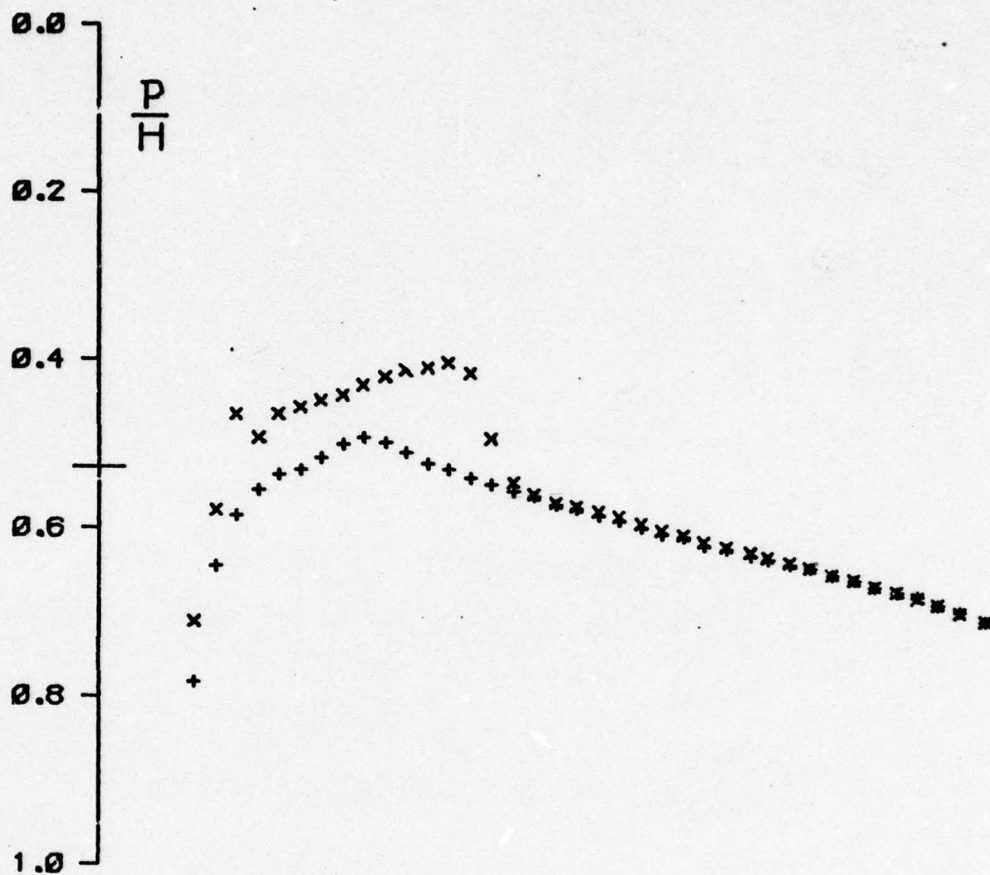
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .759 AL= 3.00 CN= 0.323 CM= 0.012 R= 1.643



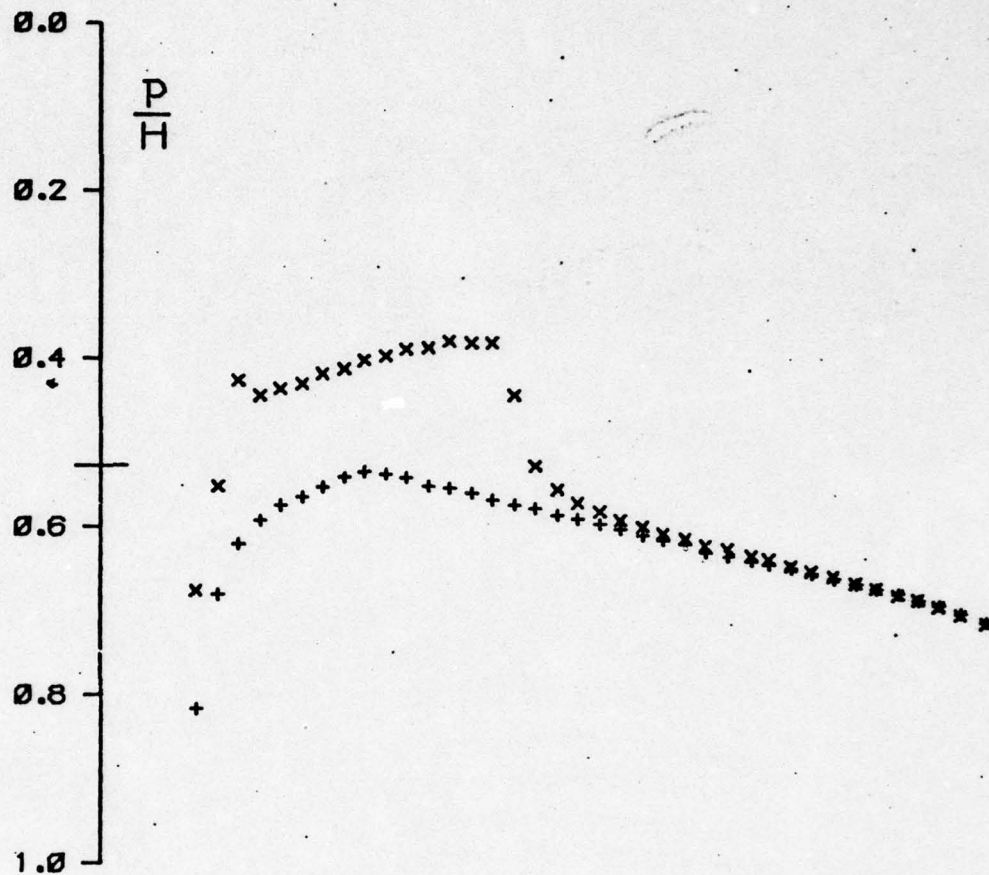
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .758 AL= 4.00 CN= 0.424 CM= 0.011 R= 1.643



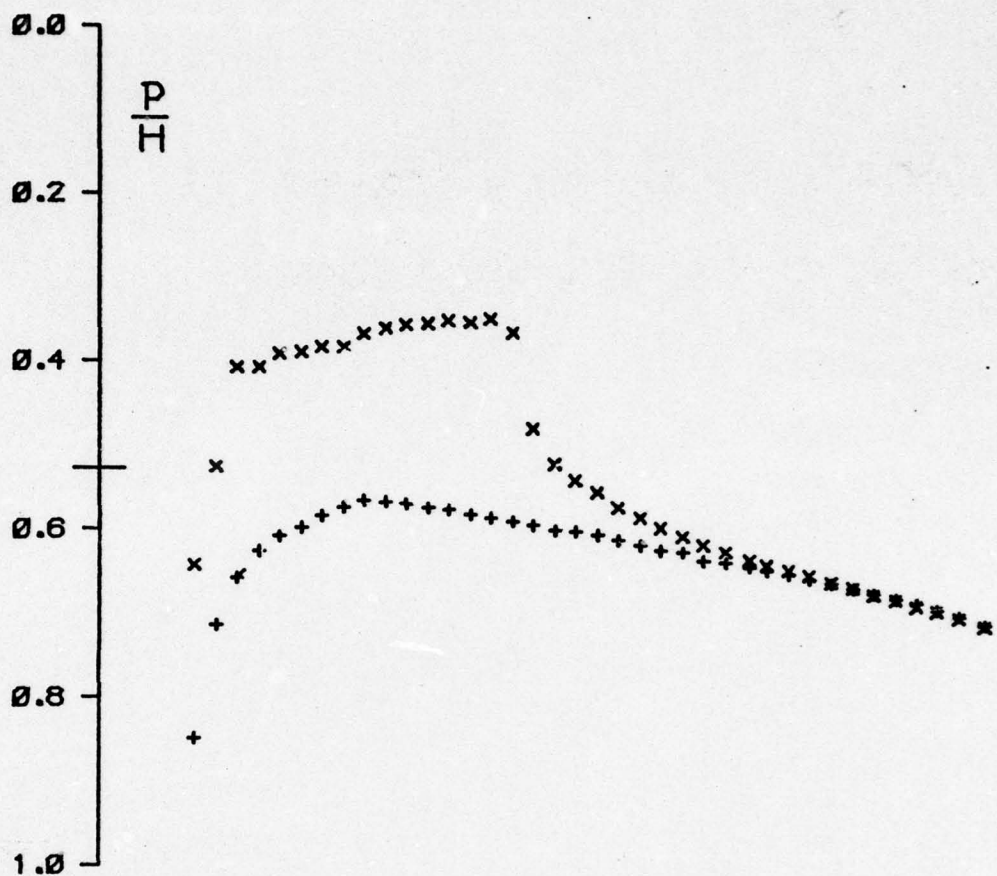
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .780 AL= 0.00 CN=-0.007 CM= 0.000 R= 1.606



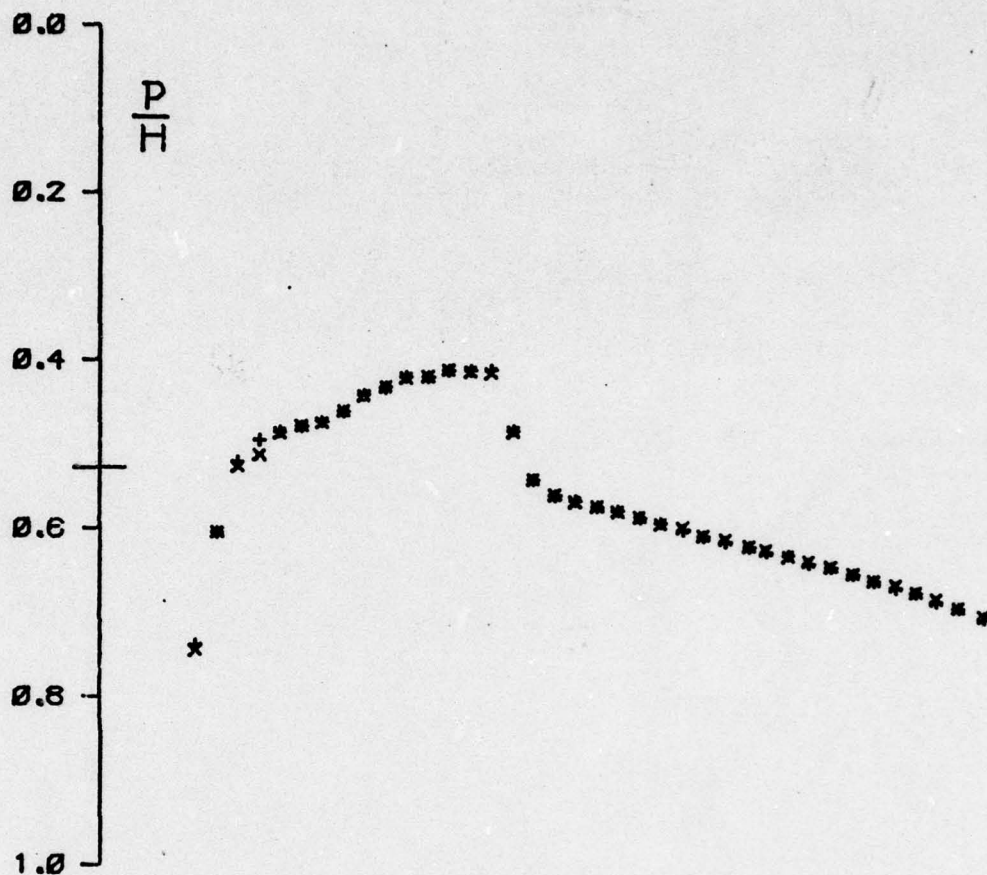
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .730$ $AL = 1.00$ $CN = 0.114$ $CM = 0.004$ $R = 1.606$



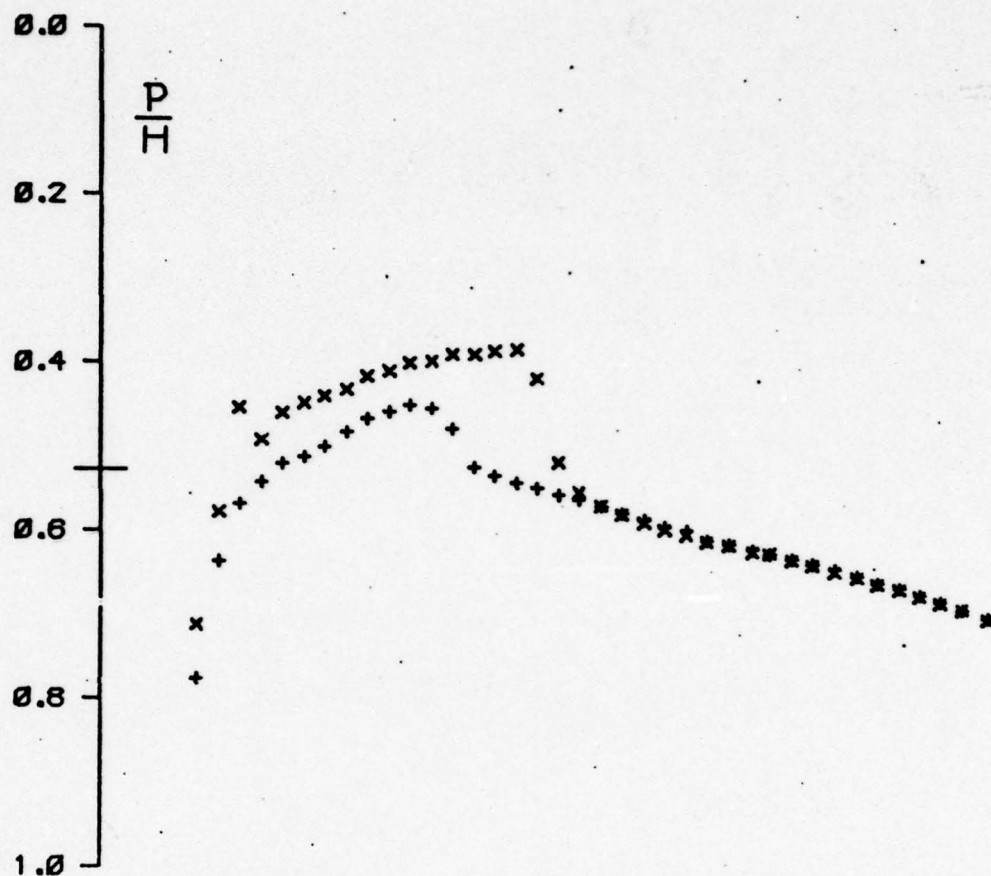
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .780$ $AL = 2.00$ $CN = 0.224$ $CM = 0.006$ $R = 1.606$



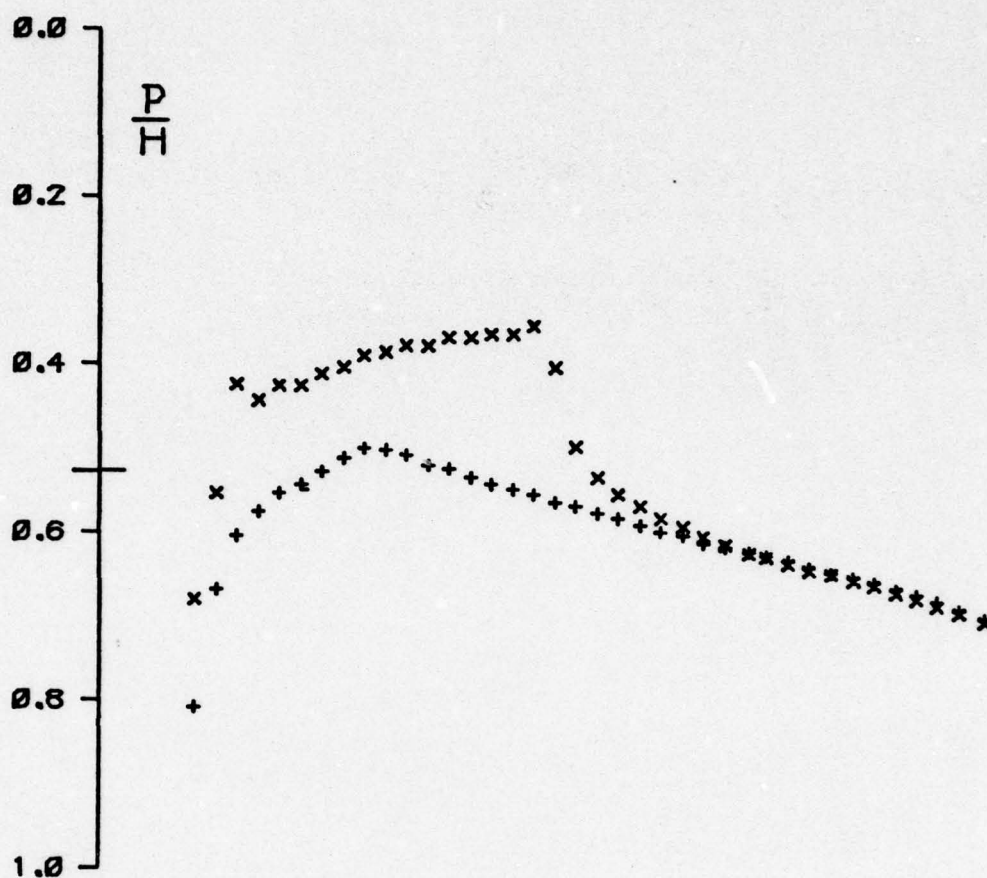
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .781 AL= 3.00 CN= 0.332 CM= 0.006 R= 1.629



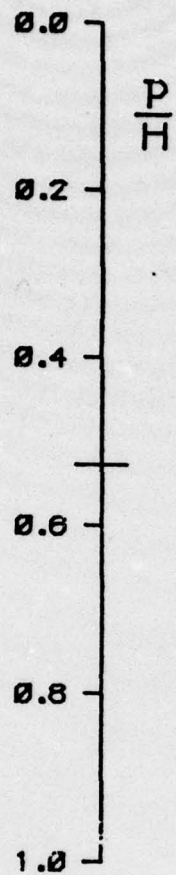
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .801$ $AL = 0.00$ $CN = -0.005$ $CM = 0.000$ $R = 1.630$



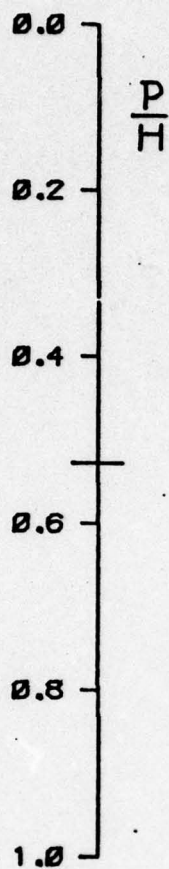
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .800$ $AL = 1.00$ $CN = 0.120$ $CM = 0.001$ $R = 1.632$



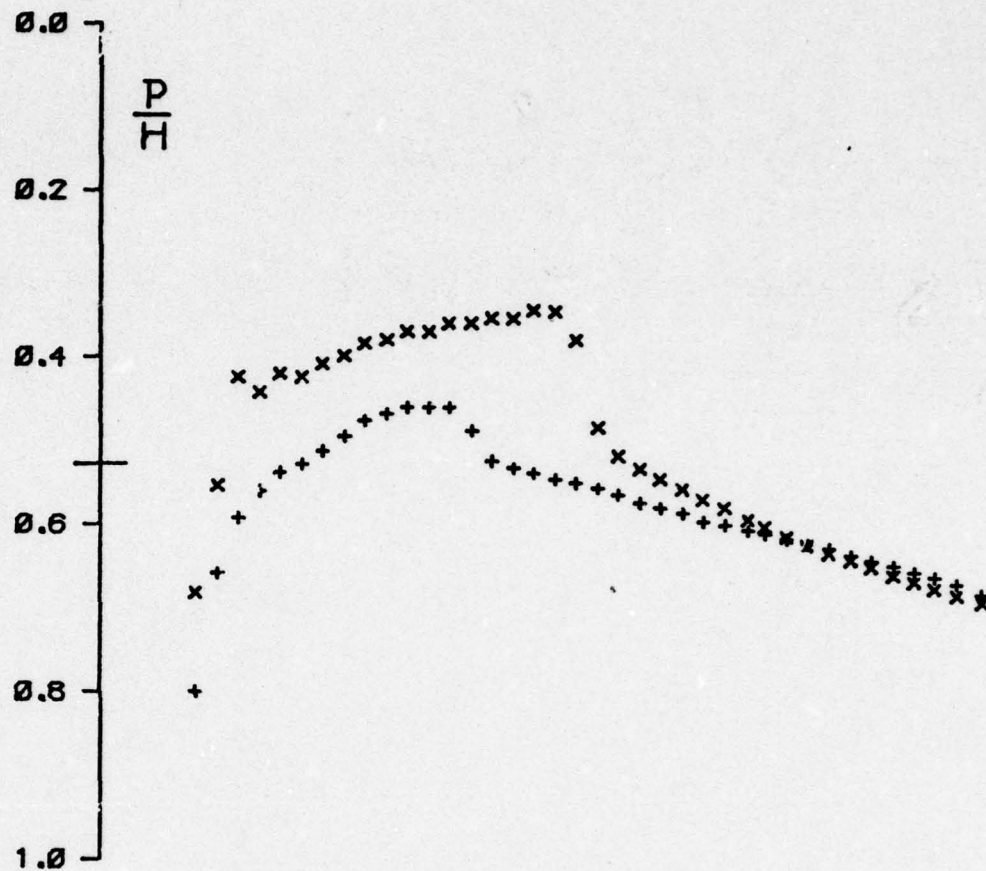
x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .801 AL= 2.00 CN= 0.230 CM= 0.001 R= 1.633



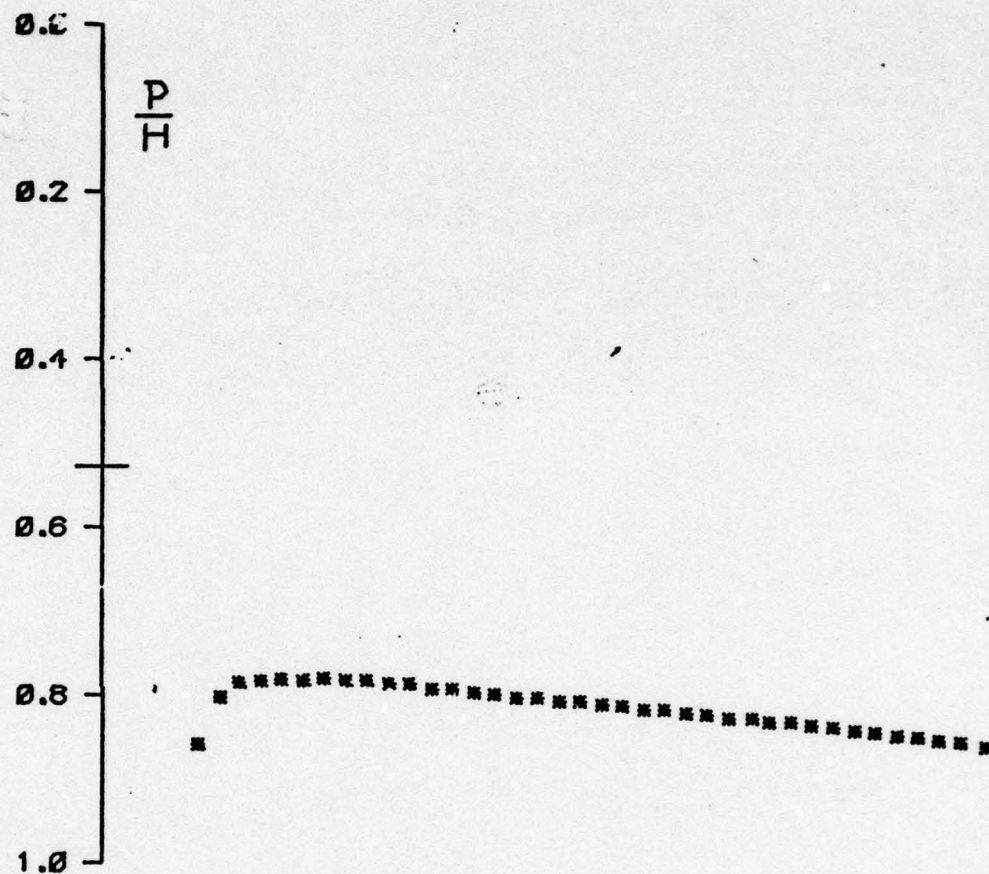
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .820$ $AL = 0.00$ $CN = -0.014$ $CM = 0.002$ $R = 1.684$



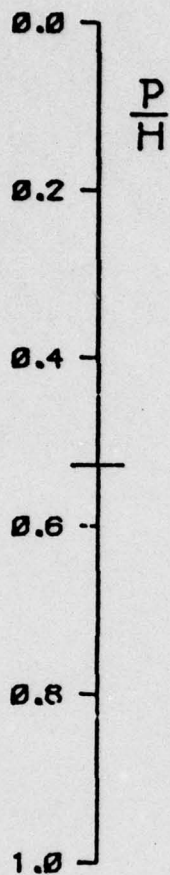
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .819$ $AL = 1.00$ $CN = 0.118$ $CM = -0.004$ $R = 1.681$



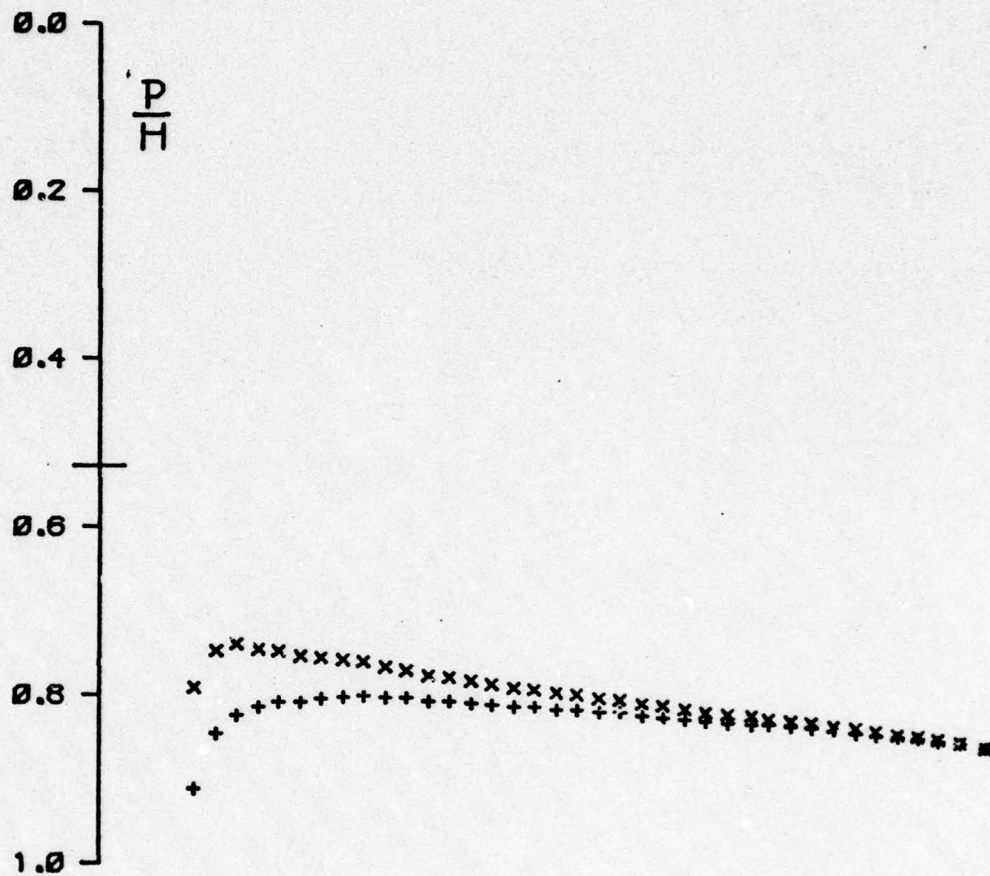
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .819$ $AL = 2.00$ $CN = 0.214$ $CM = -0.004$ $R = 1.631$



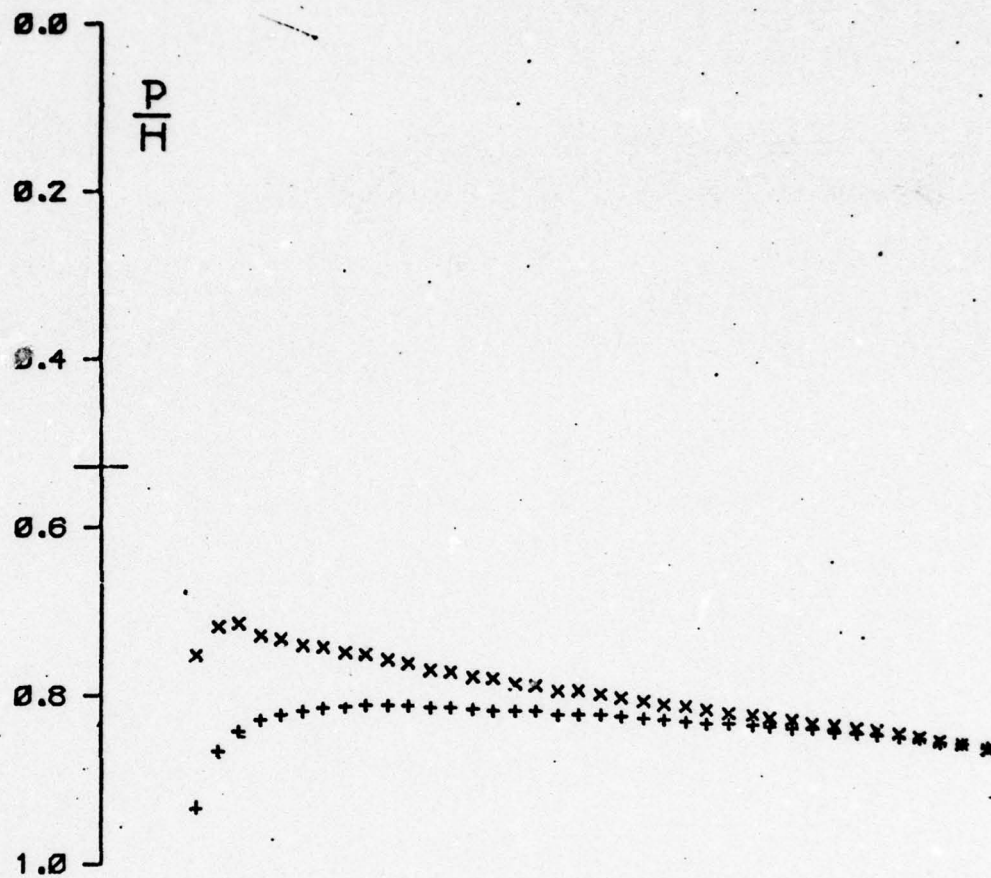
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .499 A= 2.00 CN= 0.000 CM= 0.000 R= 0.850



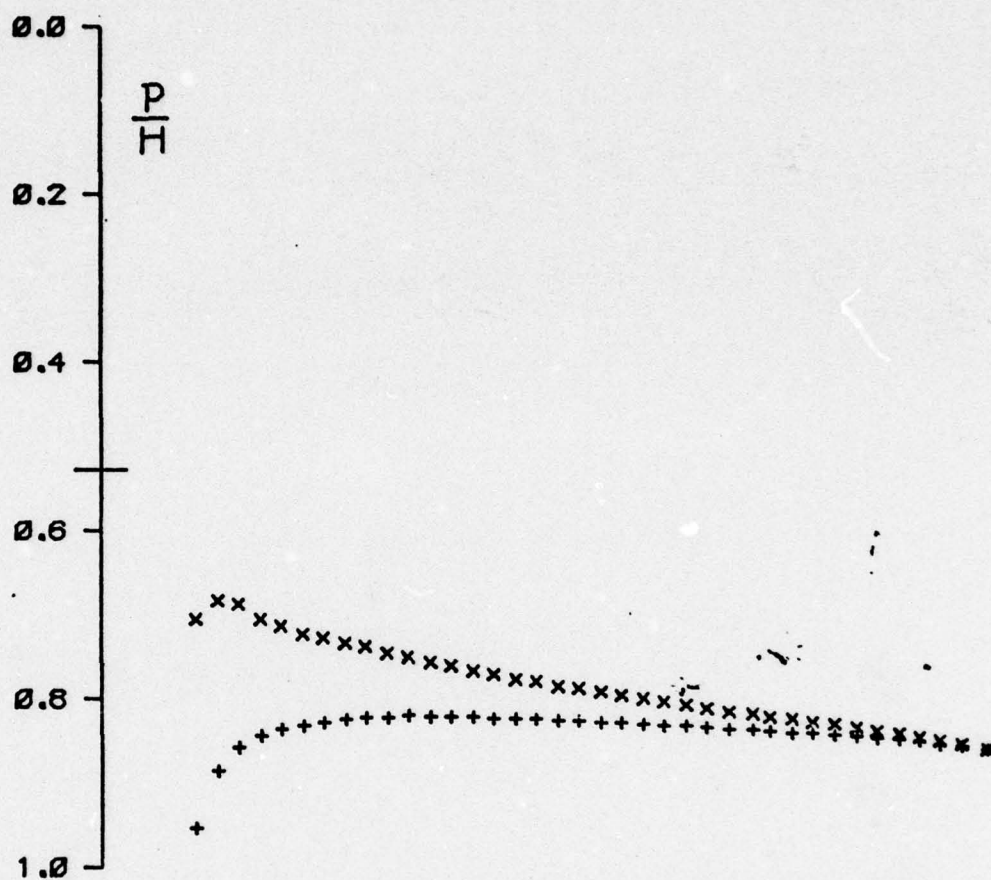
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .489$ $AL = 1.00$ $CN = 0.081$ $CM = 0.002$ $R = 0.904$



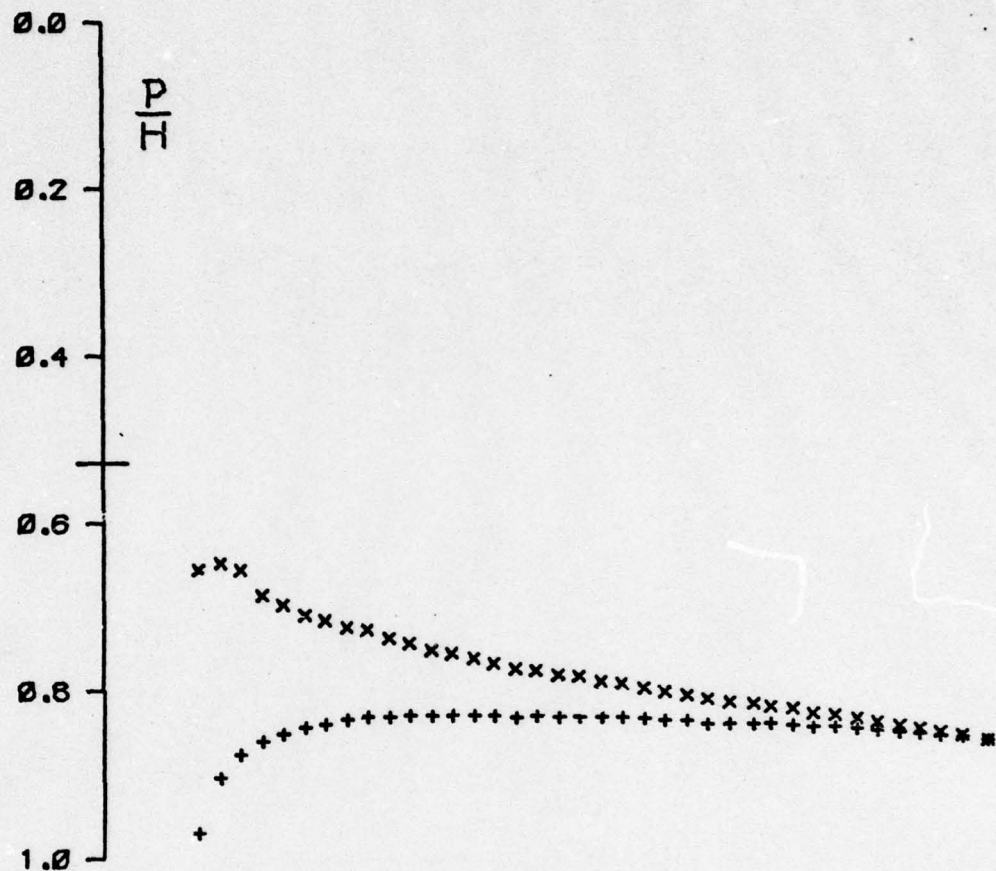
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .499 AL= 2.00 CN= 0.171 CM= 0.003 R= 0.850



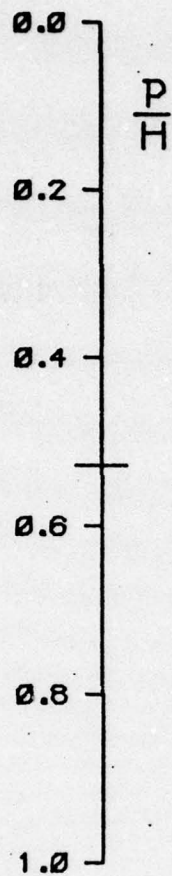
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .489 AL= 3.00 CN= 0.256 CM= 0.005 R= 0.650



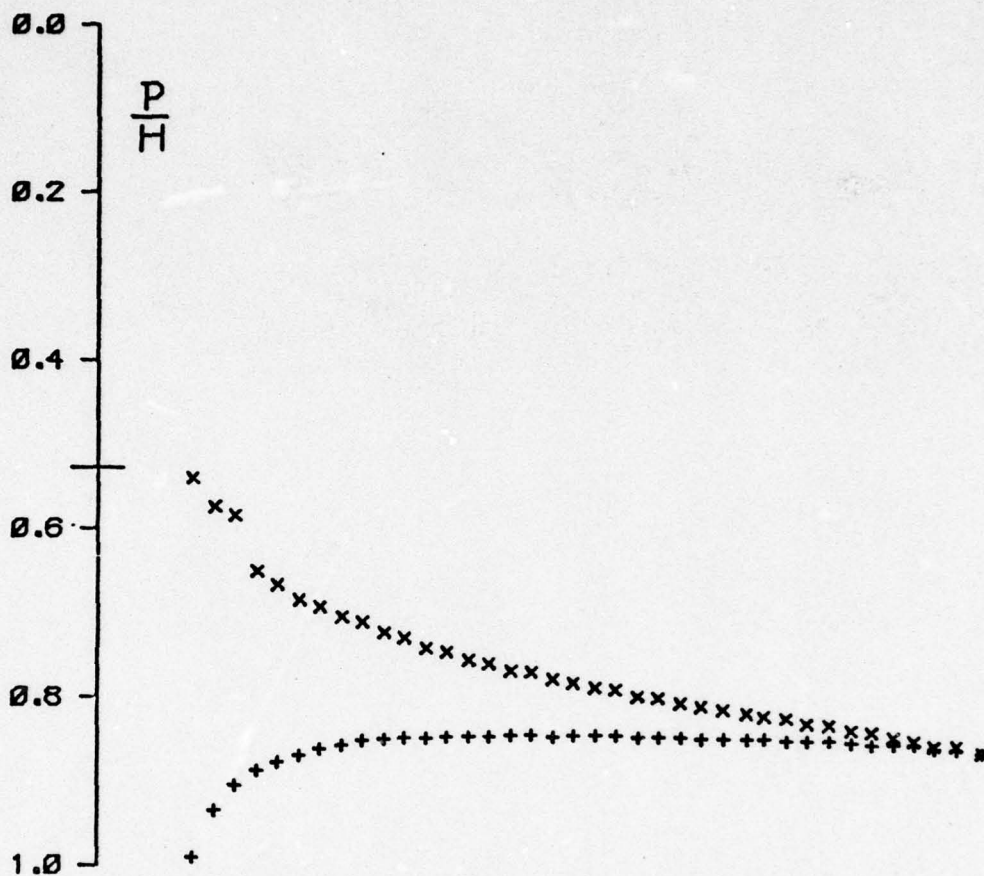
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .500$ $AL = 4.00$ $CN = 0.344$ $CM = 0.007$ $R = 0.853$



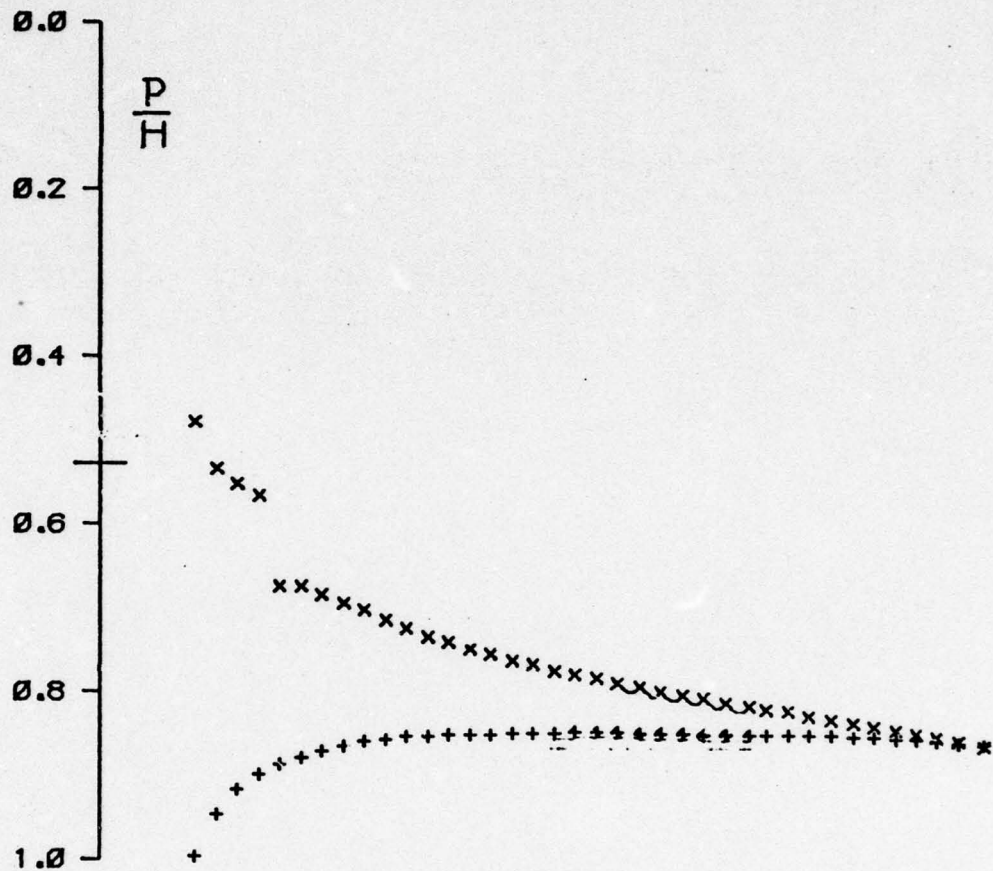
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .500$ $AL = 5.00$ $CN = 0.429$ $CN = 0.009$ $R = 0.853$



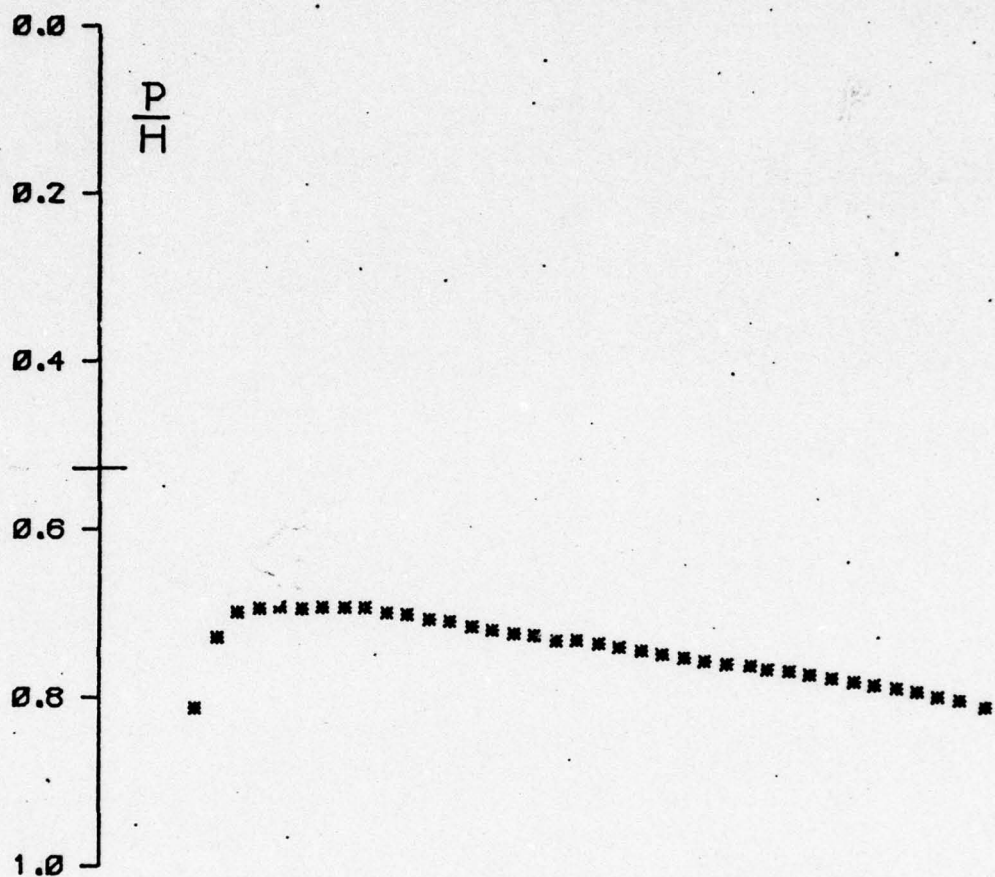
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M = .498 AL = 6.00 CN = 0.515 CM = 0.011 R = 0.850



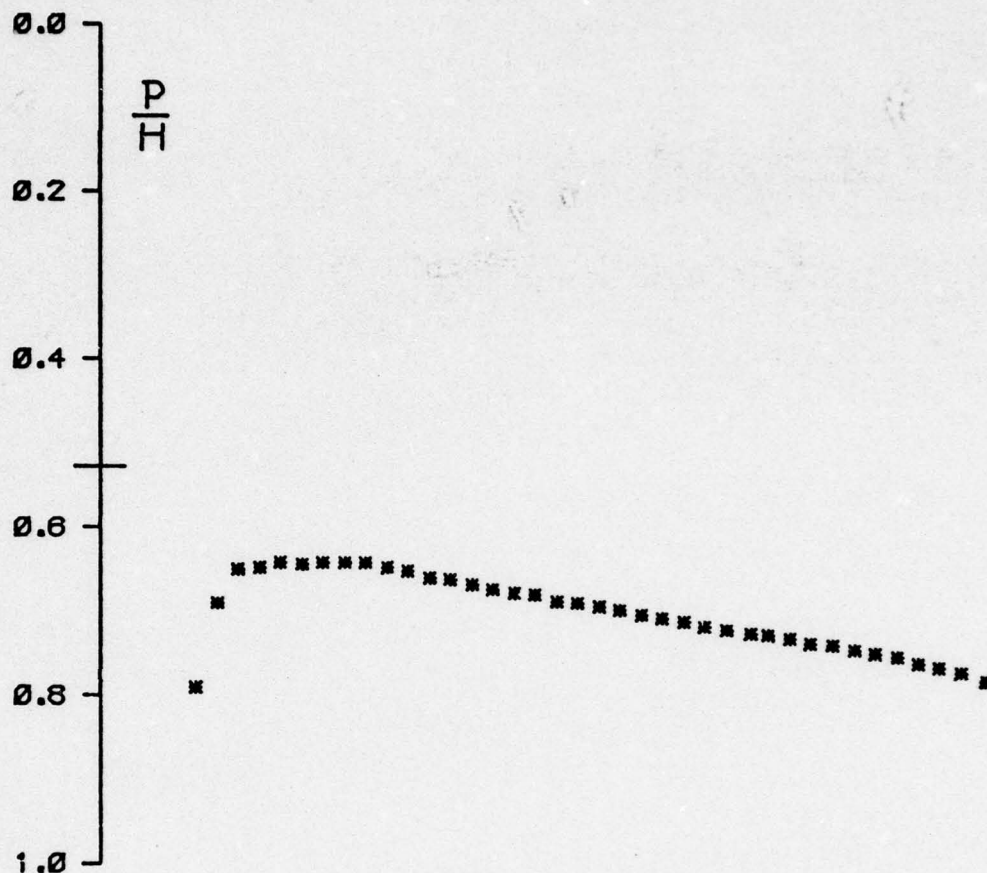
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .501 AL= 7.00 CN= 0.594 CM= 0.014 R= 0.857



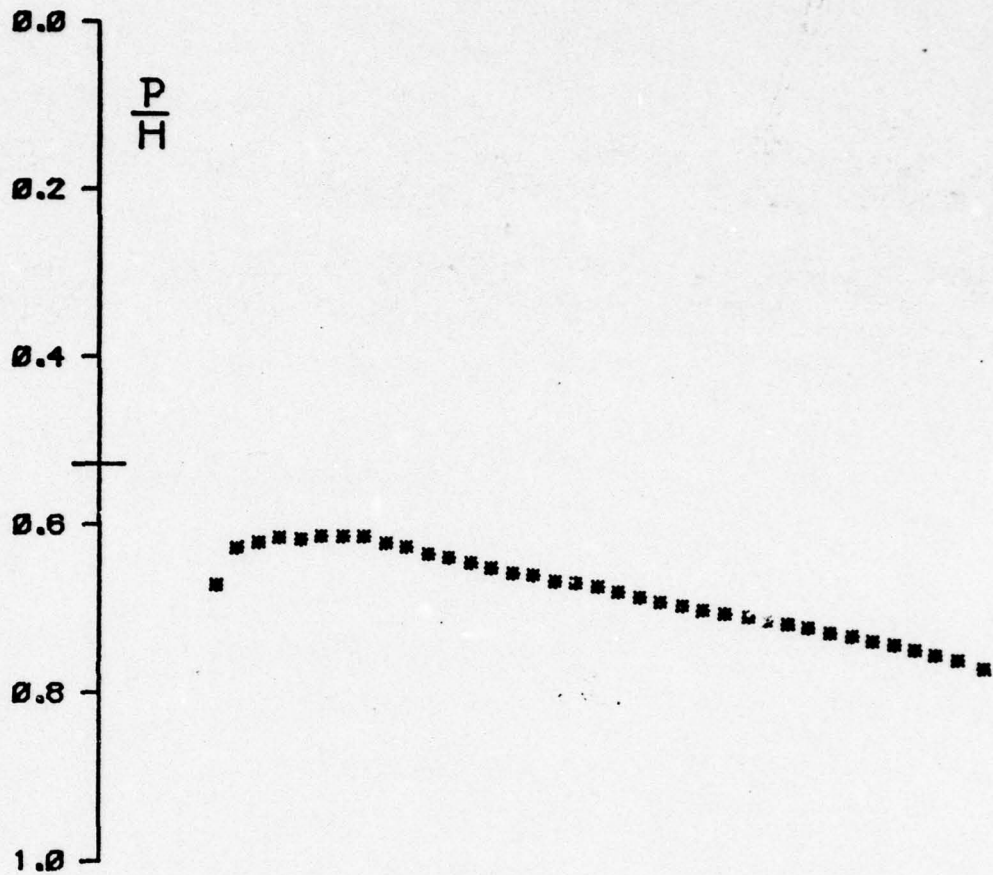
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .499$ $AL = 8.00$ $CN = 0.680$ $CM = 0.018$ $R = 0.857$



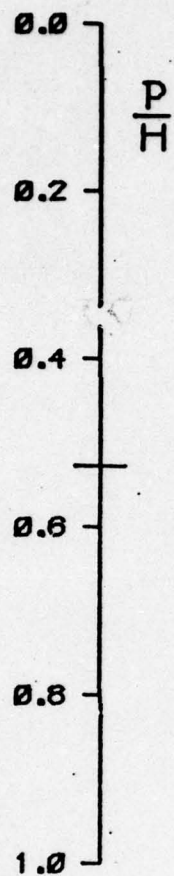
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .589 AL= 0.00 CN= 0.000 CM= 0.000 R= 0.357



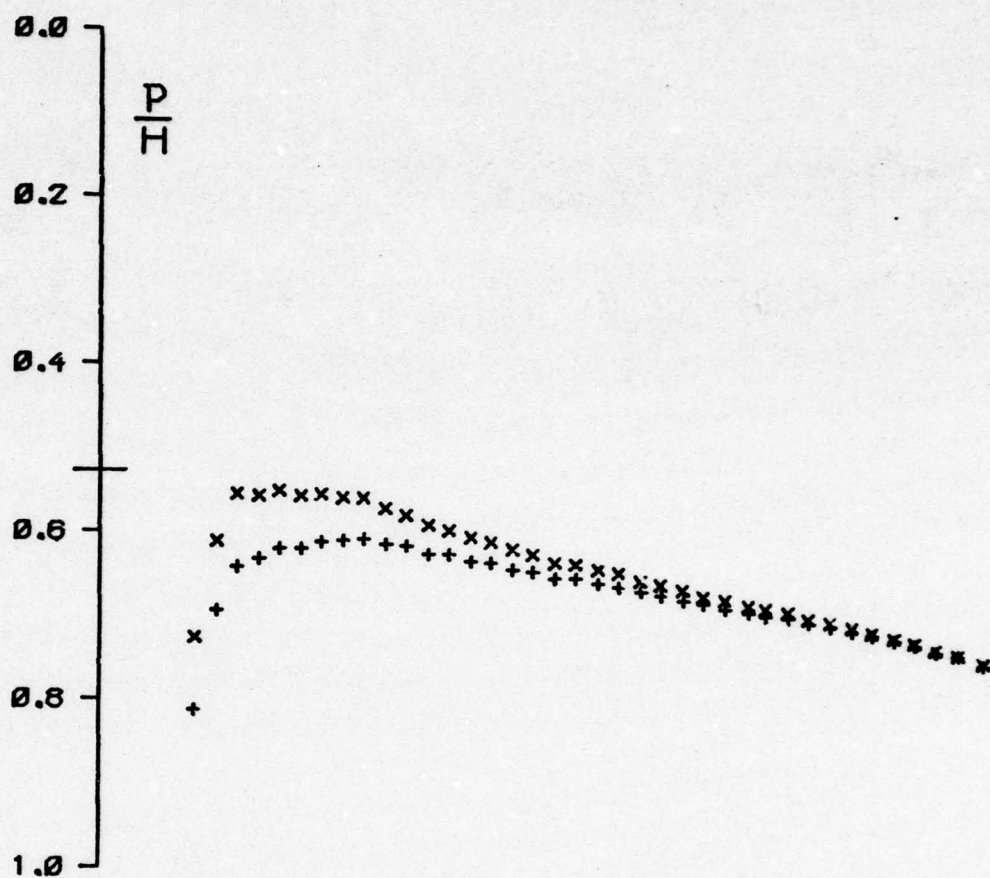
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .651 AL= 0.00 CN= 0.000 CM= 0.000 R= 0.832



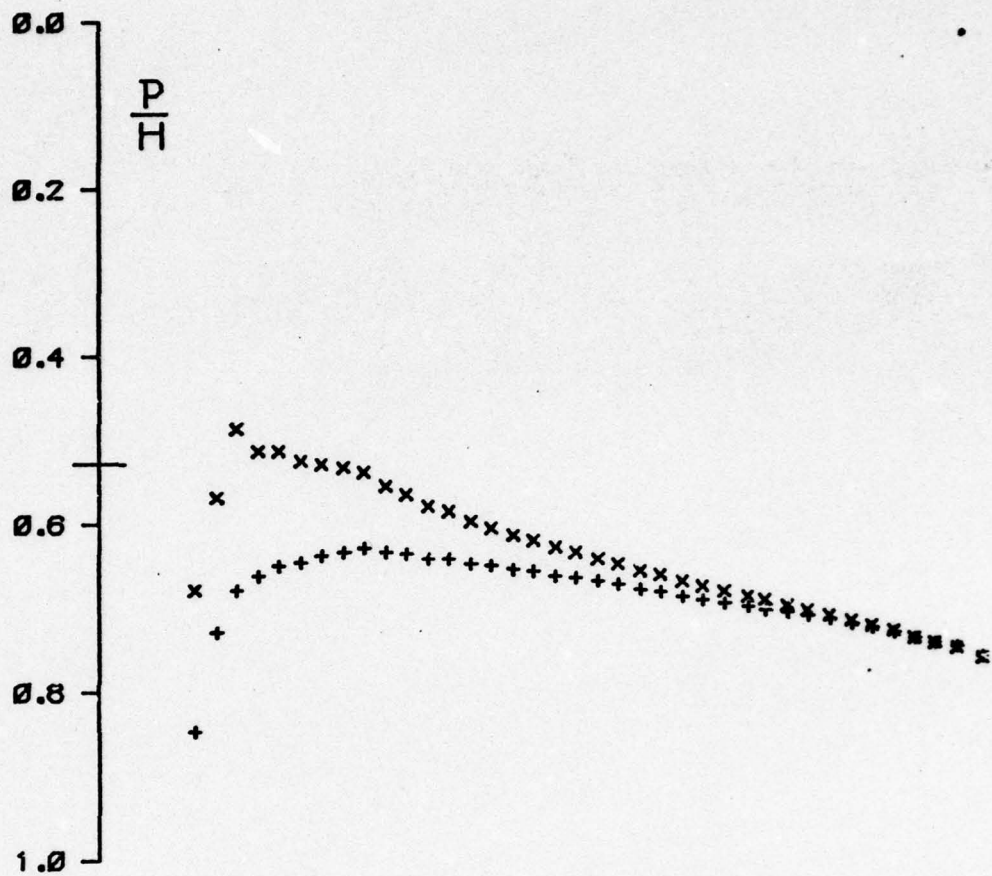
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .676 AL= 0.00 CN= 0.000 CM= 0.000 R= 0.857



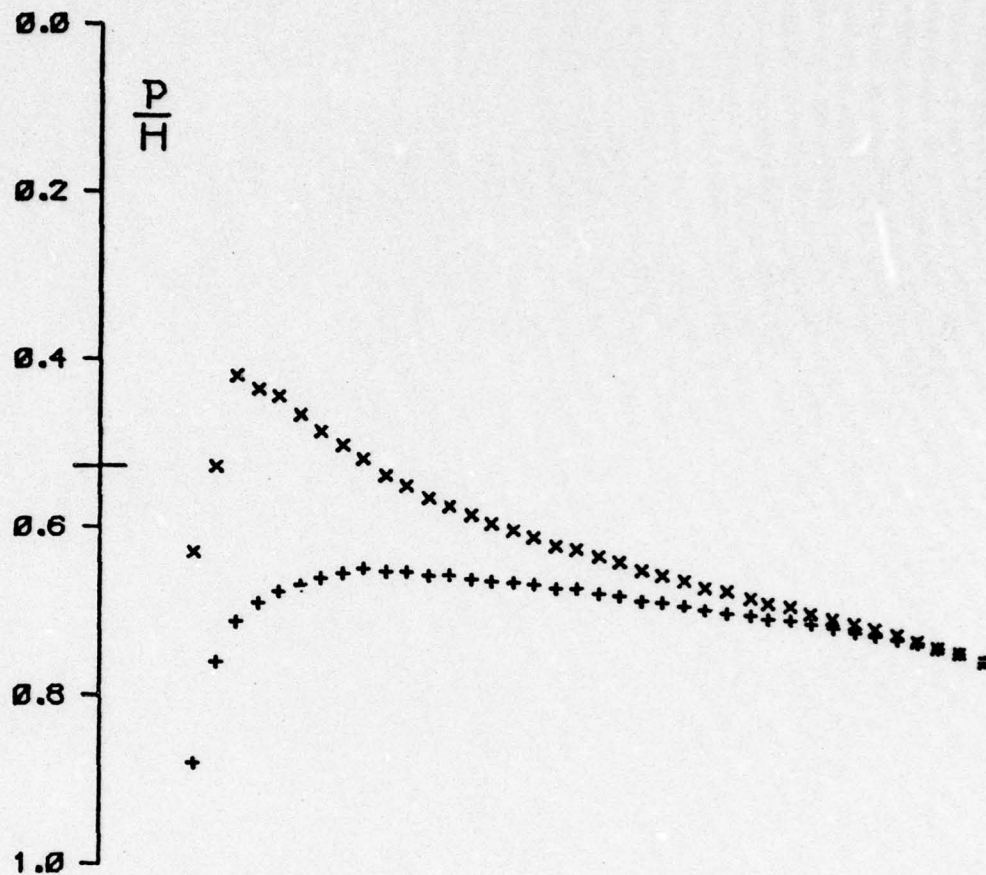
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .701 AL= 0.00 CN= 0.000 CM= 0.000 R= 0.834



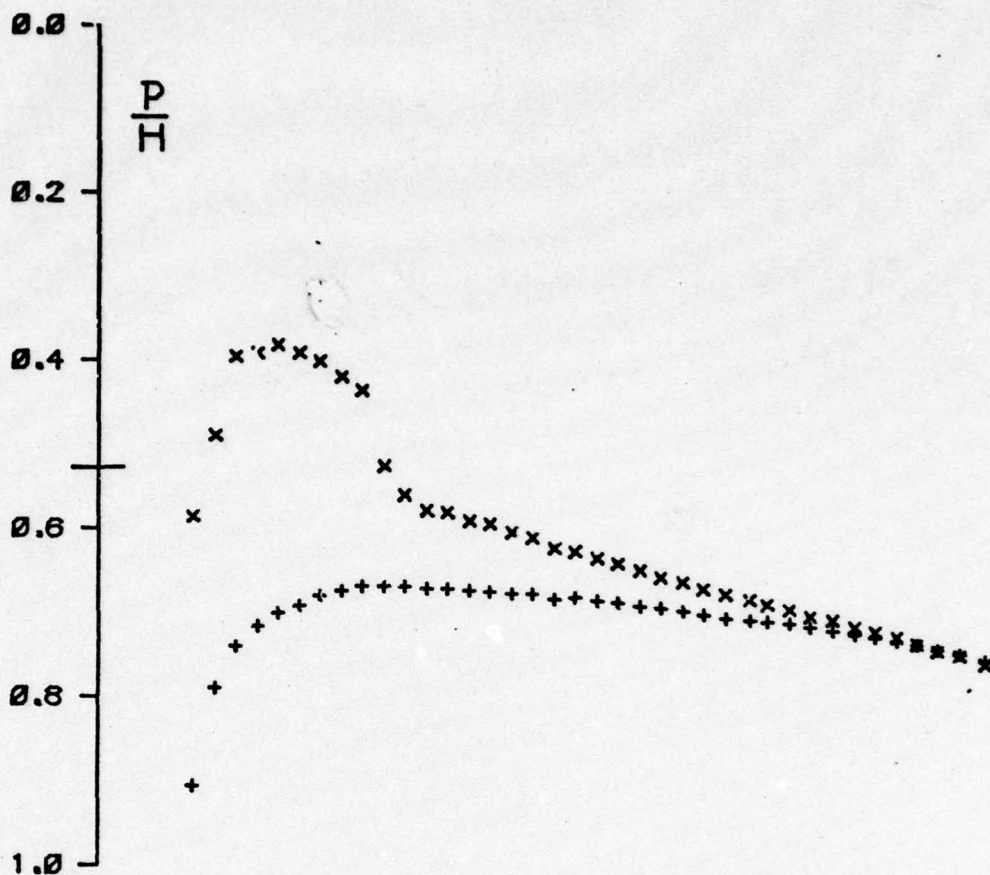
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .699$ $AL = 1.00$ $CN = 0.097$ $CM = 0.003$ $R = 0.835$



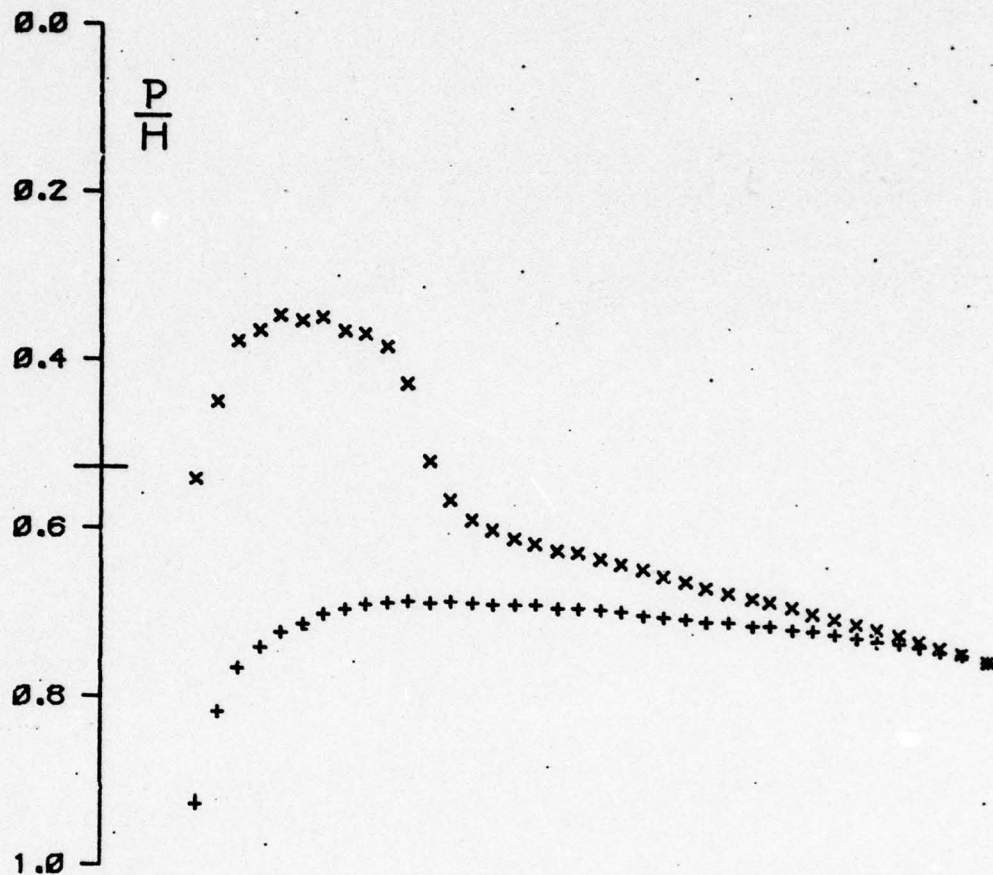
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .699$ $AL = 2.00$ $CN = 0.193$ $CM = 0.007$ $R = 0.835$



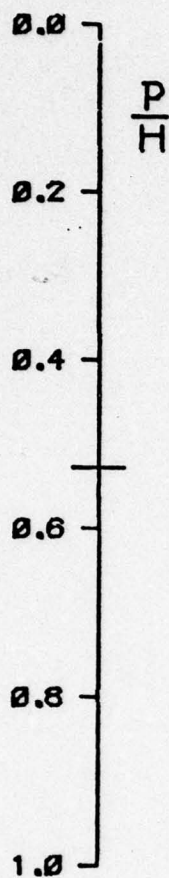
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .701$ $AL = 3.00$ $CN = 0.295$ $CM = 0.012$ $R = 0.837$



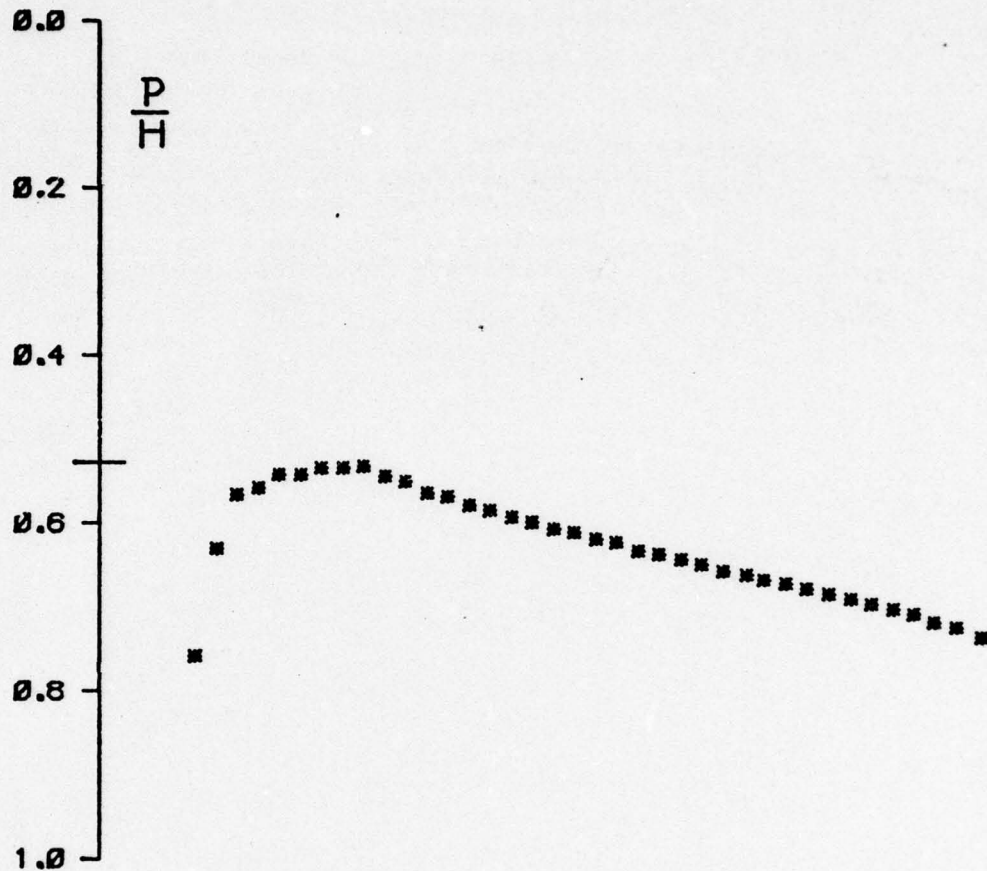
+ x NACA 0012 263.2 MM CHORD SLOTTED WALLS
 $M = .701$ $AL = 4.00$ $CN = 0.393$ $CN = 0.017$ $R = 0.837$



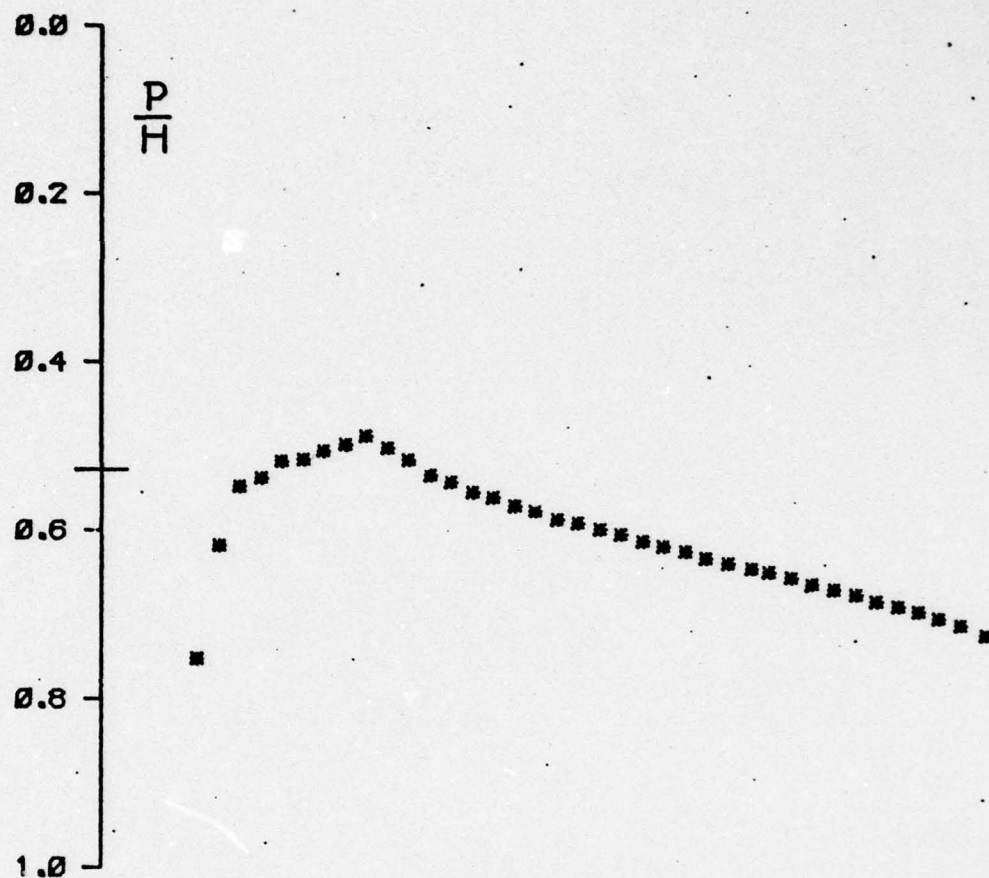
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .699$ $AL = 5.00$ $CN = 0.499$ $CM = 0.020$ $R = 0.835$



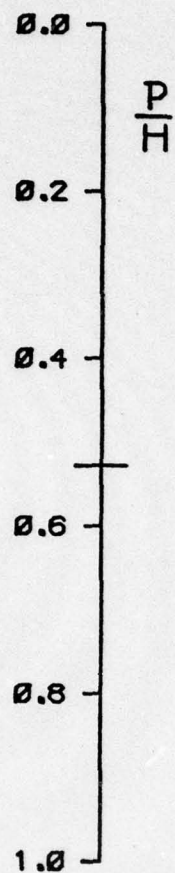
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .721 AL= 0.00 CN= 0.000 CM= 0.020 R= 0.848



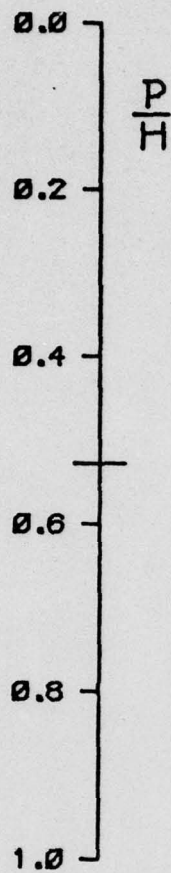
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .739 AL= 0.00 CN= 0.000 CM= 0.000 R= 0.865



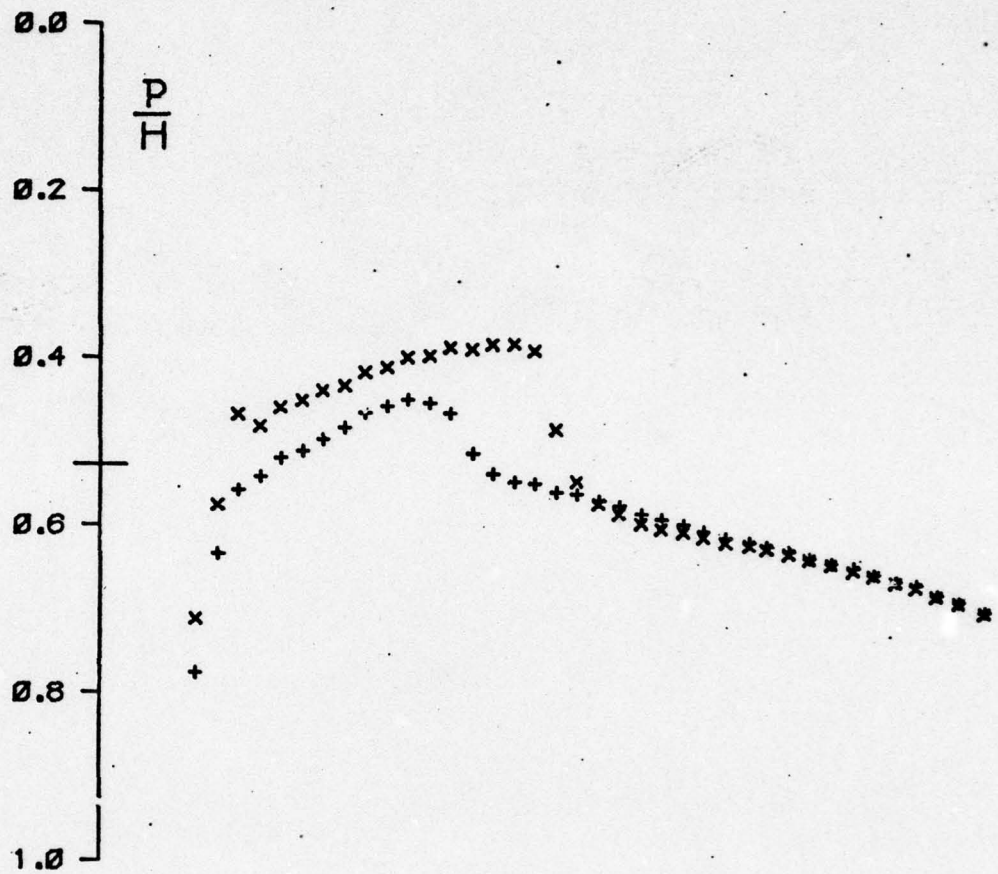
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .760 AL= 0.00 CN= 0.000 CM= 0.000 R= 0.829



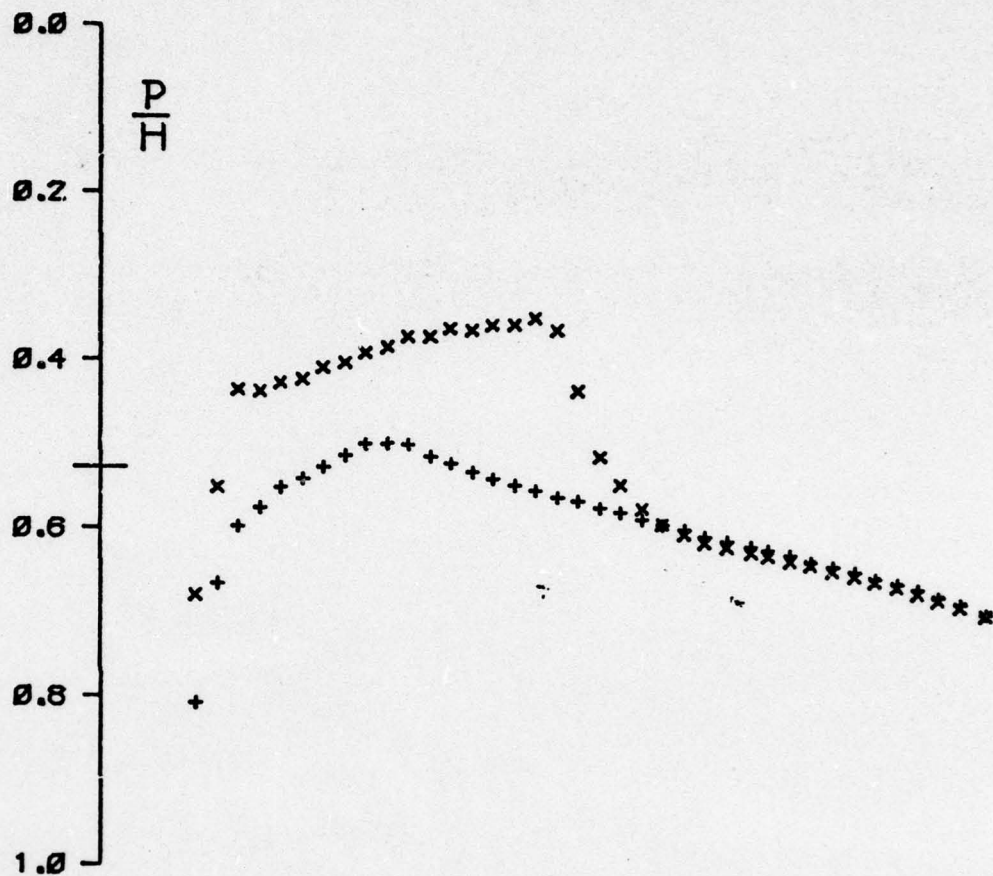
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M = .781 AL = 0.00 CN = 0.000 CM = 0.000 P = 0.839



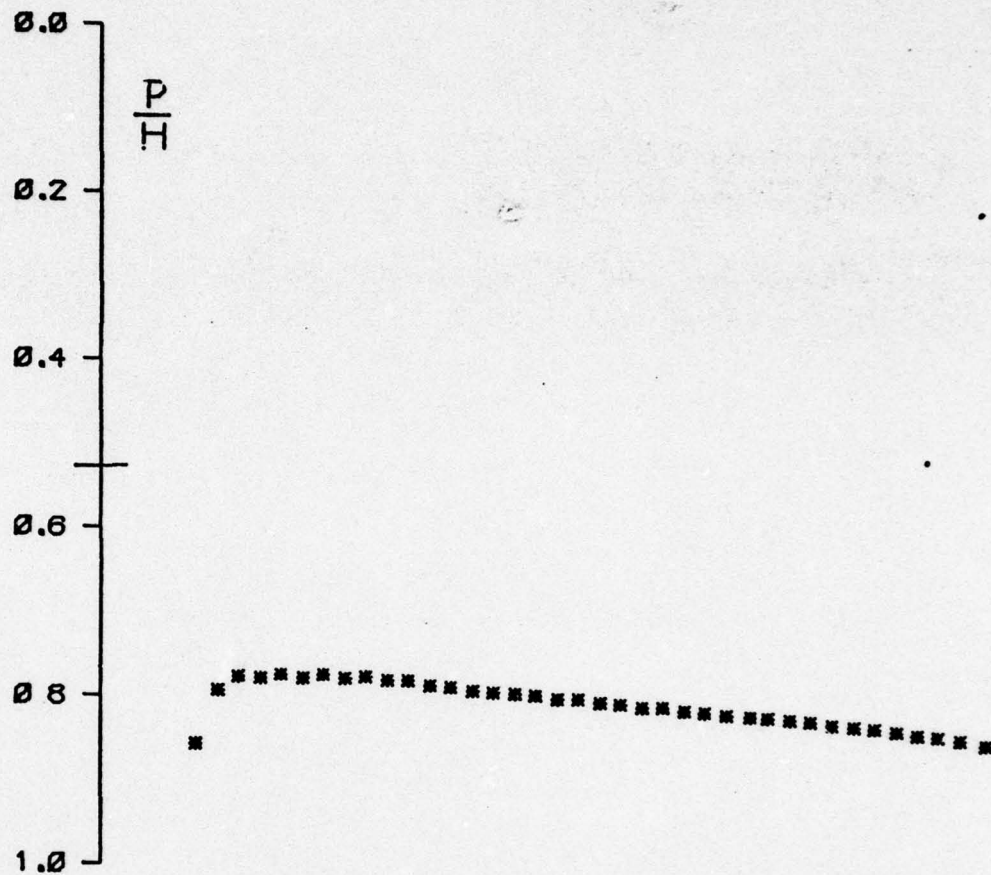
+ X NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .800 AL= 0.00 CN= 0.000 CM= 0.000 R= 0.835



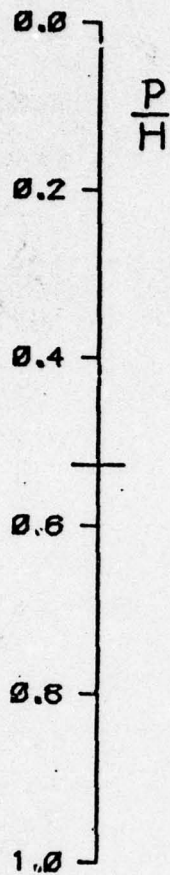
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .799$ $AL = 1.00$ $CN = 0.123$ $CM = 0.000$ $R = 0.835$



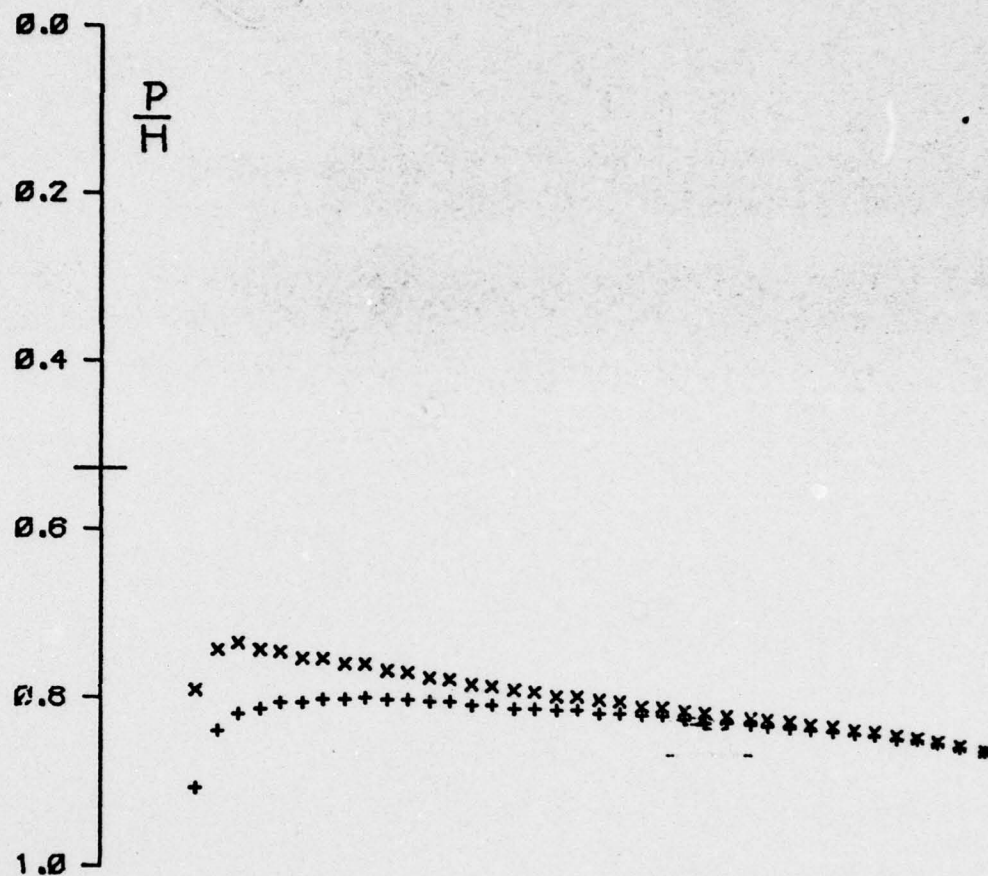
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .800$ $AL = 2.00$ $CN = 0.238$ $CN = -0.001$ $R = 0.835$



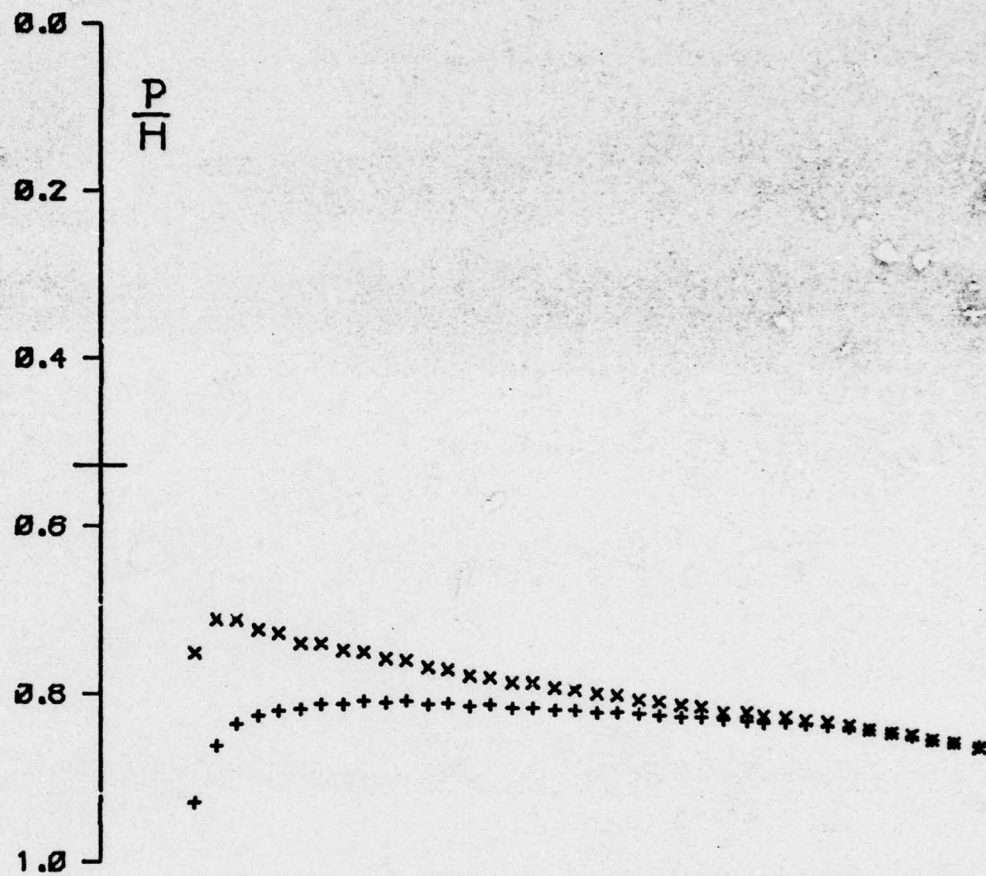
+ X NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .500$ $A_1 = 0.00$ $C_N = 0.000$ $C_M = 0.000$ $R = 0.434$



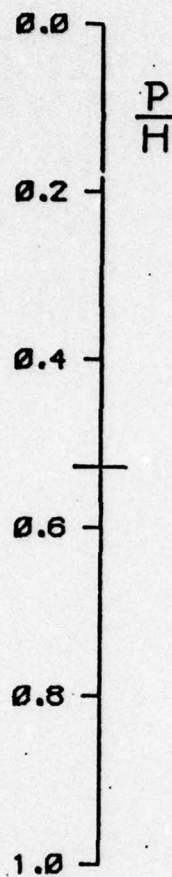
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .493$ $AL = 1.00$ $CN = 0.077$ $CM = 0.003$ $R = 0.434$



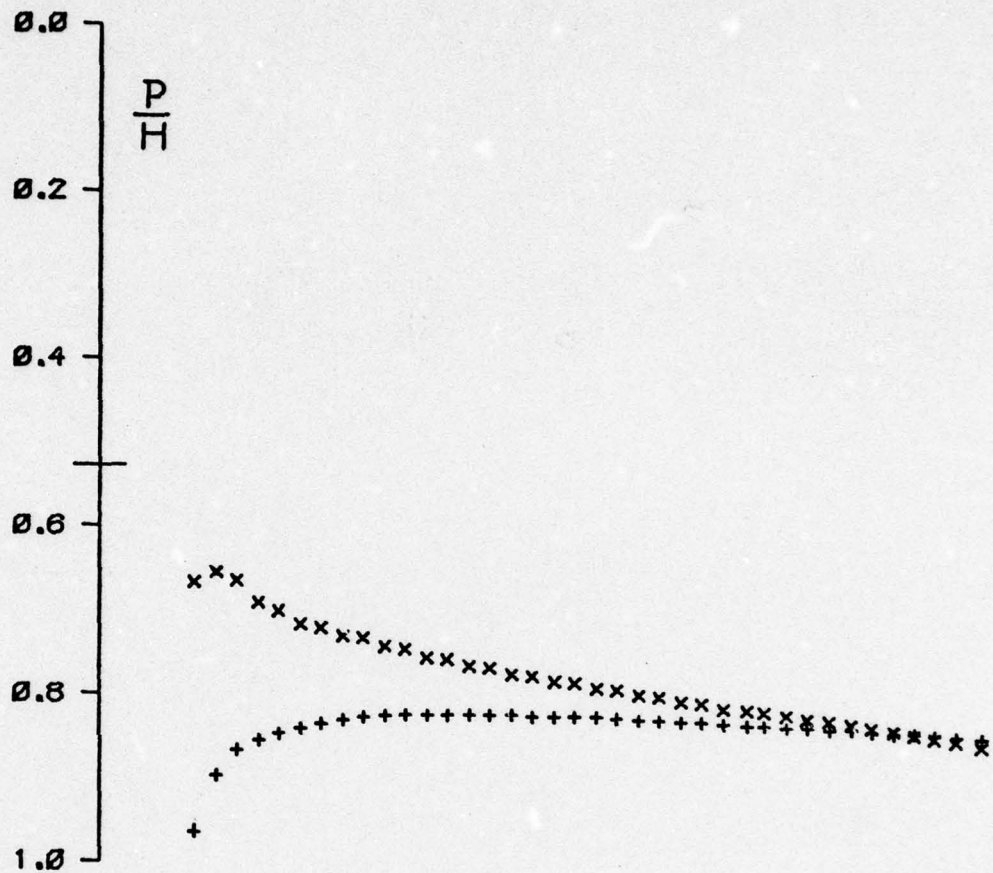
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .499$ $AL = 2.00$ $CN = 0.157$ $CM = 0.006$ $R = 0.437$



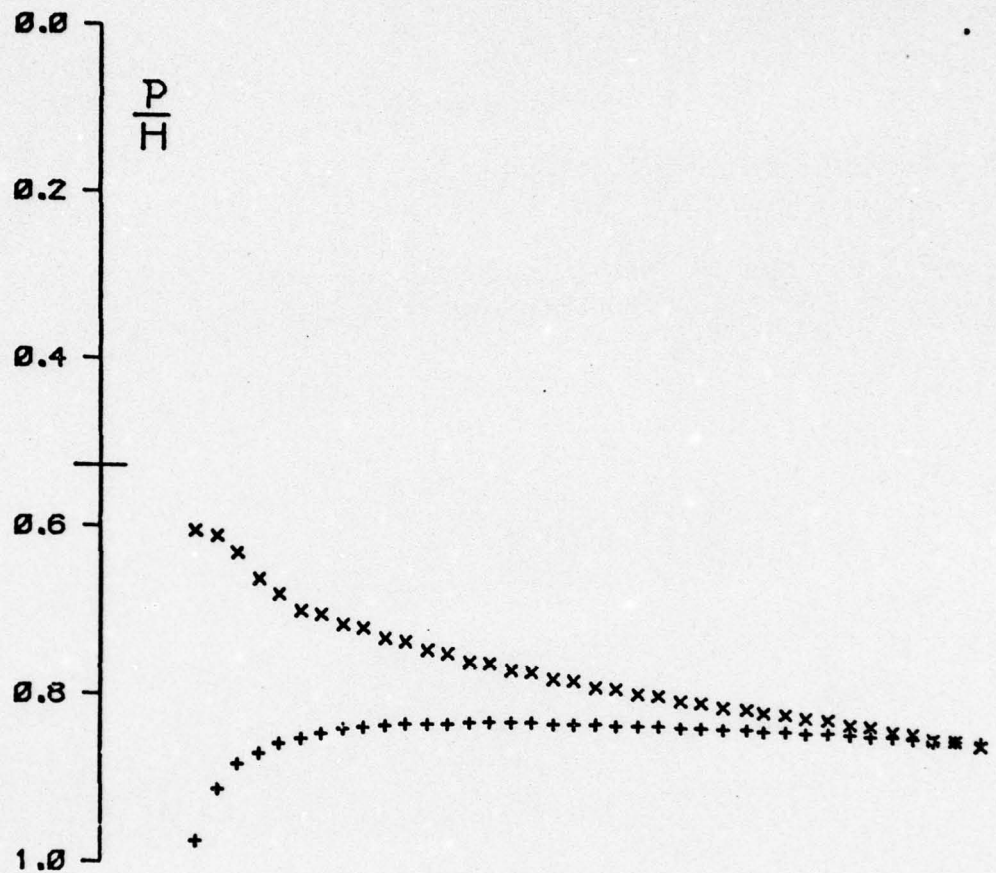
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .502$ $AL = 3.00$ $CN = 0.234$ $CM = 0.011$ $R = 0.440$



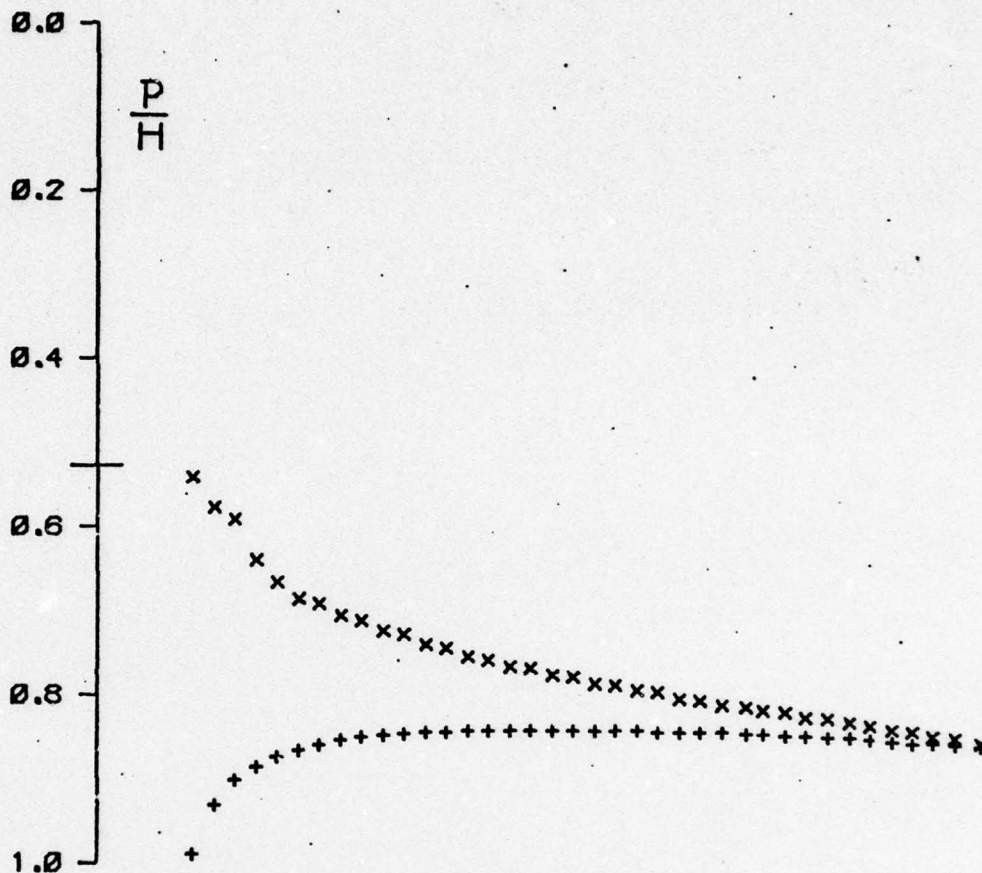
+ X NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .501$ $AL = 4.00$ $CN = 0.309$ $CM = 0.015$ $R = 0.440$



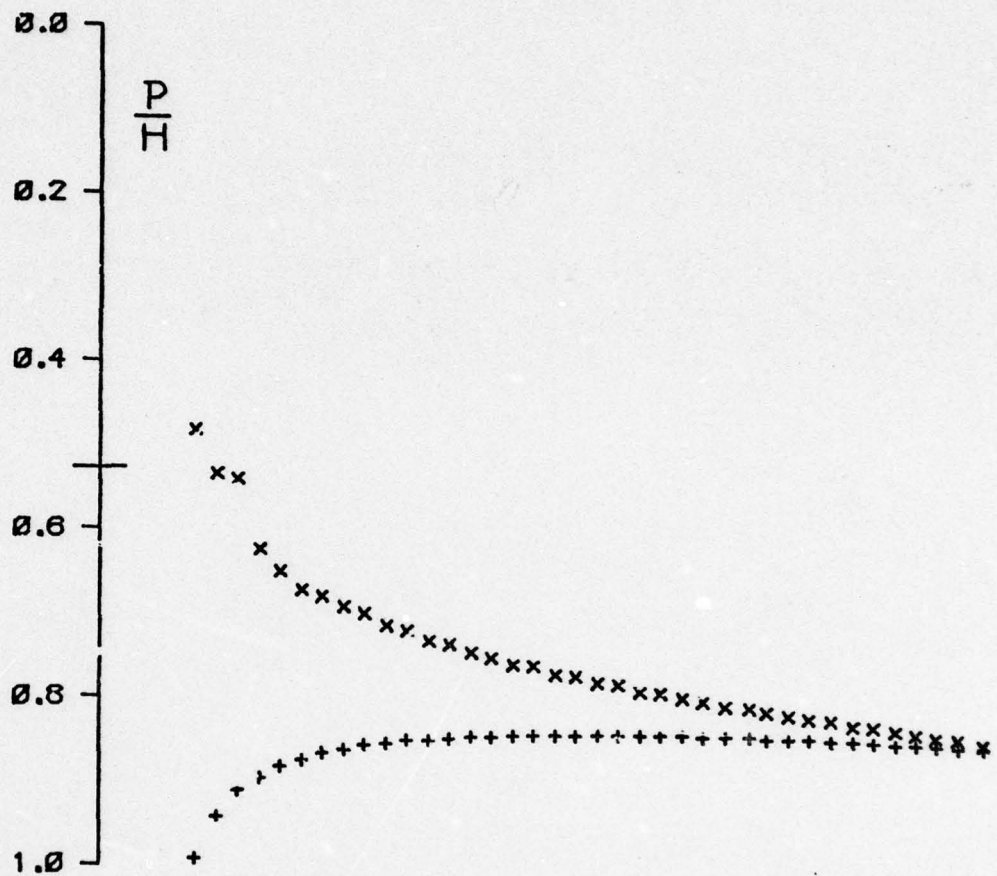
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .499$ $AL = 5.00$ $CN = 0.383$ $CM = 0.018$ $R = 0.440$



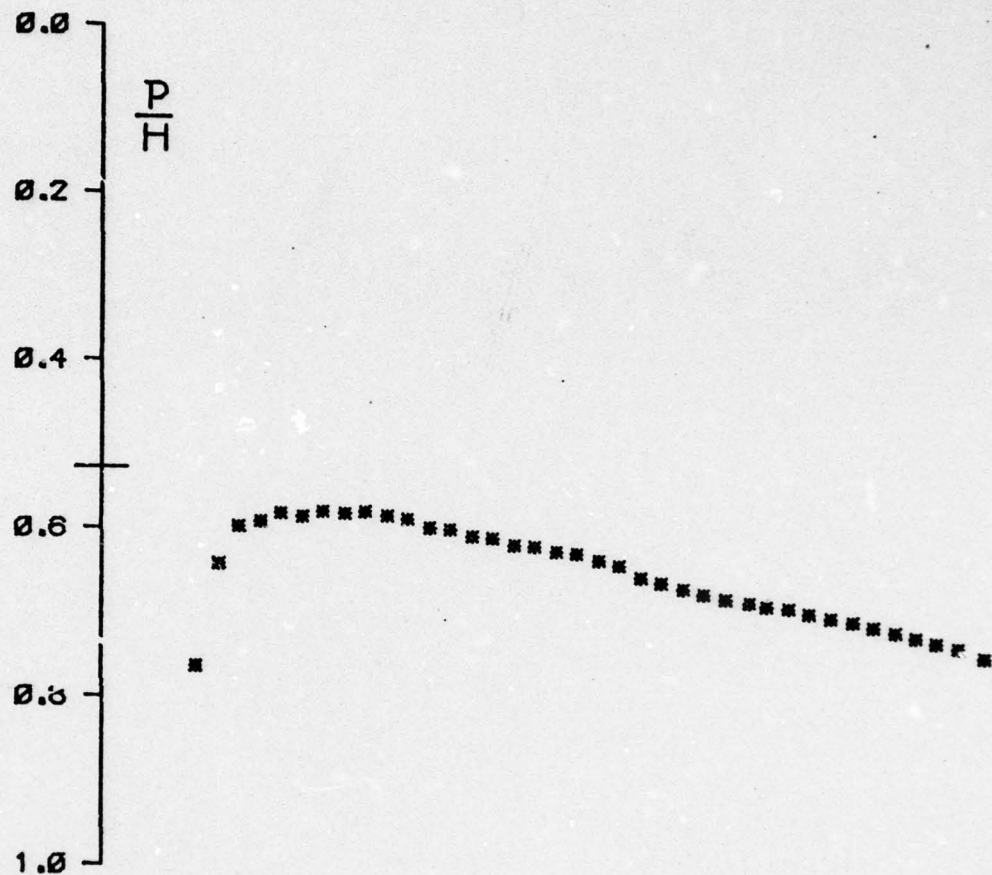
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .499 AL= 6.00 CN= 0.437 CM= 0.014 R= 0.440



+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .499 AL= 7.00 CN= 0.606 CM= 0.008 R= 0.440



+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .501 AL= 8.00 CN= 0.671 CM= 0.014 R= 0.444



+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .701 AL= 0.00 CN= 0.000 CM= 0.000 R= 0.427

AD-A055 004

AERONAUTICAL RESEARCH LABS MELBOURNE (AUSTRALIA)
TRANSONIC WIND TUNNEL TESTS ON A NACA 0012 AEROFOIL, (U)
APR 77 N POLLOCK, B D FAIRLIE
ARL/AERO-148

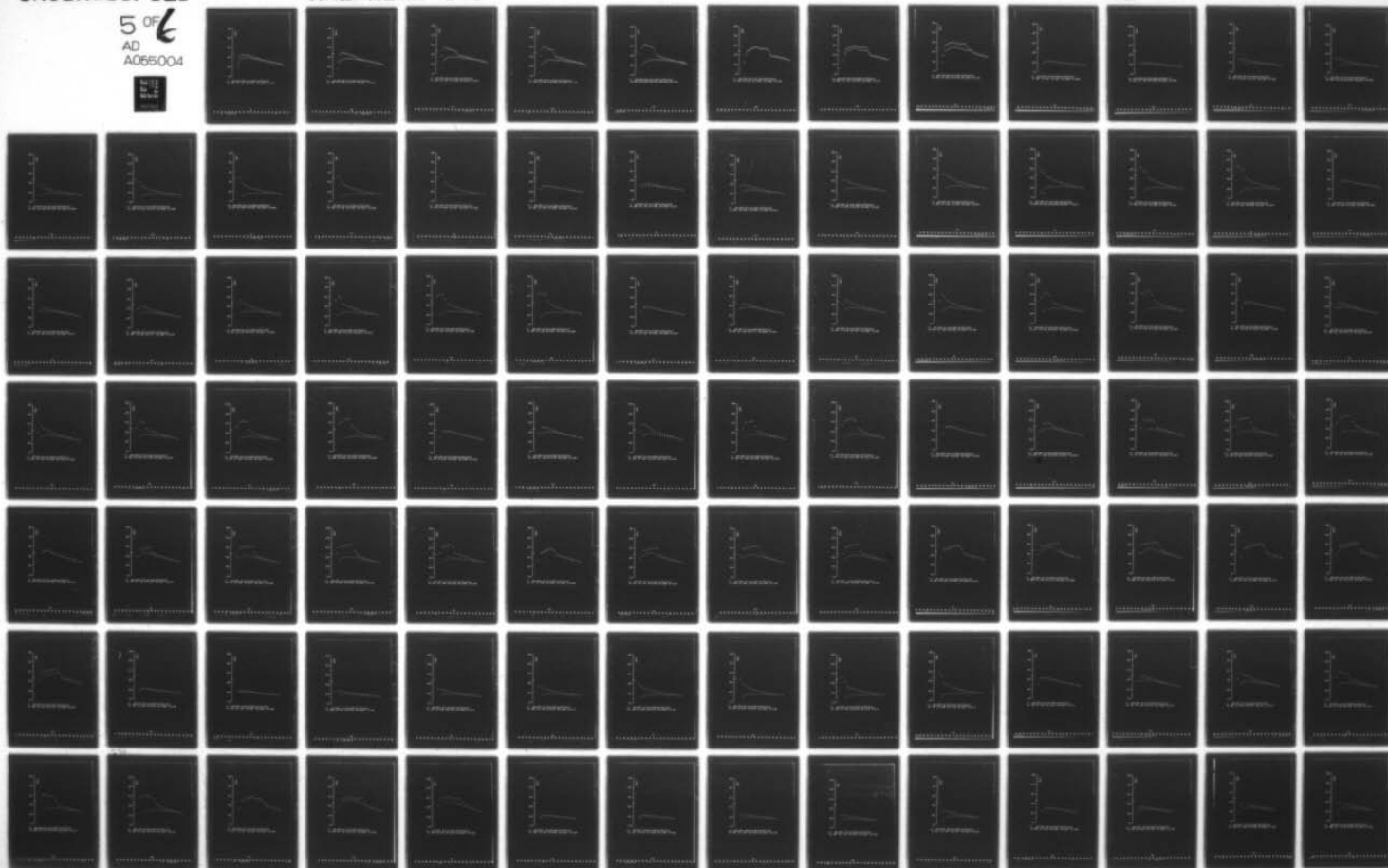
F/G 20/4

UNCLASSIFIED

5 OF 6
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A055004



NI

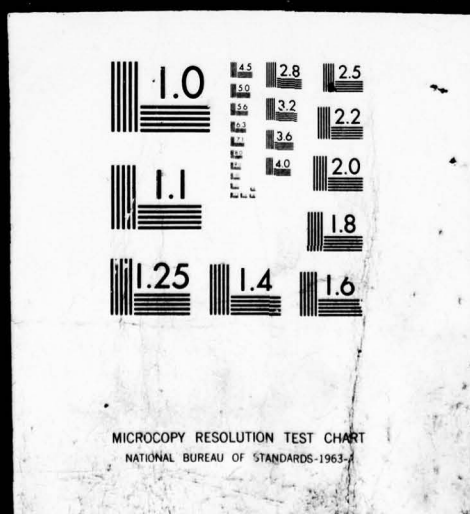


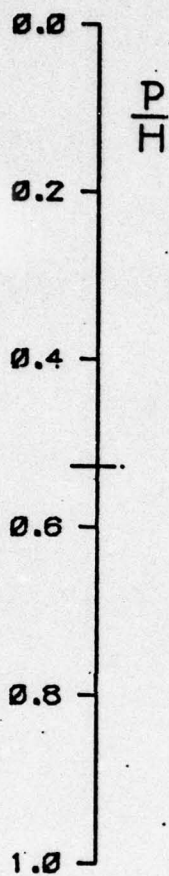
SIFIED

5 OF 6

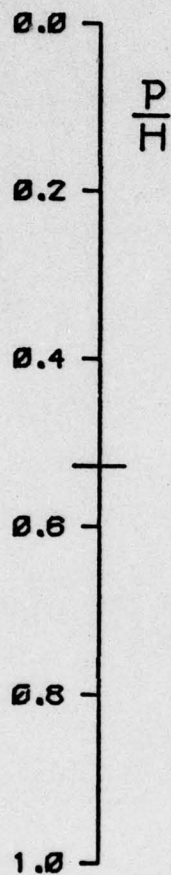
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A055004

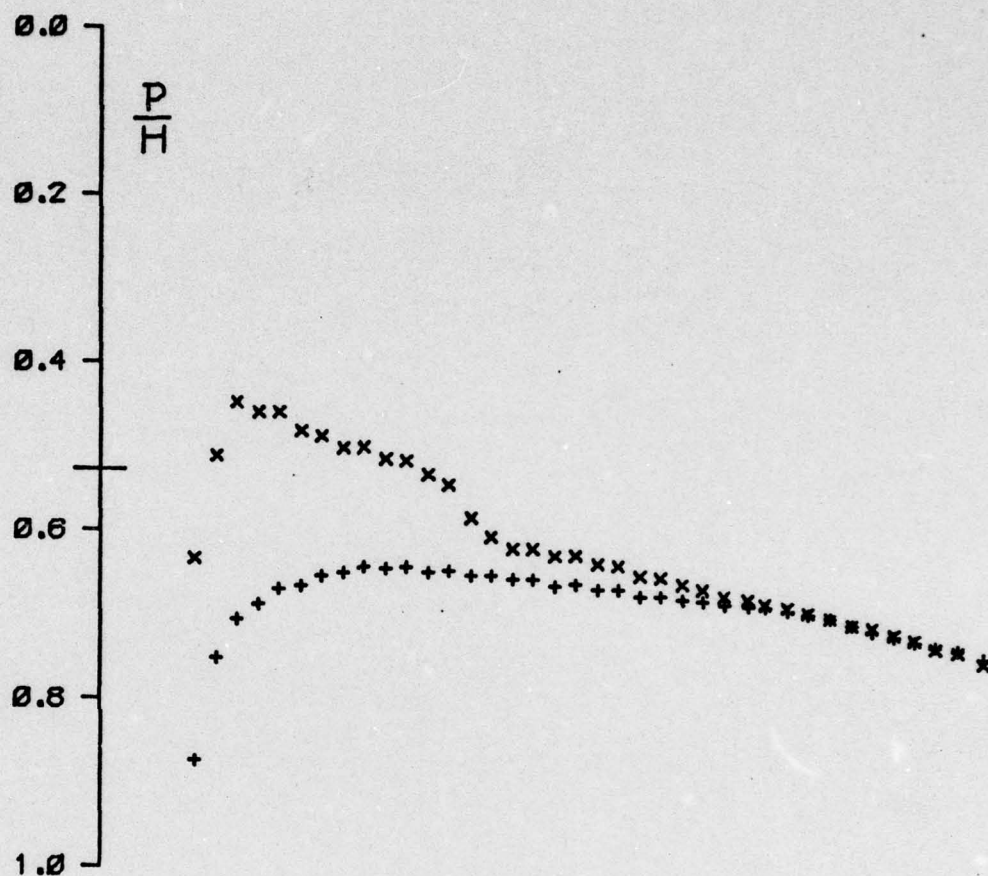




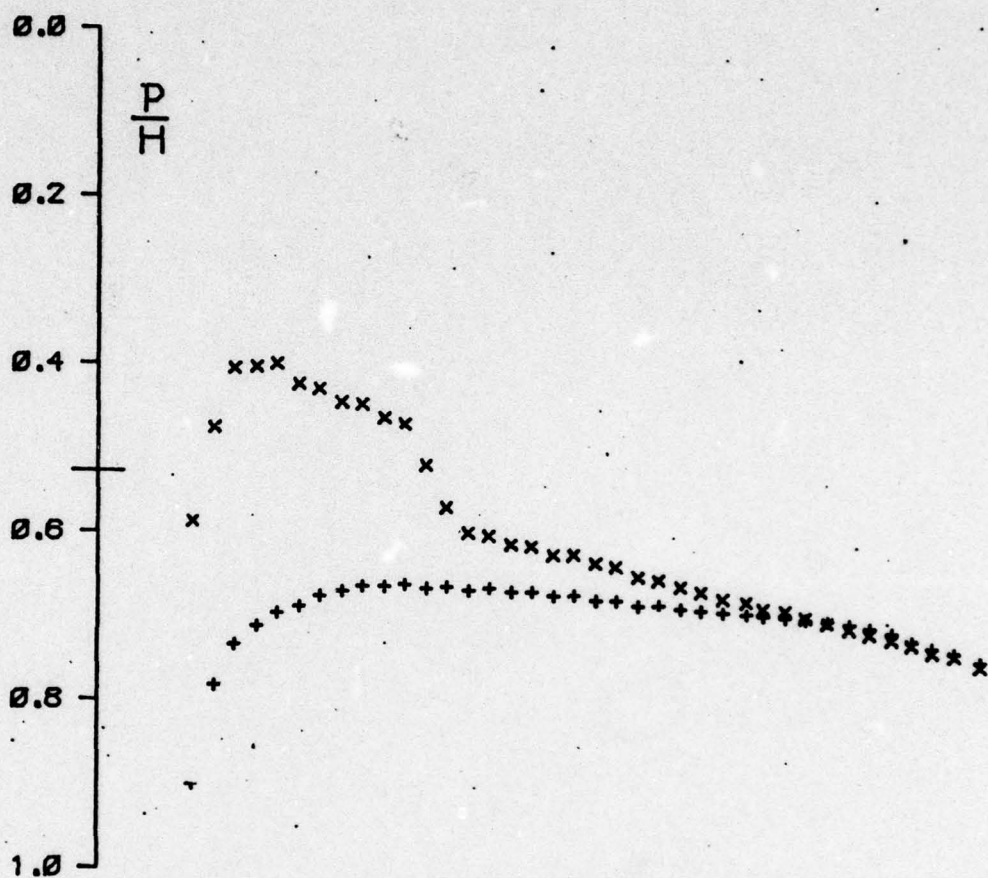
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .700$ $AL = 1.00$ $CN = 0.091$ $CM = 0.005$ $R = 0.429$



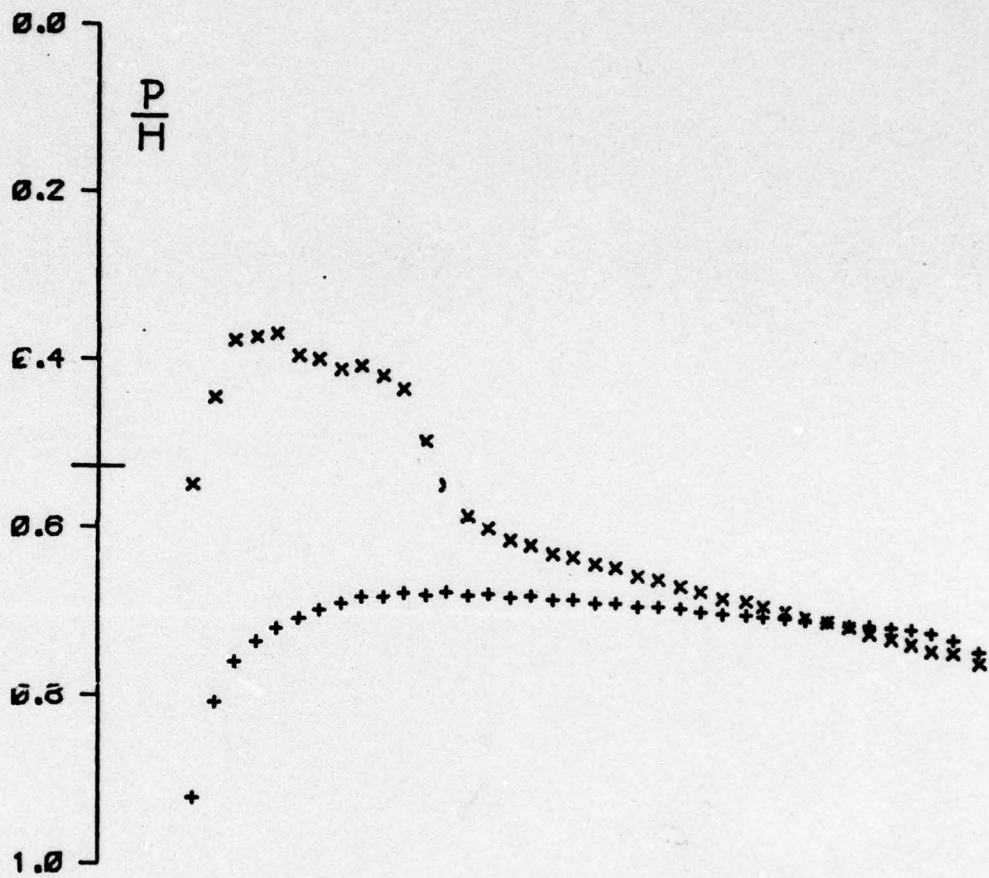
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .700$ $AL = 2.00$ $CN = 0.177$ $CM = 0.011$ $R = 0.431$



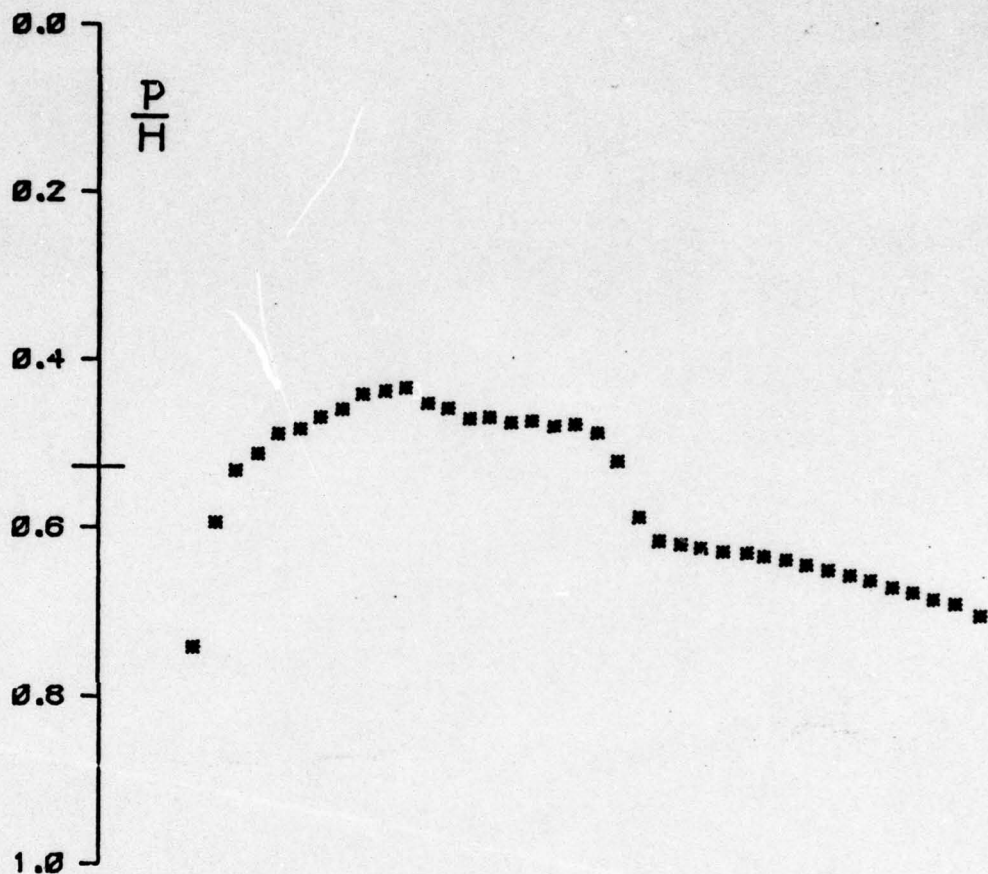
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .700$ $AL = 3.00$ $CN = 0.269$ $CM = 0.018$ $R = 0.434$



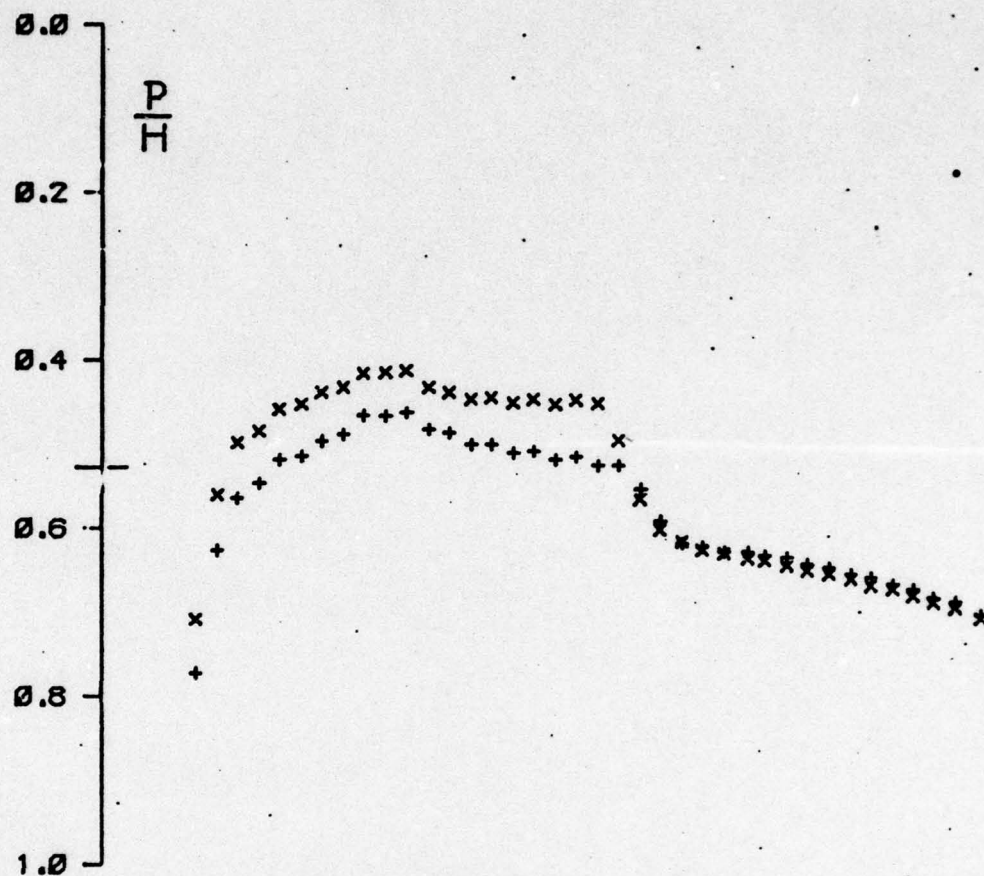
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
M= .702 AL= 4.00 CN= 0.360 CM= 0.025 R= 0.434



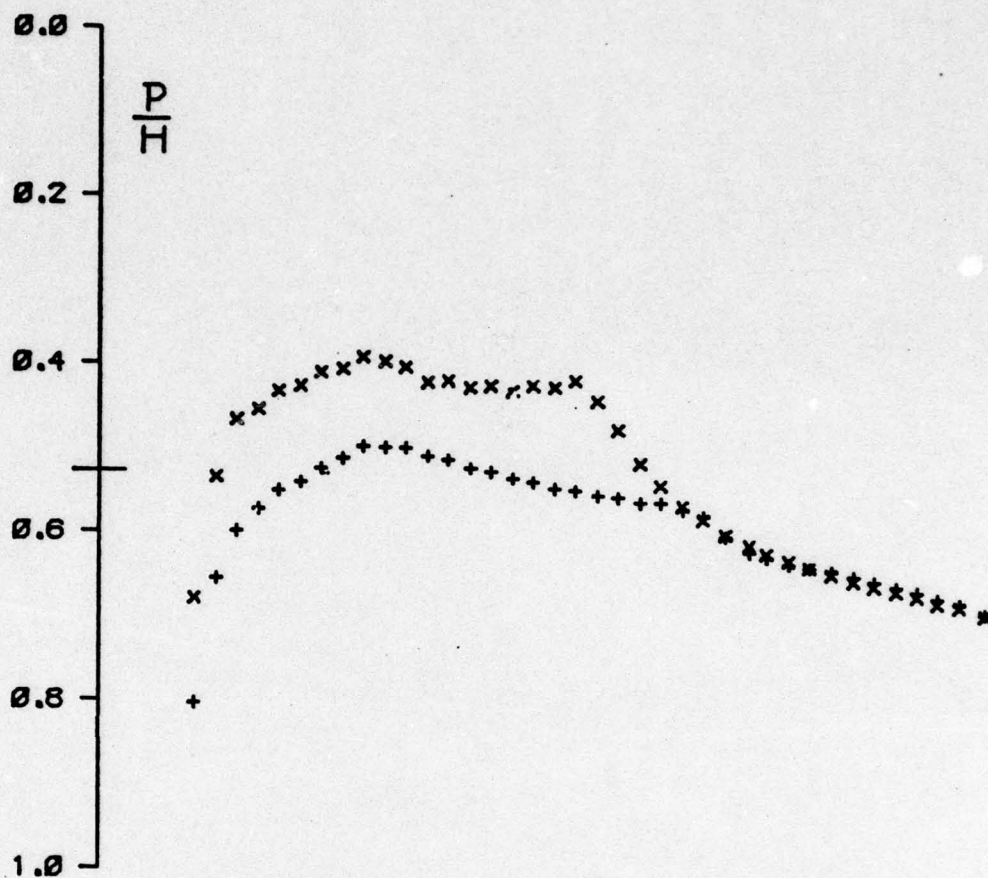
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = 0.698$ $Al = 5.00$ $CN = 0.445$ $CN = 0.031$ $R = 0.434$



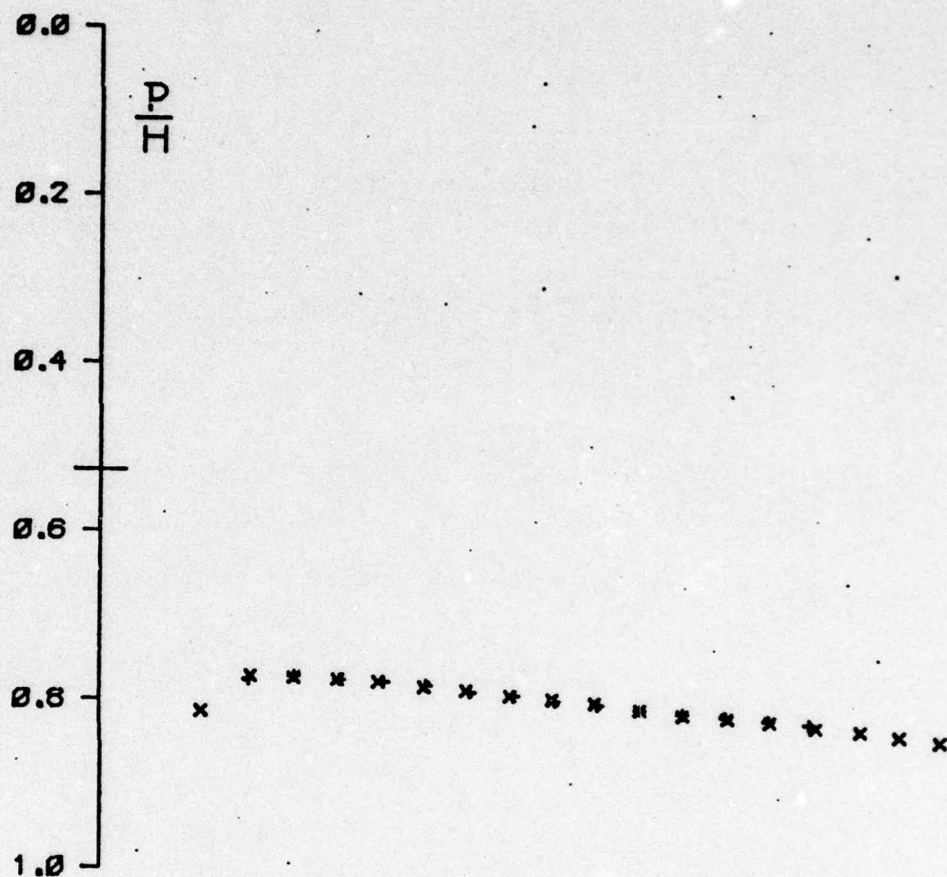
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 M= .800 AL= 0.00 CN= 0.000 CM= 0.000 R= 0.439



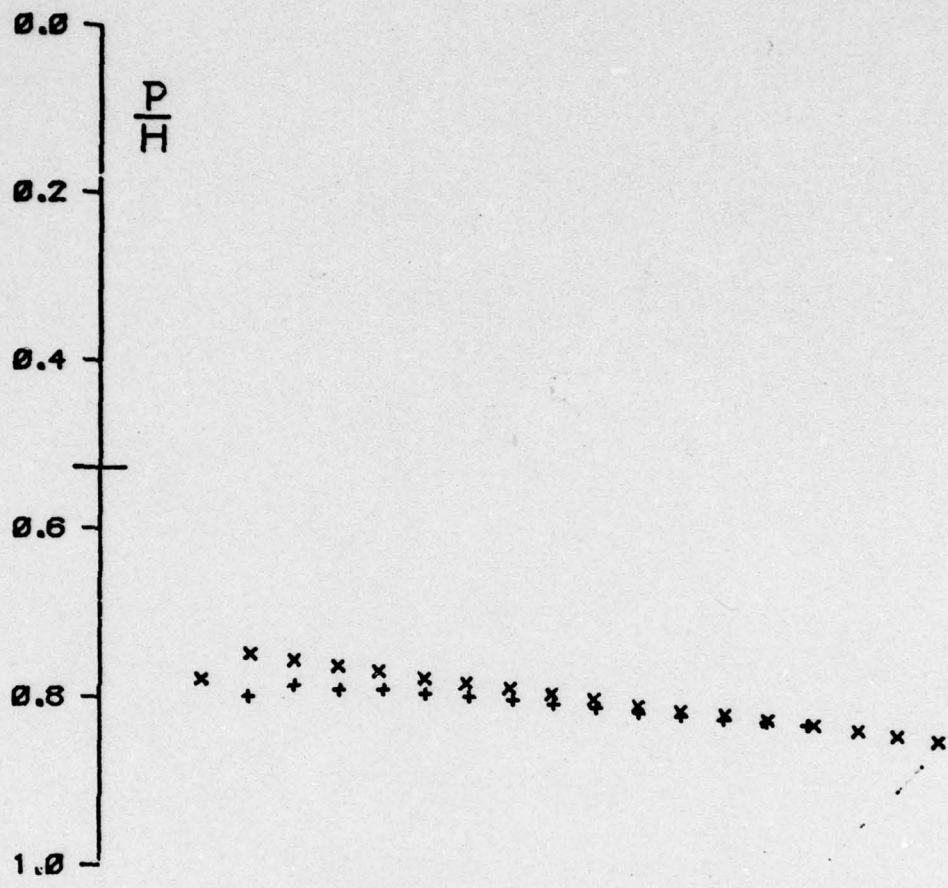
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .620$ $AL = 1.00$ $CN = 0.093$ $CM = 0.002$ $R = 0.442$



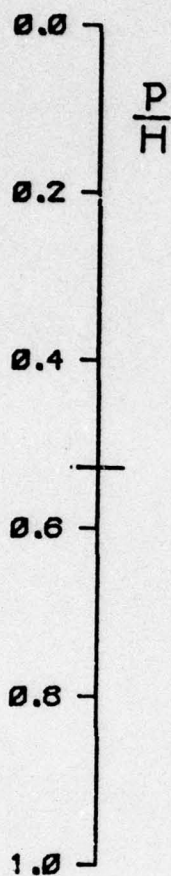
+ x NACA 0012 203.2 MM CHORD SLOTTED WALLS
 $M = .802$ $AL = 2.00$ $CN = 0.188$ $CM = 0.001$ $R = 0.442$



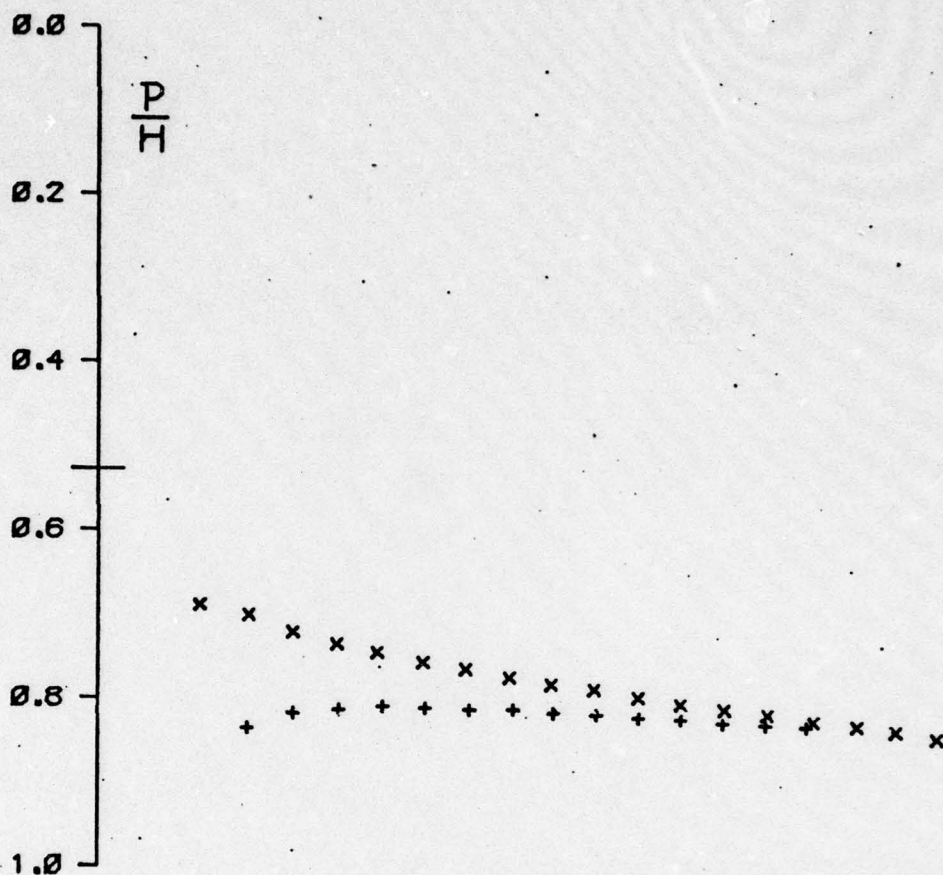
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .506$ $AL = 0.00$ $CN = 0.021$ $CM = 0.004$ $R = 0.863$



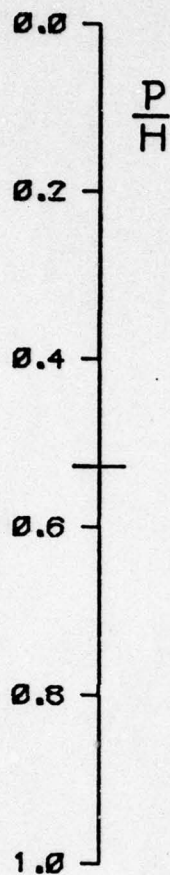
+ x NACA 0012 101.6 MM CHORD SLOTTED W. 'LS
 $M = .501$ $AL = 1.00$ $CN = 0.094$ $CM = 0.001$ $R = 0.866$



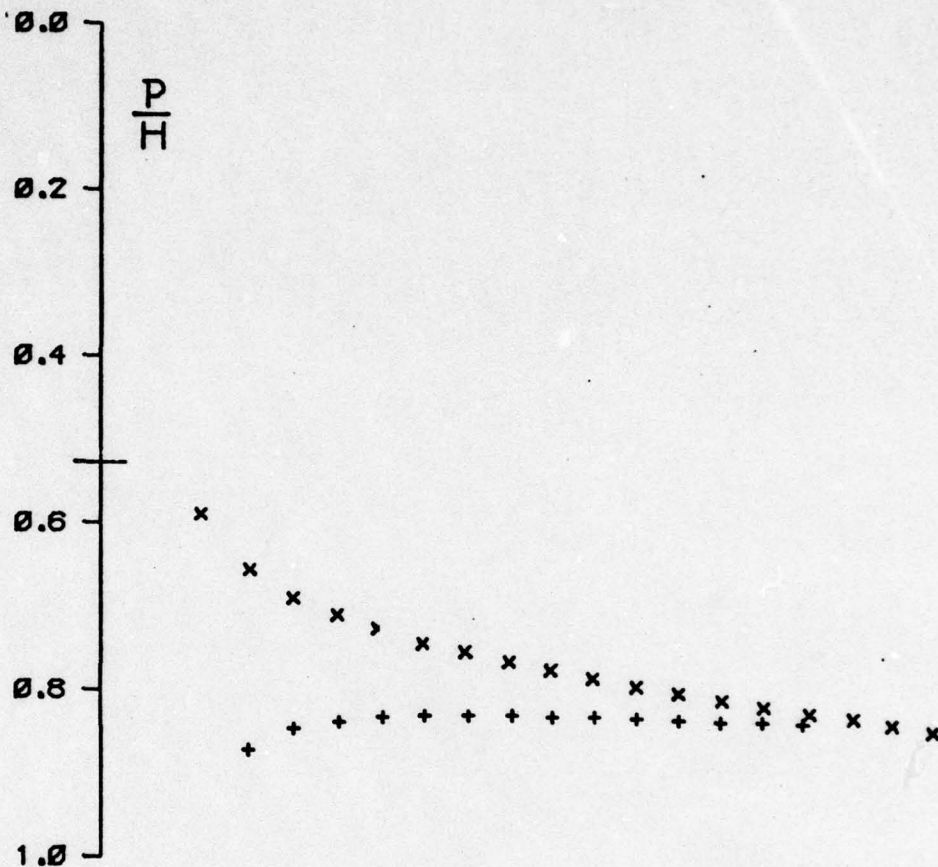
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .499$ $AL = 2.00$ $CN = 0.193$ $CM = 0.003$ $R = 0.863$



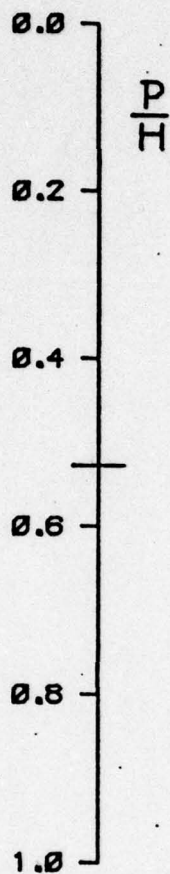
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .500$ $AL = 3.00$ $CN = 0.291$ $CME = 0.004$ $R = 0.863$



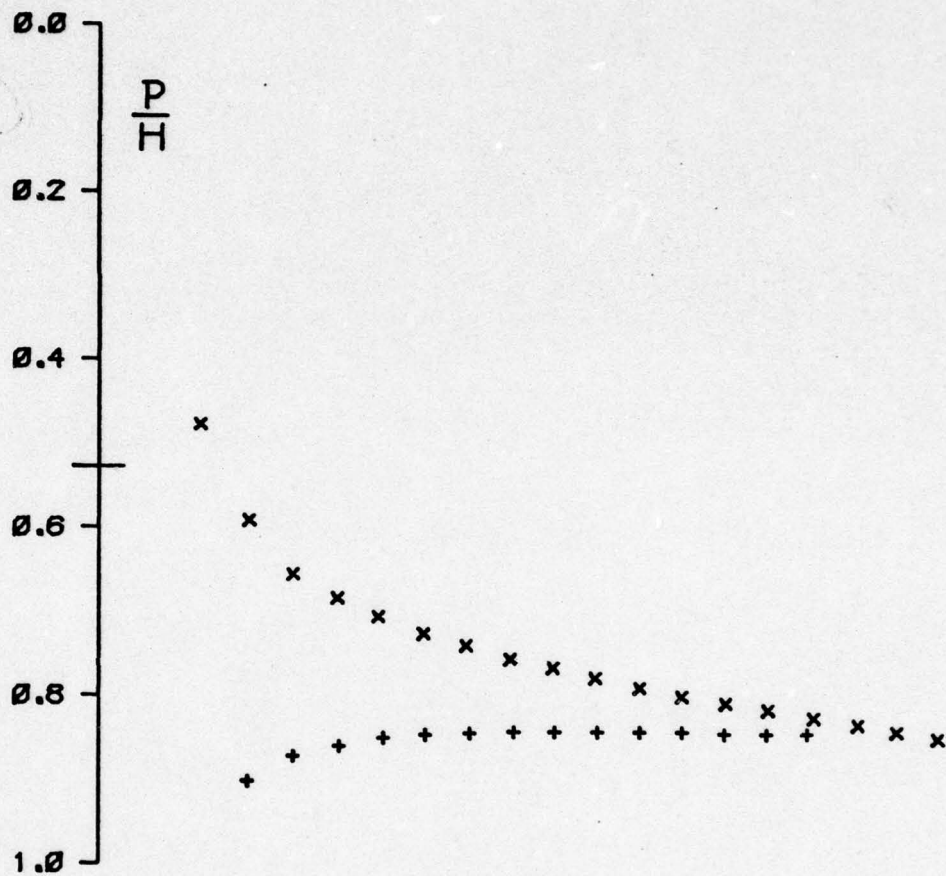
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .501$ $AL = 4.00$ $CN = 0.385$ $CM = 0.006$ $R = 0.666$



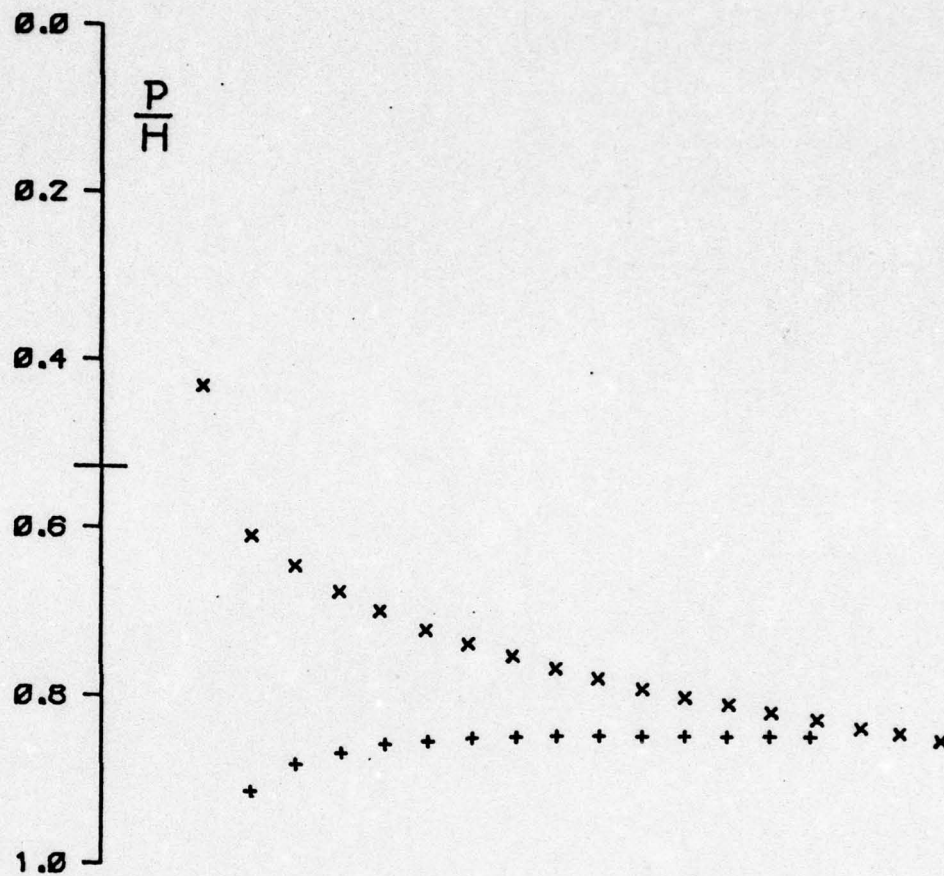
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .500 AL= 5.00 CN= 0.477 CM= 0.009 R= 0.863



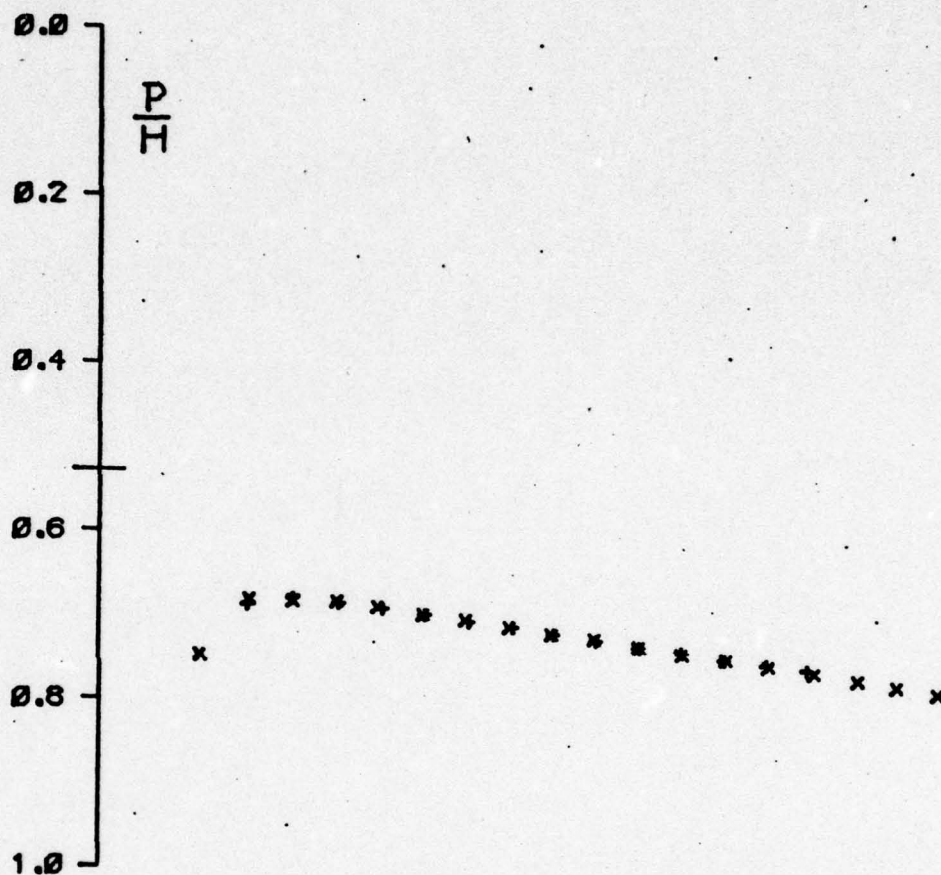
+ x NACA 0012 10% .6 MM CHORD SLOTTED WALLS
M= .499 AL= 6.00 CN= 0.577 CM= 0.013 R= 0.860



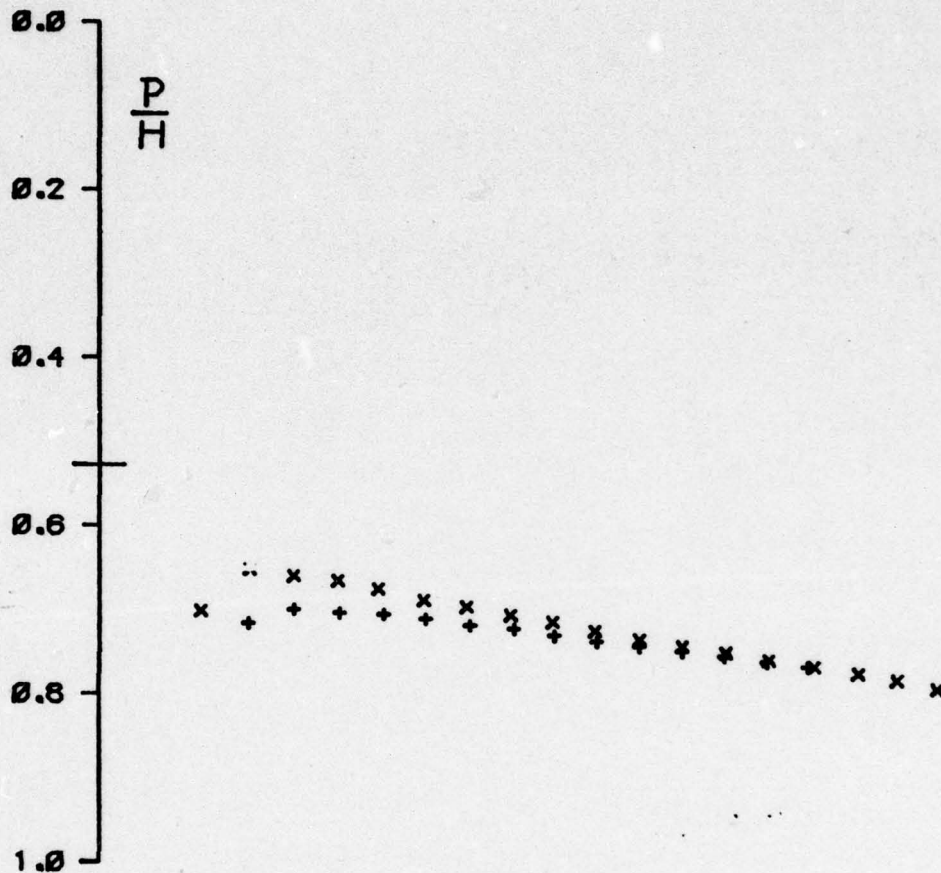
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .499 AL= 7.00 CN= 0.670 CM= 0.015 R= 0.860



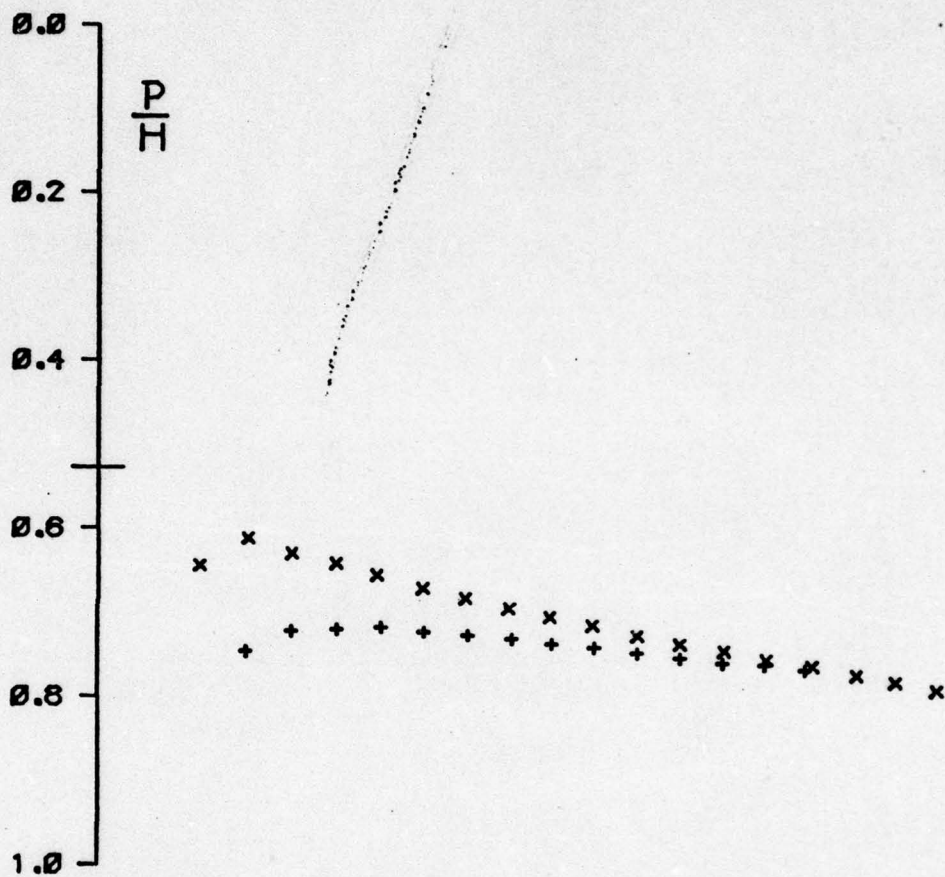
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .499 AL= 8.00 CN= 0.721 CM= 0.018 R= 0.863



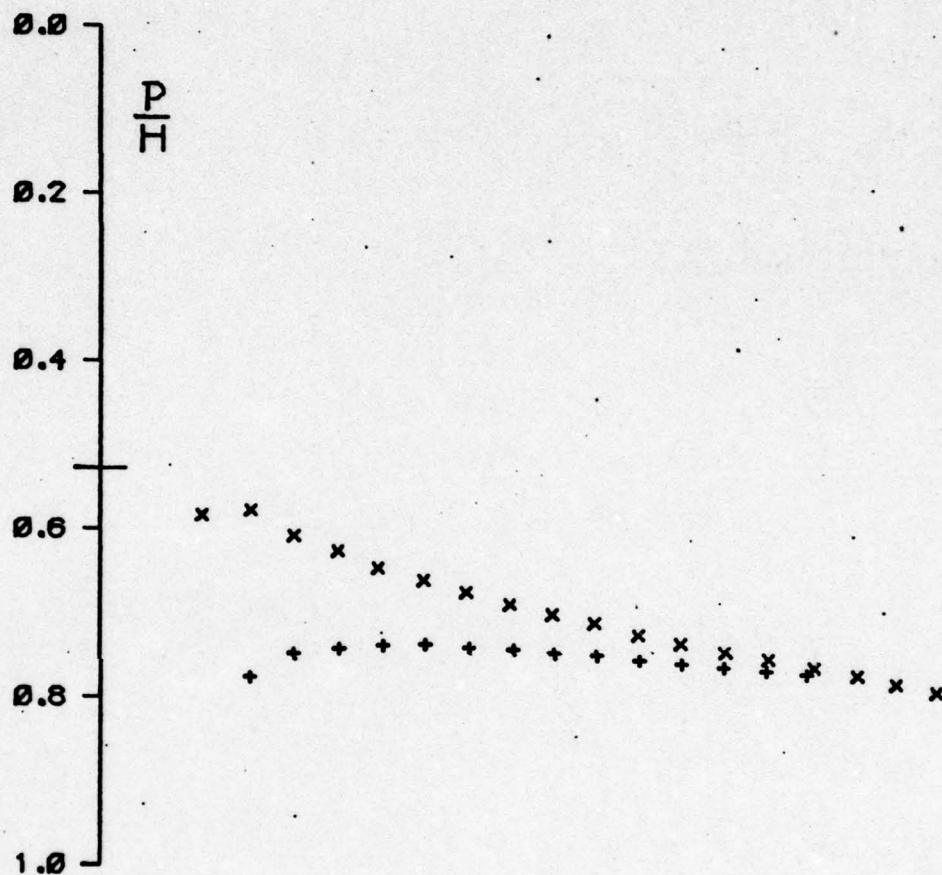
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .601 AL= 0.00 CN= 0.020 CM= 0.004 R= 0.814



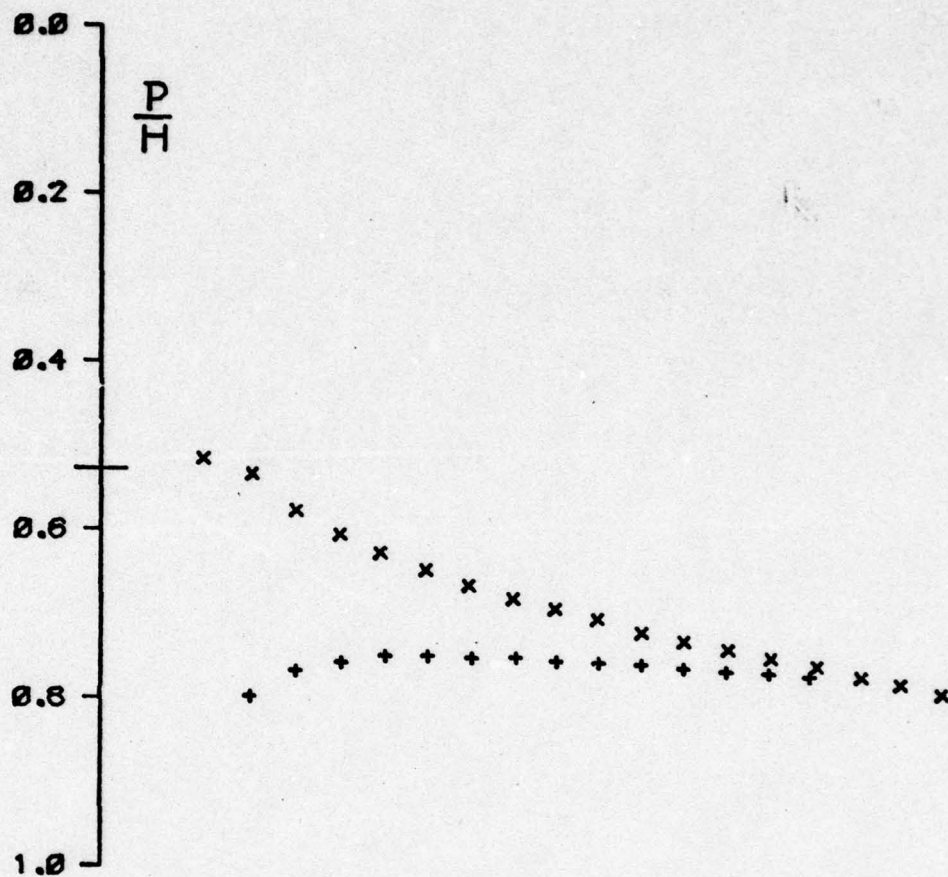
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .600 AL= 1.00 CN= 0.093 CM= 0.002 R= 0.811



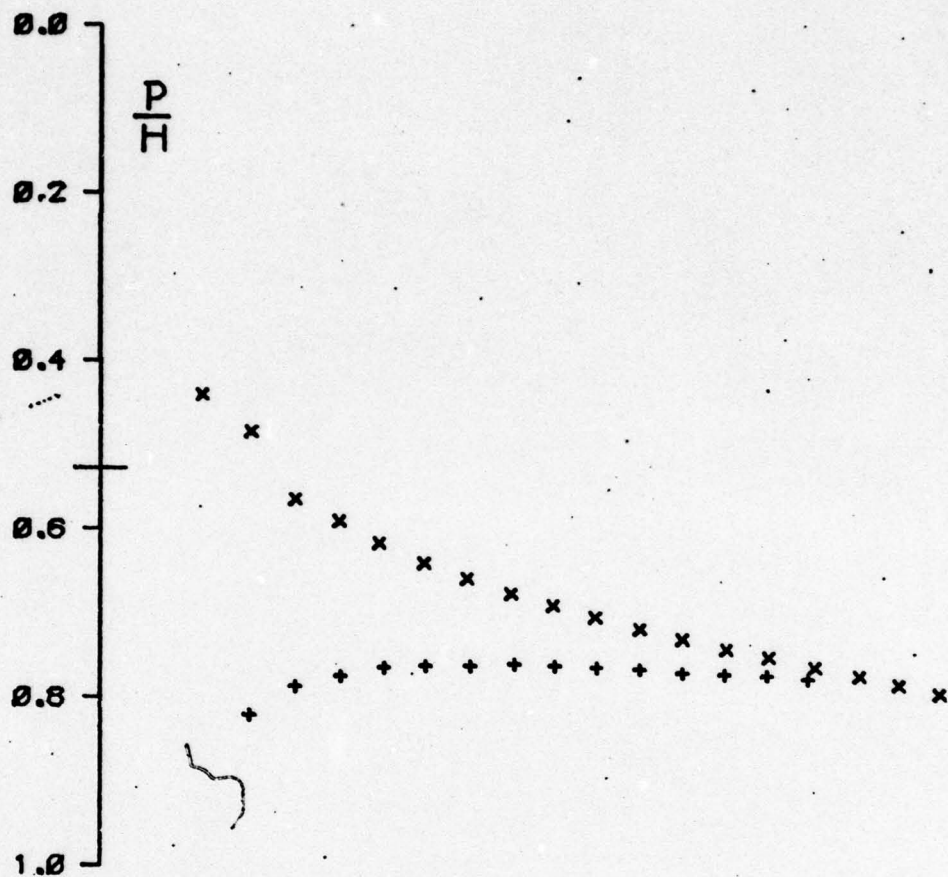
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .601$ $AL = 2.00$ $C_N = 0.202$ $C_N = 0.004$ $R = 0.814$



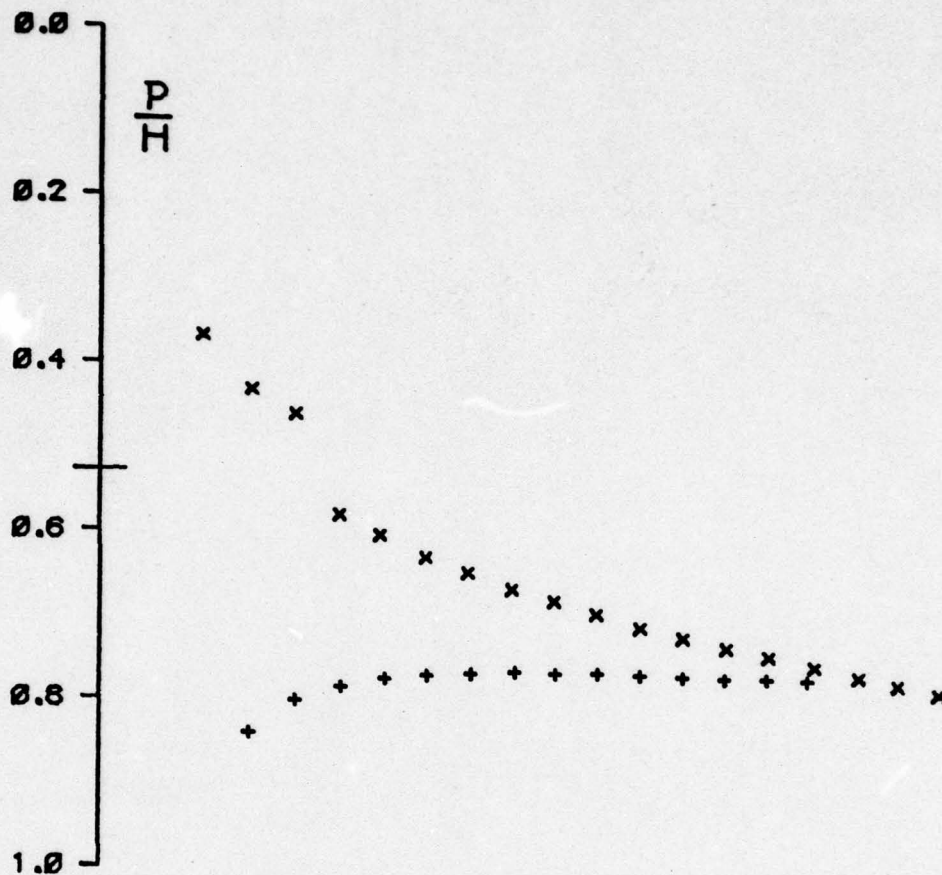
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .596$ $AL = 3.00$ $CN = 0.303$ $CM = 0.007$ $R = 0.812$



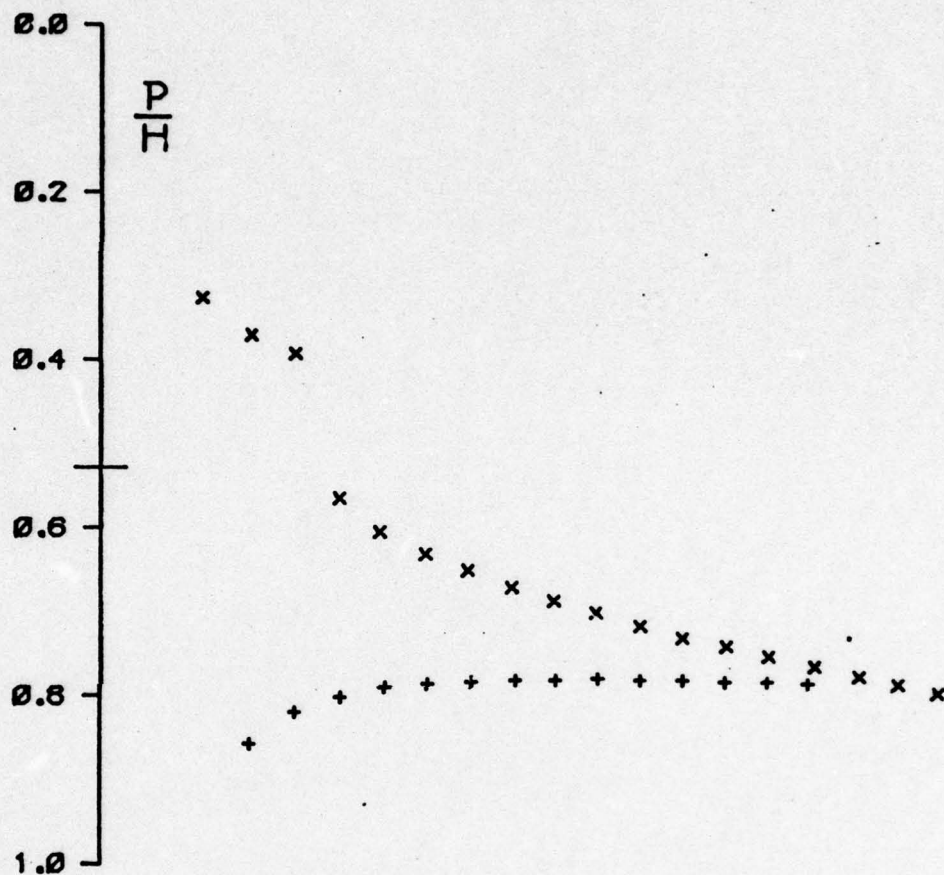
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .600 AL= 4.00 CN= 0.402 CM= 0.011 R= 0.811



+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $V = .599$ $AL = 5.06$ $CN = 0.498$ $CM = 0.015$ $R = 0.814$



+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
M= .599 AL= 6.00 CN= 0.605 CM= 0.022 R= 0.811

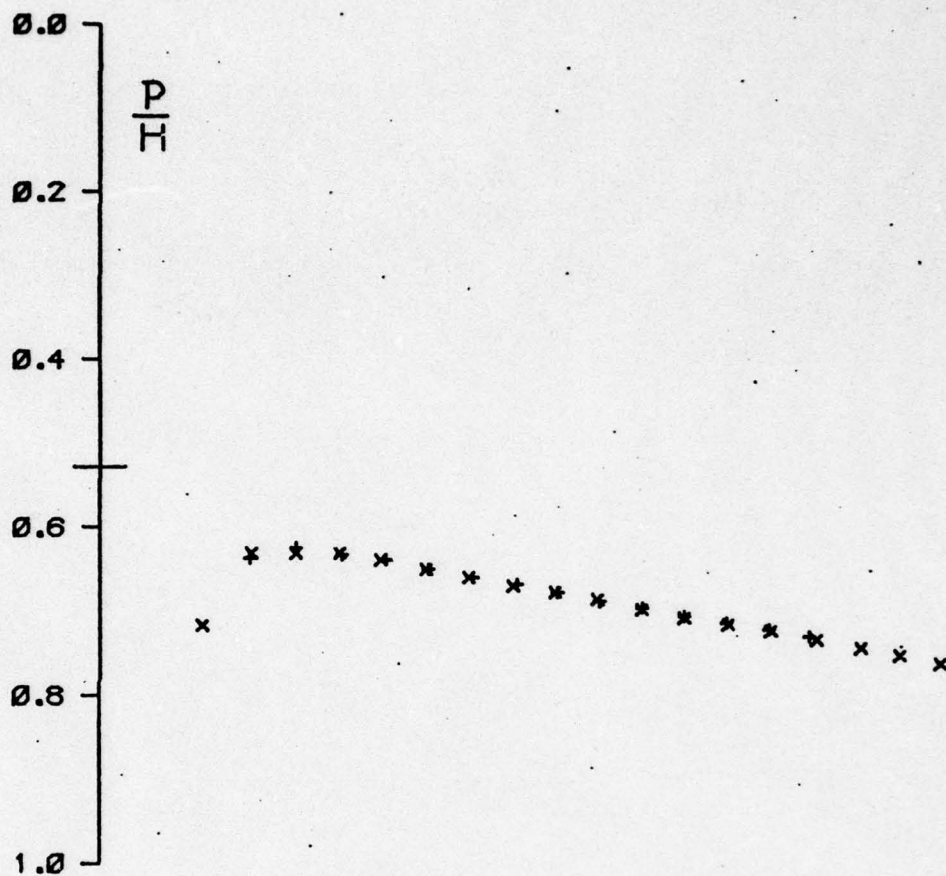


+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
M= .599 AL= 7.00 CN= 0.695 CM= 0.027 R= 0.814

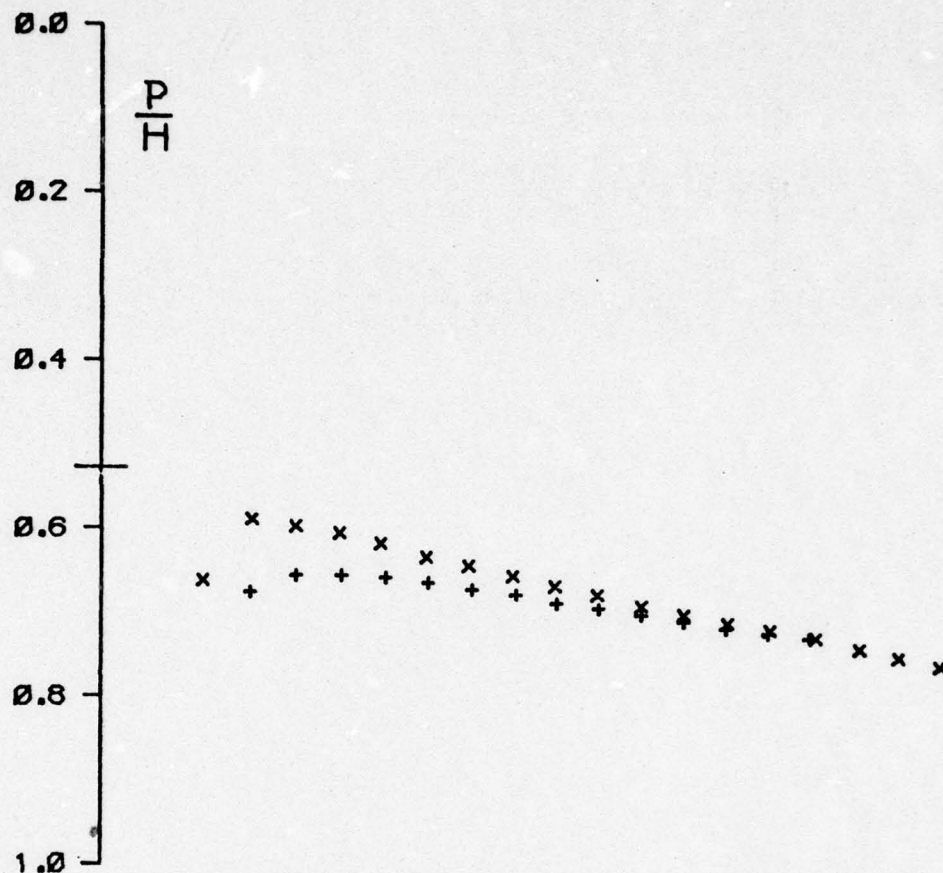
12°

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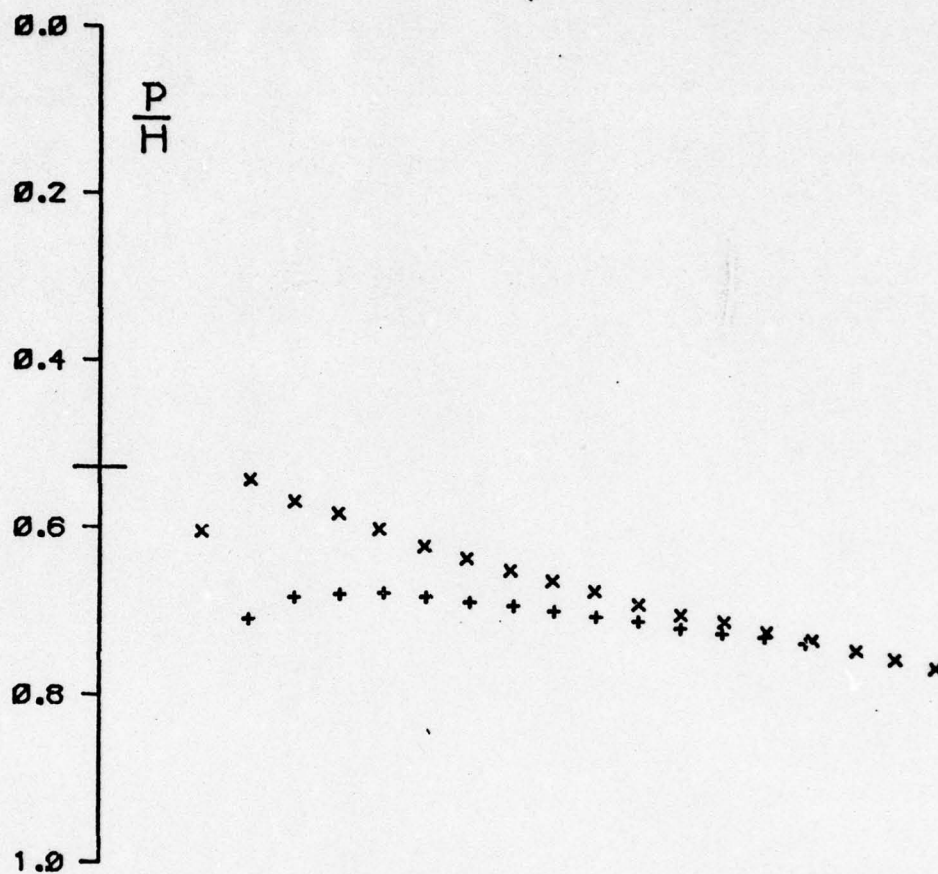
PRINTED IN AUSTRALIA



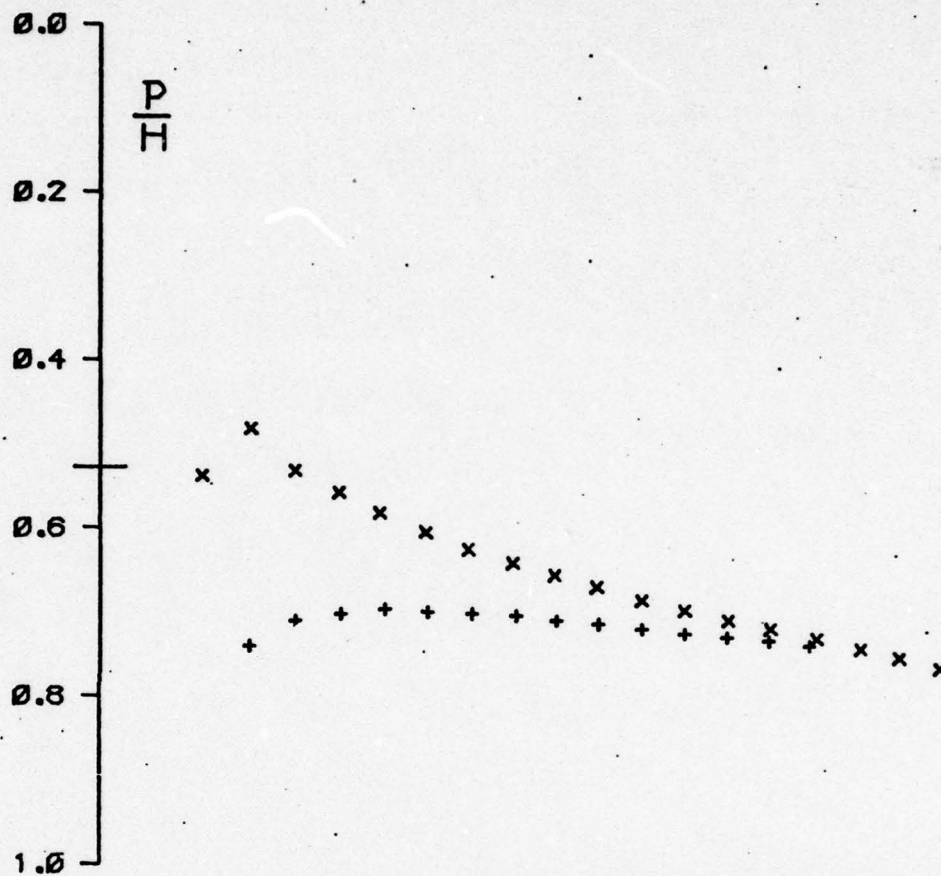
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .650 AL= 0.00 CN= 0.023 CM= 0.004 R= 0.816



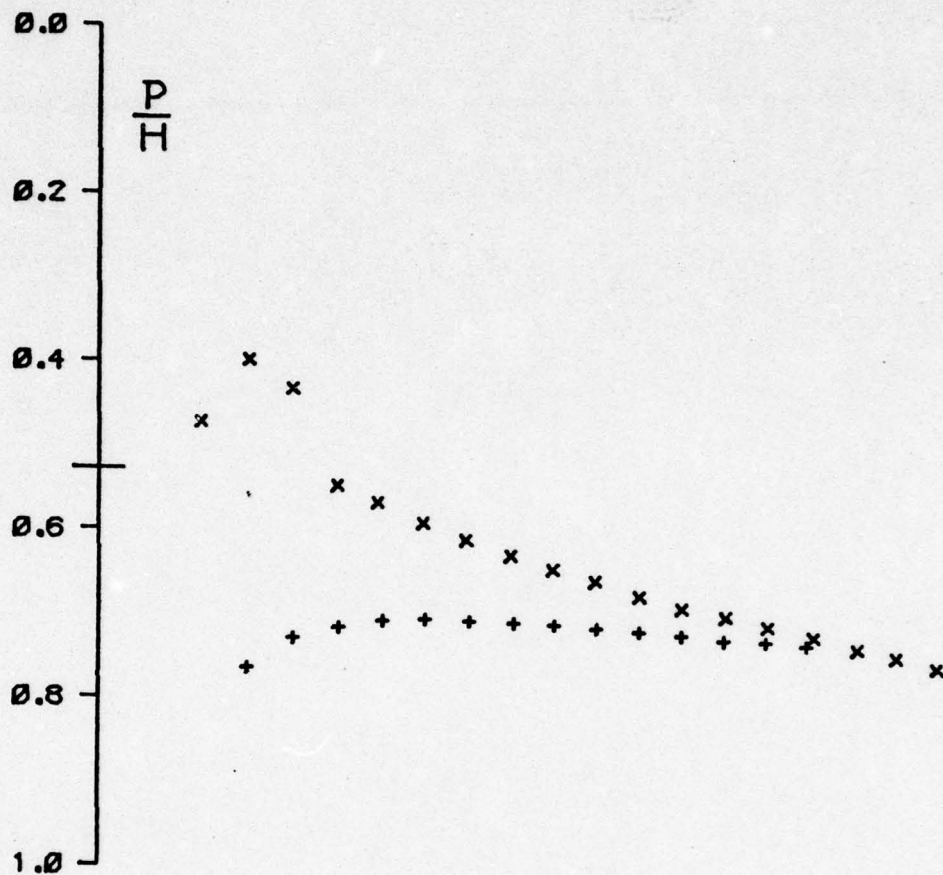
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
M= .651 AL= 1.00 CN= 0.107 CM= 0.003 R= 0.814



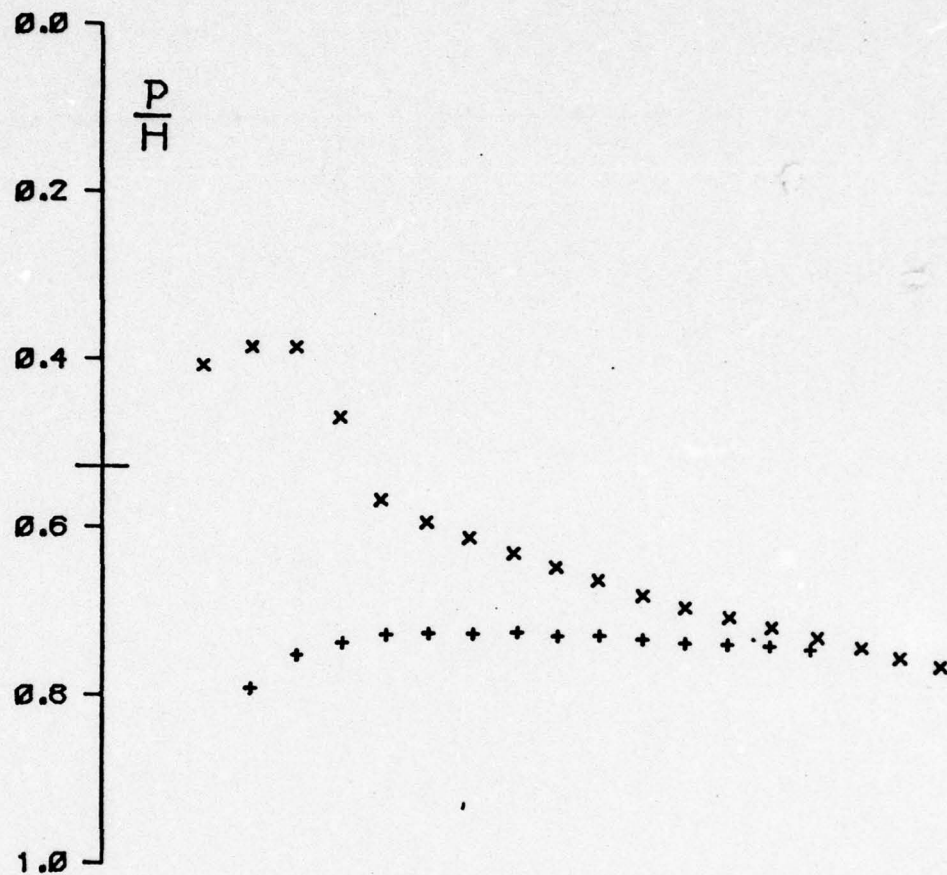
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .650 AL= 2.00 CN= 0.211 CM= 0.006 R= 0.816



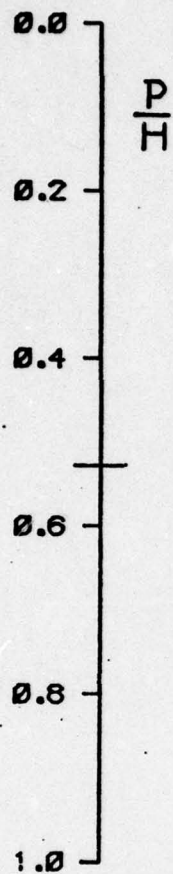
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .651$ $AL = 3.00$ $CN = 0.318$ $CM = 0.011$ $R = 0.829$



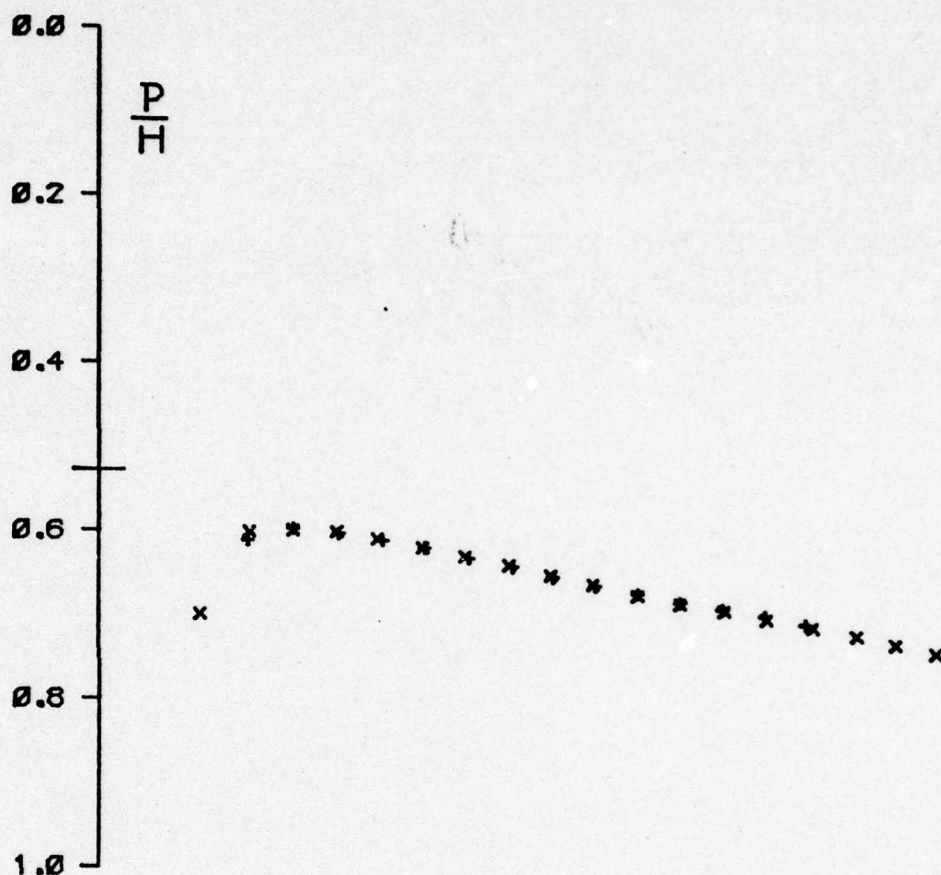
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .650$ $AL = 4.00$ $CN = 0.433$ $CM = 0.018$ $R = 0.812$



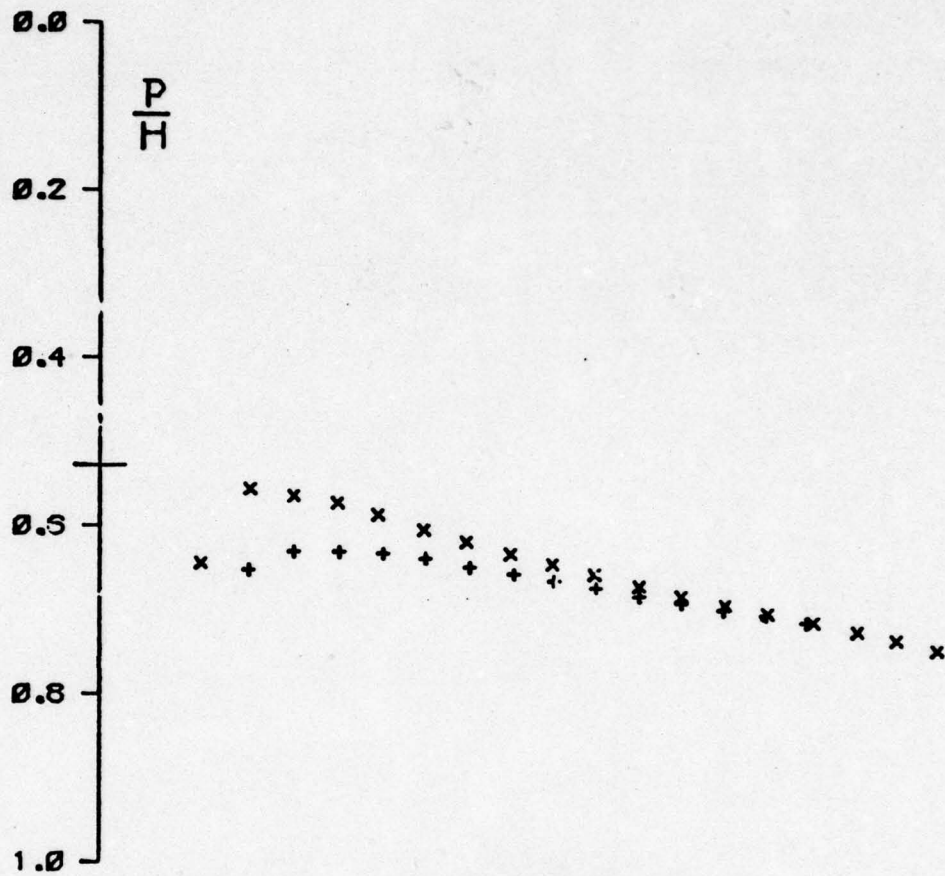
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .649 AL= 5.00 CN= 0.529 CM= 0.020 R= 0.809



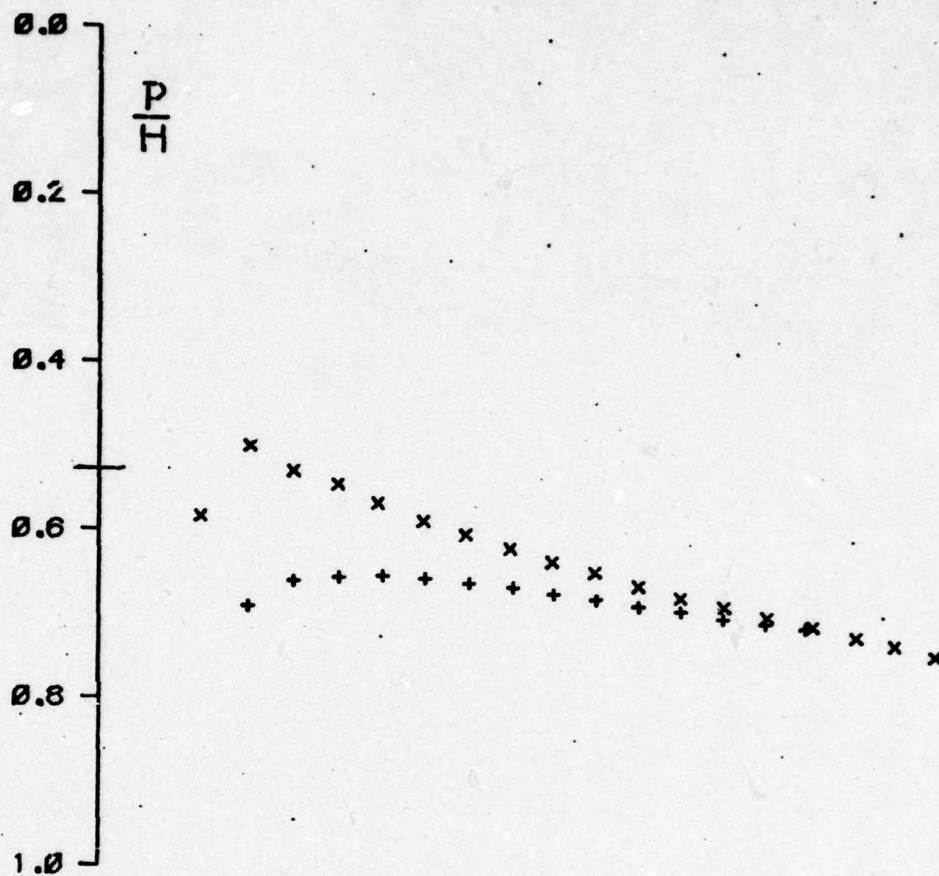
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
M= .649 AL= 6.00 CN= 0.631 CM= 0.025 R= 0.806



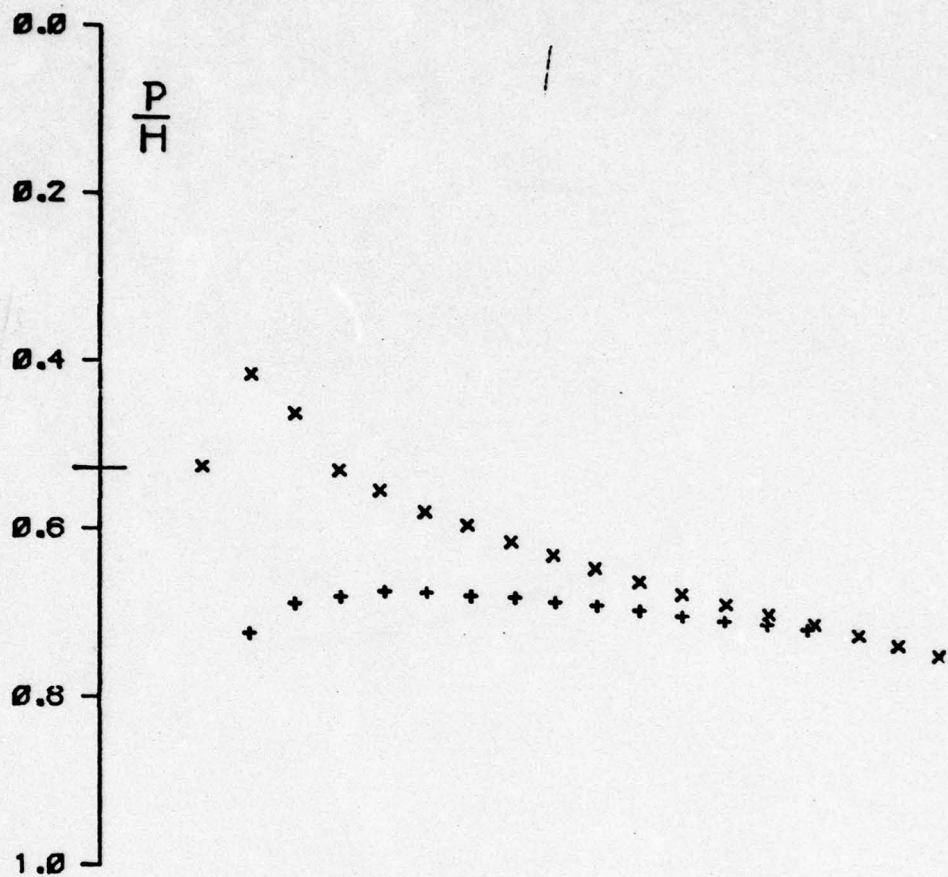
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .676$ $AL = 0.00$ $CN = 0.026$ $CM = 0.005$ $R = 0.824$



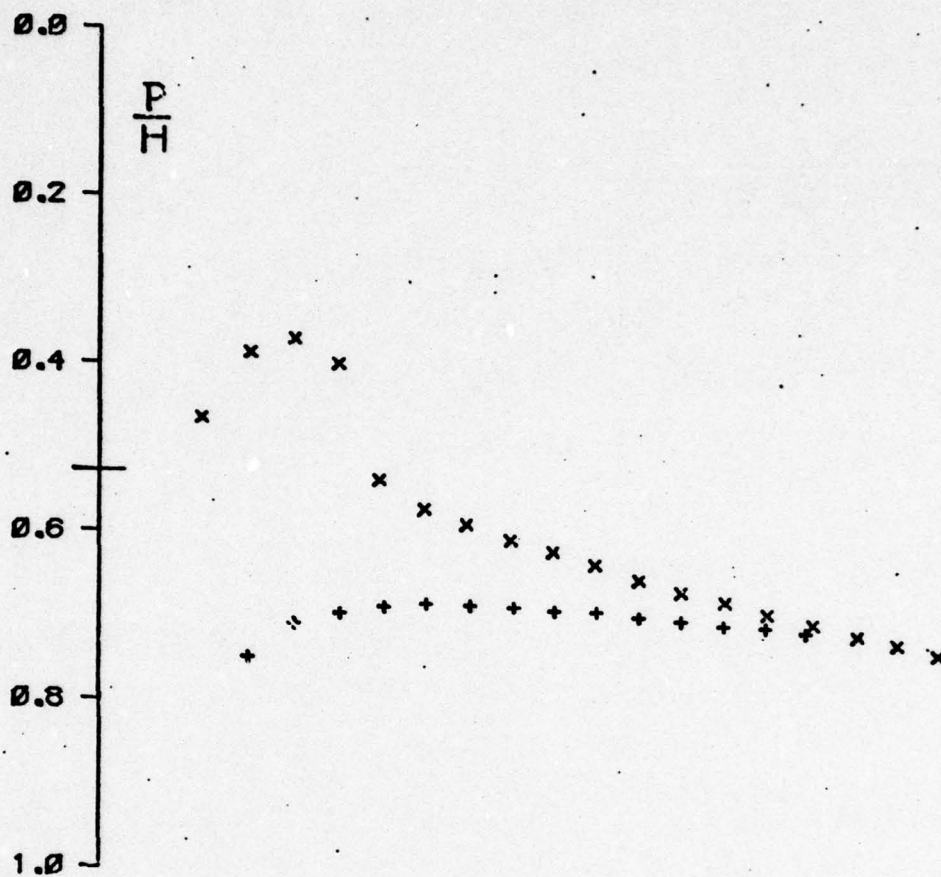
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .676$ $AL = 1.00$ $CN = 0.111$ $CM = 0.004$ $R = 0.824$



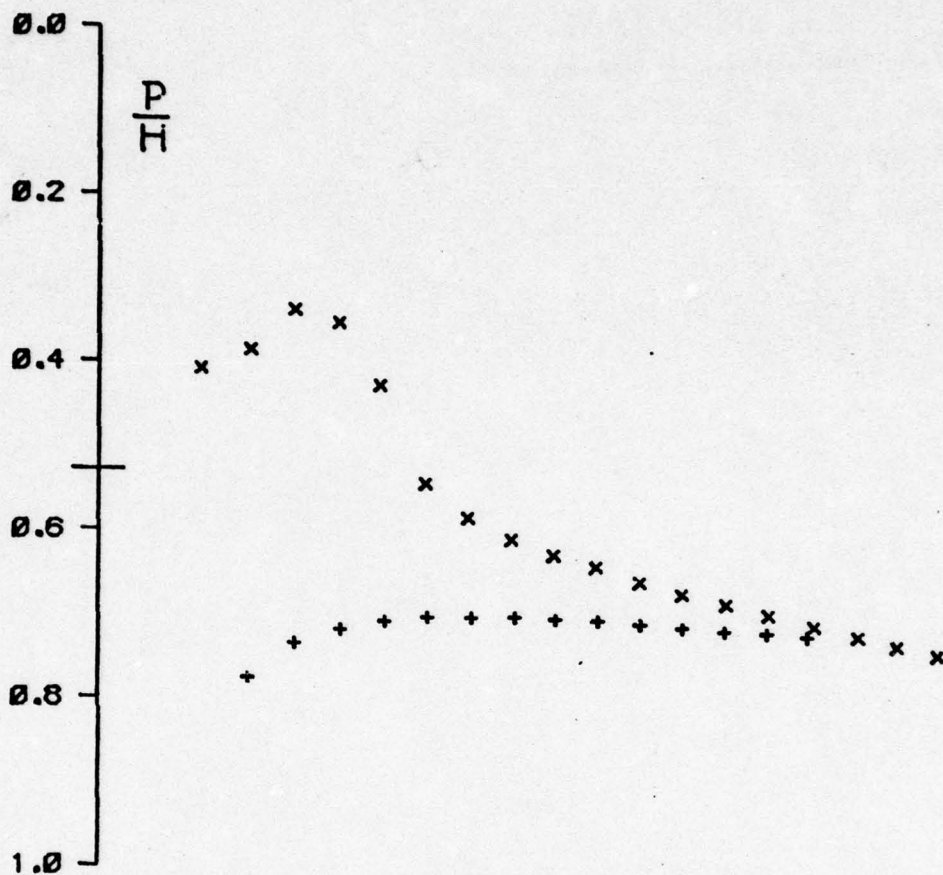
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .676$ $AL = 2.00$ $CN = 0.221$ $CM = 0.008$ $R = 0.824$



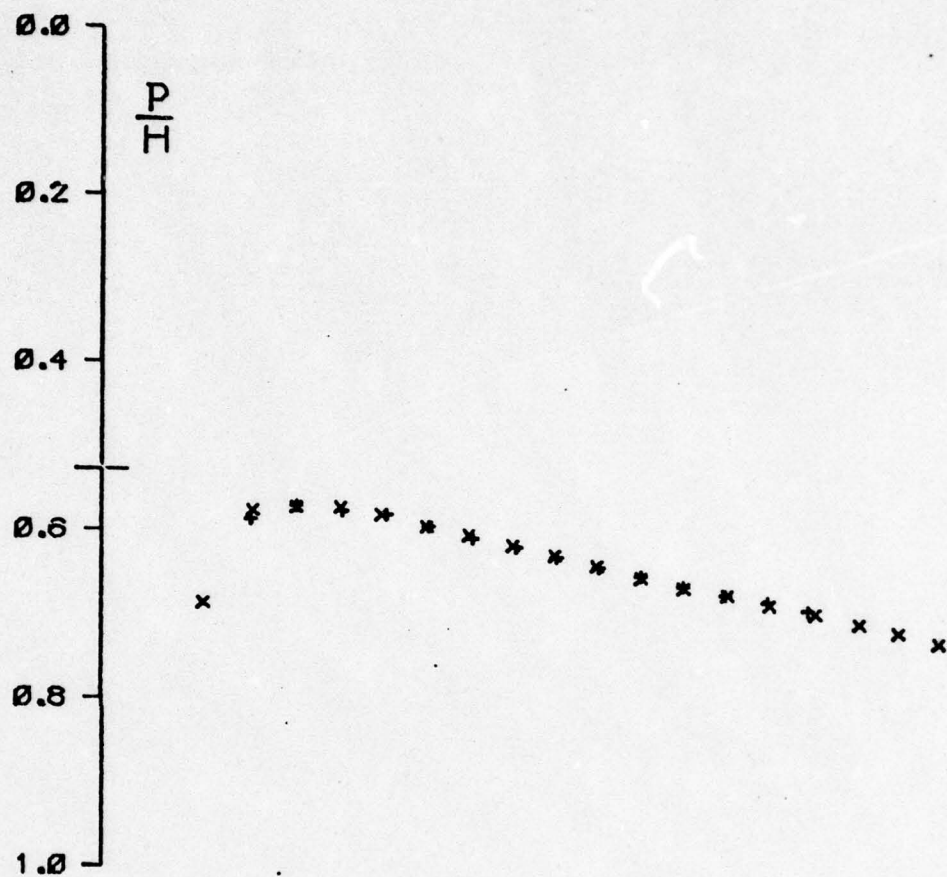
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .674$ $AL = 3.00$ $CN = 0.356$ $CM = 0.018$ $R = 0.822$



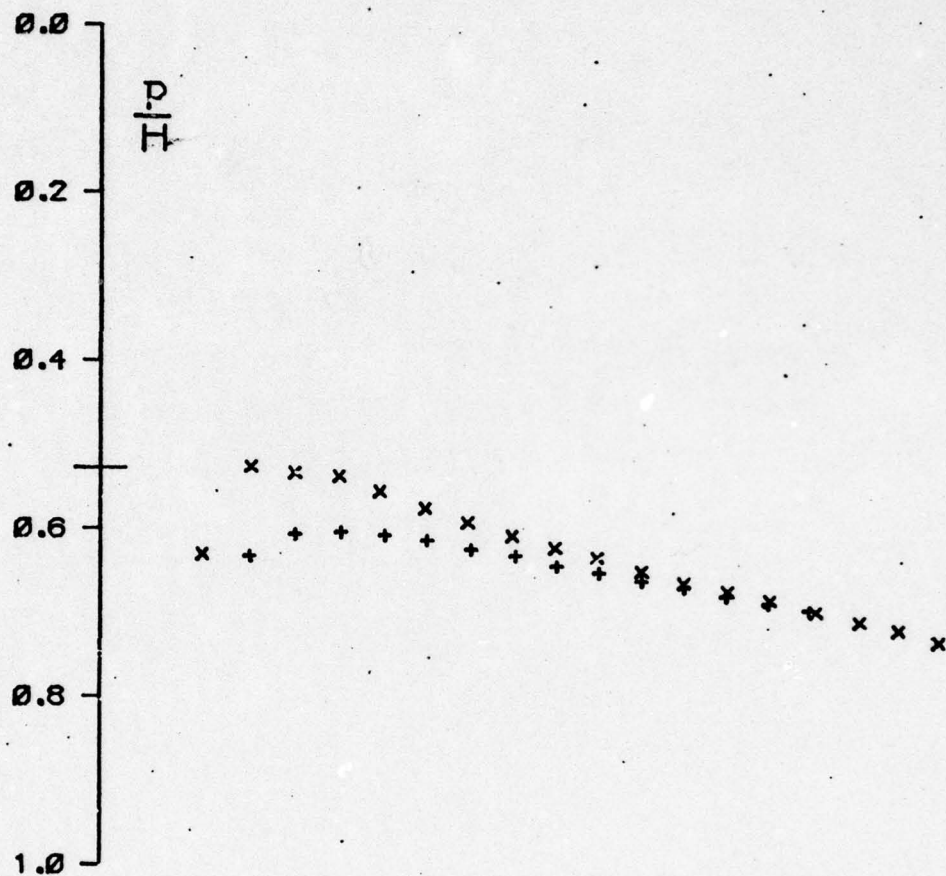
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .674$ $AL = 4.00$ $C_N = 0.447$ $C_N = 0.018$ $R = 0.822$



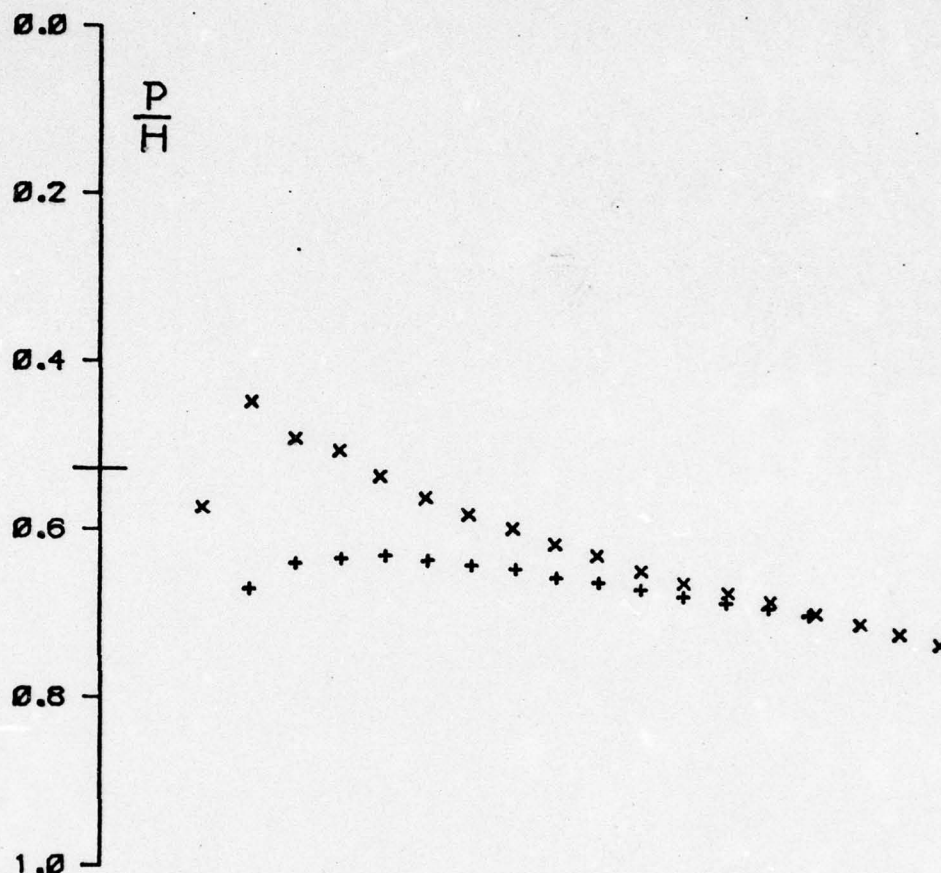
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .675 AL= 5.00 CN= 0.550 CM= 0.022 R= 0.825



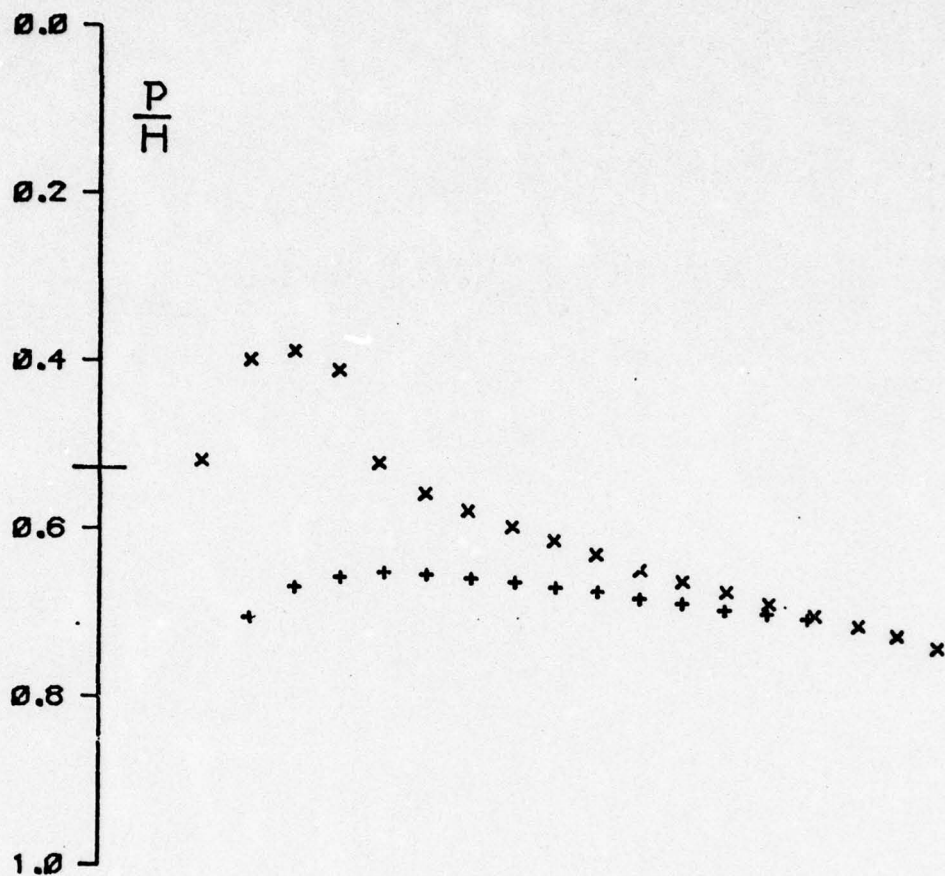
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .701$ $AL = 0.00$ $CN = 0.028$ $CM = 0.005$ $R = 0.847$



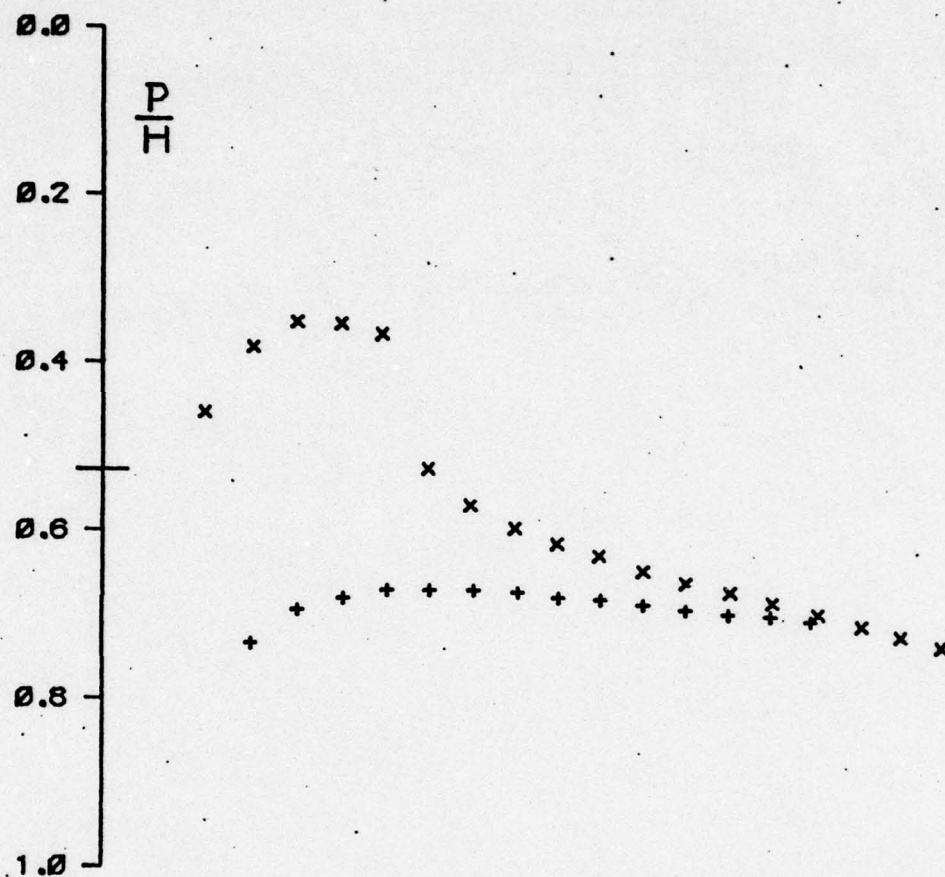
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .700$ $AL = 1.00$ $CN = 0.119$ $CM = 0.005$ $R = 0.847$



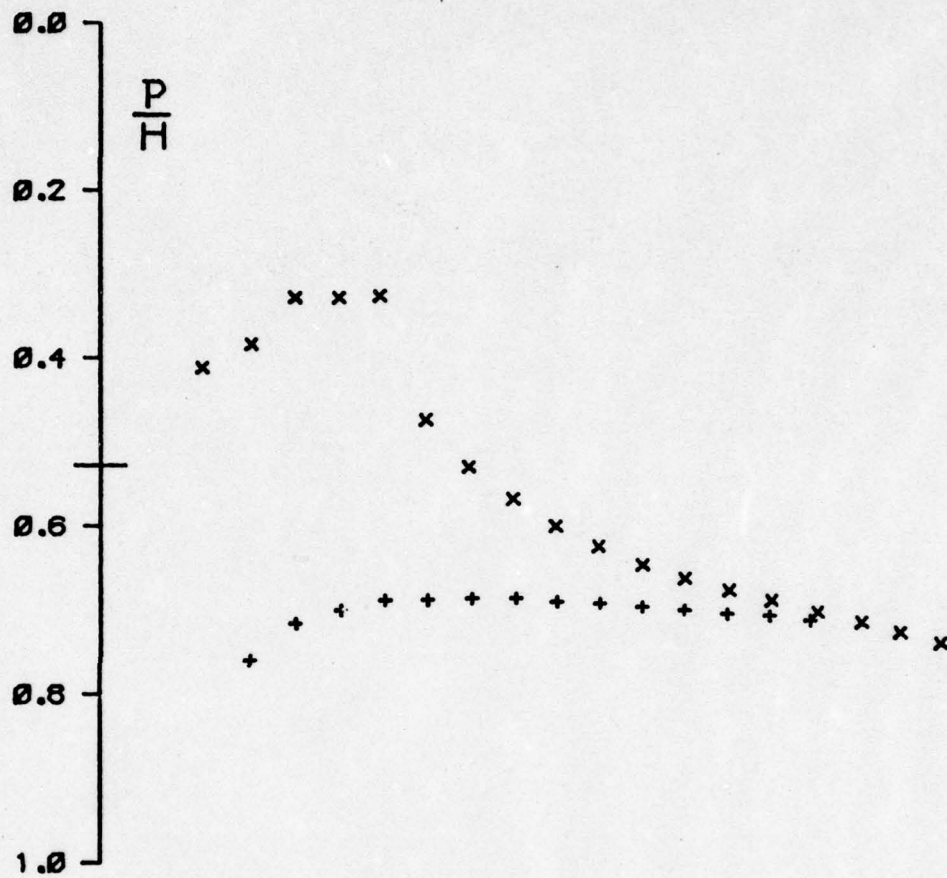
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .698 AL= 2.00 CN= 0.255 CM= 0.015 R= 0.840



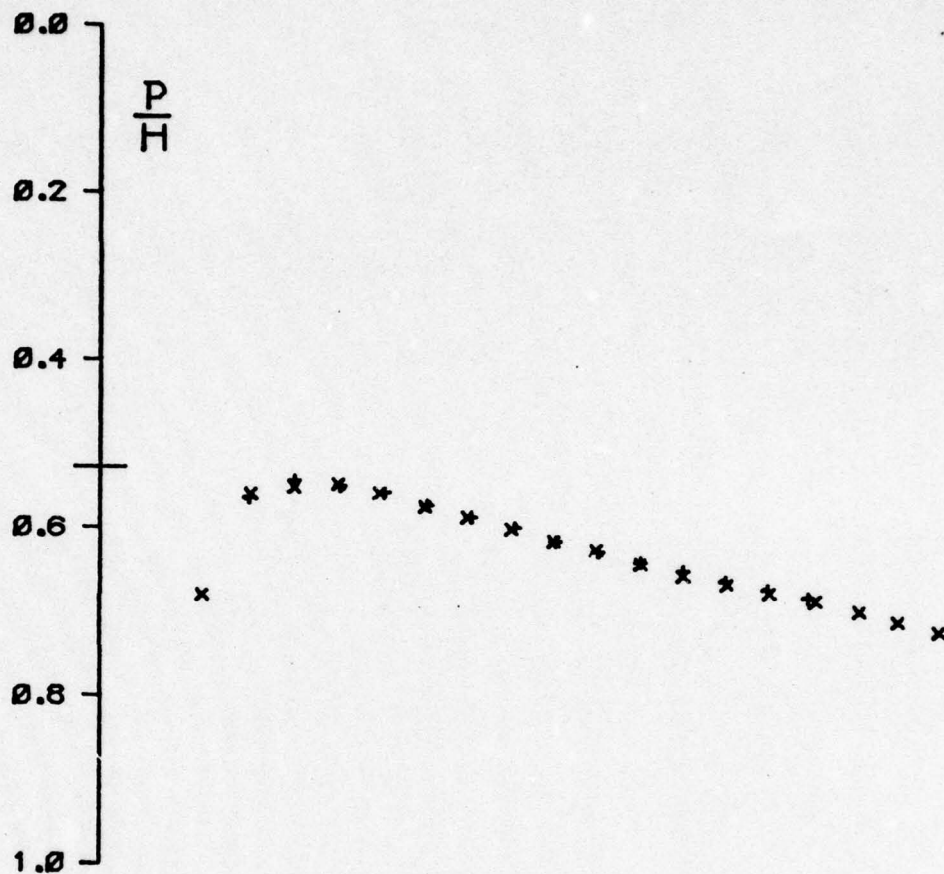
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .699$ $AL = 3.00$ $CN = 0.348$ $CM = 0.016$ $R = 0.843$



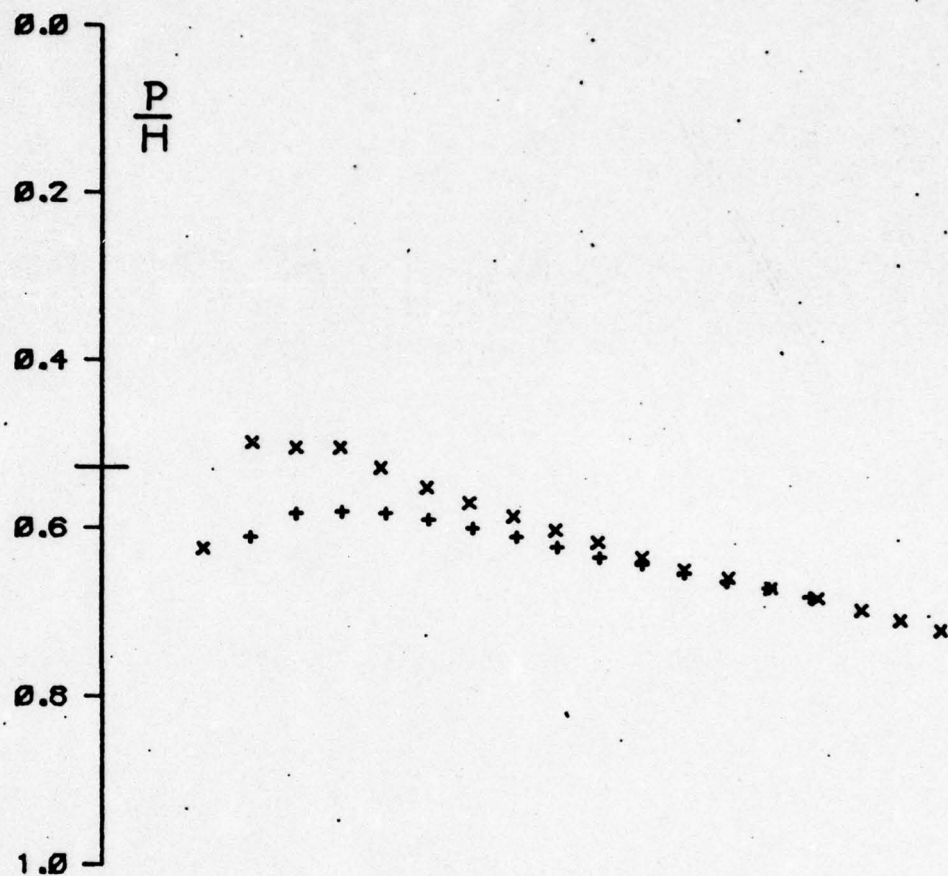
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .698$ $AL = 4.00$ $CN = 0.464$ $CM = 0.019$ $R = 0.843$



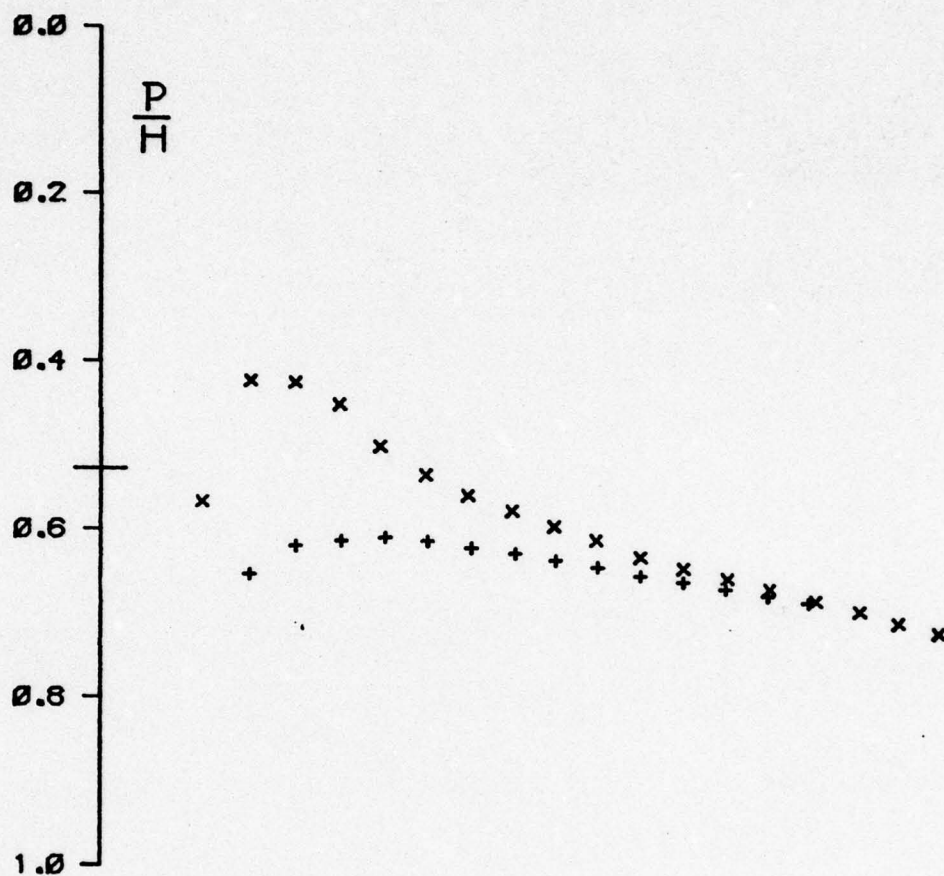
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
M= .698 AL= 5.00 CN= 0.559 CM= 0.019 R= 0.840



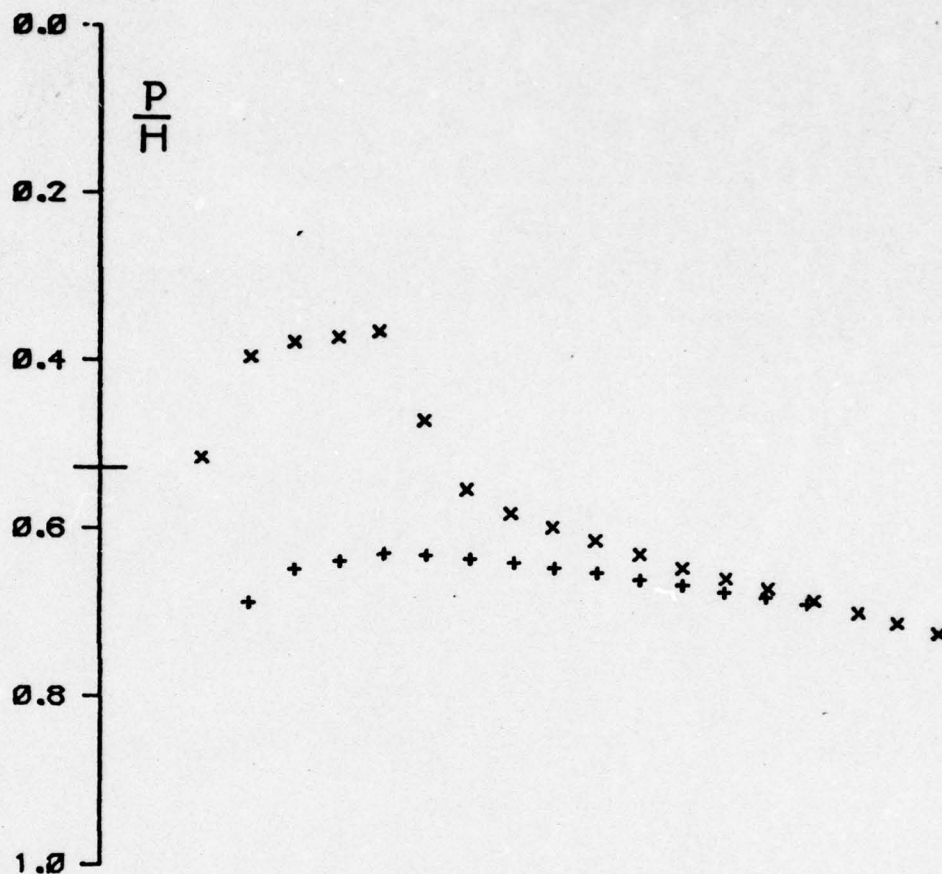
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .720$ $AL = 0.00$ $CN = 0.024$ $CM = 0.006$ $R = 0.799$



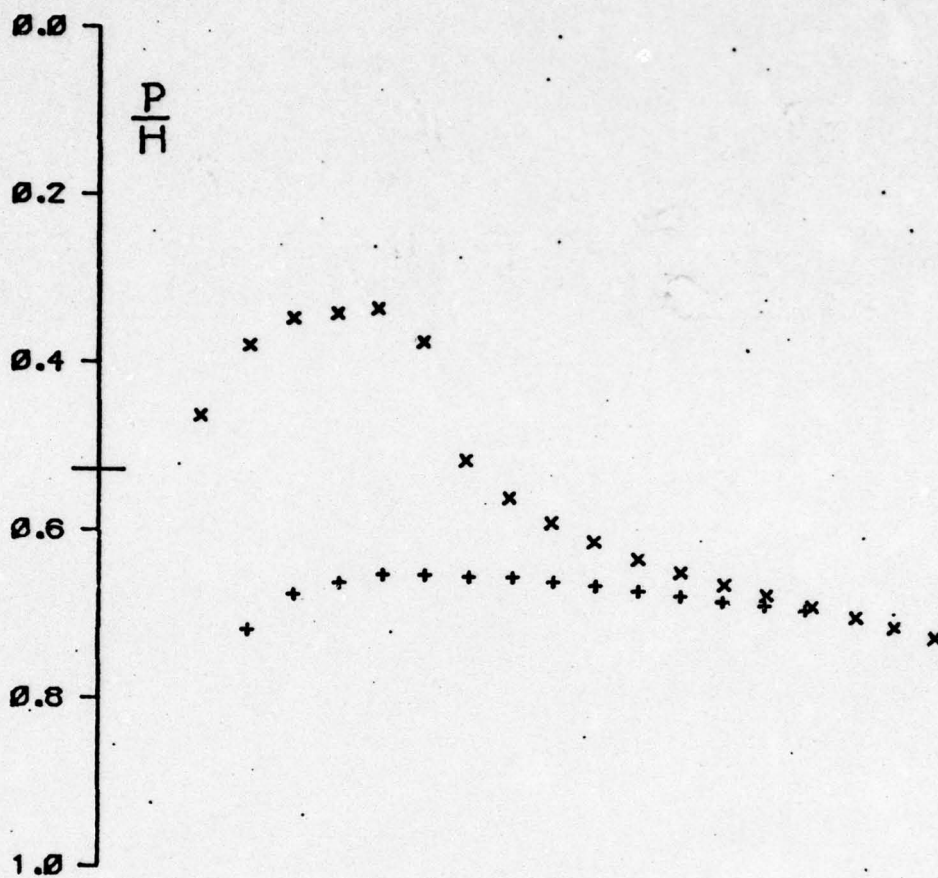
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .720 AL= 1.00 CN= 0.126 CM= 0.008 R= 0.796



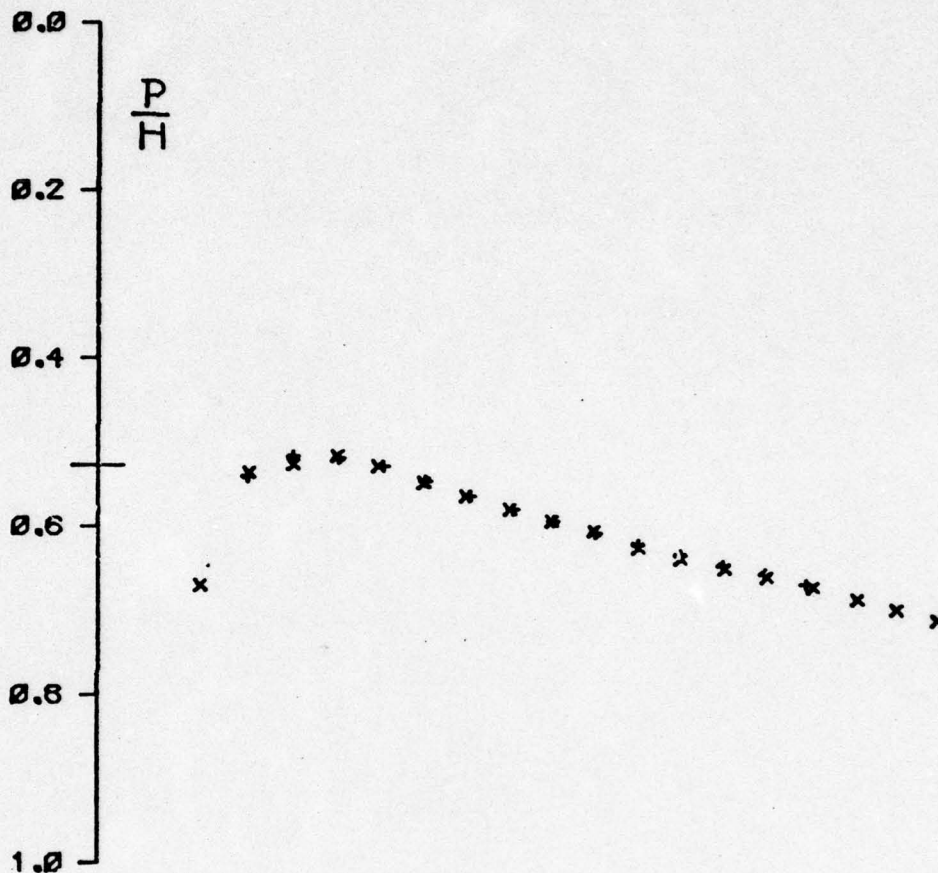
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .720 AL= 2.00 CN= 0.273 CM= 0.018 R= 0.799



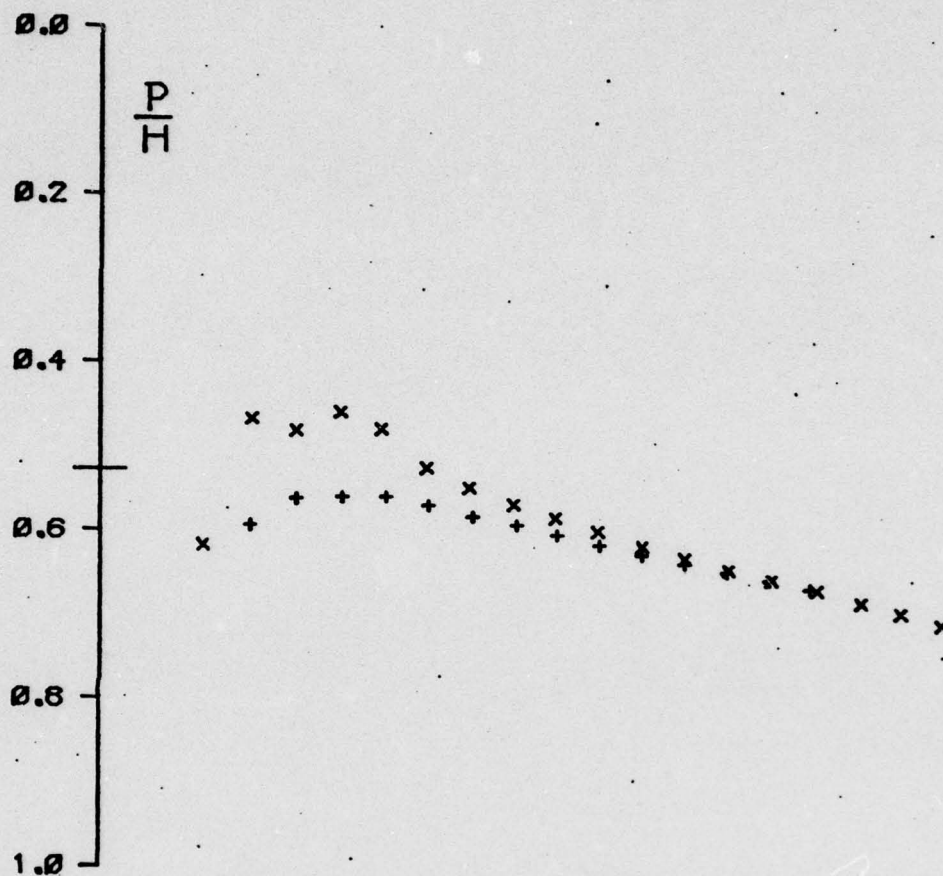
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .720 AL= 3.00 CN= 0.365 CM= 0.016 R= 0.799



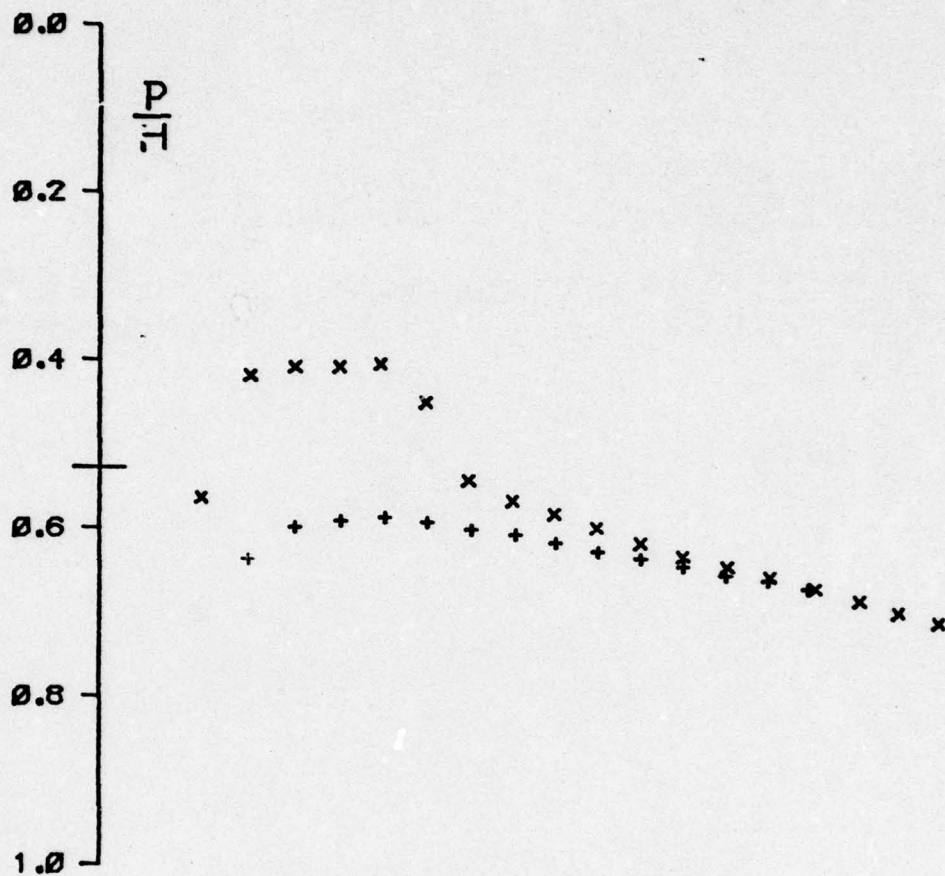
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .719 AL= 4.00 CN= 0.466 CM= 0.018 R= 0.796



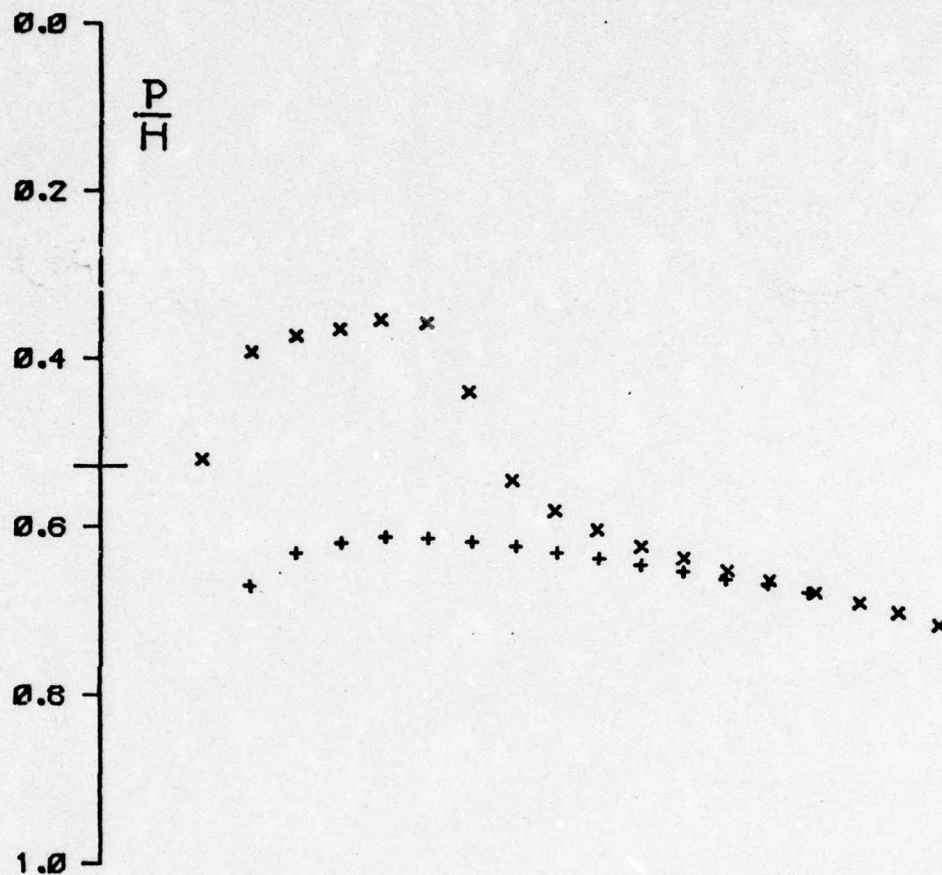
+ x NACA 0012 101 .6 MM CHORD SLOTTED WALLS
 M= .740 AL= 0.00 CN= 0.025 CM= 0.006 R= 0.799



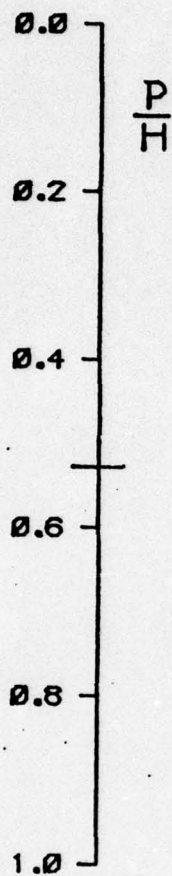
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .739$ $AL = 1.00$ $CN = 0.145$ $CM = 0.011$ $R = 0.796$



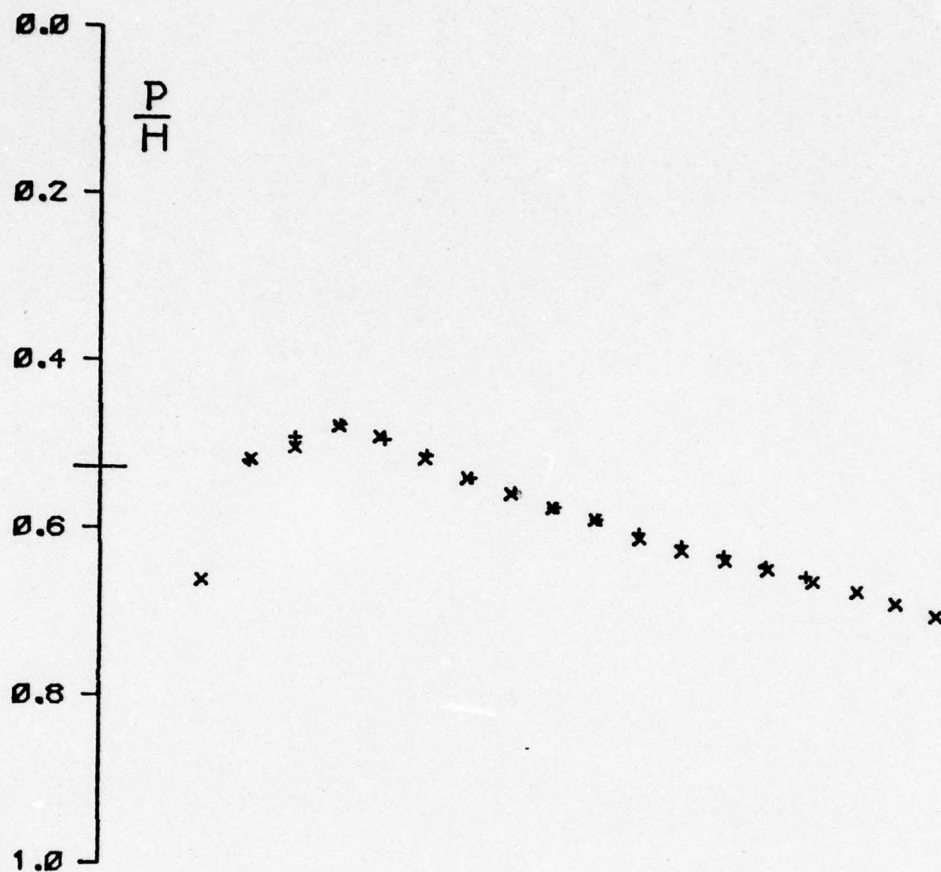
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .741$ $AL = 2.00$ $CN = 0.271$ $CM = 0.017$ $R = 0.798$



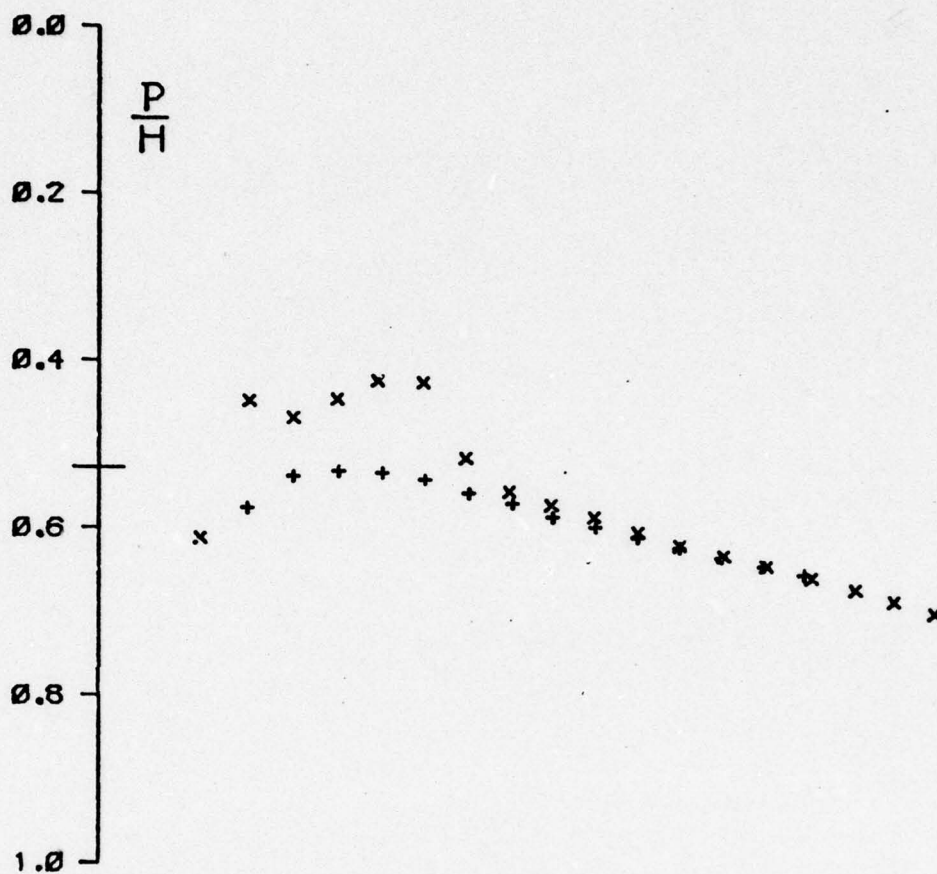
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .740 AL= 3.00 CN= 0.373 CM= 0.015 R= 0.799



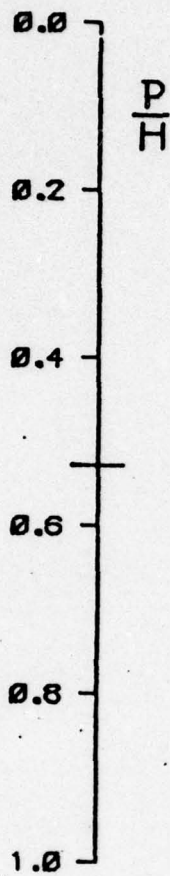
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .741 AL= 4.00 CN= 0.465 CM= 0.010 R= 0.801



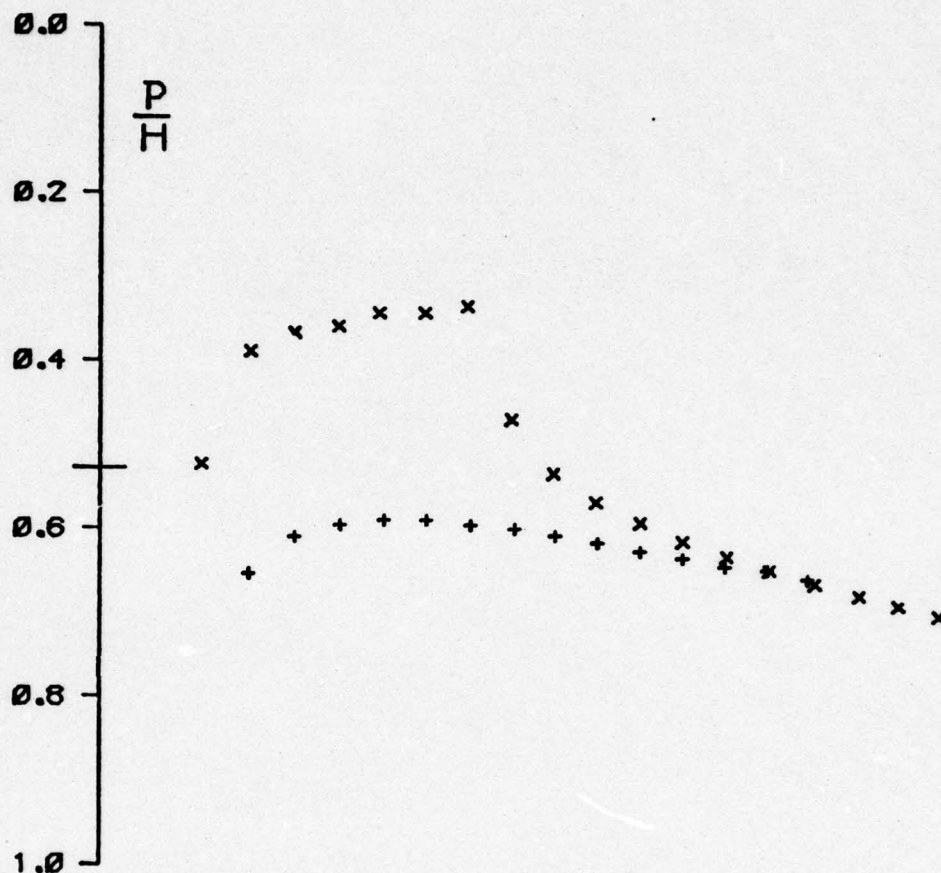
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .758 AL= 0.00 CN= 0.017 CM= 0.004 R= 0.781 .



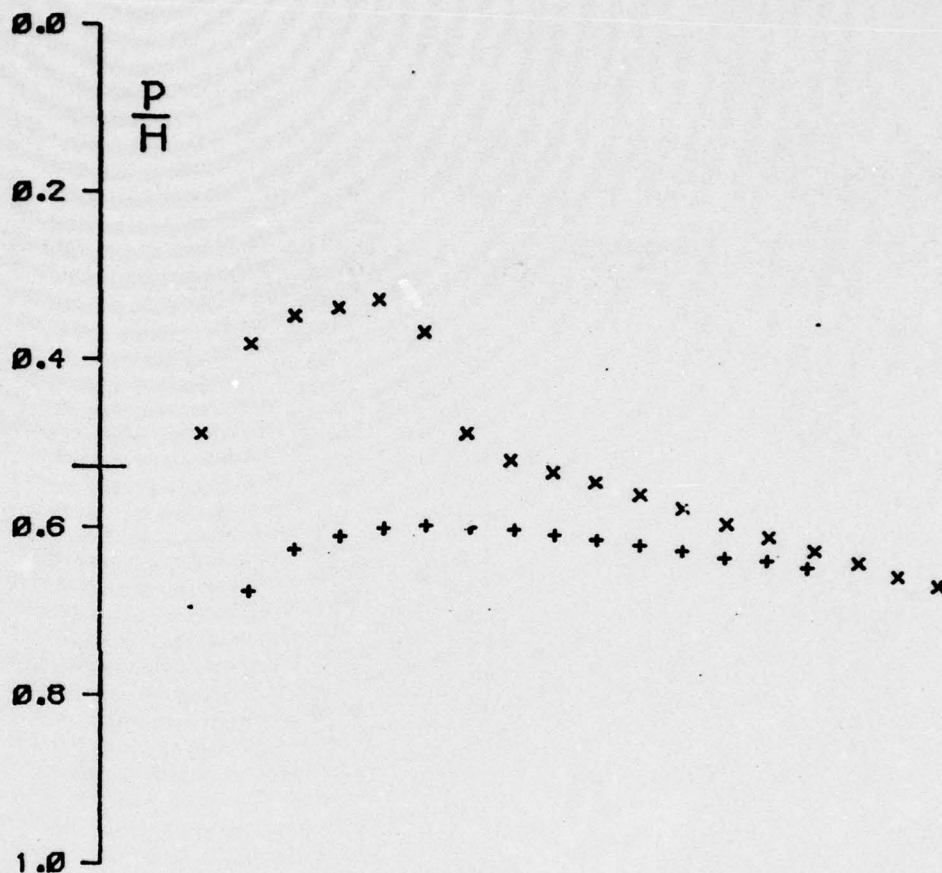
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .758 AL= 1.00 CN= 0.153 CM= 0.012 R= 0.781



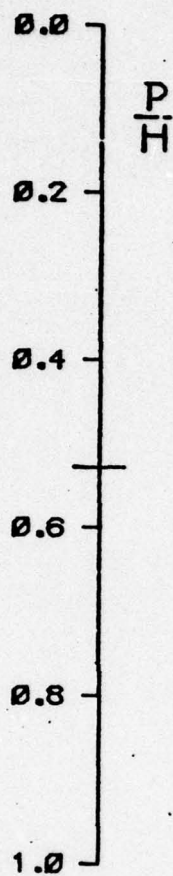
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .761 AL= 2.00 CN= 0.282 CM= 0.014 R= 0.766



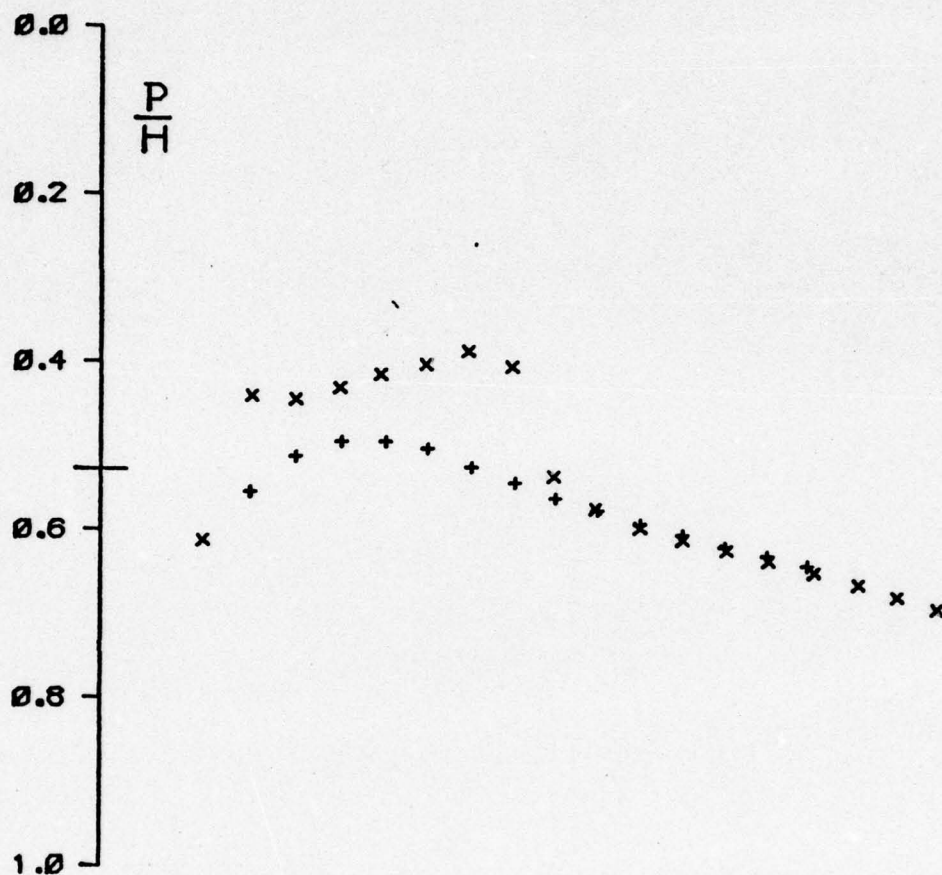
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .759$ $AL = 3.00$ $CN = 0.375$ $CM = 0.010$ $R = 0.784$



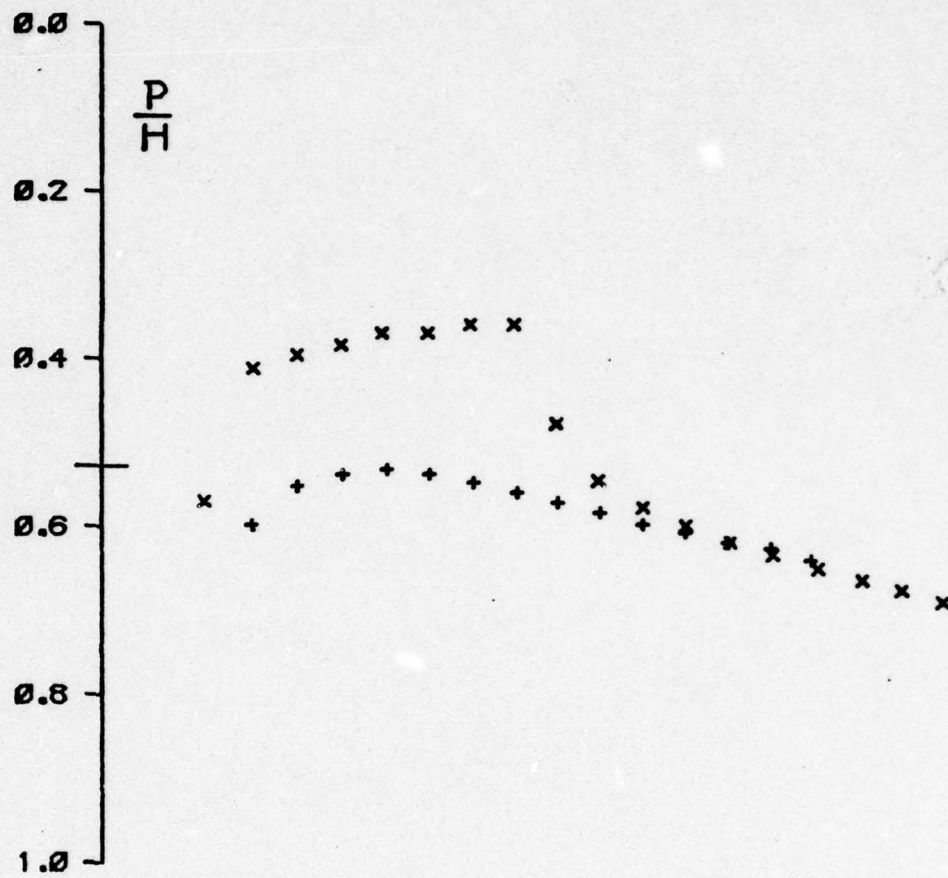
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .761 AL= 1.00 CN= 0.125 CM= 0.002 R= 0.786



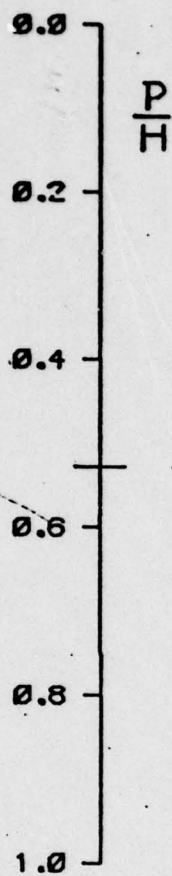
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
M= .781 AL= 0.00 CN= 0.003 CM= 0.003 R= 0.799



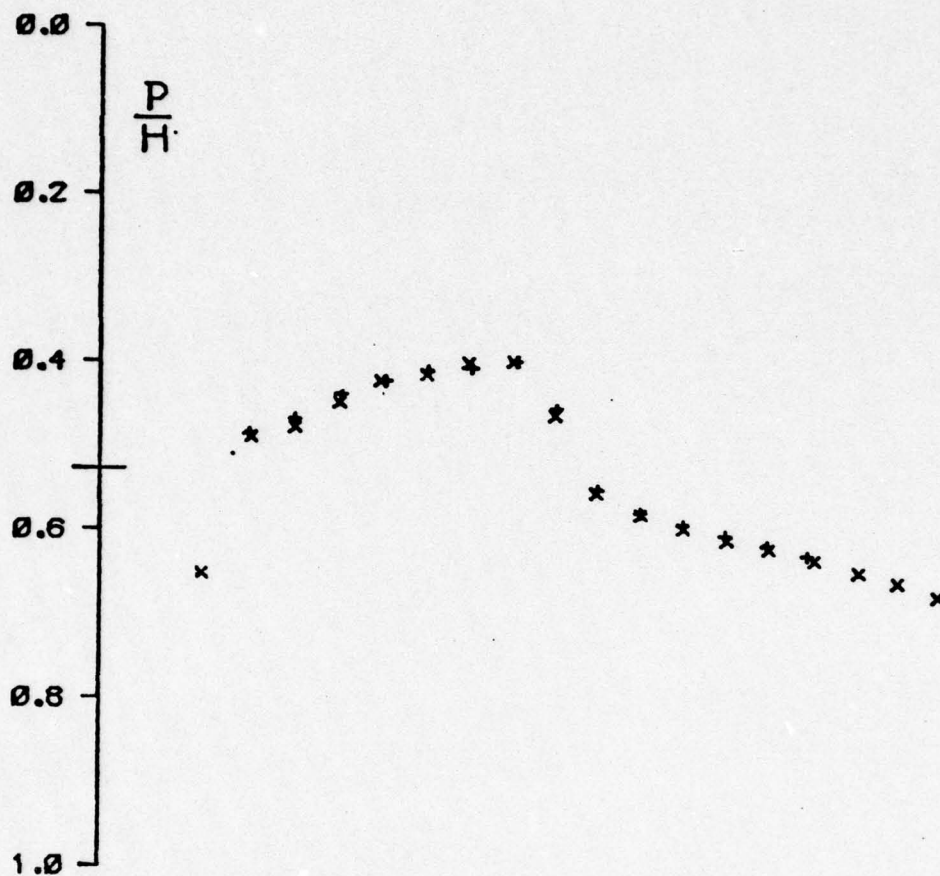
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .781$ $AL = 1.00$ $CN = 0.150$ $CM = 0.007$ $R = 0.799$



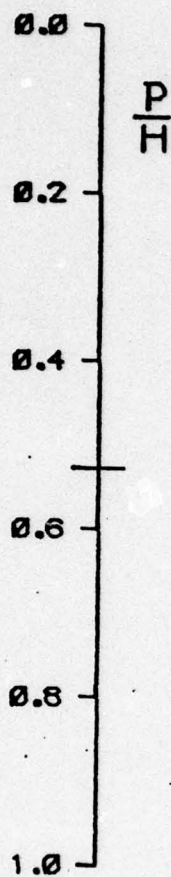
+ x NACA 0012 101.5 MM CHORD SLOTTED WALLS
 M= .780 AL= 2.00 CN= 0.287 CM= 0.006 R= 0.796



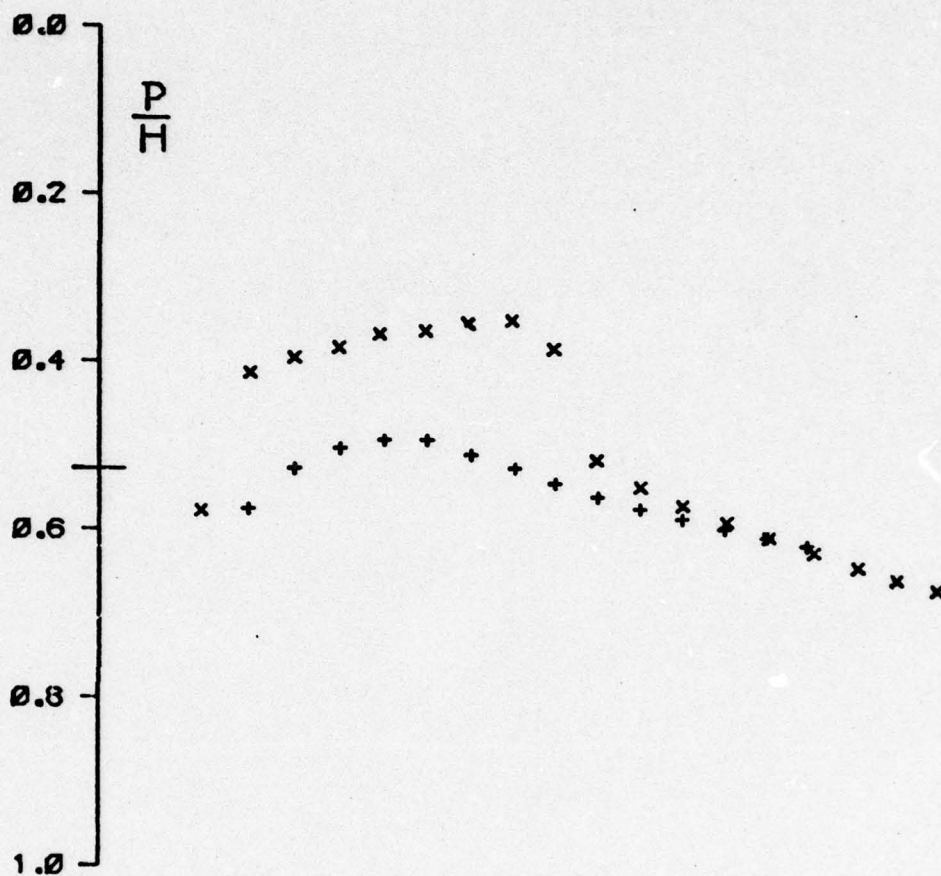
+ x NACA 001: 31.6 MM CHORD SLOTTED WALLS
 M= .779 AL= 3.00 CN= 0.339 CM= 0.001 R= 0.798



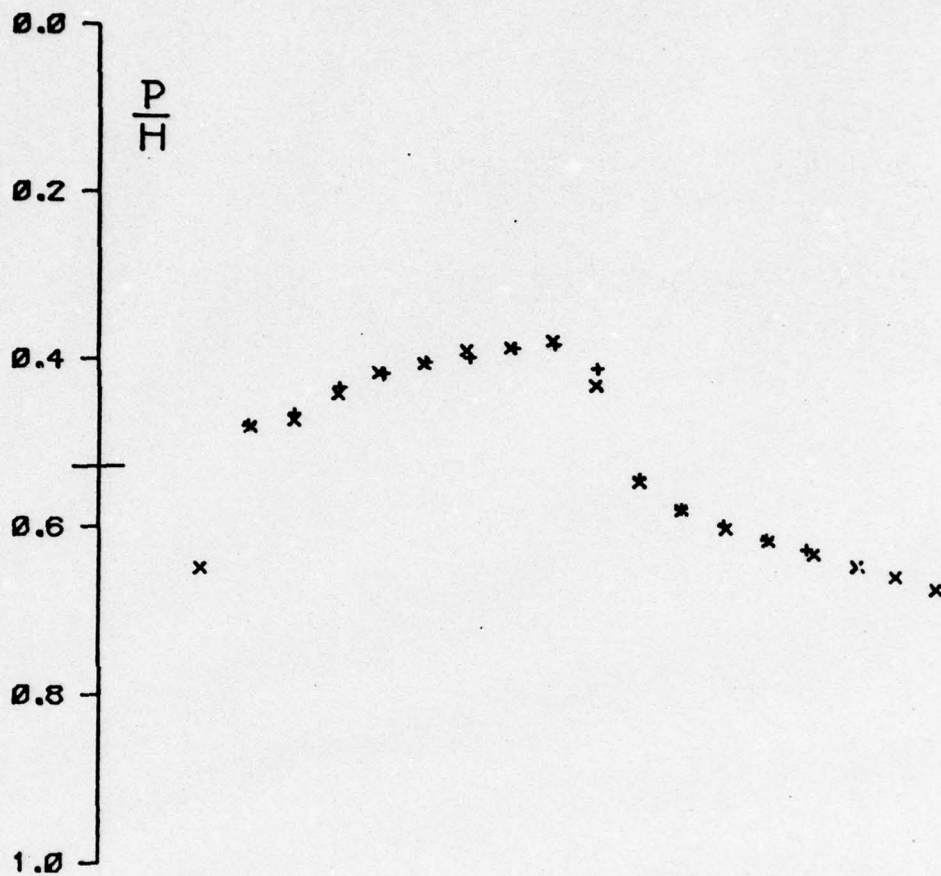
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .801$ $AL = 0.00$ $CN = 0.002$ $CM = 0.002$ $R = 0.812$



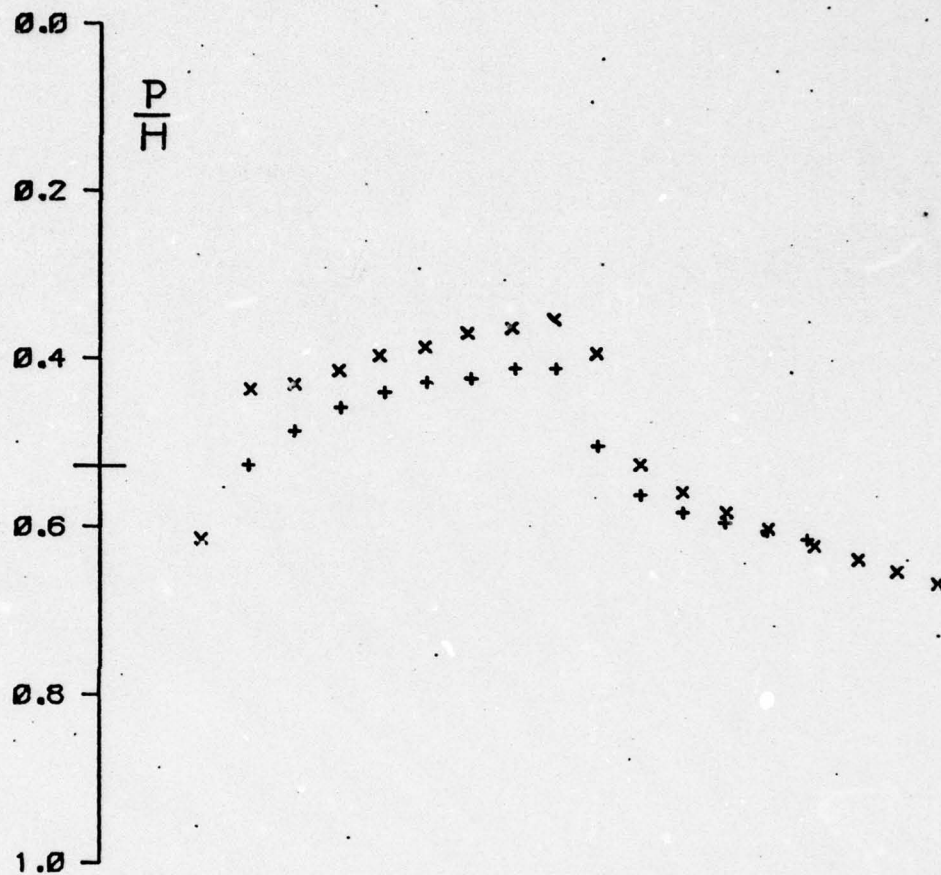
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .800$ $AL = 1.00$ $CN = 0.139$ $CM = 0.001$ $R = 0.809$



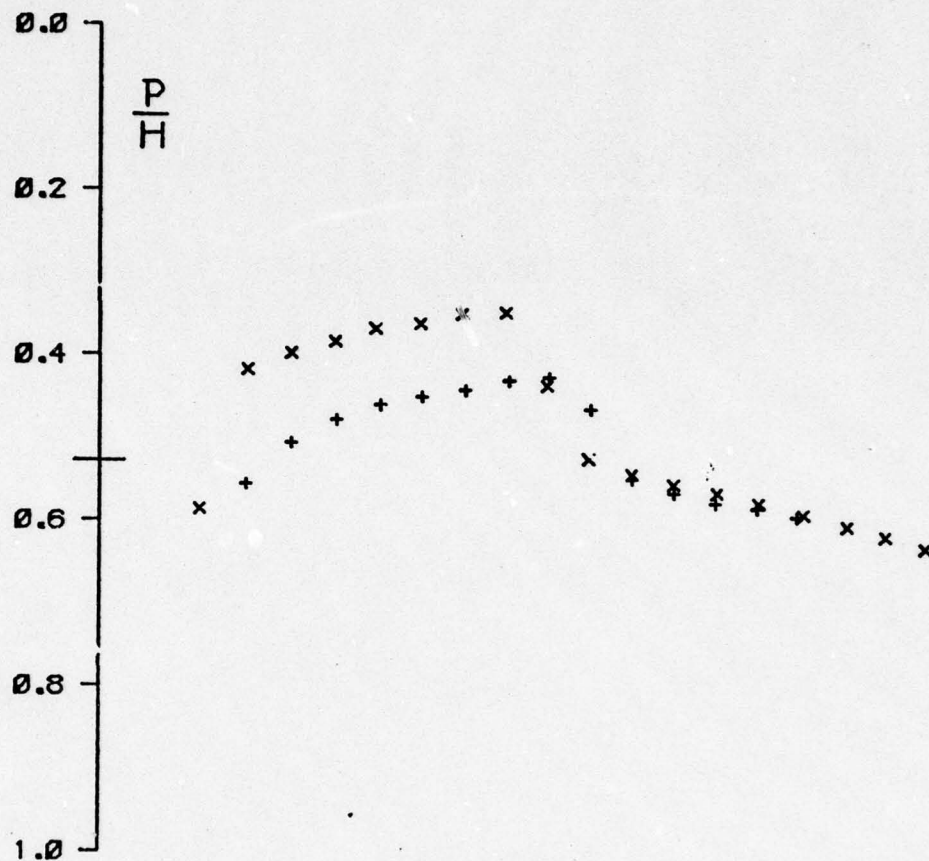
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .800$ $AL = 2.00$ $CN = 0.248$ $CM = 0.003$ $R = 0.812$



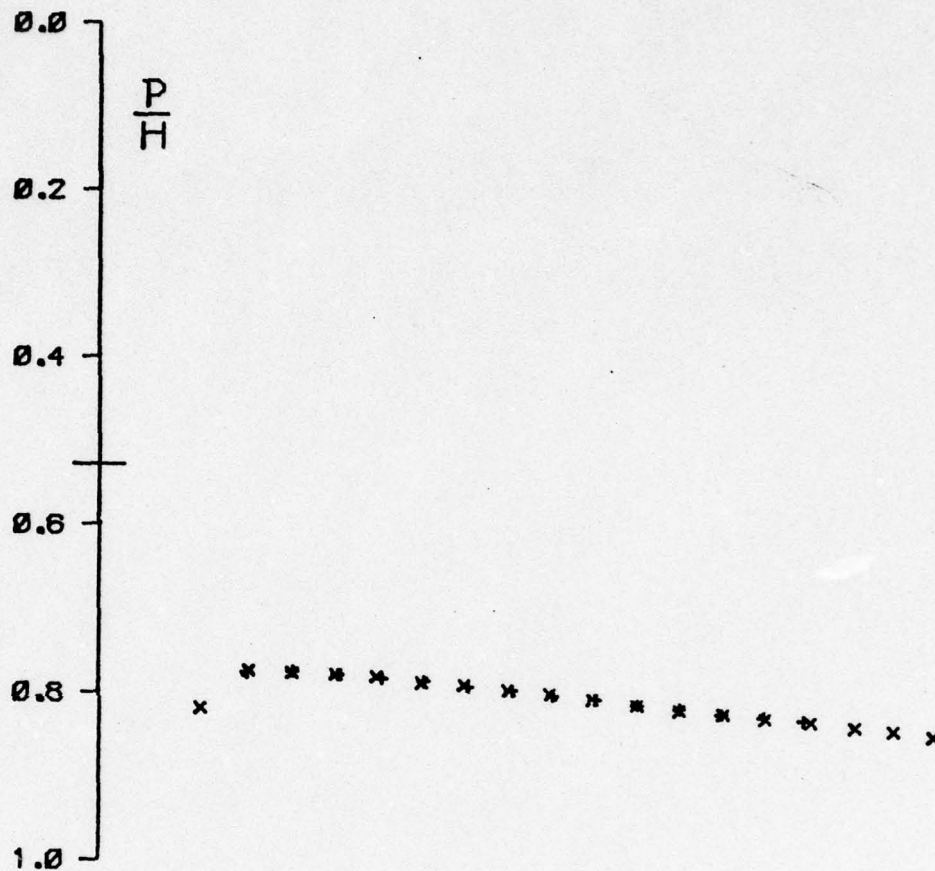
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
M= .819 AL= 0.00 CN= 0.001 CM= 0.002 R= 0.814



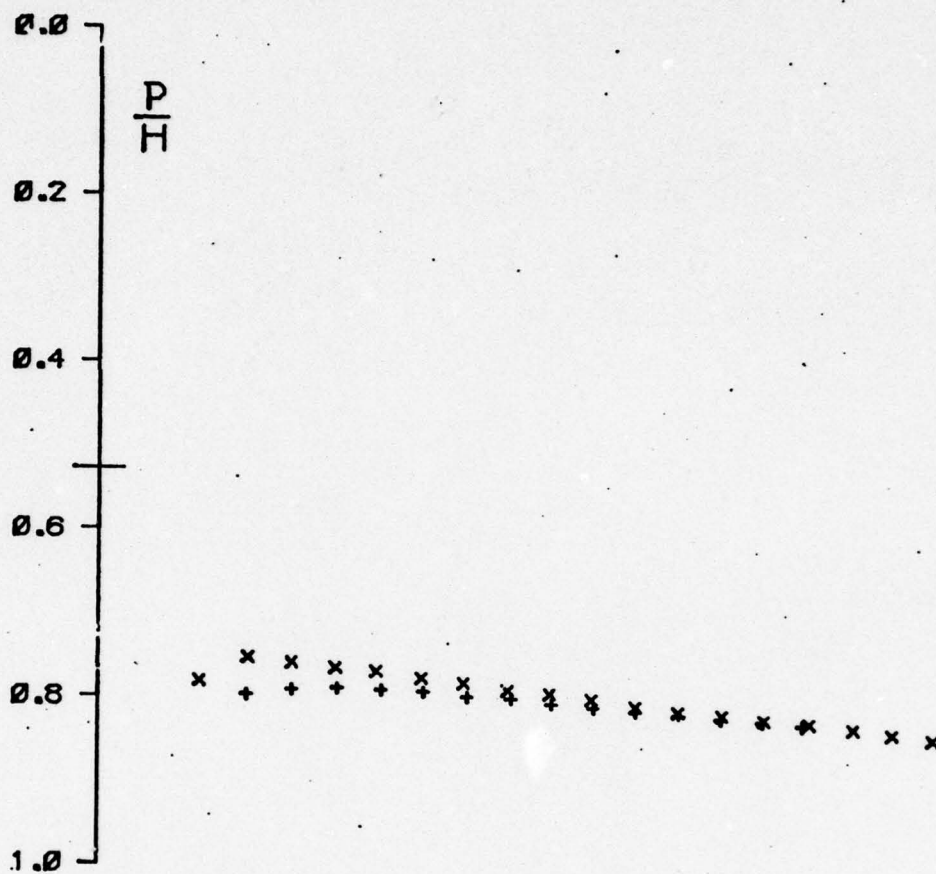
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .820 AL= 1.00 CN= 0.103 CM=-0.005 R= 0.814



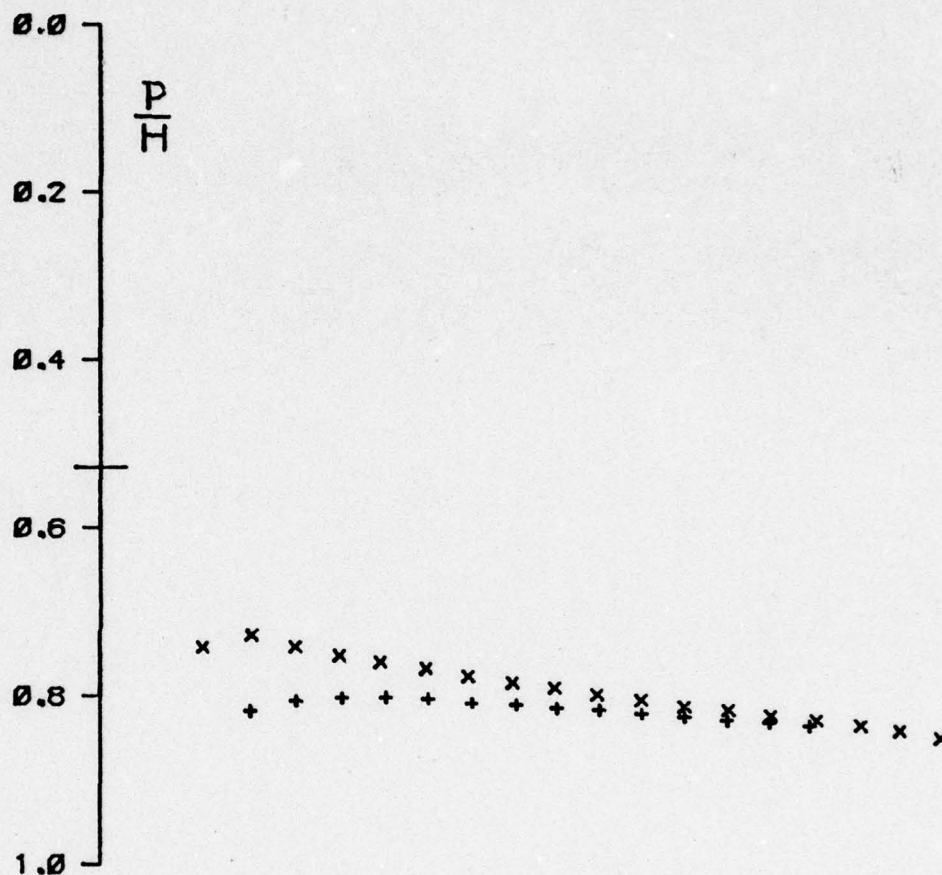
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
M= .820 AL= 2.00 CN= 0.143 CM= 0.008 R= 0.819



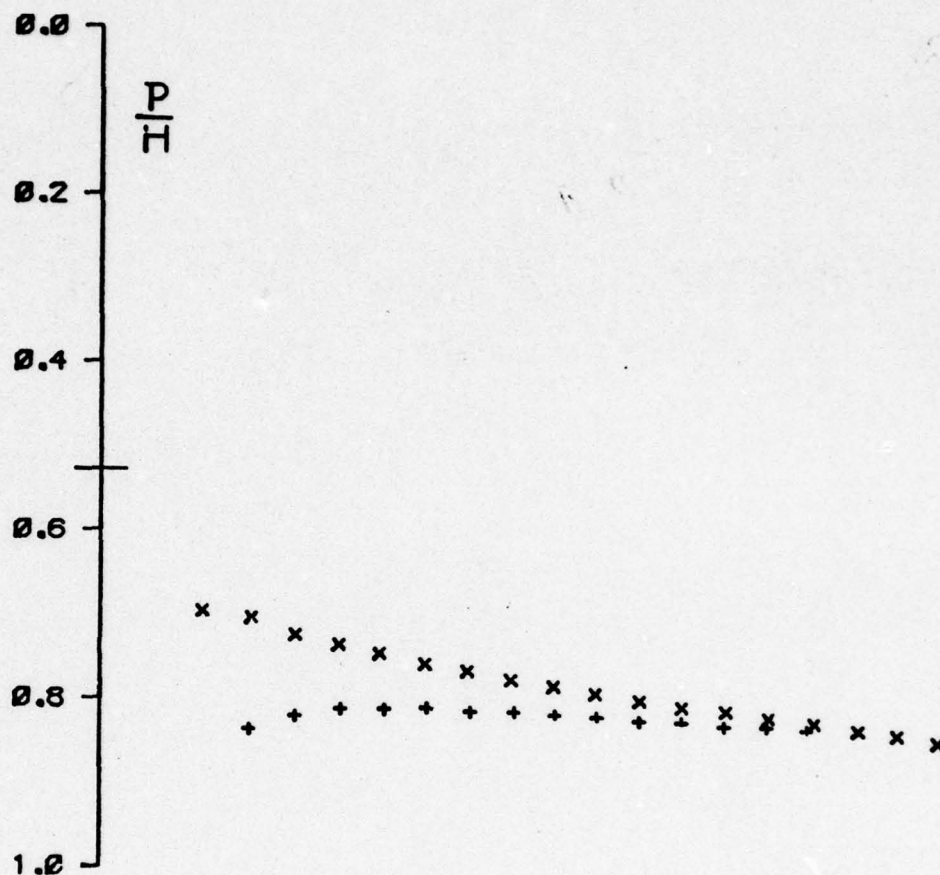
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .499 AL= 0.00 CN= 0.016 CM= 0.203 R= 0.409



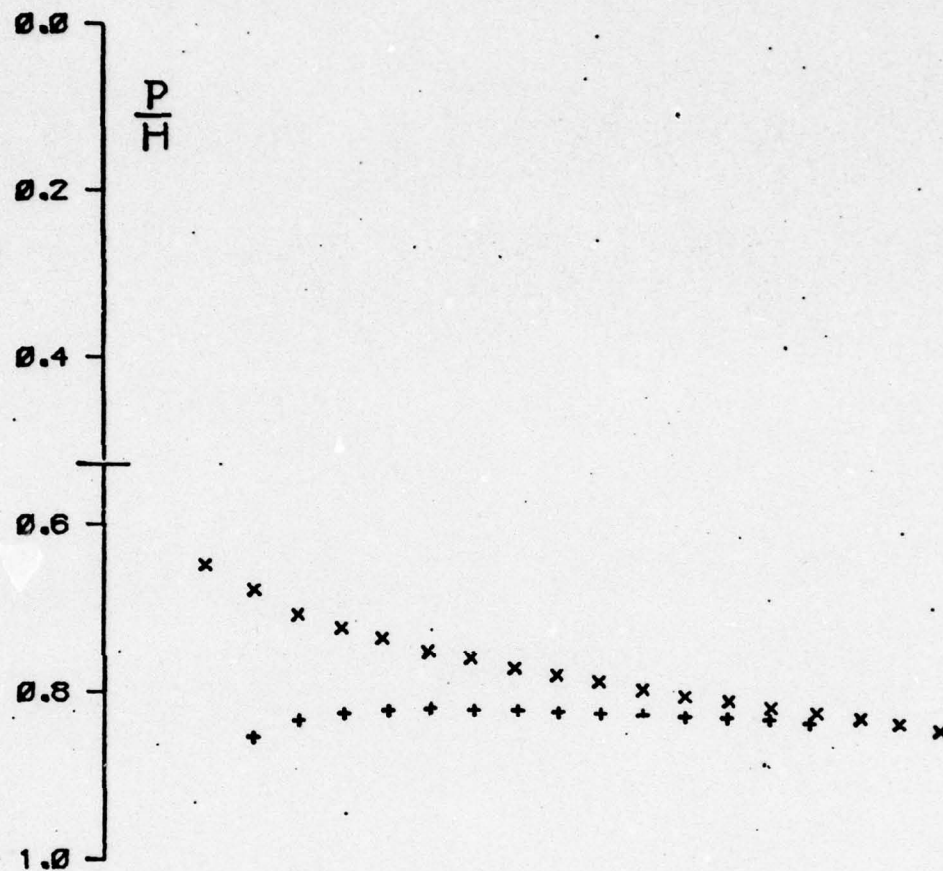
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .499 AL= 1.00 CN= 0.091 CM= 0.001 R= 0.409



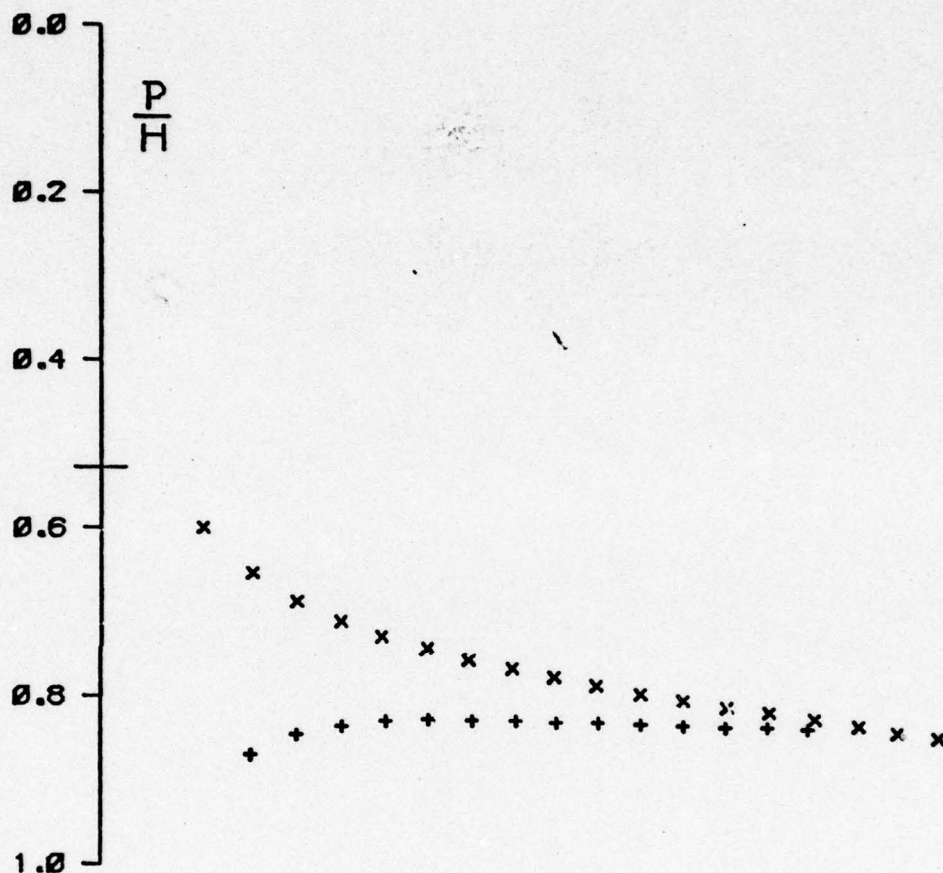
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .500 AL= 2.00 CN= 0.184 CM= 0.002 R= 0.409



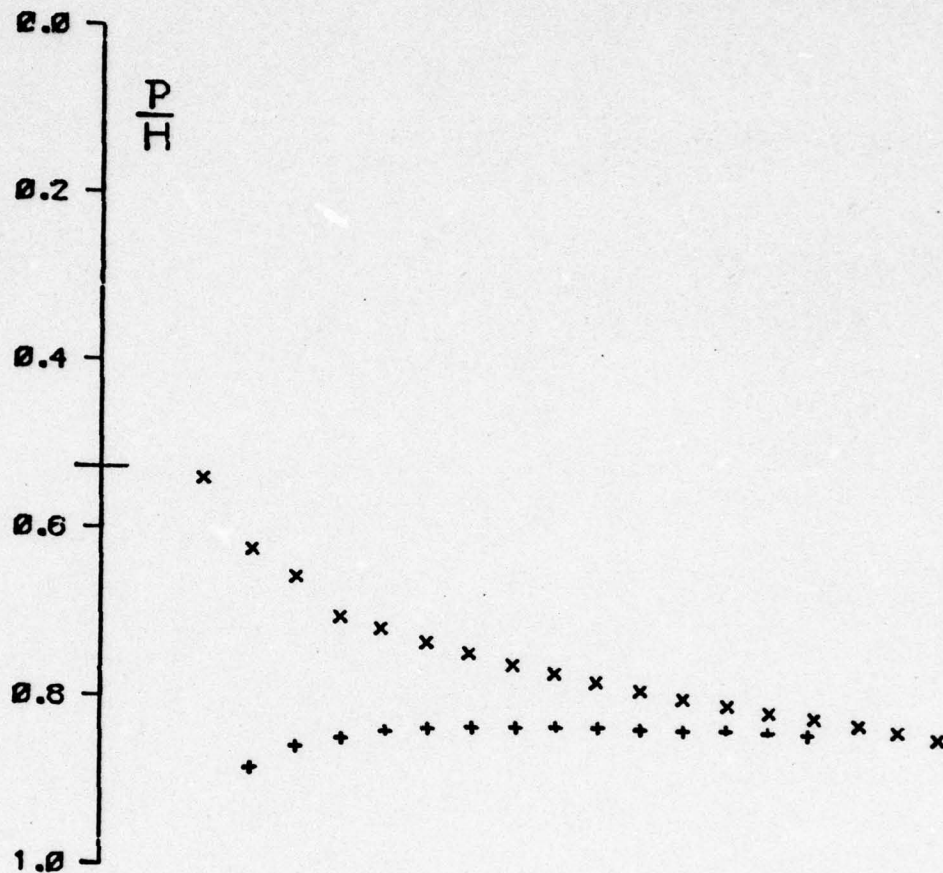
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .501$ $AL = 3.00$ $CN = 0.282$ $CM = 0.003$ $R = 0.412$



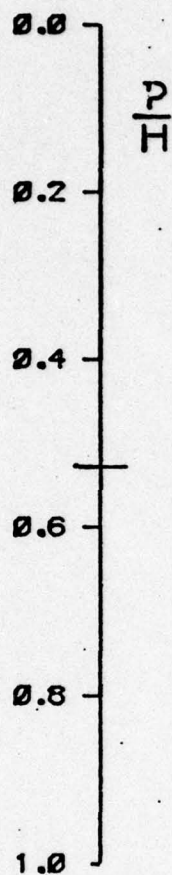
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .500 AL= 4.00 CN= 0.378 CM= 0.006 R= 0.412



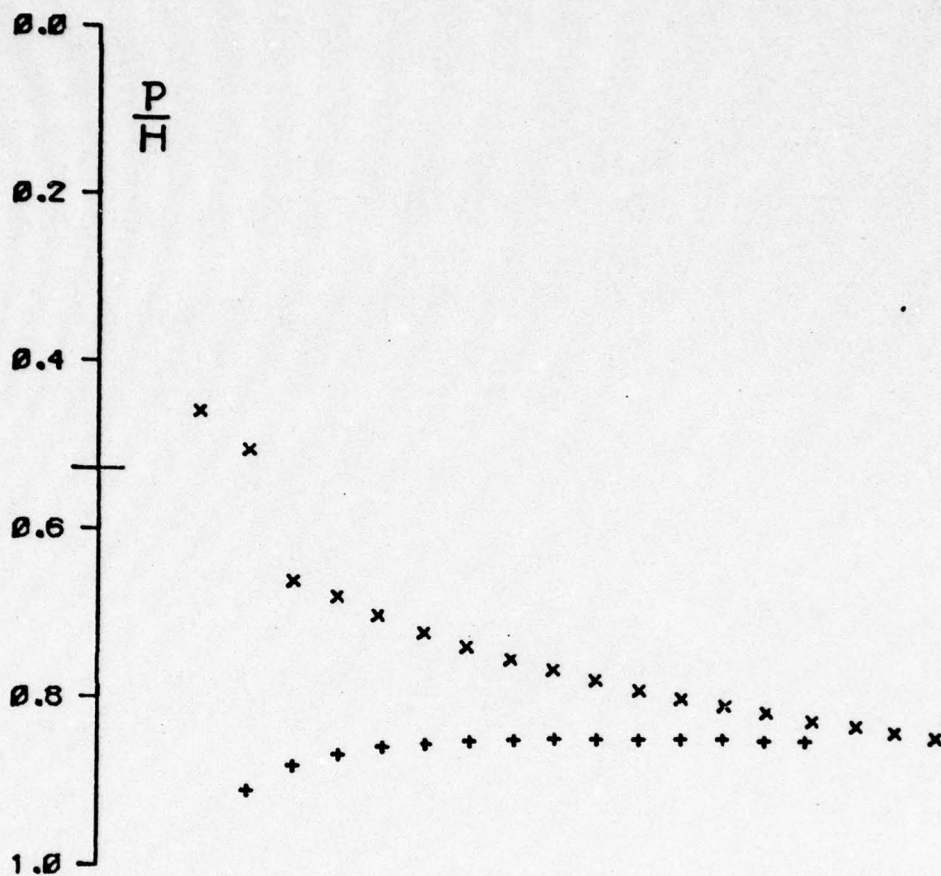
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .498 AL= 5.00 CN= 0.472 CM= 0.008 R= 0.409



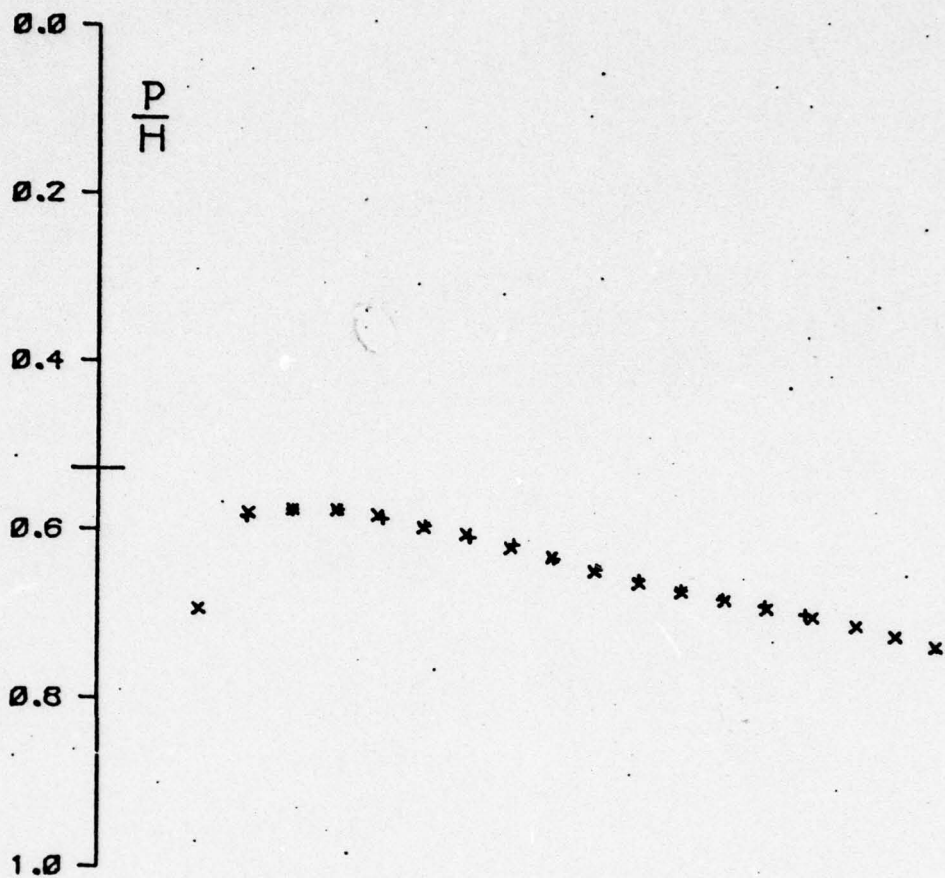
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .498$ $AL = 6.00$ $CN = 0.574$ $CM = 0.010$ $R = 0.409$



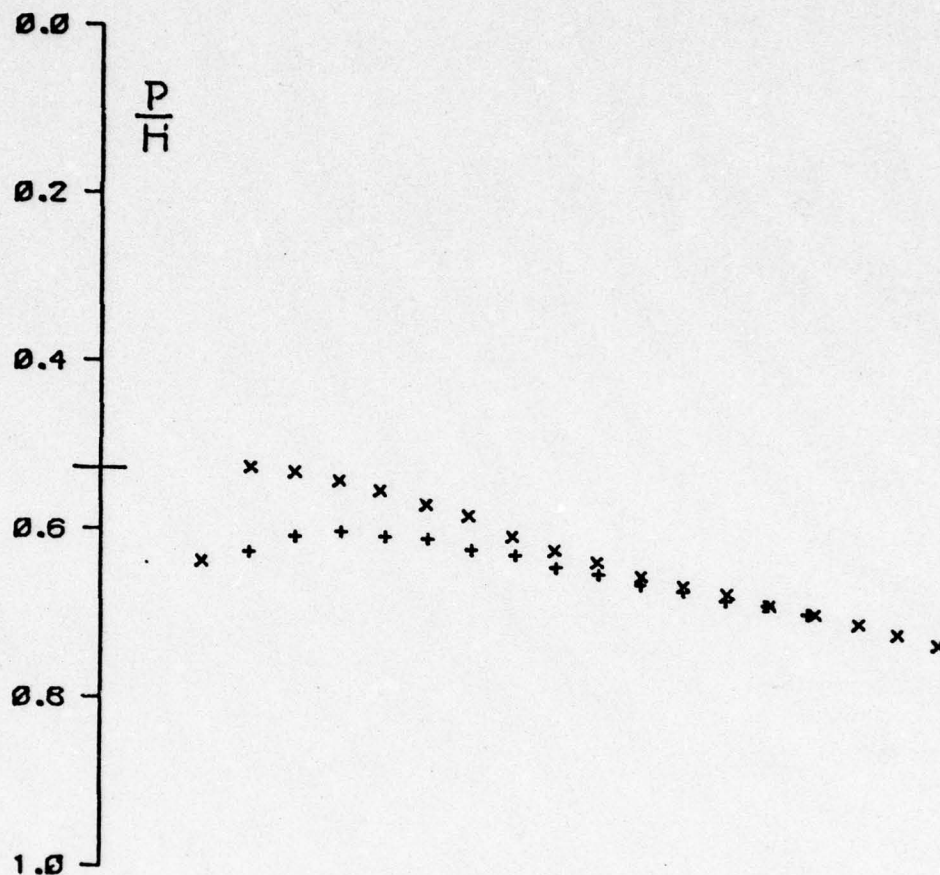
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .499 AL= 7.00 CN= 0.673 CM= 0.010 R= 0.412



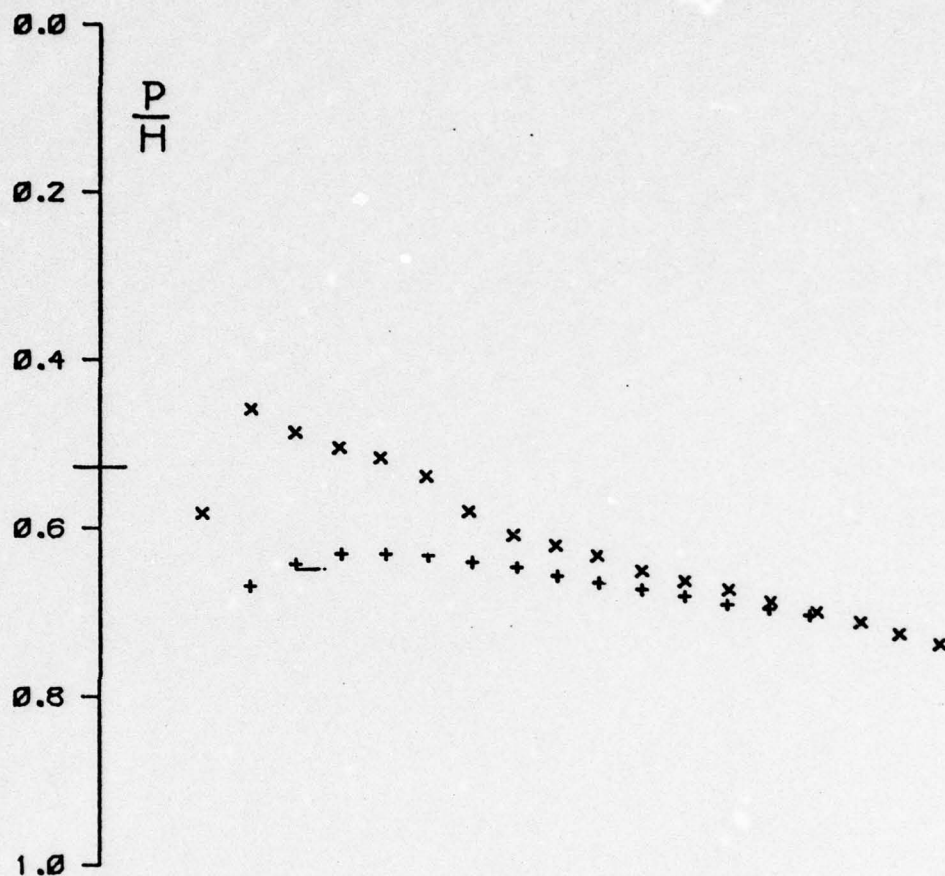
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .499 AL= 8.00 CN= 0.759 CM= 0.018 R= 0.412



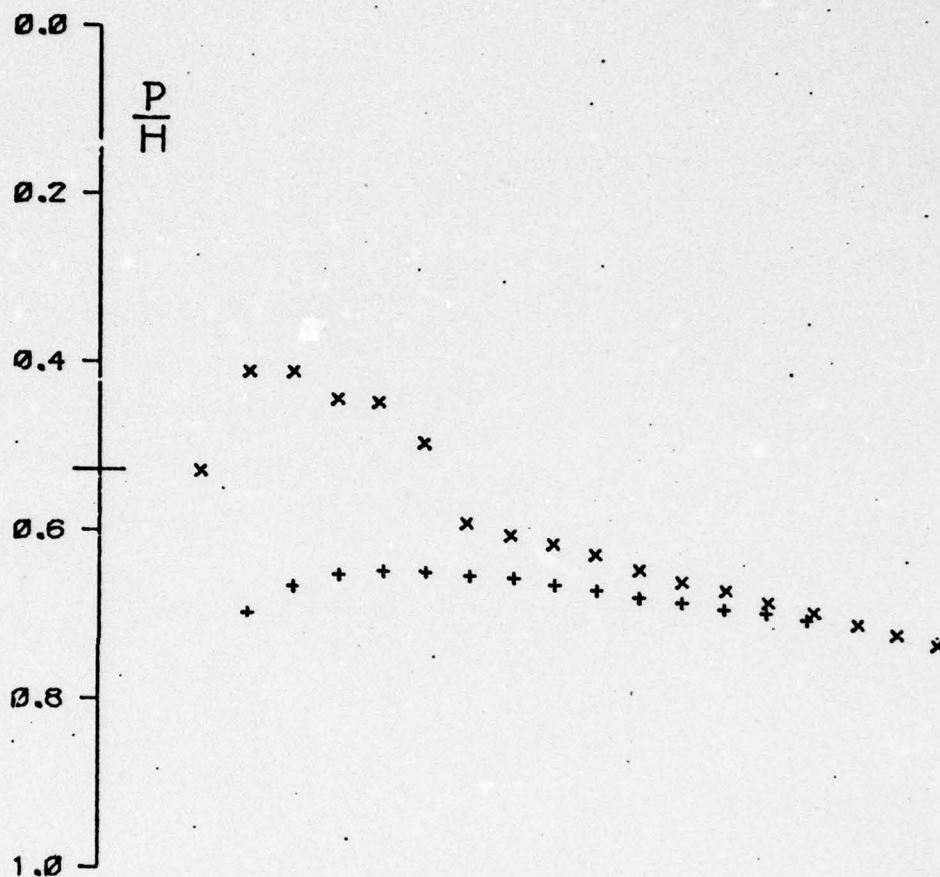
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .639 AL= 0.00 CN= 0.010 CM= 0.003 R= 0.414



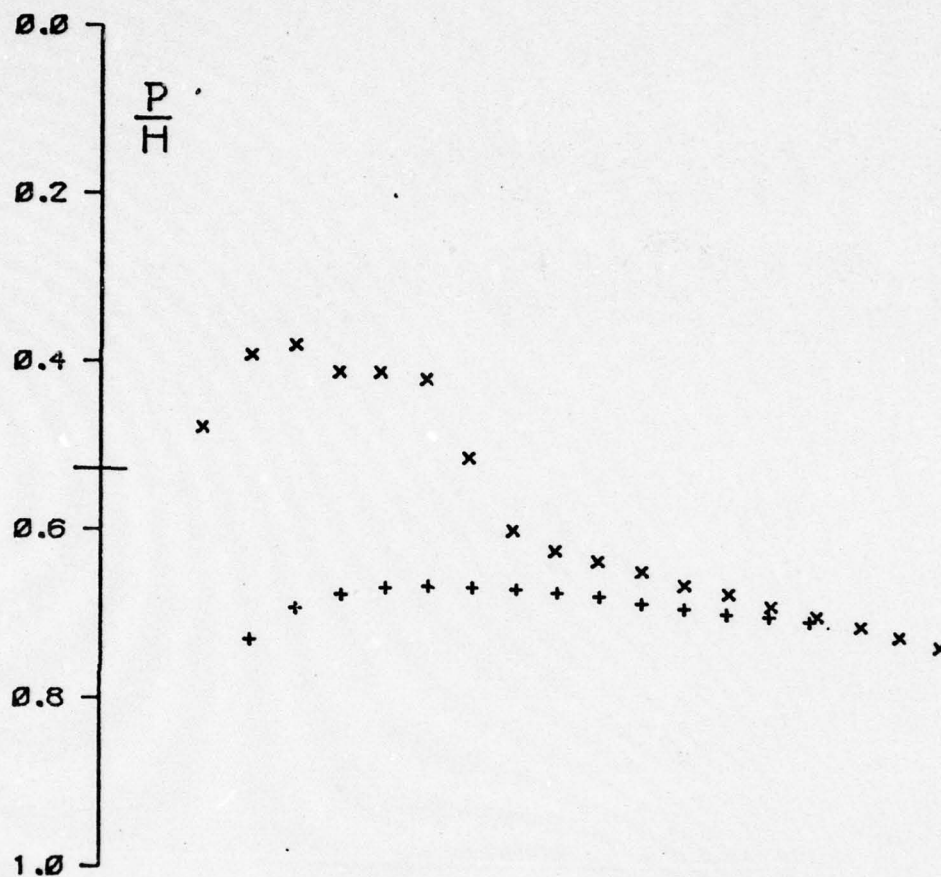
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .701$ $AL = 1.00$ $CN = 0.110$ $CM = 0.003$ $R = 0.416$



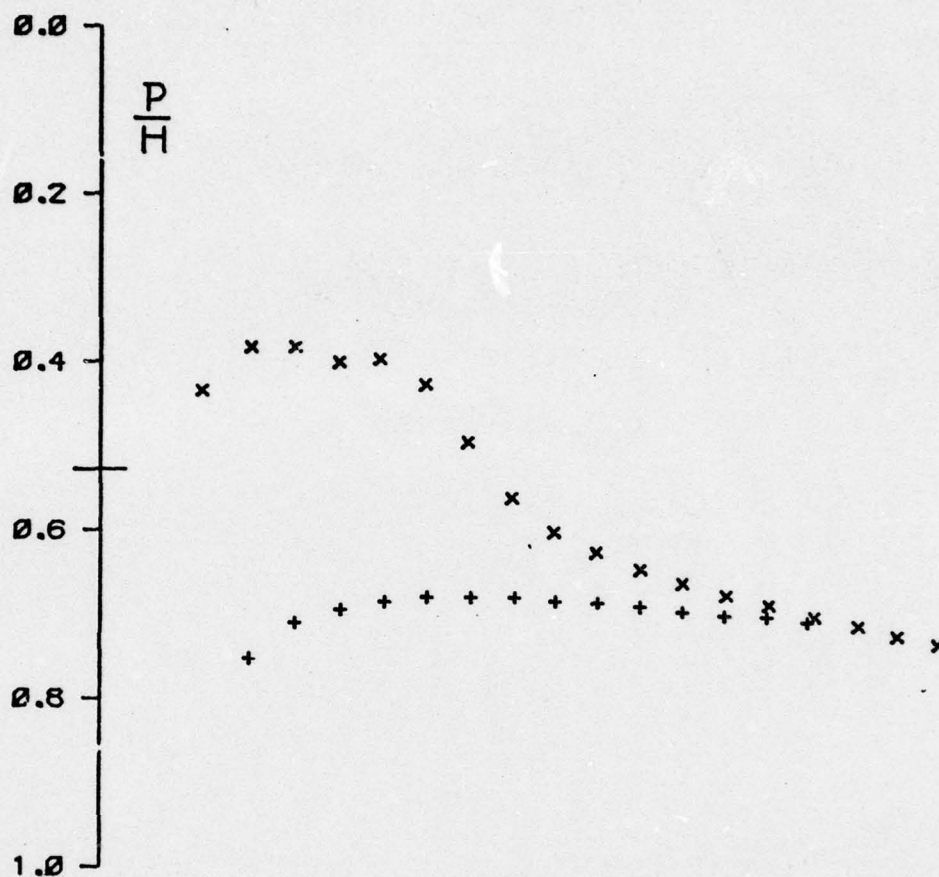
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .699$ $AL = 2.00$ $CN = 0.248$ $CM = 0.011$ $R = 0.417$



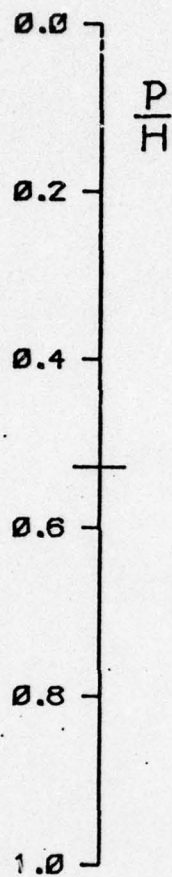
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $M = .701$ $AL = 3.00$ $CN = 0.352$ $CM = 0.015$ $R = 0.416$



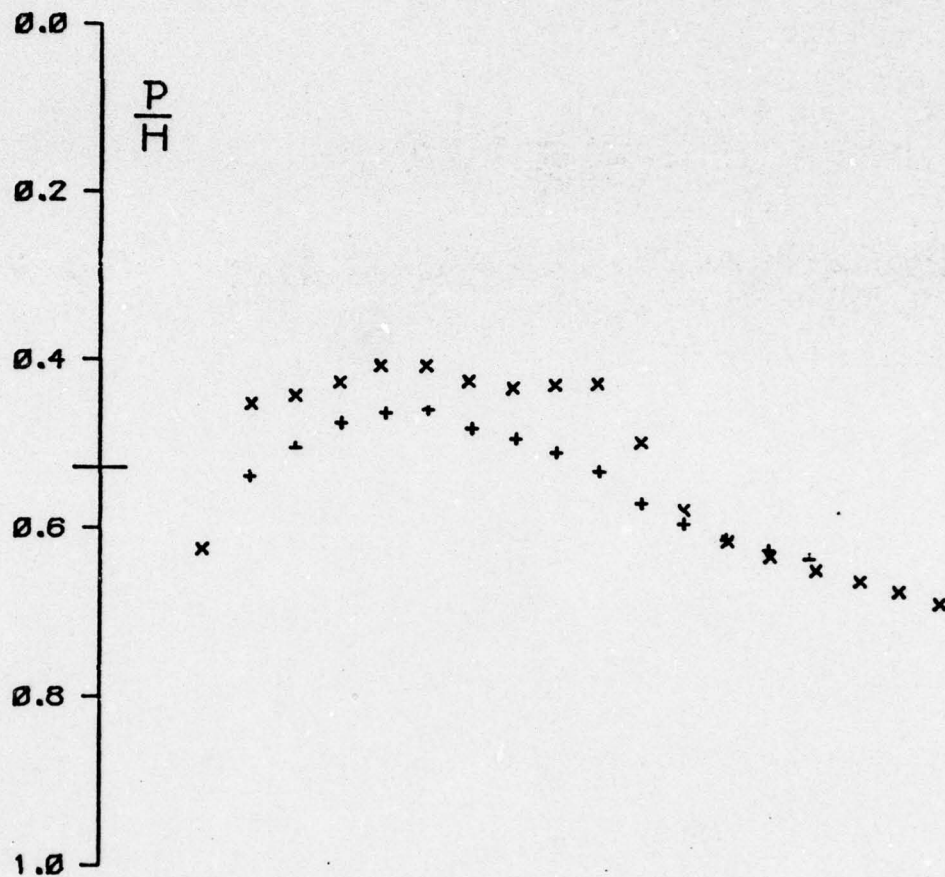
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .699 AL= 4.00 CN= 0.446 CM= 0.018 R= 0.417



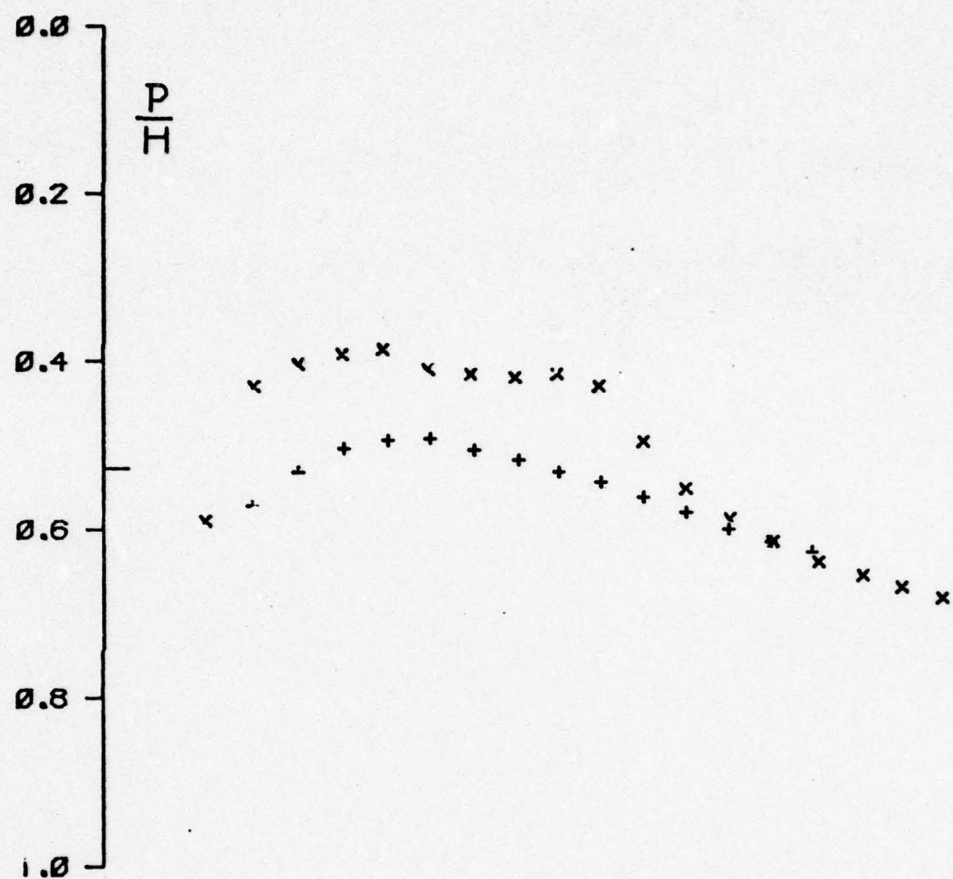
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
M= .699 AL= 5.00 CN= 0.514 CM= 0.017 R= 0.417



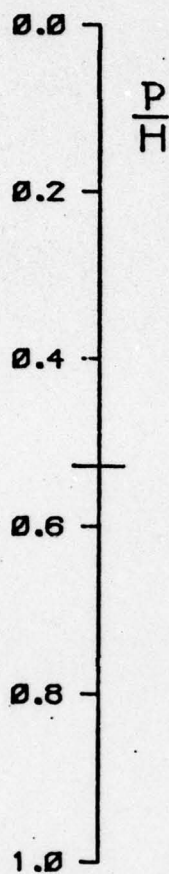
+ x . NACA 0012 101.6 MM CHORD SLOTTED WALLS
 M= .801 AL= 0.20 CN= 0.001 CM= 0.001 R= 0.406



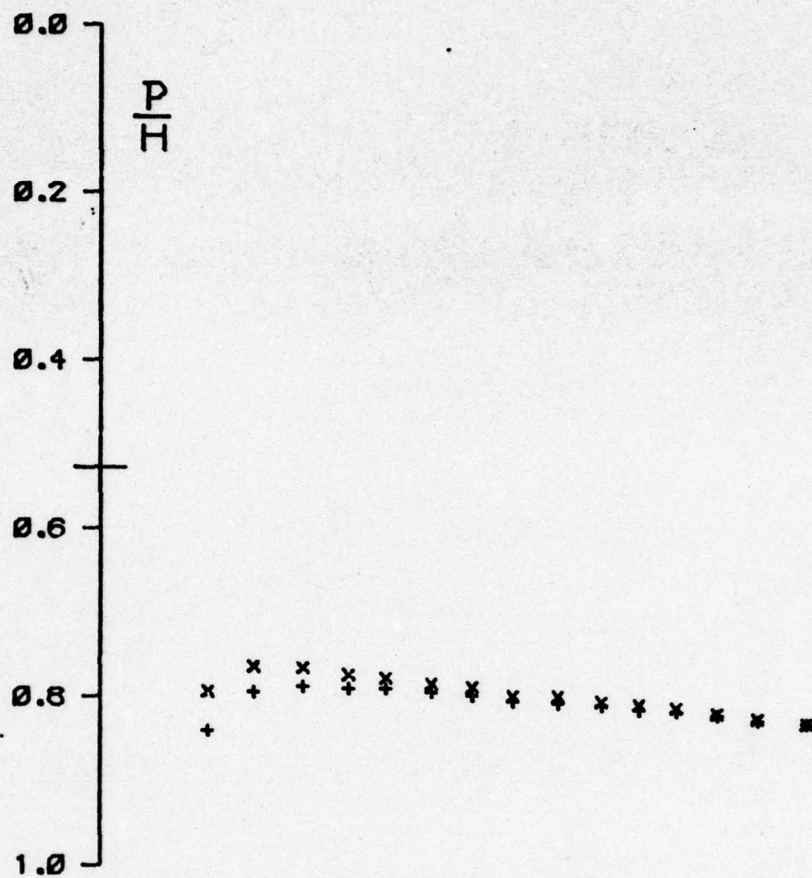
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $V = .799$ $AL = 1.00$ $CN = 0.154$ $CM = 0.002$ $R = 0.406$



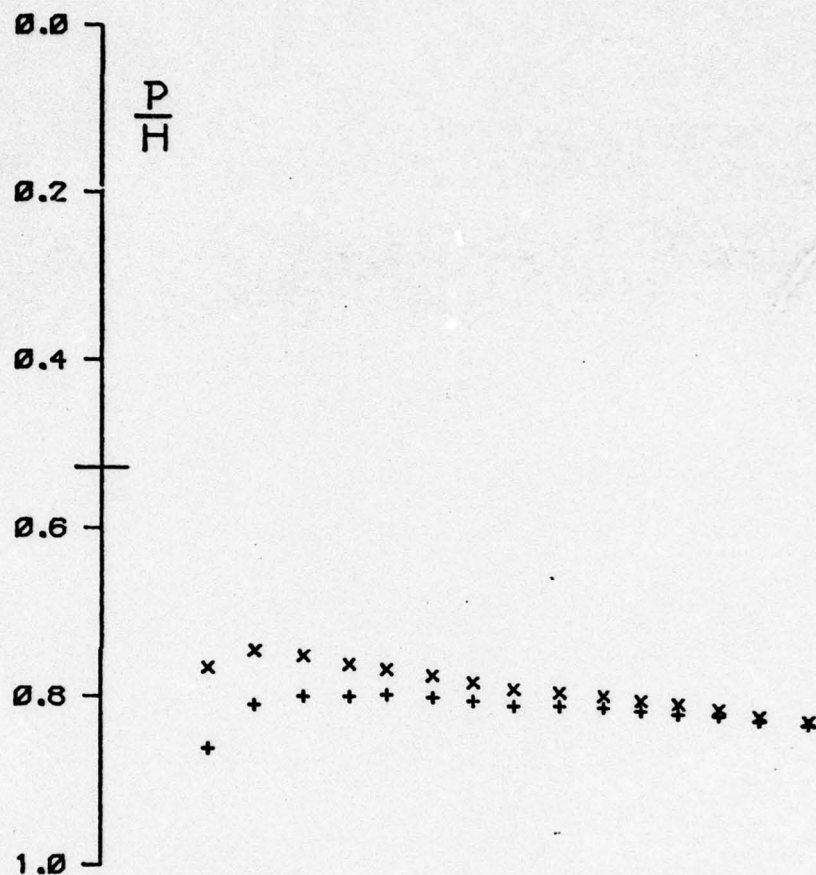
+ x NACA 0012 101.6 MM CHORD SLOTTED WALLS
 $V = .798$ $AL = 2.00$ $CN = 0.208$ $CN = -0.002$ $R = 0.406$



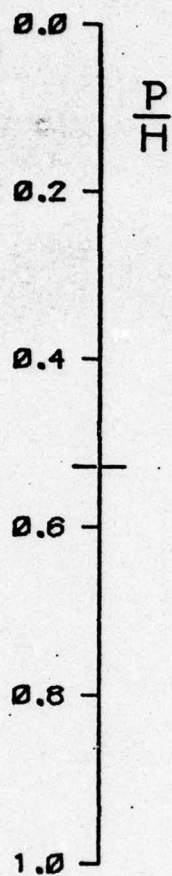
+ x NACA 0012 52.8 MM CHORD SLOTTED WALLS
 $M = .500$ $AL = 0.00$ $CN = -0.003$ $CM = -0.001$ $R = 0.399$



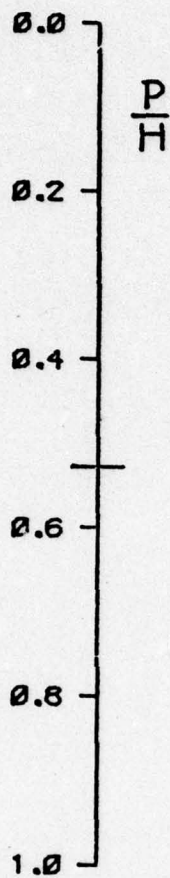
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .497 AL= 0.50 CN= 0.068 CM= 0.000 R= 0.398



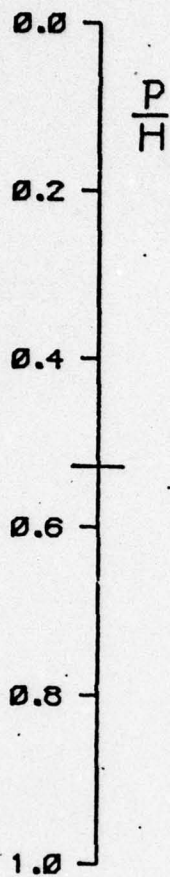
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .501 AL= 1.00 CN= 0.136 CM= 0.001 R= 0.403



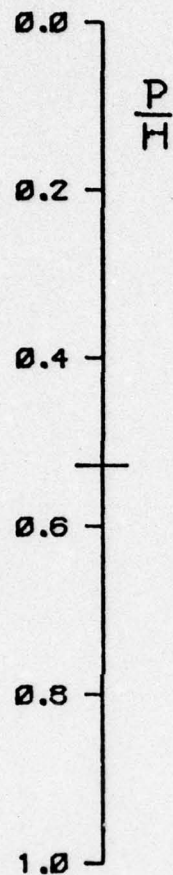
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .500$ $AL = 1.50$ $CN = 0.204$ $CM = 0.003$ $R = 0.403$



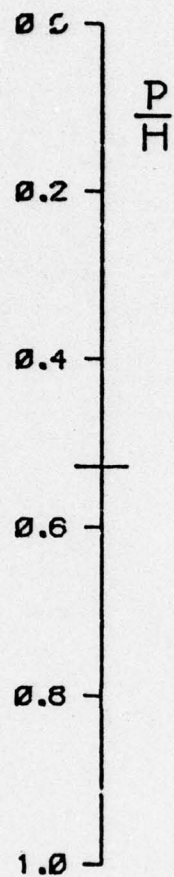
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .50$; $AL = 2.00$ $CN = 0.274$ $CM = 0.004$ $R = 0.403$



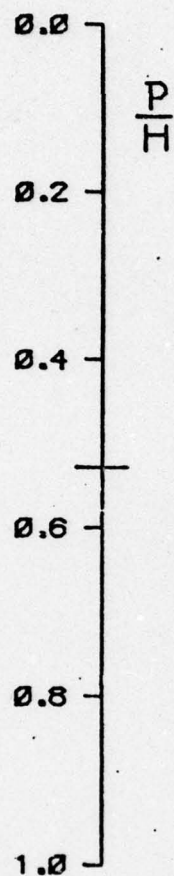
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .600 AL= 0.00 CN=-0.004 CM=-0.001 R= 0.393



+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .599 AL= 0.50 CN= 0.079 CM= 0.000 R= 0.393



+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .599$ $AL = 1.00$ $CN = 0.160$ $CM = 0.003$ $R = 0.393$



+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .600$ $AL = 1.50$ $CN = 0.239$ $CM = 0.004$ $R = 0.393$

AD-A055 004

AERONAUTICAL RESEARCH LABS MELBOURNE (AUSTRALIA)
TRANSONIC WIND TUNNEL TESTS ON A NACA 0012 AEROFOIL, (U)
APR 77 N POLLOCK, B D FAIRLIE

F/G 20/4

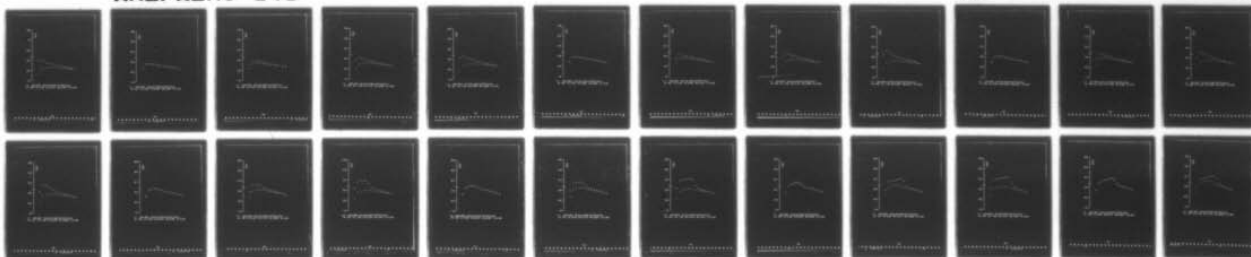
UNCLASSIFIED

ARL/AERO-148

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6 OF 6

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A055004



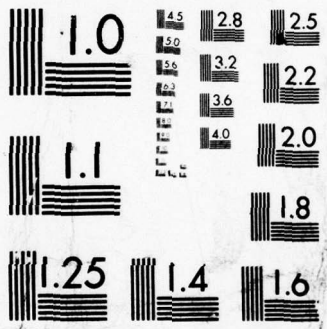
END

DATE

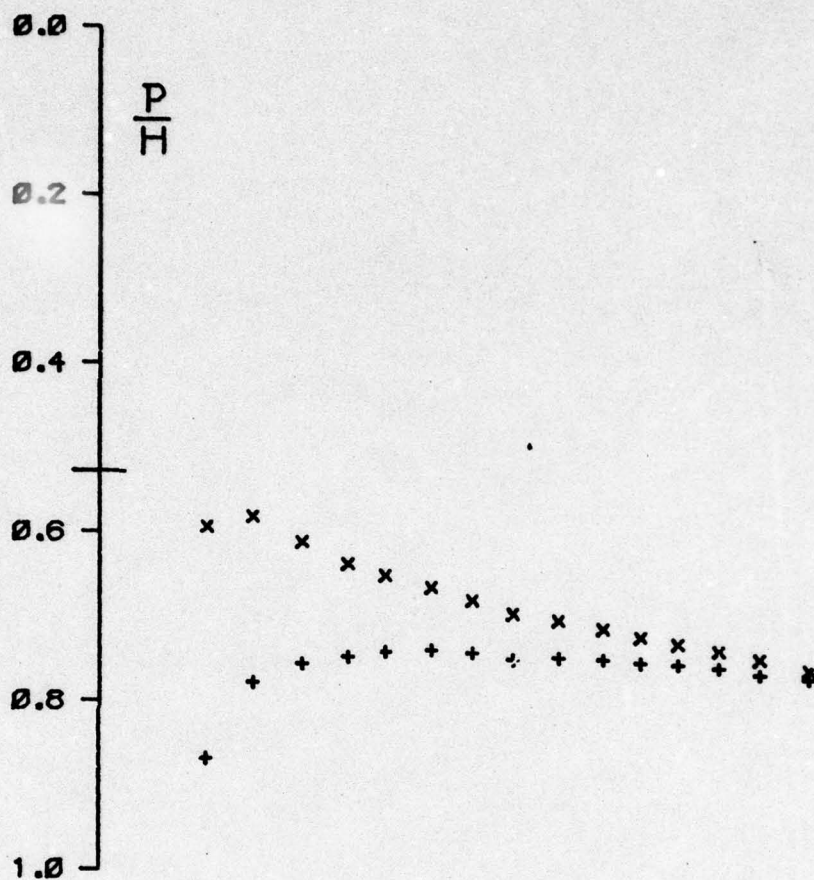
FILMED

7-78

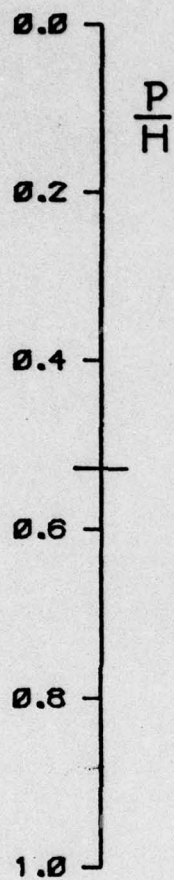
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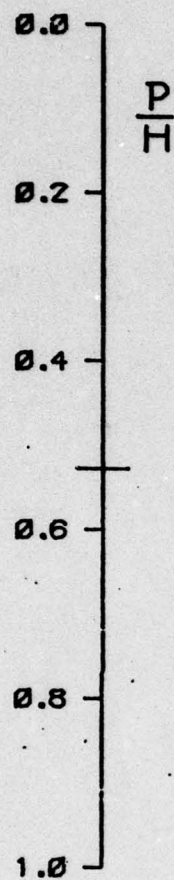
MICROCOPY RESOLUTION TEST CHART
NATIONAL BUREAU OF STANDARDS-1963-A

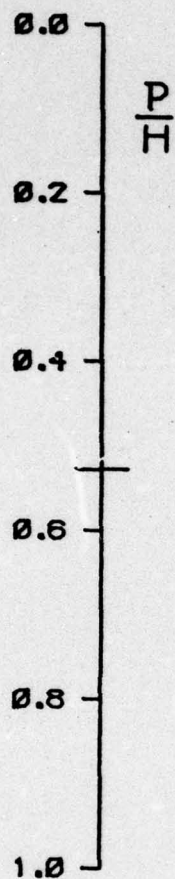


+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .598 AL= 2.00 CN= 0.314 CM= 0.007 R= 0.393

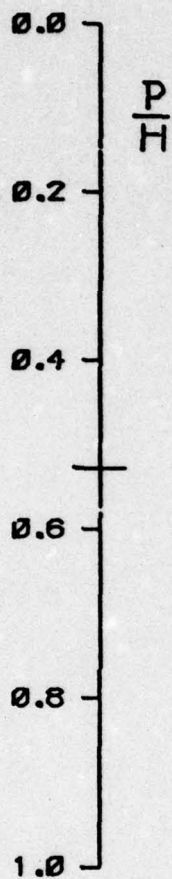


+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .651 AL= 0.00 CN=-0.003 CM=-0.001 R= 0.394

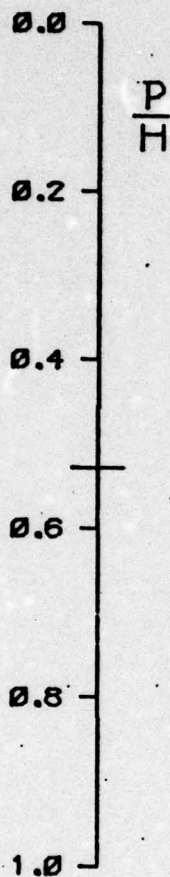




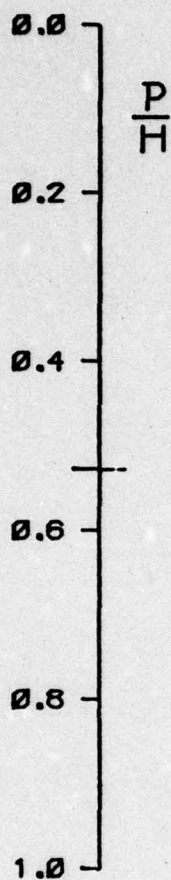
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .649$ $AL = 1.00$ $CN = 0.178$ $CM = 0.004$ $R = 0.394$



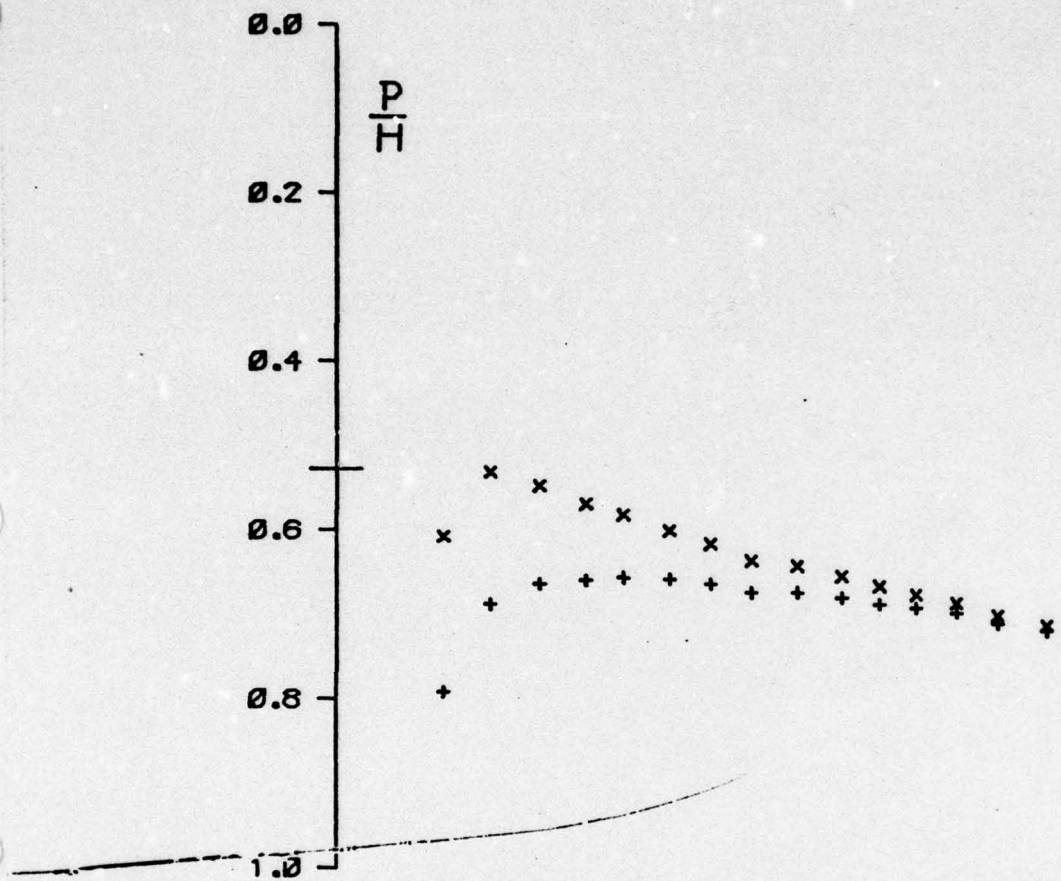
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .651$ $AL = 1.50$ $CN = 0.269$ $CM = 0.008$ $R = 0.394$



+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .676 AL= 0.00 CN=-0.005 CM=-0.001 R= 0.399



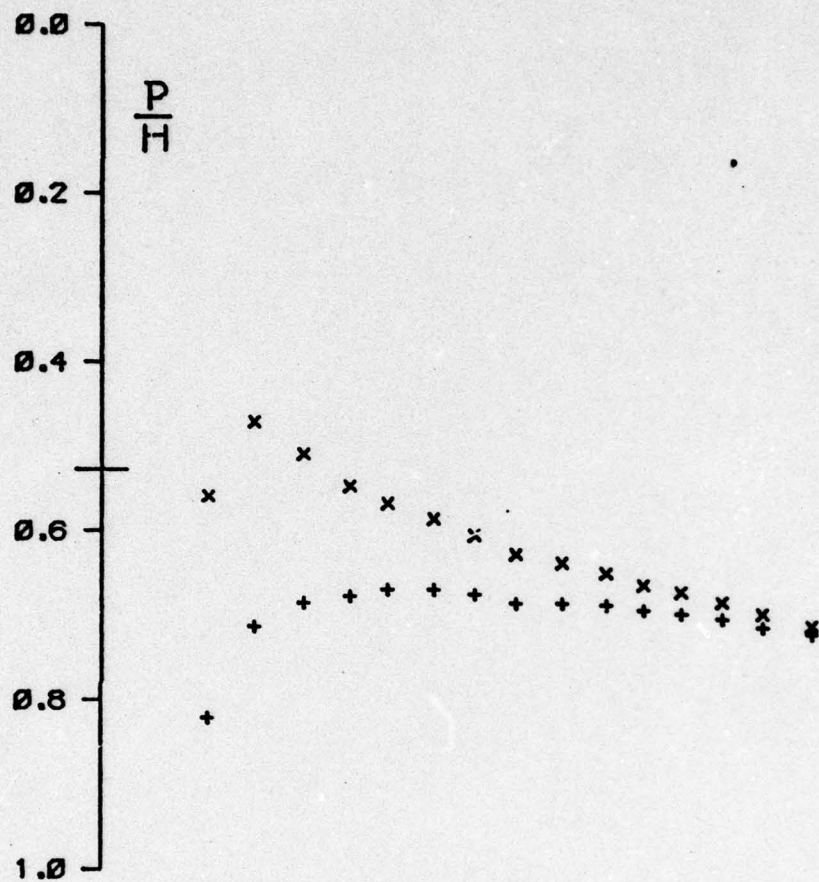
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .675$ $AL = 0.50$ $CN = 0.095$ $CM = 0.002$ $R = 0.399$



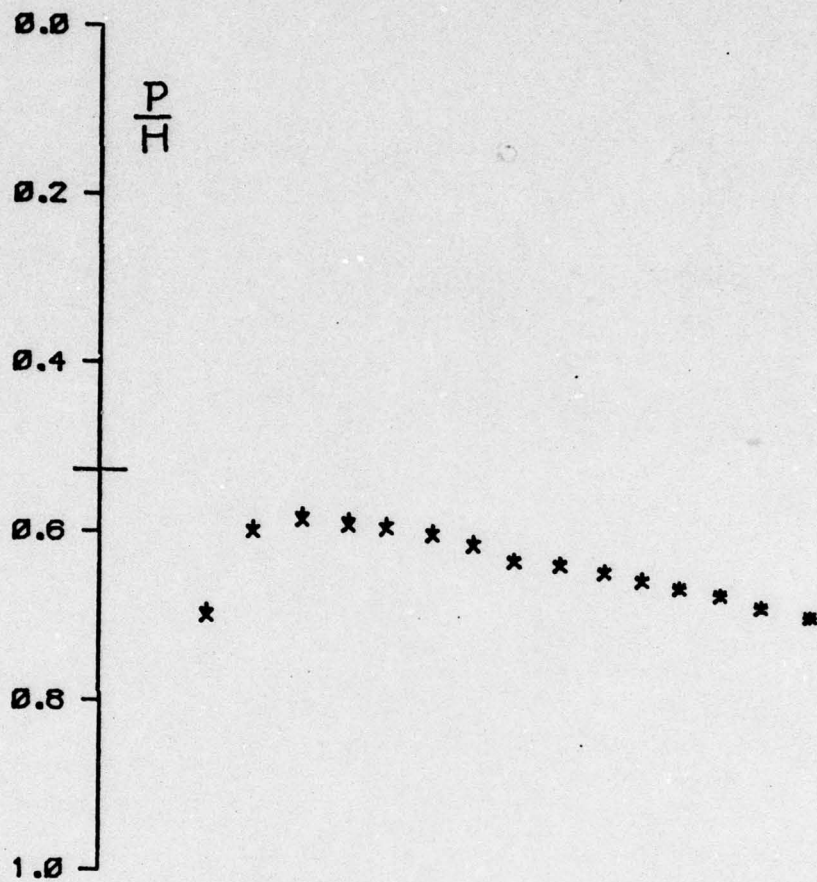
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .676$ $AL = 1.00$ $CN = 0.191$ $CM = 0.005$ $R = 0.399$

200.

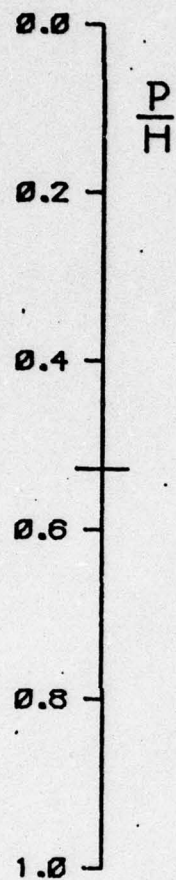
5475F



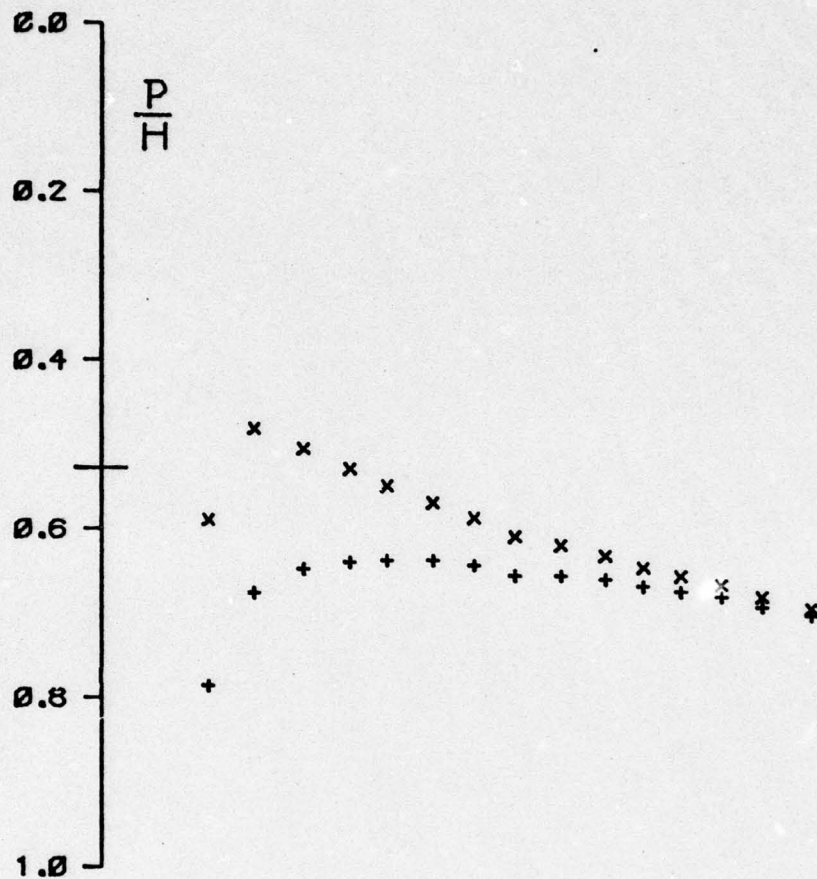
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .676 AL= 1.50 CN= 0.291 CM= 0.011 R= 0.399



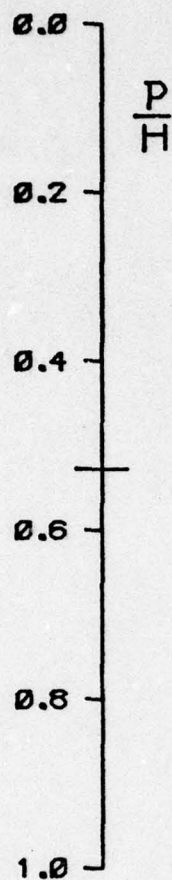
+ X : JACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .009 AL= 0.00 CN=0.010 CM=0.001 R= 0.391



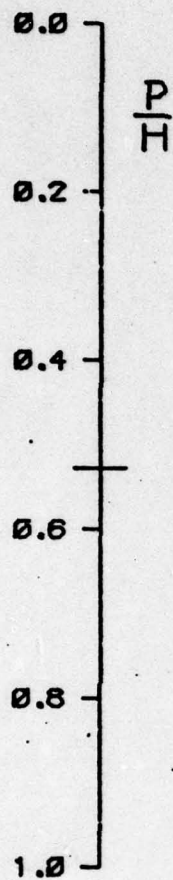
+ x NACA 0012 50.0 MM CHORD SLOTTED WALLS
 $M = .700$ $AL = 0.50$ $CN = 0.105$ $CM = 0.003$ $R = 0.391$



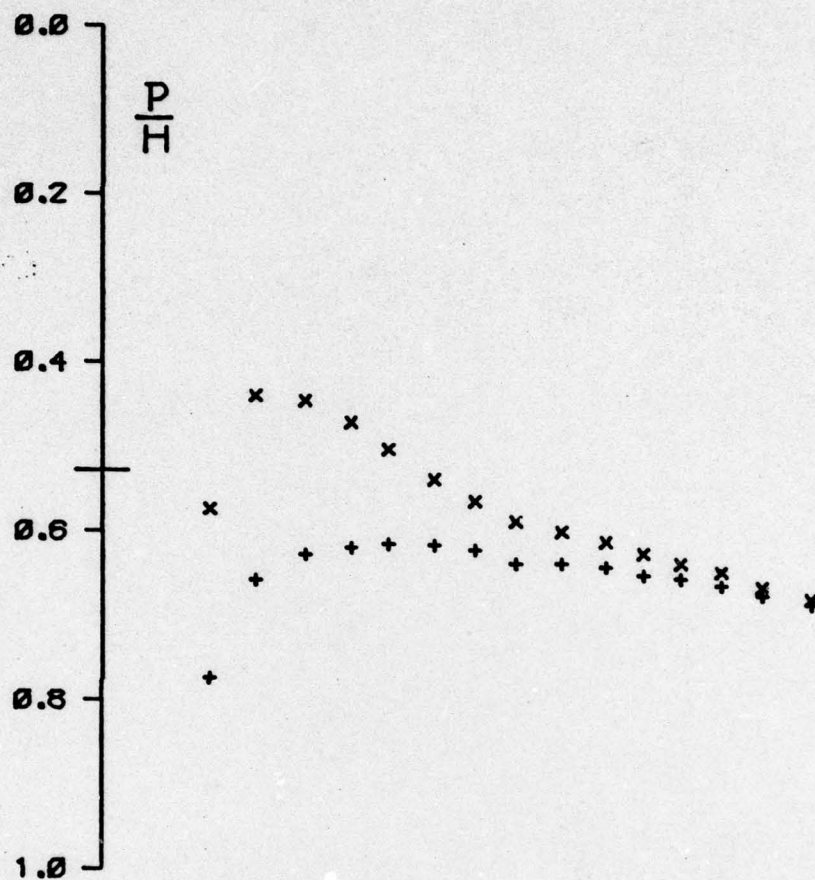
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .699 AL= 1.00 CN= 0.210 CM= 0.007 R= 0.391



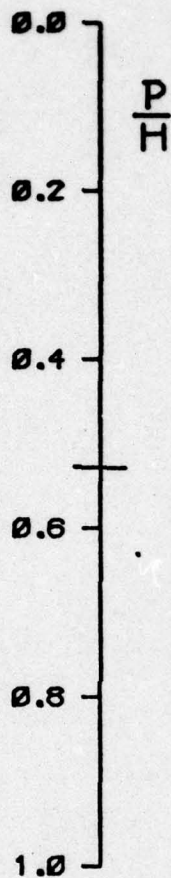
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .720 AL= 0.00 CN=-0.003 CM=-0.001 R= 0.393



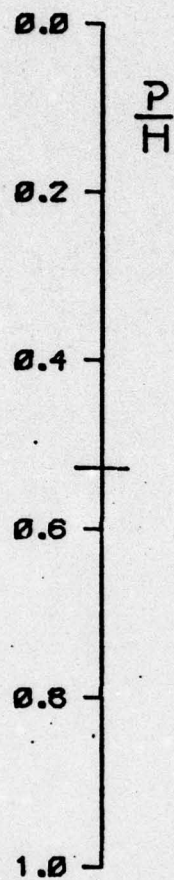
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .721$ $AL = 0.50$ $CN = 0.115$ $CM = 0.004$ $R = 0.393$



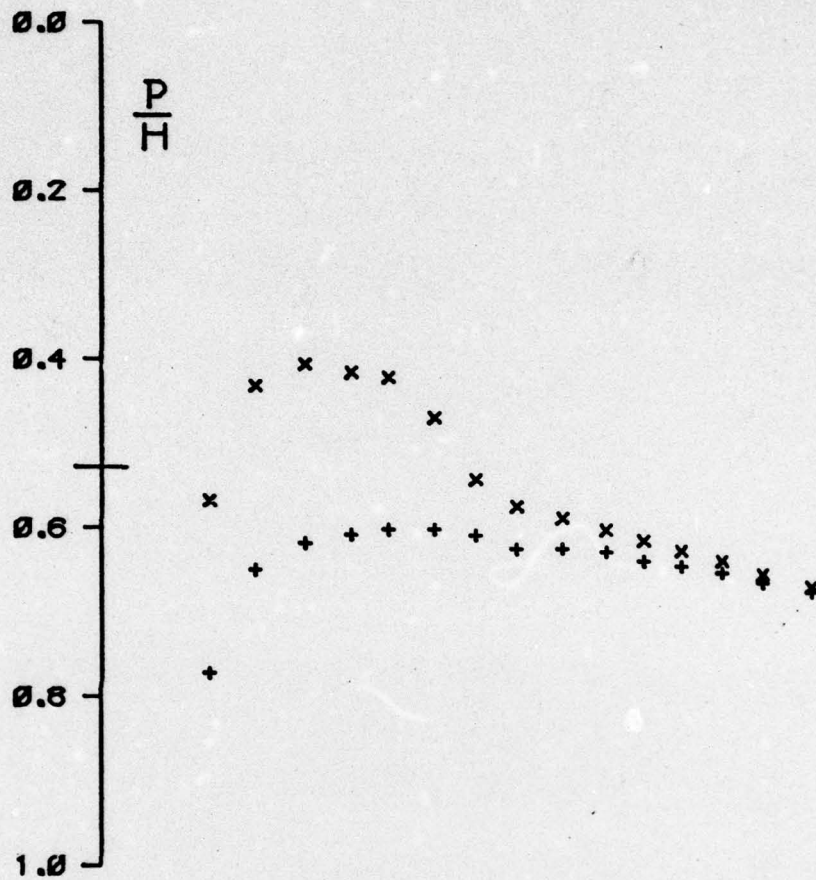
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .721$ $AL = 1.00$ $CN = 0.230$ $CM = 0.009$ $R = 0.393$



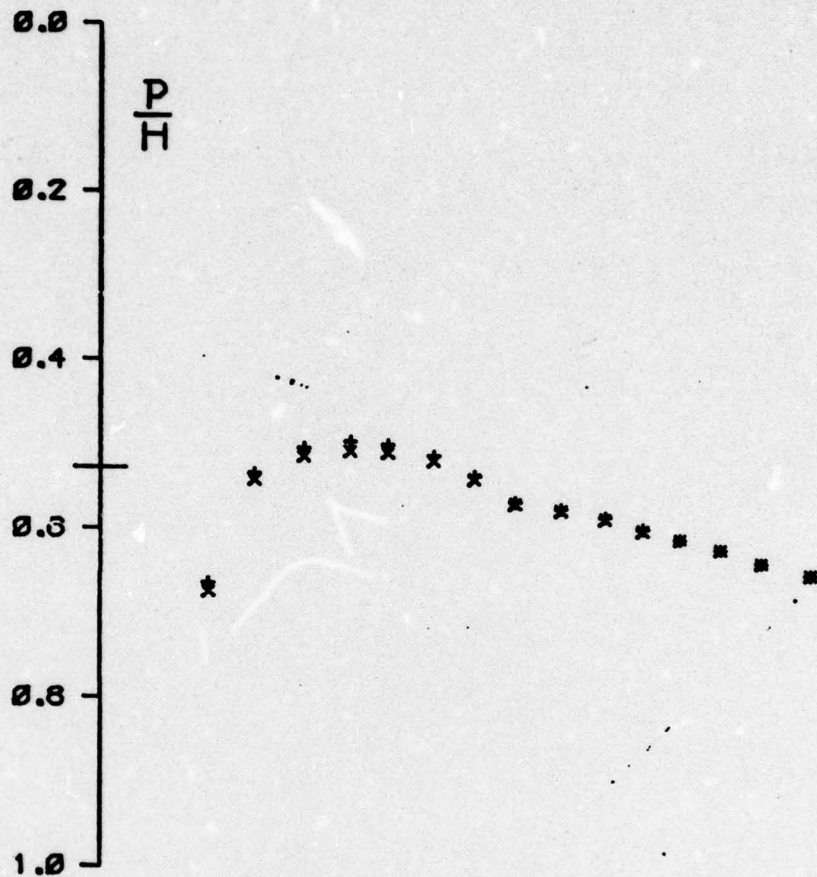
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .739 AL= 0.00 CN=-0.010 CM=-0.001 R= 0.393



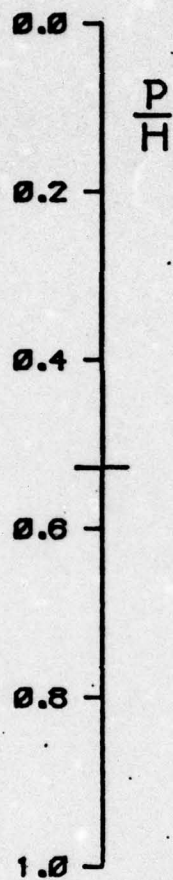
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .738$ $AL = 0.50$ $CN = 0.128$ $CM = 0.005$ $R = 0.393$



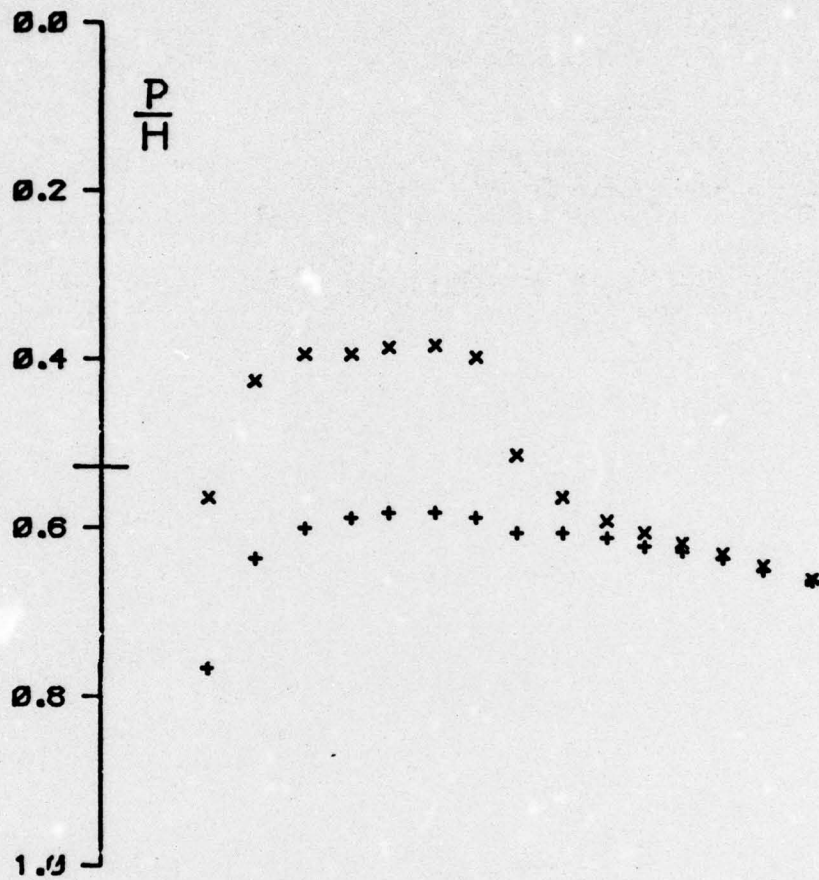
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
M= .739 AL= 1.00 CN= 0.266 CM= 0.013 R= 0.393



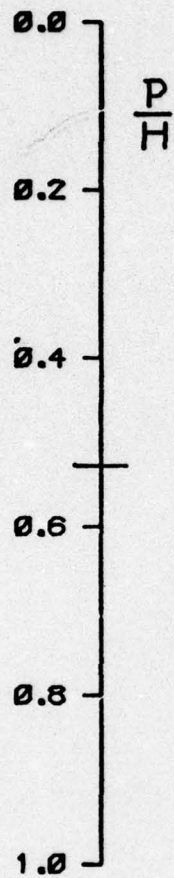
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .760 AL= 0.00 CN=0.013 CM=0.001 R= 0.396



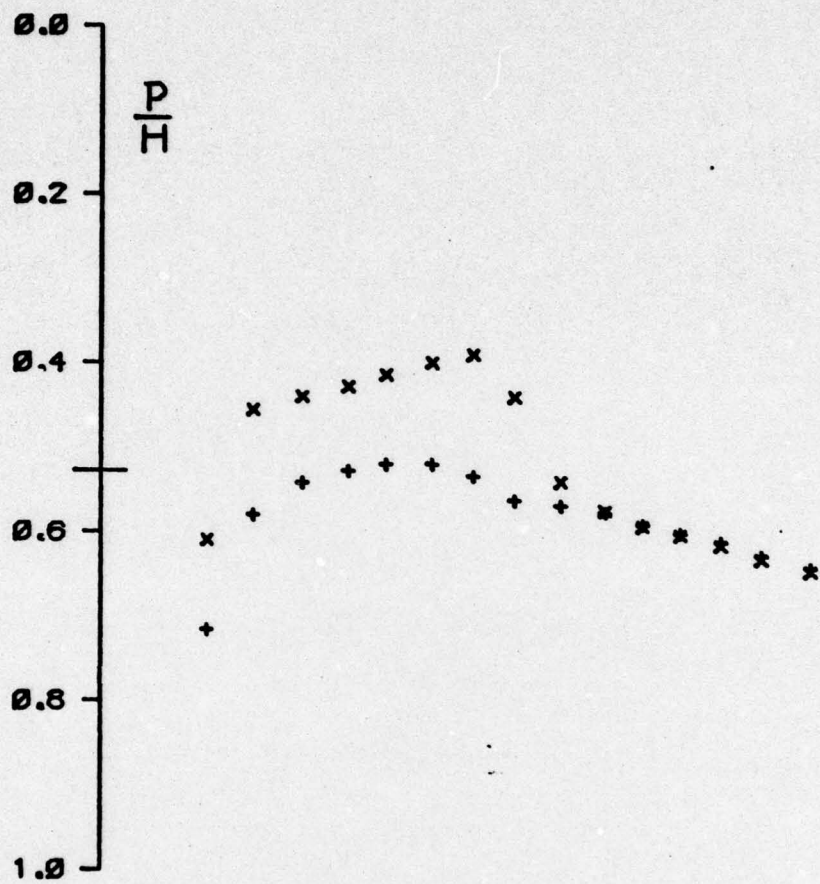
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .757$ $AL = 0.50$ $CN = 0.146$ $CM = 0.007$ $R = 0.393$



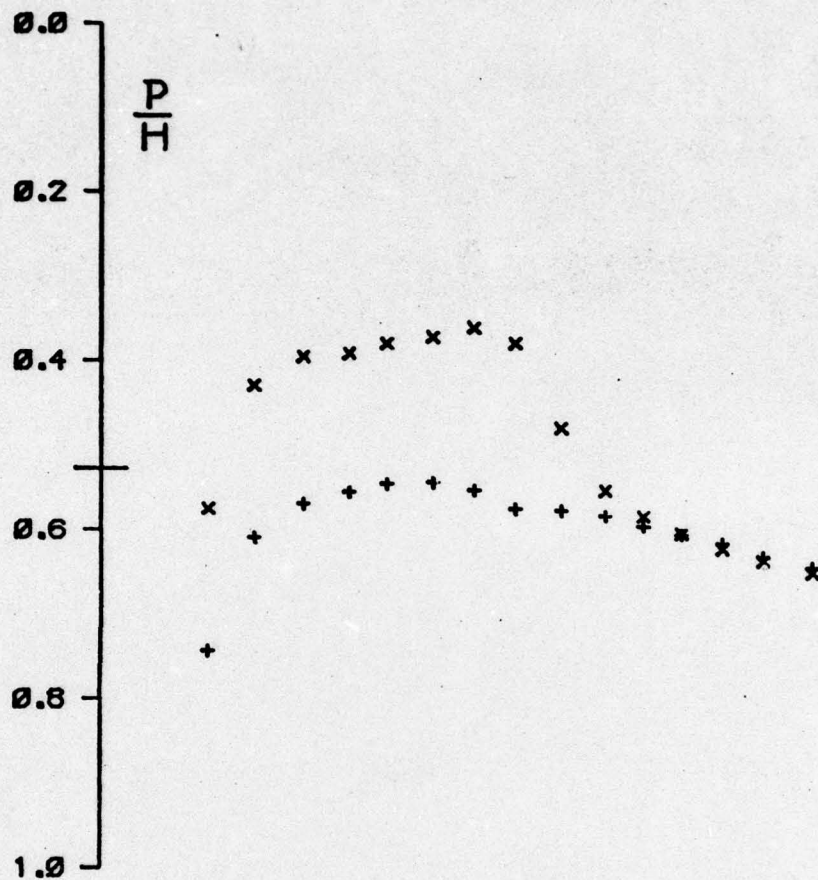
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .761 AL= 1.00 CN= 0.269 CM= 0.011 R= 0.394



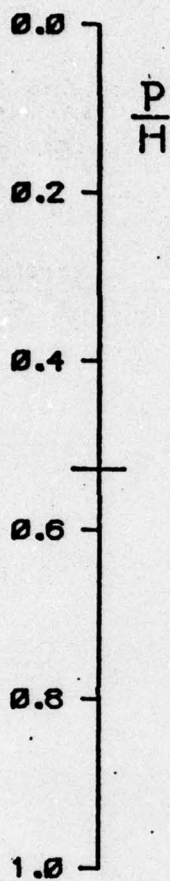
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .780 AL= 0.00 CN=-0.019 CM=-0.002 R= 0.394



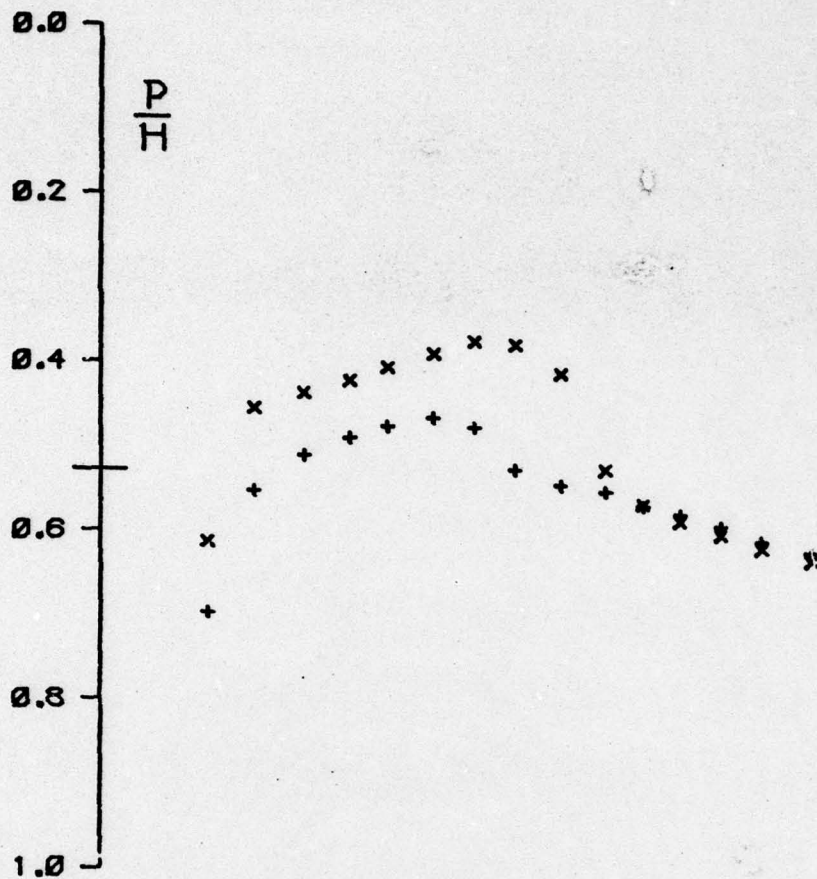
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .781$ $AL = 0.50$ $CN = 0.174$ $CM = 0.008$ $R = 0.394$



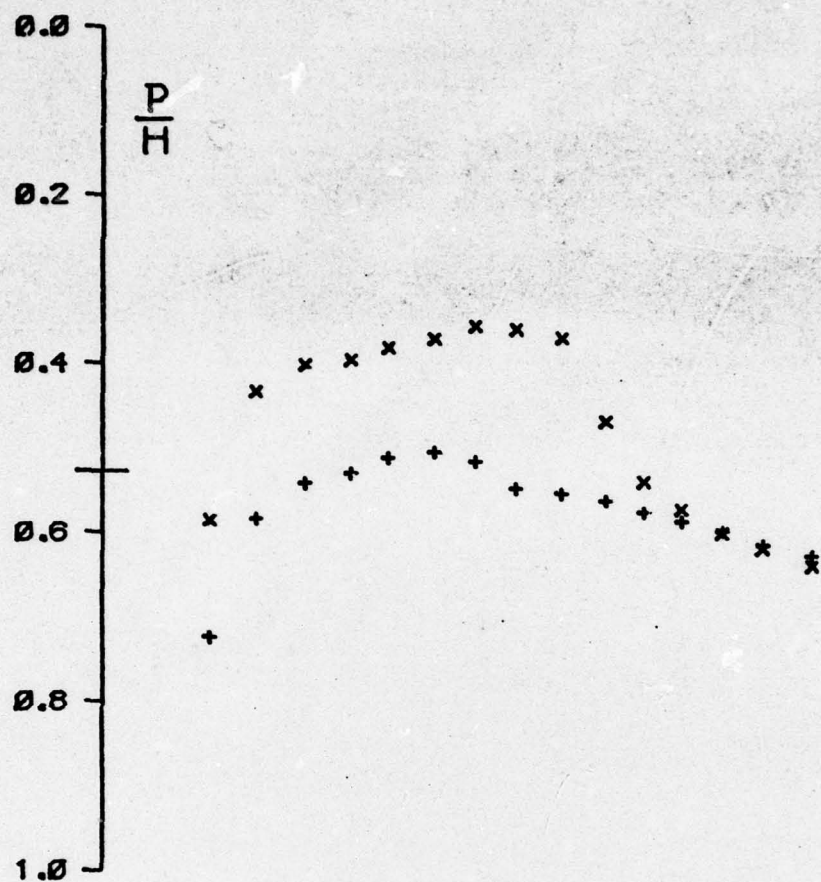
+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .781 AL= 1.00 CN= 0.275 CM= 0.007 R= 0.394



+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .799 AL= 0.00 CN=-0.017 CM=-0.001 R= 0.399



+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 M= .798 AL= 0.50 CN= 0.152 CM= 0.003 R= 0.399



+ x NACA 0012 50.8 MM CHORD SLOTTED WALLS
 $M = .800$ $AL = 1.00$ $CN = 0.251$ $CN = -0.001$ $R = 0.403$